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RESEARCH MEMORANDUM

INVESTIGATION AT TRANSONIC SPEEDS OF LOADING OVER A 30° SWEPTBACK WING OF ASPECT RATIO 3, TAPER RATIO 0.2, AND NACA 65A004 AIRFOIL SECTION MOUNTED ON A BODY

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NATIONAL ADVISORY COMMITTEE FOR AERONAUTICS

WASHINGTON
September 26, 1957

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INVESTIGATION AT TRANSONIC SPEEDS OF LOADING OVER A 30° SWEPTBACK WING OF ASPECT RATIO 3, TAPER RATIO 0.2, AND NACA 65A004 AIRFOIL SECTION MOUNTED ON A BODY

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SUMMARY

The aerodynamic load characteristics of a wing-body combination were determined experimentally from 0.80 to 1.03 Mach number for angles of attack up to 26 degrees. Two wings, both with 30° sweep of the quarter-chord line, taper ratio 0.2, aspect ratio 3, and thickness of 4 percent chord, but of different types of construction, were tested. One wing was of solid steel and the other was of plastic with an inner steel core.

The load distributions for both wings were similar, but loads on the more flexible wing were somewhat reduced. The twist distributions for both wings were calculated. Some typical flow studies of the boundary layer are presented.

INTRODUCTION

Satisfactory stability characteristics have been obtained at subsonic speeds for thin low-aspect-ratio wings with moderate leading-edge sweep (ref. 1). In order to evaluate in detail the load and stability characteristics of this type of wing on a body of revolution throughout the transonic speed range, a wing with an aspect ratio of 3, a taper ratio of 0.2, 30° sweepback of the quarter-chord line, and with NACA 65A004 airfoil sections was selected and the load characteristics are presented. The longitudinal stability characteristics are presented in reference 2. This wing is one of several wings being studied in a general program at the Langley 16-foot transonic tunnel. To date, the load characteristics of other wings in the program have been published in references 3, 4, 5, and 6. Data were obtained at Mach numbers from 0.80

to 1.03 for angles of attack up to about 26° for two wings of the same geometry but constructed of different materials. One wing was made of steel and plastic in an attempt to devise a cheaper and faster method of wing construction. The other was a solid steel wing used for comparison to check the effect of aeroelasticity and to establish the validity of data obtained with the less-rigid reinforced plastic wing. The twist distribution due to aerodynamic loading was calculated for both wings. Some typical flow studies of the boundary layer are also presented.

SYMBOLS

b	wing span
c	wing chord parallel to the plane of symmetry
$\bar{c}$	average wing chord
c'	mean aerodynamic chord
c_n	wing section normal-force coefficient
c_m	section pitching-moment coefficient about the wing mean aerodynamic chord
C_N	wing-panel normal-force coefficient, $\int_{0.16}^{1.0} c_n \frac{c}{\bar{c}} d\left(\frac{2y}{b}\right)$
C_m	wing-panel pitching-moment coefficient about 0.25c', $\int_{0.16}^{1.0} c_m \frac{c^2}{\bar{c}c'} d\left(\frac{2y}{b}\right)$
C_p	pressure coefficient, $\frac{\Delta p}{q}$
M	Mach number
Δp	local static pressure minus the free-stream static pressure
q	dynamic pressure
x	distance parallel to the center line
y	distance normal to the plane of symmetry

α model angle of attack

θ angle of twist of the chord line measured in planes parallel to the plane of symmetry

Subscripts:

L wing lower surface

U wing upper surface

MODEL DESCRIPTION

The general arrangement of the model is shown in figure 1(a). The wing was mounted to the same steel body of revolution used in references 4 and 5. The fuselage had a fineness ratio of 11, an ogive nose, cylindrical center section and a boattail afterbody. The wing was swept 30° at the quarter-chord line with a taper ratio of 0.2, and aspect ratio of 3, and NACA 65A004 sections parallel to the plane of symmetry. Two wings were constructed of different materials. Figure 1(b) shows typical cross sections of both wings. One was constructed entirely of steel with a leading-edge section and a trailing-edge section which was tongue and grooved to a center section. The spaces left in the grooves were used as ducts for the pressure tubes to the orifices. The other wing was constructed in such manner that a steel core with a thin brass plate at the trailing edge was surrounded with the wing pressure tubes, and then polyester resin was poured about the structure to form the wing contour. This wing hereinafter is called the plastic wing.

The twist characteristics for these wings were determined by the method described in appendix A. The steel wing was found to be less than half as flexible as the plastic wing. The influence coefficients A_{ij} and B_{ij} (see appendix A) used to calculate the twist were as follows:

For the steel wing:

i	$A_{ij} \times 10^{-5}$ at j = —				
	1	2	3	4	5
1	0	0	0	-1	-4
2	-2	6	-2	-8	-13
3	-2	5	9	-9	-28
4	1	4	7	11	-27
5	1	4	5	12	-5

i	$B_{ij} \times 10^{-5}$ at j = —				
	1	2	3	4	5
1	0	0.1	0.1	0.2	-0.3
2	-.1	1.3	1.1	.9	.3
3	-.1	1.7	3.9	3.2	.3
4	.3	1.7	4.9	11.5	10.5
5	.3	1.8	4.8	14.1	37.9

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For the plastic wing:

i	$A_{ij} \times 10^{-5}$ at j = –				
	1	2	3	4	5
1	0	-2	-3	-6	-11
2	-1	7	-7	-22	-34
3	1	9	8	-31	-78
4	2	9	14	-2	-100
5	2	9	14	6	-69

i	$B_{ij} \times 10^{-5}$ at j = –				
	1	2	3	4	5
1	0	0.3	1.0	0.8	1.1
2	-0.1	1.9	1.8	0	3.1
3	.1	3.3	7.0	7.3	7.6
4	.2	3.5	10.4	23.1	33.4
5	.2	3.5	11.1	30.2	90.8

where A_{ij} and B_{ij} represent the twist in degrees measured parallel to the angle-of-attack plane at the ith station due to a load or moment at the jth station, respectively. The five spanwise stations chosen were located as follows:

Station	$\frac{y}{b/2}$
1	0.245
2	.412
3	.580
4	.750
5	.915

A better comparison of the twist characteristics, however, of the steel and plastic wing is shown in figures 2(a) and (b). The plots show the effect of a unit loading applied at any spanwise station (abscissa), on the particular spanwise stations 1 through 5, for loadings at the 25- and 65-percent-chord lines. The main difference between the two plots results from a change in the stiffness and a shift of the elastic-axis location of the two wings. If the elastic axis is defined as that point of the local chord which gives zero twist when a load is applied at the point, then figure 2 indicates the position of the elastic axis. The plots show that the elastic axis of the plastic wing passes through the 0.25c at about the 0.75b/2 station, while that of the steel wing passes through the 0.25c at the 0.85b/2 station. Inboard of these spanwise stations the elastic axis lies behind the 0.25c line (positive values of twist), and outboard the axis lies ahead of the 0.25c line (negative values of twist).

The rows of pressure orifices were located at 16, 25, 40, 60, 75, and 95 percent semispan stations for both the steel and plastic wings. In each row on both the upper and lower surfaces, the orifices were located at 1, 2, 5, 7, 10 percent c and at intervals of every 5 percent chord thereafter up to the 95-percent-chord station.

TESTS AND TECHNIQUES

The tests were conducted in the Langley 16-foot transonic tunnel, which is described in reference 7. The Mach number range extended from 0.80 to 1.03, which corresponded to a Reynolds number range from about 7×10^6 to 8×10^6 (based on the wing mean aerodynamic chord). The maximum angle-of-attack range extended from -2° to 26° in 2° increments.

The pressure data were obtained simultaneously with the force data presented in reference 2. The wing pressures were recorded by photographing mercury manometer boards. The data were then processed by electronic calculating machines, which plotted and tabulated the results.

At the termination of the pressure program, a study was made of the flow in the boundary layer of the plastic wing for a reduced Mach number and angle-of-attack range. The technique used in reference 5 was employed to render the flow visible. The technique entails painting the wing surface black and then applying a white ground-glass paint similar to china clay. The wing therefore appears white when dry. Wetting with a clear fluid causes the black sublayer to become visible. Thus, by emitting fluid from a point source on the wing in a stream, the fluid path in the boundary layer is traced. As the fluid trace changes with time, the history of the trace disappears as a result of the evaporation of the fluid, so that the existing trace represents an average flow for a short interval of time. For these tests clear varsol was used as the liquid agent. The point sources were particular pressure orifices through which the fluid was forced. The locations of the sources were as follows:

$\frac{x}{c}$ at –				
$0.25\frac{b}{2}$	$0.40\frac{b}{2}$	$0.60\frac{b}{2}$	$0.75\frac{b}{2}$	$0.95\frac{b}{2}$
0.5	0.5	0.5	0.5	0.5
.10	----	----	----	----
.15	.15	----	----	----
.20	.20	----	----	----
.25	.25	.25	.25	----
----	.35	----	----	----
.45	.45	.45	.45	.45
.65	.65	.65	.65	----
.80	.80	.80	.80	.80
.90	.90	.90	.90	----

The flow studies were recorded photographically.

ACCURACY OF MEASUREMENTS

Sufficient time was allowed after a particular test condition was reached for the pressure manometer tubes to settle within about 1 percent of the ultimate value of the manometer level.

The indicated angle of attack was corrected for tunnel-flow angularity. Based on readout accuracy and repeatability, the angle of attack and Mach number are believed to be accurate within the following limits:

α, deg	±0.01
M	±0.005

RESULTS AND DISCUSSION

Flow studies.- Sample photographs of the flow on the right plastic wing are presented in figure 3. No general discussion of swept-wing flow is attempted here. Only those features of the flow studies which represent significant characteristics to be noted in the following wing pressure discussion are covered. A general discussion of the flow over swept wings may be found in references 8, 9, and 10.

Some of the features of the boundary flow which can be noted in the photographs of figure 3 are the indications of shock waves, the indications of flow separation, and the indications of vortex-type flow.

Shock waves are frequently indicated by the fluid path where there are abrupt changes of the streamlines. Note in figure 3(a) at $M = 0.94$ that shadowgraph traces of the waves are visible at angles of attack of 2° and 4° (indicated by the arrows on the figure), and note how the fluid lines are altered where they intersect the wave. The location of the waves is more obvious at the higher angles of attack by the more abrupt turning of the flow.

Separation first appears as an irregular darkened region generally increasing in area with increasing angle of attack. For this wing there appear to be two different areas where separation may commence, depending on the Mach number. At $M = 0.80$ for example, figure 3(a) at $\alpha = 4^\circ$ shows the separation to start along the leading edge near the wing tip. This results from a combination of a swept leading edge, a small leading-edge radius, and a thin wing. At the higher Mach numbers leading-edge separation as such occurs only at much higher angles of attack. (Compare 0.80 with 0.94 and 1.03 Mach numbers of fig. 3(a) for $\alpha = 6^\circ$.) However, separation does start near the tip at the trailing edge before the leading-edge separation occurs. See figure 3(a) at $\alpha = 6^\circ$ for

$M = 0.94$ and figure 3(b) at $\alpha = 10^{\circ}$ for $M = 1.03$. This separation appears to stem from the intersection, in the vicinity of the wing tip, of the shocks originating at the leading edge and near the trailing edge of the wing-body juncture. Both types of separation extend inboard with increasing angle of attack.

Once separation occurs, the existence of vorticity in the flow above the wing is indicated in the boundary-layer traces by the circulation of the fluid lines in a counterclockwise direction. For example, observe the photographs for $M = 0.80$ at the higher angles of attack. A line through the aftermost points along each of the indicated streamlines should coincide with the projection of the vortex core on the wing surface. Note that at $\alpha = 6^{\circ}$, the vortex cone sheds near the tip and the point of shedding progresses inboard with increasing angle of attack, as does the separation. At $\alpha = 19^{\circ}$ (fig. 3(c)) the vortex appears to shed at about $0.25b/2$. The vortex strength at this angle of attack is much greater than at the lower angles of attack, as is shown by the accumulation of the fluid near the vortex origin.

The origin of the vorticity at $M = 0.94$ and $M = 1.03$ appears to be in the vicinity of the intersection of the shock waves where the separation forms. Vorticity is permitted at the shock intersection since different entropy changes occur inboard and outboard of the intersection. The angle of attack at which this vortex forms increases with Mach number. With increasing angle of attack at the higher Mach numbers, the vortex flow finally reverts to a vortex generated along the leading edge once the leading-edge separation occurs at the higher angles of attack.

Chordwise pressure distributions.- A tabulation of the chordwise pressure coefficients for all test conditions for the steel wing is presented in table I. Figure 4 presents a comparison between the chordwise pressure distributions for the plastic and steel wings. As is noted, there are minor differences in the angles of attack for the two wings. In general these differences are of the order of the accuracy of measurements of these angles ($\pm 0.10^{\circ}$). The differences in the pressure coefficient with one exception may therefore be considered to be caused principally by aeroelastic effects. A significant difference in the variation of the chordwise pressure distributions exists between the two wings at angles of attack from about 2° to 8° at a Mach number of 0.80. As this Mach number was the first for which data were obtained, the discrepancy suggests a temporary difference in the leading-edge surface conditions for the two wings. The plastic-wing flow studies of figure 3(a), which were taken after the pressure tests, for angles of attack of 4° and 6° at a Mach number of 0.80 indicated separation at the outer spanwise stations, but the pressure distributions of the plastic wings indicated attached flow. The outboard stations of the plastic wing generally show the effect of decreased local angles of attack due to load when compared to the steel wing.

For either wing at the low angles of attack the increase in load coefficient progressing toward the tip illustrates the effective spanwise increase of angle of attack induced by the trailing vortices of a highly tapered swept wing. Consequently, the separation appears first at the tip and progresses inboard with increasing angle of attack as indicated by the flow studies of figure 3.

The pressure distributions on the upper surface are fairly constant over most of the wing panel at an angle of attack of about 20°, which of course indicates separation. Increasing the angle above 20° produces more negative pressure coefficients and, in addition, the innermost station shows signs of the streamlines being turned downward toward the wing surface; that is, the pressures near the trailing edge begin to recover or increase in a positive sense. As the angle of attack is further increased this effect tends to progress outboard. These pressure changes are believed to be caused by the change in location and the increasing strength of the vortex that is shown in figure 3 at $M = 0.80$. At $M = 0.94$ to 1.03 for the high angles of attack, the distributions near the root are also influenced by the strong shock wave shown by the chordwise distributions.

Spanwise load distributions.- It is apparent from the chordwise pressures that the type of wing construction, with some exceptions, has only minor effects on load distribution; therefore, the spanwise load distributions are presented only for the steel wing in figure 5. The distributions are nearly elliptical at the low angles of attack, but as the angle of attack increases, the load distributions tend to become triangular, with the triangular loading commencing at the tip. The triangular distribution spreads inboard as separation forms with further increase of angle of attack. At the angles of attack where the load distribution is elliptical inboard and triangular outboard, increasing Mach number tended to reduce the extent of the triangular loading. The implication is that increasing Mach number at a high constant angle of attack extends the attached flow region outboard. This implication is verified by the flow studies (fig. 3(c)). Note that at angles of attack of 15° and 17°, the higher the Mach number, the larger the region of attached flow.

Panel loads.- The variation of the integrated wing loads with angle of attack is shown in figure 6 for the test Mach number range. If compressibility effects are considered, the load-carrying capacity per unit angle of attack should increase to a maximum at approximately $M = 1.00$.

The data show that, for C_N values up to about 0.6, the maximum load-carrying capacity per unit angle of attack occurred at $M \approx 0.94$. For C_N values from 0.6 to the highest test value, the maximum load-carrying capacity occurred at $M \approx 0.98$.

A comparison of the variation of the panel pitching-moment coefficient about the 0.25c' with normal-force coefficient and the wing-body pitching-moment data of reference 2 is shown in figure 7. The changes of the slopes $\frac{dC_m}{dC_N}$ with normal-force coefficient agree in general with those of the data of reference 2. The absolute differences in $\frac{dC_m}{dC_N}$ at a given normal-force coefficient are due to the absence of the fuselage stability contribution in the present data.

Center of loads.- Figure 8 presents the exposed panel load centers and the local section load centers for the angle-of-attack and Mach number range of the tests. The spanwise center of load was located at approximately 50 percent of the semispan for all test conditions. The most rearward position was at about 46 percent of the mean aerodynamic chord for the panel load centers and 46 percent of the local chord for the section load centers.

Increasing angle of attack up to about 20° tended to shift the panel center of load rearward and inboard. The single data point for α above 20° shows a tendency for the center of load to become invariant with the higher angles of attack. In general the effects of changes in angle of attack on the center of load decrease with increasing Mach number; this result is to be expected since the chordwise load distribution becomes more rectangular as the flow becomes supersonic over most of the wing.

Twist distribution.- Combining the influence coefficients and the integrated normal forces and moments in the manner described in appendix A or by the method of reference 5 yields the wing spanwise twist distributions. The dynamic pressures corresponding to the measured loads are presented in figure 9 for the test Mach number range. Calculations were made for both wings at angles of attack of 4°, 8°, and 20° and for $M = 0.80$ and 1.00. A comparison of the resulting spanwise twist distributions for the steel and plastic wings is presented in figures 10(a) and 10(b). At $\alpha = 20^\circ$ and $M = 1.0$ the calculated twist angle of the tip of the plastic wing was -0.9° as compared to -0.4° for the steel wing.

CONCLUDING REMARKS

The following remarks are drawn from the loads investigation of an all steel wing and a geometrically identical reinforced plastic wing. Both wings have 30° sweepback of the quarter chord, a taper ratio of 0.2, and embody NACA 65A004 airfoil sections.

The chordwise pressure distributions for the steel and plastic wings were similar for the test range with some exceptions at Mach number 0.80.

However the type of construction had only minor effects on the chordwise and spanwise load distributions. The order of magnitude of the tip twist was calculated at a Mach number of 1.0 and an angle of attack of 20° to be -0.9° for the plastic wing as compared to -0.4° for the steel wing. The spanwise load distributions were nearly elliptical at the low angles of attack, but at the higher angles the distributions tended to become triangular commencing at the tip. The center of load on the wing panels moved rearward and inboard with increasing angle of attack for all Mach numbers. The movement of the load center with angle of attack decreased considerably with increasing Mach number.

Langley Aeronautical Laboratory,
National Advisory Committee for Aeronautics,
Langley Field, Va., June 19, 1957.

APPENDIX A

METHOD OF COMPUTING WING TWIST DUE TO AERODYNAMIC LOADING

If the spanwise and chordwise distribution of aerodynamic loading of an elastic wing are known, the twist distribution of the wing can be calculated, as follows

$$\{\theta\} = [A]\{l\} + [B]\{m\}$$

where the influence coefficients are defined as the elements of the square matrices $[A]$ and $[B]$.

The elements A_{ij} and B_{ij} represent the twist at the ith spanwise station due to a load or moment at the jth station.

The spanwise load distribution and the spanwise pitching-moment distribution are elements of the column matrices $\{l\}$ and $\{m\}$, respectively, where the elements l_j and m_j are the integrated loads and moments respectively over the jth spanwise segment; that is,

$$l_j = q\bar{c}\,\frac{b}{2}\int_{(j-1)/n}^{j/n} c_n \frac{c}{\bar{c}}\, d\left(\frac{2y}{b}\right)$$

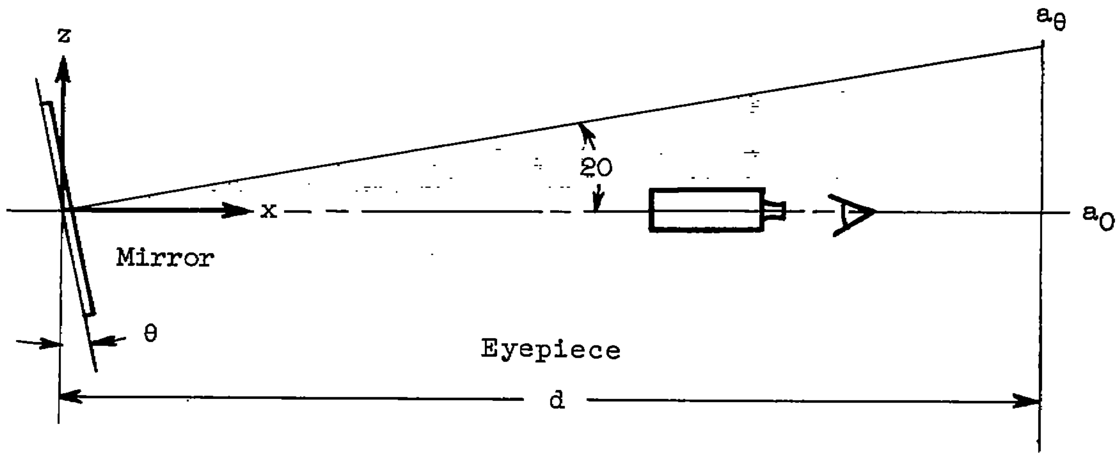
and

$$m_j = qc'\bar{c}\,\frac{b}{2}\int_{(j-1)/n}^{j/n} c_m \frac{c^2}{c'\bar{c}}\, d\left(\frac{2y}{b}\right)$$

where

n	number of spanwise stations
c'	mean aerodynamic chord
$\bar{c}$	average chord
c	local chord

The setup for measuring twist with mirrors is shown in figure 11. The technique employed for obtaining the influence coefficients involved principally the use of mirrors, linear scales, and a transit. Loads were applied at the desired points along the wing. A diagram illustrating the twist measurements is shown below:



where

a_0 zero twist reading

a_θ reading due to twist θ

A change in the angle θ of the mirror required a change in the scale reading as sighted through the eyepiece. Small translations of the mirror up or down have little effect on the scale reading. Thus, only twist about the y-axis (perpendicular to the plane of the paper) is observed.

Loading at the jth spanwise station of the 0.25c yields the influence coefficients due to normal force of the ith spanwise station. Thus

$$A_{ij} = \frac{\theta_i}{(\text{Load})_j} \frac{\text{deg}}{\text{lb}}$$

where

$$\theta_i = \frac{1}{2} \tan^{-1} \frac{\left(a_{\theta_j} - a_{0_j}\right)}{d}$$

Loading at the jth spanwise station of the 0.65c yields the influence coefficients due to a moment about the y-axis through the 0.25c of the ith station; thus,

$$B_{ij} = \frac{\dfrac{\theta_{i_{0.65c}}}{\text{Load }j} - A_{ij}}{(0.65c - 0.25c)_j} \; \frac{\text{deg}}{\text{in-lb}}$$

where

$$\theta_{i_{0.65c}} = \frac{1}{2} \tan^{-1} \frac{\left(a_{\theta_j} - a_{0_j}\right)_{0.65c}}{d}$$

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TABLE I.- STEEL WING PRESSURE COEFFICIENT DATA FOR THE TEST RANGE OF ANGLE OF ATTACK AND MACH NUMBER

		Pressure coefficient, P, at											
	Percent c	0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2	0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2
		M = 0.80	α = -1.94°					M = 0.80	α = -0.04°				
Upper surface	0.00	.041	.631	.569	.510	.454	.299	.050	.677	.640	.661	.624	.675
	1.25	.290	.223	.257	.287	.288	.288	.161	.001	-.009	-.026	-.048	.068
	2.50	.238	.132	.148	.178	.203	.193	.120	-.049	-.050	-.066	-.070	-.068
	5.00	.164	.099	.096	.109	.114	.145	.031	-.049	-.050	-.071	-.088	-.083
	7.50	.145	.065	.083	.065	.067	.083	.034	-.049	-.056	-.089	-.109	-.107
	10.00	.092	.040	.046	.038	.037	.033	-.012	-.055	-.079	-.100	-.109	-.124
	15.00	.061	.010	.003	.020	.006	-.035	-.020	-.075	-.096	-.098	-.117	-.145
	20.00	.022	-.017	-.025	-.014	-.020	-.072	-.035	-.094	-.116	-.112	-.126	-.145
	25.00	-.006	-.020	-.041	-.048	-.053	-.109	-.071	-.096	-.118	-.135	-.151	-.156
	30.00	-.024	-.042	-.051	-.061	-.068	-.103	-.075	-.104	-.124	-.137	-.152	-.136
	35.00	-.054	-.061	-.069	-.069	-.085	-.114	-.104	-.120	-.136	-.139	-.158	-.141
	40.00	-.068	-.065	-.080	-.090	-.093	-.152	-.113	-.120	-.139	-.149	-.158	-.175
	45.00	-.068	-.073	-.093	-.100	-.114	-.130	-.110	-.120	-.145	-.149	-.166	-.145
	50.00	-.054	-.084	-.099	-.108	-.119	-.134	-.092	-.120	-.144	-.149	-.161	-.141
	55.00	-.063	-.091	-.103	-.108	-.125	-.126	-.088	-.124	-.143	-.146	-.158	-.127
	60.00	-.087	-.092	-.097	-.103	-.114	-.100	-.112	-.124	-.131	-.136	-.138	-.094
	65.00	-.079	-.083	-.086	-.089	-.094	-.112	-.095	-.103	-.114	-.109	-.110	-.106
	70.00	-.072	-.074	-.072	-.075	-.082	-.067	-.087	-.092	-.092	-.089	-.092	-.058
	75.00	-.066	-.065	-.058	-.055	-.058	-.055	-.079	-.076	-.072	-.064	-.065	-.043
	80.00	-.056	-.050	-.034	-.030	-.034	-.034	-.063	-.059	-.049	-.038	-.036.	-.022
	85.00	-.051	-.028	-.017	-.011	-.007	-.015	-.056	-.033	-.024	-.011	-.005	.005
	90.00	-.021	-.016	.007	.017	.021	-.003	-.015	-.013	.001	.011	.030	.019
	95.00	.014	.006	.032	.043	.021	-.008	.023	.009	.033	.049	.037	.019
Lower surface	1.25	-.032	-.308	-.438	-.689	-.942	-.760	.135	.008	-.038	-.072	-.081	-.108
	2.50	-.077	-.262	-.331	-.436	-.547	-.646	.076	-.044	-.063	-.079	-.085	-.103
	5.00	-.094	-.188	-.238	-.317	-.403	-.545	.044	-.027	-.061	-.073	-.081	-.104
	7.50	-.107	-.184	-.229	-.287	-.346	-.441	.019	-.049	-.078	-.091	-.093	-.116
	10.00	-.137	-.178	-.216	-.280	-.319	-.376	-.016	-.060	-.092	-.108	-.106	-.126
	15.00	-.137	-.172	-.201	-.250	-.290	-.294	-.032	-.071	-.094	-.113	-.124	-.143
	20.00	-.159	-.151	-.201	-.253	-.272	-.249	-.063	-.068	-.112	-.141	-.134	-.156
	25.00	-.164	-.161	-.187	-.232	-.252	-.216	-.072	-.071	-.112	-.136	-.134	-.145
	30.00	-.134	-.171	-.190	-.228	-.243	-.206	-.057	-.093	-.121	-.137	-.139	-.145
	35.00	-.148	-.181	-.194	-.229	-.243	-.181	-.077	-.108	-.132	-.149	-.158	-.119
	40.00	-.169	-.176	-.196	-.223	-.225	-.191	-.097	-.113	-.141	-.155	-.147	-.141
	45.00	-.149	-.185	-.195	-.221	-.221	-.187	-.084	-.122	-.146	-.162	-.156	-.147
	50.00	-.178	-.186	-.193	-.213	-.207	-.177	-.121	-.132	-.141	-.162	-.147	-.141
	55.00	-.172	-.178	-.191	-.184	-.194	-.152	-.117	-.129	-.132	-.143	-.154	-.119
	60.00	-.162	-.166	-.175	-.167	-.170	-.132	-.112	-.122	-.129	-.135	-.136	-.101
	65.00	-.144	-.150	-.163	-.151	-.150	-.123	-.102	-.110	-.114	-.125	-.122	-.093
	70.00	-.124	-.124	-.138	-.118	-.116	-.098	-.083	-.093	-.096	-.098	-.084	-.071
	75.00	-.137	-.116	-.116	-.100	-.094	-.084	-.104	-.089	-.084	-.079	-.070	-.054
	80.00	-.101	-.090	-.101	-.066	-.069	-.063	-.073	-.067	-.061	-.051	-.044	-.035
	85.00	-.078	-.061	-.073	-.031	-.041	-.021	-.055	-.043	-.027	-.021	-.023	.005
	90.00	-.044	-.034	-.038	-.003	-.001	-.021	-.025	-.019	-.006	.009	.011	.005
	95.00	-.009	-.006	-.016	.024	.030	.017	.012	.004	.034	.026	.037	.037
		M = 0.80	α = 1.94°					M = 0.80	α = 3.90°				
Upper surface	0.00	.025	.561	.447	.449	.419	.606	.030	.341	.191	.124	.067	.505
	1.25	-.027	-.339	-.484	-.667	-.797	-.557	-.236	-1.001	-.909	-1.014	-1.008	-.671
	2.50	-.061	-.304	-.357	-.459	-.576	-.660	-.256	-.736	-.815	-.968	-.981	-.733
	5.00	-.168	-.249	-.285	-.352	-.423	-.543	-.367	-.578	-.715	-.902	-.932	-.718
	7.50	-.160	-.207	-.254	-.312	-.377	-.435	-.335	-.379	-.588	-.844	-.881	-.704
	10.00	-.158	-.196	-.247	-.300	-.329	-.385	-.296	-.344	-.504	-.768	-.826	-.688
	15.00	-.148	-.201	-.240	-.265	-.301	-.314	-.255	-.319	-.401	-.603	-.722	-.656
	20.00	-.143	-.211	-.245	-.254	-.285	-.268	-.226	-.307	-.367	-.466	-.607	-.619
	25.00	-.185	-.196	-.228	-.258	-.283	-.246	-.274	-.287	-.337	-.386	-.507	-.574
	30.00	-.173	-.204	-.228	-.253	-.273	-.223	-.250	-.285	-.315	-.343	-.416	-.525
	35.00	-.208	-.213	-.233	-.243	-.266	-.212	-.276	-.285	-.308	-.317	-.355	-.472
	40.00	-.204	-.208	-.228	-.248	-.248	-.234	-.264	-.264	-.289	-.304	-.313	-.421
	45.00	-.201	-.204	-.226	-.239	-.248	-.203	-.259	-.254	-.277	-.285	-.289	-.372
	50.00	-.171	-.199	-.218	-.230	-.226	-.195	-.221	-.242	-.265	-.268	-.259	-.320
	55.00	-.157	-.195	-.210	-.211	-.215	-.178	-.190	-.231	-.248	-.245	-.237	-.280
	60.00	-.182	-.188	-.194	-.187	-.187	-.147	-.214	-.215	-.218	-.221	-.206	-.243
	65.00	-.158	-.160	-.168	-.155	-.154	-.149	-.186	-.181	-.196	-.181	-.166	-.211
	70.00	-.145	-.143	-.141	-.133	-.131	-.106	-.164	-.159	-.158	-.151	-.144	-.179
	75.00	-.129	-.122	-.117	-.102	-.101	-.091	-.144	-.137	-.133	-.121	-.113	-.159
	80.00	-.107	-.093	-.086	-.078	-.070	-.066	-.116	-.105	-.096	-.083	-.077	-.137
	85.00	-.095	-.068	-.054	-.042	-.035	-.040	-.100	-.072	-.070	-.058	-.043	-.115
	90.00	-.054	-.043	-.025	-.015	-.001	-.023	-.054	-.041	-.038	-.026	-.006	-.094
	95.00	-.013	-.018	.010	.023	.013	-.018	-.010	-.013	.003	.014	.002	-.079
Lower surface	1.25	.257	.228	.230	.268	.321	.246	.403	.416	.410	.457	.511	.404
	2.50	.190	.130	.152	.181	.210	.180	.339	.301	.328	.358	.392	.331
	5.00	.147	.096	.094	.114	.132	.118	.282	.231	.234	.249	.282	.260
	7.50	.108	.052	.046	.065	.090	.065	.230	.174	.181	.191	.225	.188
	10.00	.070	.025	.017	.033	.048	.023	.183	.141	.144	.145	.174	.133
	15.00	.035	.002	-.004	-.004	.002	-.043	.142	.111	.104	.098	.114	.045
	20.00	-.001	-.008	-.029	-.041	-.029	-.087	.096	.099	.068	.049	.074	-.016
	25.00	-.017	-.017	-.042	-.061	-.055	-.103	.072	.059	.031	.024	.038	-.061
	30.00	-.018	-.045	-.061	-.071	-.072	-.117	.058	.029	.012	.001	.013	-.089
	35.00	-.045	-.064	-.077	-.098	-.097	-.101	.016	.003	-.010	-.027	-.016	-.103
	40.00	-.066	-.078	-.091	-.106	-.101	-.138	.002	-.014	-.027	-.042	-.028	-.112
	45.00	-.057	-.093	-.102	-.123	-.115	-.150	.004	-.033	-.040	-.068	-.050	-.122
	50.00	-.100	-.106	-.115	-.130	-.122	-.147	-.045	-.049	-.060	-.079	-.064	-.125
	55.00	-.106	-.112	-.112	-.118	-.135	-.128	-.054	-.060	-.062	-.074	-.082	-.110
	60.00	-.103	-.111	-.115	-.115	-.126	-.118	-.054	-.063	-.071	-.079	-.084	-.106
	65.00	-.097	-.102	-.102	-.112	-.119	-.117	-.054	-.061	-.066	-.083	-.086	-.109
	70.00	-.090	-.093	-.087	-.093	-.091	-.092	-.048	-.054	-.060	-.069	-.063	-.089
	75.00	-.113	-.095	-.085	-.082	-.050	-.080	-.080	-.059	-.058	-.064	-.059	-.081
	80.00	-.088	-.073	-.067	-.057	-.059	-.061	-.056	-.045	-.045	-.045	-.044	-.070
	85.00	-.071	-.056	-.040	-.032	-.042	-.027	-.048	-.030	-.025	-.023	-.028	-.033
	90.00	-.047	-.034	-.023	-.008	-.010	-.030	-.028	-.016	-.011	-.003	-.004	-.050
	95.00	-.009	-.015	.008	.010	.014	.002	-.001	-.004	.016	.008	.017	-.026

TABLE I.- STEEL WING PRESSURE COEFFICIENT DATA FOR THE TEST RANGE OF ANGLE OF ATTACK AND MACH NUMBER - Continued

	Percent c	Pressure coefficient, P, at: 0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2	0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2
		M = 0.80	α = 5.91°					M = 0.80	α = 7.95°				
Upper surface	0.00	.039	.107	-.054	-.160	-.197	.396	.029	-.132	-.343	-.465	-.449	.247
	1.25	-.465	-1.373	-1.130	-1.014	-.811	-.515	-.671	-1.619	-1.327	-.928	-.700	-.475
	2.50	-.504	-1.276	-1.108	-1.014	-.805	-.546	-.734	-1.574	-1.318	-.923	-.697	-.484
	5.00	-.643	-1.057	-1.088	-1.007	-.797	-.546	-.884	-1.427	-1.342	-.913	-.689	-.479
	7.50	-.510	-.811	-1.047	-1.003	-.785	-.546	-.800	-1.281	-1.323	-.903	-.682	-.479
	10.00	-.481	-.502	-1.007	-.995	-.777	-.549	-.722	-1.082	-1.312	-.890	-.677	-.482
	15.00	-.371	-.393	-.853	-.962	-.760	-.551	-.513	-.522	-1.213	-.869	-.668	-.486
	20.00	-.330	-.393	-.642	-.888	-.742	-.549	-.442	-.475	-1.087	-.847	-.657	-.486
	25.00	-.571	-.360	-.458	-.818	-.716	-.542	-.485	-.429	-.889	-.827	-.642	-.485
	30.00	-.539	-.353	-.352	-.705	-.688	-.530	-.423	-.407	-.588	-.795	-.624	-.485
	35.00	-.351	-.345	-.316	-.606	-.660	-.515	-.412	-.386	-.387	-.766	-.604	-.481
	40.00	-.322	-.315	-.299	-.486	-.620	-.499	-.372	-.352	-.283	-.753	-.585	-.475
	45.00	-.310	-.296	-.285	-.400	-.592	-.476	-.359	-.337	-.280	-.703	-.568	-.471
	50.00	-.268	-.280	-.268	-.332	-.542	-.458	-.317	-.318	-.280	-.664	-.545	-.463
	55.00	-.225	-.262	-.256	-.279	-.502	-.440	-.267	-.305	-.276	-.617	-.529	-.454
	60.00	-.241	-.240	-.233	-.247	-.462	-.415	-.283	-.285	-.257	-.580	-.511	-.444
	65.00	-.205	-.207	-.195	-.204	-.413	-.388	-.242	-.243	-.233	-.518	-.489	-.437
	70.00	-.182	-.179	-.168	-.166	-.369	-.364	-.226	-.220	-.206	-.460	-.468	-.429
	75.00	-.159	-.150	-.143	-.139	-.324	-.349	-.197	-.191	-.189	-.417	-.451	-.423
	80.00	-.127	-.119	-.106	-.089	-.281	-.328	-.160	-.157	-.150	-.374	-.432	-.418
	85.00	-.107	-.084	-.077	-.067	-.237	-.307	-.139	-.121	-.120	-.325	-.408	-.418
	90.00	-.059	-.051	-.037	-.034	-.185	-.295	-.090	-.086	-.082	-.283	-.380	-.417
	95.00	-.011	-.018	.006	.005	-.156	-.281	-.037	-.045	-.031	-.225	-.369	-.413
Lower surface	1.25	.551	.547	.536	.570	.592	.464	.637	.638	.615	.621	.629	.500
	2.50	.472	.450	.448	.475	.494	.394	.593	.529	.534	.540	.550	.437
	5.00	.406	.343	.350	.364	.384	.326	.515	.436	.436	.437	.449	.374
	7.50	.348	.282	.285	.302	.321	.254	.444	.367	.371	.375	.387	.299
	10.00	.292	.247	.249	.256	.263	.198	.383	.331	.328	.326	.327	.247
	15.00	.235	.193	.189	.196	.205	.110	.318	.267	.260	.268	.263	.151
	20.00	.182	.174	.147	.143	.155	.040	.251	.237	.211	.198	.207	.076
	25.00	.149	.129	.114	.105	.109	-.010	.212	.189	.173	.161	.157	.029
	30.00	.125	.092	.088	.083	.079	-.049	.184	.149	.135	.131	.125	-.015
	35.00	.070	.064	.059	.048	.040	-.052	.112	.116	.107	.090	.084	-.027
	40.00	.064	.043	.044	.027	.022	-.086	.112	.091	.077	.063	.056	-.074
	45.00	.057	.021	.016	.001	-.002	-.117	.103	.065	.052	.035	.026	-.114
	50.00	.006	-.002	-.005	-.026	-.024	-.126	.043	.039	.030	.009	-.003	-.130
	55.00	-.010	-.016	-.011	-.025	-.050	-.124	.022	.020	.015	-.003	-.037	-.133
	60.00	-.010	-.025	-.026	-.039	-.059	-.124	.022	.004	-.002	-.020	-.056	-.141
	65.00	-.020	-.029	-.026	-.049	-.072	-.133	.005	-.006	-.011	-.038	-.075	-.154
	70.00	-.015	-.025	-.024	-.041	-.058	-.121	.001	-.011	-.017	-.039	-.075	-.153
	75.00	-.049	-.031	-.028	-.040	-.065	-.124	-.027	-.021	-.025	-.049	-.086	-.161
	80.00	-.036	-.027	-.018	-.031	-.056	-.121	-.028	-.021	-.025	-.049	-.101	-.171
	85.00	-.028	-.017	.000	-.015	-.059	-.101	-.028	-.015	-.014	-.053	-.121	-.149
	90.00	-.015	-.006	.007	-.005	-.054	-.134	-.020	-.012	-.012	-.060	-.137	-.206
	95.00	.010	.005	.028	.009	-.062	-.145	.000	-.009	.009	-.085	-.187	-.226
		M = 0.80	α = 9.89°					M = 0.80	α = 11.84°				
Upper surface	0.00	.036	-.351	-.605	-.666	-.599	.087	.047	-.575	-.835	-.922	-.763	-.094
	1.25	-.821	-1.733	-1.270	-.783	-.626	-.452	-.957	-1.640	-1.026	-.726	-.600	-.440
	2.50	-.905	-1.714	-1.252	-.777	-.625	-.453	-1.050	-1.611	-1.019	-.718	-.595	-.438
	5.00	-1.043	-1.598	-1.241	-.771	-.617	-.453	-1.182	-1.531	-1.013	-.716	-.586	-.438
	7.50	-.950	-1.481	-1.208	-.765	-.612	-.453	-1.100	-1.449	-.983	-.718	-.579	-.438
	10.00	-.902	-1.341	-1.207	-.756	-.612	-.450	-1.000	-1.366	-.972	-.715	-.579	-.438
	15.00	-.666	-.992	-1.160	-.742	-.608	-.455	-.673	-1.226	-.945	-.712	-.576	-.440
	20.00	-.508	-.414	-1.090	-.727	-.604	-.455	-.552	-.939	-.930	-.699	-.571	-.440
	25.00	-.483	-.392	-1.023	-.705	-.599	-.457	-.539	-.662	-.926	-.689	-.566	-.440
	30.00	-.511	-.409	-.920	-.689	-.593	-.460	-.565	-.516	-.906	-.671	-.563	-.441
	35.00	-.430	-.409	-.814	-.667	-.584	-.462	-.489	-.475	-.862	-.654	-.560	-.444
	40.00	-.400	-.378	-.696	-.656	-.574	-.463	-.450	-.435	-.822	-.643	-.555	-.447
	45.00	-.384	-.367	-.591	-.642	-.567	-.463	-.435	-.420	-.772	-.626	-.553	-.450
	50.00	-.361	-.357	-.513	-.626	-.555	-.463	-.420	-.406	-.716	-.616	-.549	-.453
	55.00	-.311	-.349	-.445	-.614	-.544	-.463	-.357	-.394	-.651	-.599	-.543	-.454
	60.00	-.326	-.334	-.393	-.588	-.536	-.463	-.372	-.380	-.607	-.583	-.537	-.457
	65.00	-.289	-.298	-.355	-.570	-.522	-.463	-.341	-.344	-.554	-.574	-.529	-.461
	70.00	-.276	-.275	-.318	-.547	-.507	-.465	-.324	-.326	-.515	-.555	-.521	-.466
	75.00	-.248	-.248	-.280	-.518	-.495	-.466	-.300	-.304	-.472	-.534	-.517	-.468
	80.00	-.207	-.225	-.251	-.493	-.484	-.468	-.262	-.273	-.438	-.520	-.509	-.472
	85.00	-.179	-.174	-.212	-.472	-.470	-.473	-.235	-.237	-.386	-.507	-.497	-.478
	90.00	-.126	-.131	-.167	-.437	-.449	-.475	-.177	-.194	-.348	-.491	-.483	-.488
	95.00	-.068	-.086	-.117	-.419	-.441	-.477	-.104	-.141	-.293	-.479	-.481	-.491
Lower surface	1.25	.733	.709	.687	.657	.645	.531	.805	.761	.708	.689	.660	.556
	2.50	.703	.615	.607	.600	.588	.477	.801	.687	.658	.644	.625	.509
	5.00	.625	.520	.513	.503	.501	.418	.720	.598	.579	.554	.553	.460
	7.50	.545	.449	.442	.445	.440	.347	.631	.527	.517	.499	.494	.399
	10.00	.476	.415	.397	.395	.385	.291	.558	.484	.471	.452	.441	.349
	15.00	.397	.340	.332	.331	.317	.203	.471	.411	.400	.386	.375	.255
	20.00	.325	.303	.272	.259	.260	.130	.398	.375	.343	.316	.313	.179
	25.00	.285	.252	.232	.217	.209	.075	.351	.318	.293	.276	.265	.129
	30.00	.250	.210	.197	.185	.169	.022	.311	.271	.250	.242	.224	.073
	35.00	.172	.174	.162	.139	.123	-.002	.231	.230	.214	.199	.177	.047
	40.00	.171	.145	.130	.109	.100	-.049	.226	.199	.187	.153	.149	-.006
	45.00	.156	.113	.103	.073	.064	-.090	.209	.164	.149	.125	.114	-.055
	50.00	.097	.084	.069	.042	.030	-.115	.146	.132	.119	.087	.077	-.081
	55.00	.066	.058	.049	.029	-.006	-.123	.112	.104	.091	.068	.034	-.094
	60.00	.066	.040	.026	.008	-.035	-.135	.112	.081	.068	.039	.002	-.114
	65.00	.041	.022	.013	-.021	-.062	-.156	.080	.057	.049	.015	-.028	-.135
	70.00	.029	.015	-.001	-.033	-.065	-.159	.059	.048	.025	-.005	-.036	-.141
	75.00	.004	-.003	-.015	-.047	-.089	-.173	.032	.018	.005	-.029	-.069	-.157
	80.00	-.013	-.010	-.022	-.056	-.104	-.186	.010	.005	-.005	-.046	-.083	-.172
	85.00	-.020	-.014	-.024	-.067	-.132	-.172	-.008	-.006	-.023	-.072	-.118	-.162
	90.00	-.020	-.018	-.033	-.104	-.159	-.228	-.019	-.021	-.054	-.109	-.155	-.221
	95.00	-.015	-.025	-.031	-.167	-.225	-.252	-.031	-.048	-.084	-.189	-.223	-.247

TABLE I.- STEEL WING PRESSURE COEFFICIENT DATA FOR THE TEST RANGE OF ANGLE OF ATTACK AND MACH NUMBER - Continued

	Percent c	Pressure coefficient, P, at: 0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2	0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2
		M = 0.80	α = 13.95°					M = 0.80	α = 17.24°				
Upper surface	0.00	.022	-.849	-1.079	-1.081	-.665	-.333	.002	-1.047	-1.017	-.753	-.646	-.573
	1.25	-1.106	-1.338	-1.009	-.724	-.622	-.495	-.849	-.879	-.795	-.723	-.623	-.541
	2.50	-1.265	-1.380	-1.035	-.703	-.621	-.491	-.826	-.884	-.785	-.701	-.627	-.541
	5.00	-1.293	-1.328	-.973	-.699	-.617	-.491	-.840	-.885	-.795	-.692	-.624	-.540
	7.50	-1.250	-1.279	-.947	-.698	-.611	-.491	-.856	-.906	-.799	-.690	-.617	-.541
	10.00	-1.098	-1.248	-.957	-.698	-.611	-.491	-.867	-.896	-.809	-.687	-.621	-.544
	15.00	-.847	-1.187	-.920	-.696	-.604	-.492	-.853	-.878	-.801	-.686	-.617	-.548
	20.00	-.740	-1.089	-.896	-.695	-.600	-.492	-.814	-.864	-.785	-.684	-.610	-.547
	25.00	-.674	-.979	-.873	-.689	-.598	-.492	-.772	-.855	-.773	-.680	-.608	-.549
	30.00	-.677	-.863	-.855	-.680	-.595	-.494	-.735	-.829	-.761	-.674	-.608	-.552
	35.00	-.576	-.776	-.836	-.673	-.593	-.495	-.675	-.808	-.756	-.674	-.606	-.557
	40.00	-.533	-.675	-.810	-.666	-.593	-.498	-.635	-.773	-.750	-.674	-.606	-.554
	45.00	-.511	-.611	-.789	-.661	-.591	-.500	-.598	-.748	-.743	-.670	-.607	-.559
	50.00	-.504	-.560	-.749	-.650	-.590	-.504	-.581	-.715	-.739	-.670	-.608	-.560
	55.00	-.441	-.525	-.725	-.645	-.588	-.507	-.572	-.688	-.729	-.666	-.608	-.562
	60.00	-.465	-.507	-.691	-.630	-.588	-.509	-.560	-.673	-.719	-.658	-.612	-.566
	65.00	-.444	-.477	-.662	-.625	-.585	-.513	-.542	-.651	-.709	-.657	-.612	-.572
	70.00	-.438	-.460	-.636	-.613	-.581	-.516	-.550	-.638	-.698	-.656	-.611	-.575
	75.00	-.409	-.447	-.614	-.601	-.579	-.523	-.523	-.622	-.686	-.645	-.615	-.574
	80.00	-.380	-.417	-.588	-.589	-.574	-.527	-.508	-.601	-.673	-.639	-.615	-.580
	85.00	-.362	-.387	-.555	-.582	-.566	-.533	-.502	-.569	-.659	-.636	-.610	-.586
	90.00	-.307	-.356	-.527	-.568	-.557	-.537	-.481	-.545	-.639	-.628	-.602	-.588
	95.00	-.217	-.293	-.490	-.561	-.557	-.541	-.414	-.505	-.621	-.623	-.606	-.590
Lower surface	1.25	.846	.790	.722	.674	.642	.550	.906	.826	.738	.675	.626	.541
	2.50	.871	.731	.690	.663	.634	.514	.965	.800	.736	.690	.656	.528
	5.00	.780	.647	.618	.586	.573	.474	.872	.731	.677	.646	.620	.506
	7.50	.685	.576	.544	.526	.522	.416	.776	.666	.621	.599	.578	.459
	10.00	.607	.528	.508	.485	.466	.359	.699	.616	.580	.556	.538	.413
	15.00	.514	.457	.435	.414	.400	.273	.604	.542	.513	.486	.461	.326
	20.00	.440	.415	.372	.345	.340	.195	.524	.492	.451	.419	.405	.252
	25.00	.385	.354	.328	.295	.289	.137	.469	.437	.401	.373	.356	.186
	30.00	.340	.305	.280	.258	.247	.082	.423	.384	.354	.329	.308	.123
	35.00	.254	.259	.238	.215	.197	.049	.334	.337	.309	.282	.257	.092
	40.00	.247	.225	.205	.176	.166	-.005	.319	.298	.269	.241	.225	.034
	45.00	.228	.188	.170	.132	.123	-.058	.297	.256	.230	.198	.183	-.023
	50.00	.163	.149	.131	.097	.085	-.089	.228	.215	.188	.160	.141	-.055
	55.00	.123	.118	.104	.077	.043	-.100	.181	.172	.159	.130	.092	-.079
	60.00	.120	.089	.069	.038	.006	-.125	.173	.147	.120	.095	.051	-.103
	65.00	.077	.061	.047	.005	-.026	-.152	.122	.108	.092	.055	.012	-.129
	70.00	.047	.042	.025	-.012	-.042	-.161	.081	.084	.057	.030	-.003	-.147
	75.00	.025	.012	-.009	-.042	-.074	-.182	.050	.041	.022	-.005	-.042	-.170
	80.00	-.006	-.007	-.031	-.065	-.097	-.197	.019	.010	-.009	-.037	-.073	-.196
	85.00	-.038	-.035	-.053	-.095	-.137	-.188	-.013	-.024	-.041	-.072	-.119	-.194
	90.00	-.068	-.067	-.100	-.142	-.176	-.256	-.067	-.068	-.100	-.122	-.162	-.257
	95.00	-.099	-.111	-.144	-.226	-.246	-.280	-.163	-.151	-.171	-.217	-.240	-.286
		M = 0.80	α = 19.29°					M = 0.80	α = 21.30°				
Upper surface	0.00	-.017	-.711	-.719	-.697	-.627	-.601	-.059	-.779	-.771	-.744	-.697	-.660
	1.25	-.698	-.687	-.700	-.691	-.618	-.574	-.786	-.768	-.762	-.738	-.691	-.635
	2.50	-.691	-.693	-.694	-.673	-.621	-.572	-.771	-.769	-.758	-.734	-.692	-.634
	5.00	-.657	-.699	-.701	-.670	-.621	-.571	-.742	-.775	-.762	-.729	-.690	-.633
	7.50	-.658	-.702	-.694	-.666	-.616	-.574	-.753	-.780	-.756	-.727	-.686	-.635
	10.00	-.667	-.707	-.702	-.665	-.620	-.577	-.758	-.780	-.762	-.727	-.688	-.636
	15.00	-.683	-.713	-.705	-.663	-.618	-.580	-.768	-.785	-.765	-.726	-.687	-.637
	20.00	-.695	-.713	-.704	-.662	-.613	-.581	-.772	-.787	-.766	-.724	-.684	-.638
	25.00	-.699	-.717	-.710	-.657	-.614	-.583	-.772	-.788	-.769	-.723	-.684	-.639
	30.00	-.697	-.717	-.709	-.657	-.613	-.585	-.767	-.789	-.767	-.720	-.683	-.640
	35.00	-.681	-.717	-.707	-.657	-.614	-.589	-.742	-.785	-.769	-.720	-.684	-.642
	40.00	-.669	-.714	-.707	-.660	-.618	-.589	-.723	-.780	-.766	-.723	-.684	-.643
	45.00	-.648	-.714	-.704	-.659	-.618	-.591	-.693	-.778	-.766	-.722	-.686	-.645
	50.00	-.648	-.706	-.704	-.661	-.622	-.596	-.701	-.766	-.766	-.723	-.688	-.649
	55.00	-.633	-.700	-.700	-.661	-.623	-.598	-.687	-.760	-.763	-.723	-.688	-.651
	60.00	-.623	-.693	-.698	-.655	-.628	-.602	-.686	-.756	-.761	-.715	-.690	-.654
	65.00	-.615	-.686	-.696	-.657	-.630	-.607	-.681	-.746	-.756	-.715	-.692	-.657
	70.00	-.623	-.682	-.692	-.662	-.632	-.611	-.686	-.740	-.751	-.717	-.689	-.657
	75.00	-.599	-.676	-.688	-.654	-.634	-.614	-.658	-.727	-.742	-.711	-.692	-.658
	80.00	-.597	-.670	-.680	-.652	-.635	-.617	-.647	-.715	-.734	-.705	-.691	-.660
	85.00	-.594	-.649	-.672	-.651	-.632	-.623	-.638	-.693	-.726	-.705	-.688	-.662
	90.00	-.588	-.637	-.666	-.647	-.624	-.626	-.616	-.672	-.715	-.700	-.677	-.661
	95.00	-.531	-.610	-.658	-.641	-.629	-.629	-.534	-.627	-.706	-.693	-.686	-.660
Lower surface	1.25	.920	.842	.741	.666	.603	.525	.899	.848	.736	.646	.566	.502
	2.50	.999	.834	.758	.696	.656	.525	1.005	.858	.768	.702	.651	.514
	5.00	.922	.771	.715	.668	.636	.513	.944	.813	.743	.691	.655	.517
	7.50	.823	.710	.666	.626	.603	.476	.858	.757	.704	.658	.600	.489
	10.00	.754	.665	.623	.590	.562	.432	.796	.711	.666	.628	.593	.452
	15.00	.657	.593	.557	.525	.494	.353	.695	.641	.603	.568	.530	.380
	20.00	.580	.525	.501	.460	.444	.281	.615	.573	.544	.507	.480	.312
	25.00	.523	.487	.446	.411	.395	.218	.564	.530	.490	.461	.433	.247
	30.00	.477	.432	.402	.372	.348	.155	.514	.476	.443	.415	.386	.184
	35.00	.387	.385	.353	.323	.297	.120	.431	.428	.401	.368	.336	.143
	40.00	.375	.340	.314	.281	.264	.063	.418	.392	.357	.324	.301	.091
	45.00	.343	.306	.273	.234	.221	.001	.385	.342	.318	.282	.259	.026
	50.00	.275	.263	.232	.197	.178	-.030	.310	.300	.274	.240	.216	-.006
	55.00	.229	.221	.200	.169	.128	-.056	.268	.263	.240	.209	.164	-.036
	60.00	.214	.185	.157	.130	.084	-.083	.246	.219	.198	.164	.117	-.063
	65.00	.163	.141	.128	.086	.046	-.112	.193	.180	.161	.123	.081	-.096
	70.00	.115	.116	.092	.058	.027	-.137	.137	.148	.124	.094	.057	-.125
	75.00	.085	.071	.052	.019	-.017	-.160	.106	.097	.082	.052	.012	-.151
	80.00	.042	.036	.016	-.015	-.053	-.191	.066	.063	.042	.011	-.027	-.183
	85.00	.001	-.003	-.020	-.054	-.095	-.193	.013	.023	.005	-.024	-.073	-.182
	90.00	-.036	-.055	-.079	-.106	-.145	-.260	-.042	-.033	-.060	-.085	-.129	-.259
	95.00	-.163	-.147	-.161	-.206	-.227	-.287	-.096	-.121	-.145	-.192	-.214	-.289

TABLE I.- STEEL WING PRESSURE COEFFICIENT DATA FOR THE TEST RANGE OF ANGLE OF ATTACK AND MACH NUMBER - Continued

		Pressure coefficient, P, at:											
	Percent c	0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2	0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2
		M = 0.80		α = 23.46°				M = 0.80		α = 25.45°			
Upper surface	0.00	-.134	-.955	-.946	[illegible]	-.875	-.825	-.191	-1.032	-1.028	-1.029	-.964	-.865
	1.25	-.957	-.951	-.942	-.923	-.865	-.778	-1.049	-1.032	-1.029	-1.017	-.950	-.812
	2.50	-.949	-.951	-.936	-.913	-.865	-.776	-1.032	-1.032	-1.023	-1.001	-.950	-.811
	5.00	-.932	-.952	-.941	-.909	-.863	-.775	-1.021	-1.035	-1.029	-.995	-.947	-.809
	7.50	-.942	-.953	-.935	-.907	-.858	-.774	-1.026	-1.035	-1.022	-.995	-.940	-.808
	10.00	-.945	-.955	-.941	-.908	-.858	-.773	-1.030	-1.039	-1.030	-.994	-.939	-.807
	15.00	-.944	-.958	-.944	-.905	-.856	-.773	-1.030	-1.043	-1.032	-.991	-.936	-.807
	20.00	-.940	-.958	-.944	-.902	-.854	-.772	-1.022	-1.041	-1.031	-.987	-.934	-.804
	25.00	-.928	-.959	-.944	-.900	-.852	-.772	-.986	-1.039	-1.031	-.984	-.930	-.801
	30.00	-.909	-.953	-.944	-.898	-.849	-.773	-.932	-1.032	-1.030	-.983	-.927	-.803
	35.00	-.841	-.948	-.942	-.898	-.848	-.777	-.865	-1.021	-1.030	-.981	-.925	-.805
	40.00	-.801	-.931	-.943	-.899	-.847	-.778	-.830	-.991	-1.028	-.981	-.920	-.806
	45.00	-.775	-.919	-.941	-.895	-.847	-.779	-.800	-.954	-1.025	-.977	-.919	-.807
	50.00	-.753	-.887	-.940	-.895	-.846	-.780	-.732	-.888	-1.019	-.975	-.916	-.807
	55.00	-.724	-.860	-.934	-.892	-.845	-.781	-.687	-.814	-1.007	-.971	-.912	-.806
	60.00	-.717	-.831	-.927	-.881	-.845	-.780	-.637	-.736	-.981	-.958	-.909	-.803
	65.00	-.680	-.780	-.918	-.881	-.843	-.781	-.573	-.645	-.959	-.956	-.905	-.801
	70.00	-.640	-.740	-.903	-.881	-.836	-.778	-.548	-.586	-.935	-.952	-.900	-.796
	75.00	-.596	-.692	-.886	-.869	-.837	-.778	-.511	-.518	-.886	-.941	-.898	-.792
	80.00	-.555	-.632	-.862	-.866	-.836	-.774	-.467	-.459	-.835	-.933	-.895	-.786
	85.00	-.525	-.563	-.830	-.862	-.827	-.774	-.462	-.400	-.774	-.918	-.885	-.783
	90.00	-.448	-.506	-.781	-.858	-.820	-.771	-.415	-.384	-.678	-.896	-.871	-.781
	95.00	-.340	-.405	-.712	-.854	-.854	-.766	-.363	-.351	-.566	-.871	-.863	-.775
Lower surface	1.25	.852	.837	.715	.609	.511	.471	.813	.833	.702	.581	.476	.452
	2.50	.997	.873	.772	.685	.635	.497	.987	.894	.778	.677	.626	.489
	5.00	.970	.849	.768	.704	.666	.519	.983	.879	.795	.726	.679	.526
	7.50	.898	.799	.740	.684	.619	.503	.926	.839	.772	.711	.677	.522
	10.00	.834	.761	.705	.657	.625	.472	.867	.803	.746	.691	.659	.498
	15.00	.737	.690	.644	.602	.570	.409	.779	.737	.693	.643	.607	.441
	20.00	.669	.637	.588	.545	.521	.342	.716	.682	.641	.592	.567	.382
	25.00	.612	.577	.541	.501	.477	.282	.662	.630	.592	.548	.524	.323
	30.00	.569	.527	.491	.461	.432	.215	.611	.583	.546	.505	.477	.263
	35.00	.483	.479	.447	.410	.381	.179	.538	.536	.496	.459	.432	.224
	40.00	.461	.437	.409	.367	.347	.119	.519	.493	.463	.418	.399	.164
	45.00	.434	.395	.365	.326	.305	.055	.482	.448	.419	.379	.357	.101
	50.00	.363	.349	.324	.283	.261	.021	.417	.402	.381	.335	.325	.066
	55.00	.318	.308	.289	.290	.206	-.012	.372	.365	.345	.305	.263	.032
	60.00	.301	.269	.245	.206	.161	-.044	.352	.320	.298	.260	.215	-.001
	65.00	.239	.225	.209	.165	.118	-.076	.300	.283	.262	.214	.170	-.032
	70.00	.186	.196	.174	.132	.096	-.107	.239	.252	.225	.181	.147	-.064
	75.00	.154	.145	.127	.090	.047	-.137	.206	.197	.183	.139	.096	-.100
	80.00	.116	.114	.088	.049	.004	-.172	.157	.162	.140	.093	.051	-.134
	85.00	.062	.074	.049	.007	-.047	-.175	.106	.121	.099	.046	-.002	-.141
	90.00	.010	.018	-.011	-.059	-.105	-.234	.043	.066	.037	-.019	-.064	-.225
	95.00	-.049	-.047	-.100	-.173	-.200	-.292	-.031	.000	-.044	-.135	-.164	-.264

TABLE I. - STEEL WING PRESSURE COEFFICIENT DATA FOR THE TEST RANGE OF ANGLE OF ATTACK AND MACH NUMBER - Continued

		Pressure coefficient, P, at:											
	Percent c	0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2	0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2
		M = 0.90		α = -2.02°				M = 0.90		α = -0.04°			
Upper surface	0.00	.037	.676	.613	.574	.509	.371	.057	.702	.669	.654	.605	.623
	1.25	.311	.231	.248	.273	.269	.288	.236	.101	.099	.110	.075	.130
	2.50	.249	.139	.146	.167	.183	.171	.172	.047	.032	.029	.035	-.019
	5.00	.181	.110	.094	.101	.094	.111	.107	.033	.010	.006	-.016	-.037
	7.50	.166	.072	.059	.055	.049	.062	.101	.019	-.008	-.021	-.054	-.073
	10.00	.101	.049	.027	.033	.017	.014	.043	.003	-.034	-.047	-.056	-.106
	15.00	.073	.017	-.014	.004	-.021	-.073	.026	-.029	-.061	-.058	-.089	-.157
	20.00	.030	-.017	-.039	-.033	-.043	-.151	-.005	-.060	-.086	-.082	-.114	-.204
	25.00	-.001	-.029	-.057	-.067	-.080	-.232	-.037	-.066	-.091	-.121	-.145	-.251
	30.00	-.061	-.043	-.073	-.090	-.099	-.223	-.110	-.083	-.104	-.124	-.154	-.221
	35.00	-.056	-.071	-.093	-.107	-.125	-.214	-.089	-.106	-.127	-.140	-.174	-.206
	40.00	-.075	-.082	-.113	-.130	-.145	-.247	-.104	-.119	-.144	-.164	-.186	-.246
	45.00	-.088	-.095	-.135	-.148	-.171	-.205	-.119	-.125	-.162	-.176	-.210	-.212
	50.00	-.073	-.114	-.146	-.164	-.188	-.192	-.101	-.136	-.174	-.190	-.219	-.193
	55.00	-.082	-.131	-.161	-.177	-.200	-.169	-.101	-.156	-.184	-.195	-.230	-.166
	60.00	-.126	-.136	-.151	-.167	-.183	-.133	-.150	-.159	-.171	-.155	-.200	-.116
	65.00	-.115	-.121	-.142	-.145	-.141	-.138	-.129	-.139	-.153	-.146	-.142	-.116
	70.00	-.108	-.112	-.120	-.120	-.116	-.079	-.111	-.117	-.121	-.105	-.102	-.063
	75.00	-.093	-.097	-.098	-.088	-.082	-.062	-.094	-.093	-.088	-.064	-.062	-.041
	80.00	-.076	-.069	-.067	-.053	-.044	-.035	-.068	-.062	-.051	-.018	-.025	-.016
	85.00	-.063	-.039	-.035	-.021	-.005	-.013	-.050	-.026	-.016	.002	.011	.010
	90.00	-.020	-.020	-.008	.009	.029	.005	-.009	-.007	.014	.034	.048	.028
	95.00	.017	.010	.026	.045	.038	.005	.030	.024	.048	.071	.057	.030
Lower surface	1.25	.018	-.297	-.464	-.822	-.987	-1.067	.138	-.062	-.147	-.283	-.433	-.493
	2.50	-.036	-.234	-.308	-.528	-.832	-.999	.068	-.079	-.125	-.216	-.386	-.435
	5.00	-.060	-.168	-.253	-.310	-.424	-.899	.037	-.042	-.094	-.145	-.195	-.392
	7.50	-.078	-.172	-.243	-.313	-.332	-.751	.009	-.069	-.118	-.168	-.181	-.335
	10.00	-.118	-.174	-.233	-.321	-.347	-.576	-.025	-.082	-.134	-.174	-.184	-.274
	15.00	-.126	-.177	-.226	-.297	-.335	-.415	-.047	-.091	-.128	-.169	-.193	-.261
	20.00	-.152	-.162	-.235	-.297	-.342	-.399	-.083	-.091	-.147	-.200	-.209	-.288
	25.00	-.171	-.173	-.230	-.282	-.324	-.291	-.095	-.094	-.147	-.195	-.215	-.246
	30.00	-.147	-.192	-.238	-.288	-.327	-.252	-.080	-.124	-.156	-.200	-.224	-.236
	35.00	-.171	-.219	-.245	-.307	-.342	-.230	-.103	-.149	-.176	-.221	-.243	-.205
	40.00	-.207	-.212	-.257	-.320	-.345	-.247	-.135	-.153	-.188	-.231	-.247	-.229
	45.00	-.180	-.231	-.280	-.339	-.356	-.247	-.119	-.175	-.210	-.253	-.264	-.232
	50.00	-.234	-.259	-.301	-.349	-.350	-.228	-.175	-.196	-.223	-.265	-.264	-.207
	55.00	-.245	-.268	-.298	-.326	-.320	-.186	-.182	-.207	-.223	-.250	-.264	-.163
	60.00	-.254	-.264	-.284	-.288	-.233	-.145	-.194	-.207	-.211	-.220	-.212	-.127
	65.00	-.228	-.225	-.234	-.220	-.161	-.128	-.169	-.178	-.178	-.180	-.158	-.104
	70.00	-.192	-.176	-.168	-.138	-.098	-.104	-.145	-.141	-.137	-.122	-.091	-.076
	75.00	-.172	-.146	-.125	-.102	-.075	-.086	-.145	-.122	-.101	-.082	-.065	-.056
	80.00	-.115	-.103	-.085	-.059	-.041	-.065	-.094	-.082	-.065	-.042	-.031	-.033
	85.00	-.077	-.060	-.042	-.024	-.016	-.011	-.061	-.045	-.019	-.007	-.005	.022
	90.00	-.036	-.026	-.011	.011	.021	-.016	-.019	-.012	.016	.032	.034	.017
	95.00	.006	.007	.026	.039	.055	.026	.022	.019	.046	.056	.067	.056
		M = 0.90		α = 1.96°				M = 0.90		α = 3.91°			
Upper surface	0.00	.060	.642	.542	.557	.528	.642	.041	.531	.390	.388	.342	.566
	1.25	.071	-.231	-.314	-.644	-.765	-.737	-.065	-.697	-.831	-.986	-1.065	-.970
	2.50	.011	-.214	-.289	-.382	-.644	-.896	-.129	-.416	-.739	-.833	-.959	-1.156
	5.00	-.082	-.174	-.221	-.280	-.345	-.814	-.248	-.332	-.496	-.674	-.777	-1.082
	7.50	-.095	-.140	-.213	-.271	-.341	-.707	-.240	-.265	-.314	-.634	-.749	-1.003
	10.00	-.104	-.137	-.210	-.275	-.311	-.582	-.232	-.248	-.316	-.567	-.657	-.943
	15.00	-.088	-.157	-.209	-.254	-.311	-.398	-.182	-.252	-.329	-.362	-.641	-.862
	20.00	-.091	-.176	-.225	-.254	-.313	-.352	-.175	-.280	-.339	-.362	-.543	-.826
	25.00	-.148	-.170	-.222	-.261	-.308	-.297	-.242	-.259	-.322	-.372	-.425	-.737
	30.00	-.238	-.192	-.228	-.262	-.315	-.246	-.335	-.283	-.334	-.372	-.424	-.613
	35.00	-.190	-.213	-.235	-.273	-.325	-.224	-.283	-.308	-.344	-.382	-.438	-.483
	40.00	-.202	-.210	-.248	-.287	-.326	-.258	-.301	-.312	-.356	-.403	-.446	-.449
	45.00	-.219	-.214	-.257	-.295	-.342	-.225	-.317	-.312	-.360	-.409	-.467	-.326
	50.00	-.197	-.226	-.270	-.302	-.332	-.205	-.294	-.317	-.365	-.419	-.472	-.257
	55.00	-.177	-.248	-.283	-.299	-.302	-.174	-.265	-.338	-.383	-.418	-.451	-.208
	60.00	-.234	-.248	-.262	-.269	-.221	-.125	-.336	-.352	-.382	-.400	-.323	-.151
	65.00	-.199	-.204	-.209	-.170	-.131	-.126	-.299	-.308	-.326	-.260	-.148	-.126
	70.00	-.163	-.157	-.148	-.104	-.088	-.077	-.241	-.227	-.202	-.129	-.080	-.087
	75.00	-.119	-.112	-.094	-.066	-.056	-.057	-.161	-.143	-.115	-.065	-.046	-.077
	80.00	-.078	-.073	-.054	-.026	-.023	-.051	-.103	-.091	-.072	-.030	-.020	-.061
	85.00	-.057	-.035	-.014	.004	.016	-.005	-.075	-.052	-.052	-.001	.013	-.042
	90.00	-.010	-.008	.016	.035	.049	.016	-.030	-.025	-.002	.027	.037	-.029
	95.00	.029	.024	.051	.074	.063	.027	.010	.009	.032	.060	.048	-.021
Lower surface	1.25	.299	.247	.233	.257	.293	.212	.389	.368	.352	.393	.429	.332
	2.50	.236	.161	.165	.181	.182	.156	.321	.268	.276	.293	.307	.262
	5.00	.190	.129	.120	.124	.130	.100	.267	.216	.211	.208	.228	.193
	7.50	.147	.082	.070	.078	.093	.045	.215	.155	.148	.152	.177	.125
	10.00	.111	.057	.034	.048	.053	.010	.175	.123	.103	.119	.127	.080
	15.00	.073	.032	.015	.005	.007	-.066	.127	.091	.081	.076	.074	-.010
	20.00	.033	.007	-.012	-.038	-.029	-.140	.081	.080	.042	.014	.034	-.092
	25.00	.016	.002	-.021	-.056	-.057	-.193	.059	.048	.015	-.009	-.010	-.167
	30.00	.016	-.022	-.042	-.071	-.079	-.202	.051	.012	-.010	-.030	-.035	-.199
	35.00	-.016	-.048	-.069	-.094	-.109	-.173	.004	-.016	-.035	-.057	-.068	-.196
	40.00	-.045	-.059	-.083	-.114	-.122	-.194	-.015	-.033	-.058	-.079	-.087	-.248
	45.00	-.038	-.087	-.105	-.141	-.144	-.194	-.010	-.062	-.080	-.107	-.116	-.274
	50.00	-.093	-.108	-.124	-.154	-.164	-.176	-.070	-.084	-.104	-.128	-.138	-.247
	55.00	-.102	-.121	-.128	-.149	-.181	-.143	-.082	-.099	-.108	-.128	-.163	-.229
	60.00	-.116	-.125	-.130	-.145	-.164	-.119	-.097	-.106	-.117	-.128	-.163	-.179
	65.00	-.102	-.113	-.117	-.133	-.146	-.109	-.090	-.105	-.111	-.128	-.161	-.139
	70.00	-.092	-.094	-.098	-.103	-.094	-.082	-.083	-.088	-.095	-.110	-.117	-.100
	75.00	-.116	-.094	-.085	-.083	-.075	-.061	-.116	-.095	-.090	-.093	-.095	-.081
	80.00	-.078	-.064	-.052	-.045	-.037	-.041	-.085	-.072	-.065	-.063	-.062	-.061
	85.00	-.051	-.040	-.017	-.007	-.011	.013	-.062	-.046	-.033	-.025	-.035	-.005
	90.00	-.013	-.007	.011	.029	.025	.003	-.031	-.021	-.002	.010	.006	-.020
	95.00	.028	.020	.042	.047	.057	.042	.014	.004	.024	.033	.035	.018

TABLE I.- STEEL WING PRESSURE COEFFICIENT DATA FOR THE TEST RANGE OF ANGLE OF ATTACK AND MACH NUMBER - Continued

	Percent c	Pressure coefficient, P, at: 0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2	0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2
		M = 0.90		α = 5.88°				M = 0.90		α = 7.82°			
Upper surface	0.00	.048	.395	.235	.184	.100	.448	.044	.222	.044	-.046	-.197	.308
	1.25	-.205	-1.051	-1.139	-1.198	-1.240	-.879	-.355	-1.262	-1.309	-1.318	-1.059	-.618
	2.50	-.266	-.817	-1.055	-1.110	-1.151	-1.007	-.426	-1.111	-1.239	-1.261	-1.036	-.656
	5.00	-.410	-.616	-.909	-.990	-1.049	-.968	-.580	-.863	-1.152	-1.177	-.976	-.652
	7.50	-.390	-.392	-.724	-.947	-1.022	-.941	-.540	-.600	-1.039	-1.126	-.950	-.648
	10.00	-.364	-.360	-.601	-.897	-.943	-.917	-.512	-.537	-.956	-1.090	-.927	-.645
	15.00	-.272	-.346	-.422	-.718	-.931	-.895	-.399	-.488	-.693	-.984	-.912	-.650
	20.00	-.253	-.359	-.419	-.621	-.878	-.874	-.368	-.468	-.579	-.900	-.872	-.645
	25.00	-.315	-.356	-.399	-.528	-.772	-.844	-.409	-.433	-.504	-.762	-.841	-.657
	30.00	-.402	-.359	-.407	-.467	-.748	-.780	-.484	-.452	-.492	-.616	-.820	-.659
	35.00	-.348	-.378	-.417	-.464	-.697	-.686	-.438	-.466	-.490	-.569	-.785	-.646
	40.00	-.374	-.386	-.456	-.479	-.587	-.665	-.466	-.475	-.500	-.552	-.706	-.634
	45.00	-.389	-.389	-.444	-.492	-.551	-.567	-.466	-.475	-.505	-.540	-.674	-.612
	50.00	-.370	-.389	-.445	-.504	-.497	-.504	-.461	-.471	-.495	-.544	-.629	-.593
	55.00	-.332	-.408	-.463	-.514	-.469	-.475	-.415	-.483	-.483	-.544	-.599	-.581
	60.00	-.412	-.433	-.476	-.514	-.398	-.428	-.491	-.503	-.476	-.541	-.571	-.560
	65.00	-.398	-.416	-.458	-.399	-.283	-.368	-.477	-.470	-.463	-.480	-.528	-.526
	70.00	-.346	-.343	-.353	-.209	-.190	-.336	-.413	-.395	-.420	-.384	-.473	-.513
	75.00	-.245	-.215	-.163	-.095	-.132	-.357	-.301	-.280	-.305	-.344	-.430	-.523
	80.00	-.126	-.107	-.065	-.040	-.093	-.315	-.184	-.180	-.176	-.283	-.391	-.499
	85.00	-.070	-.052	-.020	-.004	-.050	-.260	-.128	-.111	-.093	-.232	-.357	-.463
	90.00	-.017	-.018	.015	.024	-.015	-.260	-.058	-.069	-.045	-.179	-.322	-.472
	95.00	.026	.017	.043	.059	.012	-.239	-.009	-.023	-.007	-.147	-.287	-.459
Lower surface	1.25	.516	.508	.489	.515	.545	.443	.630	.621	.590	.603	.617	.508
	2.50	.453	.403	.406	.416	.432	.369	.582	.521	.515	.521	.521	.442
	5.00	.394	.332	.322	.329	.345	.301	.510	.436	.426	.423	.435	.377
	7.50	.334	.264	.261	.252	.288	.253	.440	.363	.356	.356	.377	.308
	10.00	.284	.229	.206	.220	.233	.180	.387	.329	.309	.314	.323	.254
	15.00	.227	.188	.170	.167	.174	.081	.317	.272	.256	.255	.258	.157
	20.00	.269	.166	.125	.094	.123	-.001	.248	.243	.209	.183	.202	.073
	25.00	.140	.125	.086	.068	.078	-.077	.216	.192	.163	.148	.154	.002
	30.00	.124	.086	.065	.040	.046	-.125	.192	.154	.133	.119	.120	-.047
	35.00	.064	.054	.036	.012	.009	-.134	.118	.121	.103	.085	.077	-.064
	40.00	.055	.034	.013	-.014	-.013	-.190	.113	.096	.078	.053	.051	-.123
	45.00	.055	.004	-.011	-.045	-.040	-.230	.107	.059	.050	.018	.020	-.171
	50.00	-.009	-.021	-.036	-.068	-.067	-.244	.038	.034	.018	-.006	-.016	-.196
	55.00	-.025	-.040	-.049	-.071	-.098	-.232	.019	.009	.004	-.019	-.051	-.193
	60.00	-.039	-.054	-.063	-.081	-.111	-.212	.005	-.009	-.018	-.034	-.076	-.203
	65.00	-.039	-.054	-.059	-.090	-.123	-.201	-.004	-.019	-.025	-.057	-.097	-.218
	70.00	-.032	-.045	-.059	-.084	-.100	-.160	-.004	-.019	-.033	-.057	-.085	-.193
	75.00	-.072	-.057	-.056	-.075	-.091	-.145	-.047	-.037	-.043	-.062	-.100	-.193
	80.00	-.049	-.041	-.043	-.050	-.070	-.124	-.042	-.032	-.034	-.058	-.095	-.189
	85.00	-.038	-.021	-.015	-.021	-.049	-.069	-.037	-.025	-.020	-.046	-.102	-.141
	90.00	-.008	-.001	.004	.004	-.017	-.098	-.023	-.015	-.013	-.041	-.094	-.181
	95.00	.028	.018	.037	.024	.005	-.088	.007	-.004	.009	-.051	-.114	-.174
		M = 0.90		α = 9.87°				M = 0.90		α = 11.82°			
Upper surface	0.00	.037	.026	-.174	-.324	-.481	.124	.025	-.211	-.442	-.634	-.784	-.111
	1.25	-.492	-1.377	-1.387	-1.093	-.826	-.584	-.651	-1.435	-1.171	-.907	-.790	-.579
	2.50	-.572	-1.251	-1.333	-1.065	-.832	-.592	-.752	-1.374	-1.149	-.892	-.793	-.577
	5.00	-.728	-1.076	-1.275	-1.053	-.809	-.584	-.901	-1.260	-1.143	-.895	-.757	-.573
	7.50	-.690	-.874	-1.181	-1.032	-.798	-.579	-.850	-1.147	-1.115	-.890	-.748	-.565
	10.00	-.653	-.799	-1.126	-1.002	-.811	-.579	-.808	-1.081	-1.101	-.882	-.760	-.565
	15.00	-.525	-.730	-.928	-.963	-.800	-.579	-.675	-1.014	-1.007	-.882	-.760	-.565
	20.00	-.476	-.619	-.805	-.914	-.782	-.579	-.586	-.861	-.948	-.873	-.753	-.565
	25.00	-.489	-.519	-.712	-.848	-.774	-.584	-.558	-.610	-.877	-.857	-.757	-.566
	30.00	-.551	-.492	-.670	-.781	-.765	-.592	-.599	-.528	-.836	-.839	-.751	-.570
	35.00	-.501	-.499	-.620	-.746	-.757	-.596	-.557	-.542	-.801	-.817	-.740	-.574
	40.00	-.522	-.502	-.599	-.713	-.735	-.593	-.579	-.542	-.776	-.790	-.727	-.576
	45.00	-.497	-.498	-.585	-.676	-.718	-.593	-.548	-.542	-.758	-.754	-.720	-.580
	50.00	-.497	-.481	-.575	-.650	-.692	-.591	-.548	-.534	-.733	-.734	-.711	-.582
	55.00	-.437	-.487	-.568	-.640	-.677	-.589	-.500	-.541	-.713	-.714	-.706	-.583
	60.00	-.494	-.509	-.551	-.621	-.659	-.583	-.557	-.554	-.688	-.692	-.699	-.584
	65.00	-.487	-.498	-.523	-.588	-.639	-.573	-.535	-.535	-.664	-.671	-.692	-.585
	70.00	-.465	-.465	-.497	-.545	-.609	-.568	-.520	-.515	-.617	-.649	-.677	-.588
	75.00	-.413	-.403	-.440	-.539	-.584	-.576	-.480	-.490	-.576	-.630	-.667	-.588
	80.00	-.325	-.320	-.372	-.501	-.563	-.566	-.440	-.448	-.530	-.600	-.650	-.590
	85.00	-.229	-.235	-.285	-.456	-.536	-.554	-.375	-.389	-.486	-.580	-.629	-.590
	90.00	-.141	-.163	-.225	-.416	-.508	-.559	-.280	-.304	-.432	-.546	-.607	-.594
	95.00	-.073	-.106	-.163	-.393	-.484	-.550	-.184	-.230	-.374	-.527	-.598	-.587
Lower surface	1.25	.733	.709	.660	.653	.651	.542	.815	.773	.713	.689	.668	.565
	2.50	.701	.610	.593	.585	.584	.483	.799	.690	.663	.643	.626	.518
	5.00	.624	.525	.508	.493	.495	.428	.719	.600	.576	.557	.551	.471
	7.50	.546	.454	.440	.423	.440	.360	.631	.529	.511	.493	.494	.409
	10.00	.480	.414	.394	.381	.384	.309	.559	.485	.472	.452	.441	.359
	15.00	.403	.346	.334	.320	.317	.214	.474	.414	.396	.388	.374	.264
	20.00	.327	.311	.279	.254	.263	.131	.395	.379	.345	.320	.316	.181
	25.00	.290	.259	.231	.203	.212	.066	.352	.320	.297	.276	.269	.119
	30.00	.259	.218	.196	.177	.174	.010	.314	.275	.260	.237	.228	.063
	35.00	.179	.182	.160	.138	.130	-.019	.227	.235	.221	.196	.180	.032
	40.00	.173	.152	.128	.108	.100	-.077	.225	.204	.188	.159	.150	-.034
	45.00	.165	.114	.100	.071	.067	-.127	.211	.163	.149	.120	.114	-.082
	50.00	.095	.081	.071	.038	.029	-.156	.143	.129	.117	.090	.072	-.115
	55.00	.066	.057	.046	.021	-.014	-.164	.105	.100	.091	.069	.031	-.120
	60.00	.056	.035	.021	-.003	-.041	-.183	.101	.073	.065	.040	-.001	-.151
	65.00	.035	.014	.009	-.031	-.067	-.204	.068	.049	.048	.005	-.029	-.175
	70.00	.020	.010	-.008	-.043	-.087	-.192	.042	.037	.025	-.008	-.037	-.170
	75.00	-.006	-.015	-.026	-.056	-.093	-.198	.022	.011	.005	-.030	-.063	-.181
	80.00	-.022	-.020	-.034	-.059	-.096	-.202	-.001	-.006	-.013	-.040	-.074	-.191
	85.00	-.034	-.024	-.031	-.062	-.109	-.160	-.030	-.018	-.024	-.051	-.090	-.155
	90.00	-.030	-.030	-.042	-.075	-.115	-.209	-.044	-.037	-.049	-.077	-.107	-.207
	95.00	-.024	-.035	-.047	-.117	-.148	-.195	-.078	-.075	-.082	-.117	-.149	-.197

TABLE I. - STEEL WING PRESSURE COEFFICIENT DATA FOR THE TEST RANGE OF ANGLE OF ATTACK AND MACH NUMBER - Continued

		Pressure coefficient, P, at:											
	Percent c	0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2	0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2
		M = 0.90	α = 15.46°					M = 0.90	α = 17.43°				
Upper surface	0.00	.026	-.674	-.899	-1.011	-.691	-.492	.010	-.814	-.967	-.787	-.667	-.601
	1.25	-.880	-1.177	-.924	-.785	-.659	-.560	-.813	-.862	-.789	-.757	-.650	-.583
	2.50	-1.004	-1.200	-.916	-.730	-.659	-.560	-.799	-.867	-.786	-.715	-.647	-.584
	5.00	-1.169	-1.180	-.930	-.716	-.654	-.558	-.807	-.872	-.794	-.696	-.645	-.582
	7.50	-1.076	-1.159	-.923	-.710	-.648	-.559	-.810	-.875	-.792	-.691	-.644	-.583
	10.00	-.982	-1.120	-.930	-.710	-.647	-.559	-.821	-.871	-.809	-.691	-.638	-.585
	15.00	-.795	-1.086	-.909	-.717	-.645	-.561	-.810	-.870	-.818	-.687	-.643	-.586
	20.00	-.723	-1.028	-.882	-.717	-.640	-.561	-.780	-.857	-.797	-.681	-.643	-.586
	25.00	-.679	-.961	-.852	-.717	-.638	-.560	-.742	-.838	-.766	-.685	-.639	-.585
	30.00	-.641	-.874	-.826	-.710	-.635	-.560	-.714	-.822	-.760	-.681	-.639	-.583
	35.00	-.602	-.803	-.807	-.704	-.635	-.562	-.654	-.806	-.756	-.680	-.638	-.580
	40.00	-.583	-.724	-.788	-.699	-.632	-.563	-.625	-.779	-.751	-.680	-.636	-.588
	45.00	-.541	-.678	-.775	-.692	-.632	-.566	-.586	-.753	-.745	-.678	-.638	-.586
	50.00	-.552	-.634	-.764	-.688	-.631	-.567	-.589	-.722	-.736	-.677	-.638	-.588
	55.00	-.567	-.610	-.750	-.682	-.628	-.568	-.583	-.698	-.729	-.672	-.638	-.593
	60.00	-.560	-.599	-.729	-.671	-.628	-.570	-.589	-.684	-.724	-.666	-.639	-.596
	65.00	-.531	-.573	-.708	-.663	-.625	-.572	-.573	-.670	-.714	-.659	-.638	-.593
	70.00	-.521	-.565	-.696	-.663	-.623	-.574	-.565	-.658	-.708	-.658	-.632	-.597
	75.00	-.488	-.557	-.682	-.653	-.622	-.575	-.559	-.644	-.697	-.658	-.633	-.604
	80.00	-.488	-.547	-.665	-.648	-.620	-.578	-.559	-.638	-.691	-.655	-.632	-.604
	85.00	-.494	-.533	-.654	-.645	-.613	-.581	-.560	-.625	-.684	-.657	-.624	-.602
	90.00	-.471	-.512	-.635	-.641	-.604	-.582	-.554	-.612	-.675	-.653	-.618	-.608
	95.00	-.371	-.464	-.615	-.637	-.612	-.581	-.484	-.582	-.663	-.658	-.625	-.609
Lower surface	1.25	.928	.851	.775	.719	.678	.586	.959	.876	.788	.724	.669	.585
	2.50	.964	.799	.751	.712	.684	.562	1.014	.844	.779	.737	.701	.572
	5.00	.872	.724	.682	.650	.634	.534	.918	.772	.726	.693	.667	.557
	7.50	.769	.660	.625	.598	.588	.486	.825	.709	.673	.646	.629	.515
	10.00	.697	.611	.583	.560	.544	.439	.750	.660	.631	.608	.589	.471
	15.00	.603	.543	.513	.428	.478	.354	.654	.590	.565	.540	.516	.390
	20.00	.527	.500	.451	.422	.422	.277	.578	.546	.504	.476	.460	.314
	25.00	.471	.435	.402	.378	.373	.211	.522	.487	.456	.431	.414	.250
	30.00	.431	.384	.360	.338	.328	.148	.477	.441	.414	.392	.370	.184
	35.00	.336	.343	.319	.294	.260	.119	.388	.393	.370	.343	.323	.155
	40.00	.336	.304	.281	.257	.251	.055	.382	.354	.335	.306	.292	.093
	45.00	.312	.269	.246	.217	.210	-.005	.355	.318	.297	.267	.251	.032
	50.00	.245	.227	.207	.179	.172	-.033	.288	.273	.257	.226	.212	.001
	55.00	.204	.194	.179	.156	.125	-.055	.242	.238	.228	.200	.164	-.025
	60.00	.194	.165	.145	.123	.087	-.079	.232	.206	.190	.164	.127	-.051
	65.00	.150	.128	.120	.088	.052	-.101	.188	.170	.162	.133	.092	-.072
	70.00	.111	.111	.096	.068	.042	-.113	.142	.150	.134	.111	.080	-.090
	75.00	.088	.075	.063	.041	.012	-.127	.118	.108	.100	.078	.044	-.107
	80.00	.061	.052	.039	.020	-.009	-.144	.086	.084	.077	.057	.020	-.127
	85.00	.030	.028	.021	-.003	-.037	-.131	.052	.058	.052	.031	-.009	-.116
	90.00	.007	-.002	-.022	-.033	-.061	-.174	.014	.022	.008	-.006	-.042	-.166
	95.00	-.053	-.059	-.068	-.098	-.112	-.180	-.018	-.040	-.043	-.078	-.098	-.174
		M = 0.90	α = 19.50°					M = 0.90	α = 21.59°				
Upper surface	0.00	-.014	-.814	-.778	-.771	-.706	-.667	-.092	-.830	-.828	-.814	-.784	-.750
	1.25	-.752	-.755	-.758	-.762	-.695	-.653	-.819	-.819	-.821	-.807	-.777	-.722
	2.50	-.735	-.752	-.755	-.746	-.694	-.654	-.808	-.824	-.813	-.800	-.778	-.721
	5.00	-.725	-.752	-.759	-.730	-.694	-.654	-.800	-.825	-.819	-.795	-.776	-.718
	7.50	-.730	-.755	-.749	-.730	-.692	-.654	-.808	-.827	-.810	-.795	-.771	-.720
	10.00	-.734	-.755	-.753	-.730	-.685	-.655	-.806	-.828	-.817	-.795	-.772	-.719
	15.00	-.742	-.769	-.759	-.728	-.692	-.658	-.814	-.833	-.819	-.794	-.772	-.720
	20.00	-.752	-.769	-.759	-.723	-.694	-.658	-.817	-.831	-.821	-.789	-.771	-.720
	25.00	-.752	-.764	-.757	-.726	-.690	-.657	-.811	-.833	-.821	-.790	-.768	-.720
	30.00	-.752	-.768	-.760	-.724	-.691	-.653	-.815	-.833	-.822	-.789	-.768	-.720
	35.00	-.713	-.768	-.760	-.724	-.691	-.651	-.763	-.831	-.822	-.789	-.768	-.721
	40.00	-.692	-.765	-.760	-.726	-.691	-.661	-.738	-.824	-.821	-.791	-.767	-.723
	45.00	-.661	-.759	-.759	-.726	-.694	-.658	-.709	-.820	-.821	-.788	-.767	-.725
	50.00	-.666	-.749	-.755	-.725	-.695	-.661	-.709	-.811	-.821	-.788	-.767	-.726
	55.00	-.647	-.740	-.754	-.723	-.697	-.667	-.689	-.801	-.817	-.786	-.766	-.728
	60.00	-.662	-.738	-.754	-.718	-.700	-.667	-.718	-.801	-.815	-.776	-.765	-.728
	65.00	-.647	-.731	-.749	-.714	-.697	-.664	-.705	-.787	-.813	-.777	-.765	-.727
	70.00	-.635	-.718	-.746	-.714	-.694	-.669	-.693	-.780	-.810	-.779	-.761	-.729
	75.00	-.621	-.709	-.739	-.714	-.694	-.678	-.676	-.767	-.805	-.776	-.762	-.730
	80.00	-.624	-.703	-.737	-.714	-.695	-.677	-.681	-.754	-.799	-.776	-.762	-.730
	85.00	-.622	-.694	-.733	-.714	-.688	-.674	-.684	-.733	-.793	-.776	-.754	-.729
	90.00	-.626	-.688	-.726	-.709	-.681	-.681	-.675	-.717	-.783	-.773	-.745	-.729
	95.00	-.553	-.651	-.719	-.713	-.689	-.683	-.562	-.670	-.770	-.771	-.757	-.726
Lower surface	1.25	.971	.898	.793	.712	.642	.567	.948	.891	.783	.695	.608	.546
	2.50	1.048	.885	.807	.741	.698	.570	1.050	.908	.816	.743	.692	.560
	5.00	.968	.830	.768	.714	.684	.567	.990	.860	.793	.733	.698	.570
	7.50	.879	.769	.719	.677	.652	.533	.908	.803	.754	.704	.713	.547
	10.00	.806	.725	.679	.643	.613	.495	.840	.767	.719	.676	.656	.514
	15.00	.714	.653	.616	.584	.553	.422	.749	.691	.659	.622	.588	.449
	20.00	.638	.589	.557	.523	.506	.353	.678	.629	.605	.563	.541	.381
	25.00	.581	.549	.510	.476	.459	.288	.622	.587	.556	.518	.497	.321
	30.00	.538	.502	.466	.438	.416	.225	.576	.542	.508	.479	.452	.258
	35.00	.448	.452	.421	.389	.368	.192	.489	.494	.463	.431	.405	.219
	40.00	.441	.413	.385	.351	.337	.133	.480	.452	.420	.387	.372	.163
	45.00	.414	.374	.343	.310	.296	.069	.448	.413	.389	.350	.324	.100
	50.00	.347	.332	.306	.270	.257	.036	.381	.365	.346	.312	.293	.065
	55.00	.304	.293	.274	.246	.210	.007	.339	.330	.316	.286	.245	.037
	60.00	.291	.260	.236	.209	.169	-.021	.325	.297	.277	.247	.203	.003
	65.00	.237	.224	.205	.171	.132	-.043	.271	.254	.242	.210	.166	-.022
	70.00	.192	.201	.176	.148	.118	-.068	.220	.227	.215	.184	.149	-.048
	75.00	.167	.160	.143	.117	.080	-.088	.188	.184	.177	.149	.111	-.069
	80.00	.132	.131	.116	.088	.052	-.112	.156	.156	.143	.117	.079	-.098
	85.00	.092	.096	.087	.055	.021	-.103	.105	.121	.112	.086	.040	-.091
	90.00	.043	.059	.039	.022	-.016	-.159	.058	.074	.063	.043	.001	-.154
	95.00	-.005	-.013	-.020	-.055	-.079	-.167	-.012	.002	-.004	-.045	-.065	-.167

TABLE I.- STEEL WING PRESSURE COEFFICIENT DATA FOR THE TEST RANGE OF ANGLE OF ATTACK AND MACH NUMBER - Continued

	Percent c	Pressure coefficient, P, at:											
		0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2	0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2
		M = 0.90		α = 23.68°				M = 0.90		α = 25.78°			
Upper surface	0.00	-.246	-.998	-.991	-.975	-.947	-.898	-.310	-1.093	-1.092	-1.062	-1.033	-.956
	1.25	-.983	-.991	-.986	-.970	-.934	-.859	-1.082	-1.093	-1.086	-1.061	-1.021	-.922
	2.50	-.970	-.990	-.979	-.960	-.931	-.857	-1.070	-1.086	-1.083	-1.058	-1.014	-.920
	5.00	-.969	-.993	-.985	-.951	-.928	-.853	-1.070	-1.087	-1.089	-1.047	-1.010	-.916
	7.50	-.971	-.992	-.974	-.951	-.923	-.854	-1.075	-1.088	-1.076	-1.047	-1.007	-.917
	10.00	-.972	-.996	-.981	-.951	-.920	-.854	-1.079	-1.088	-1.084	-1.049	-.997	-.915
	15.00	-.979	-1.006	-.985	-.950	-.923	-.853	-1.079	-1.101	-1.091	-1.048	-1.005	-.914
	20.00	-.975	-1.004	-.985	-.942	-.925	-.850	-1.064	-1.098	-1.091	-1.039	-1.009	-.913
	25.00	-.931	-.998	-.981	-.946	-.921	-.850	-.952	-1.087	-1.082	-1.044	-1.002	-.908
	30.00	-.889	-.994	-.983	-.945	-.920	-.849	-.894	-1.085	-1.087	-1.039	-1.002	-.903
	35.00	-.820	-.990	-.980	-.941	-.919	-.849	-.857	-1.075	-1.087	-1.037	-1.000	-.900
	40.00	-.795	-.974	-.980	-.941	-.915	-.854	-.835	-1.047	-1.086	-1.037	-.993	-.909
	45.00	-.773	-.948	-.977	-.938	-.915	-.853	-.787	-.990	-1.081	-1.031	-.995	-.904
	50.00	-.738	-.918	-.973	-.937	-.913	-.853	-.740	-.923	-1.071	-1.032	-.990	-.905
	55.00	-.714	-.888	-.970	-.932	-.910	-.856	-.715	-.853	-1.062	-1.024	-.987	-.906
	60.00	-.732	-.854	-.964	-.918	-.909	-.856	-.695	-.803	-1.051	-1.012	-.985	-.904
	65.00	-.709	-.804	-.958	-.918	-.905	-.852	-.663	-.724	-1.032	-1.004	-.979	-.896
	70.00	-.685	-.780	-.951	-.918	-.901	-.852	-.634	-.677	-1.007	-.998	-.969	-.893
	75.00	-.665	-.753	-.938	-.916	-.898	-.855	-.598	-.628	-.966	-1.000	-.965	-.897
	80.00	-.656	-.735	-.920	-.914	-.896	-.851	-.574	-.578	-.926	-.992	-.959	-.892
	85.00	-.652	-.694	-.901	-.914	-.884	-.848	-.545	-.511	-.876	-.982	-.945	-.880
	90.00	-.611	-.649	-.866	-.908	-.876	-.846	-.468	-.455	-.817	-.968	-.931	-.883
	95.00	-.465	-.551	-.786	-.900	-.890	-.842	-.357	-.375	-.681	-.943	-.940	-.878
Lower surface	1.25	.913	.893	.774	.669	.569	.516	.871	.890	.762	.641	.534	.489
	2.50	1.045	.928	.830	.740	.684	.544	1.034	.942	.831	.731	.671	.531
	5.00	1.012	.899	.825	.760	.712	.568	1.026	.927	.842	.771	.721	.567
	7.50	.948	.851	.793	.738	.730	.558	.971	.890	.822	.759	.718	.566
	10.00	.881	.812	.763	.713	.677	.531	.913	.855	.799	.741	.702	.548
	15.00	.792	.748	.703	.661	.622	.471	.831	.791	.745	.695	.654	.495
	20.00	.723	.686	.651	.609	.577	.411	.767	.736	.693	.646	.614	.438
	25.00	.669	.640	.604	.567	.537	.352	.714	.686	.646	.603	.578	.385
	30.00	.622	.591	.560	.528	.494	.293	.669	.638	.605	.567	.534	.327
	35.00	.545	.542	.515	.480	.447	.255	.592	.595	.563	.523	.491	.288
	40.00	.529	.506	.476	.439	.417	.196	.574	.556	.527	.486	.459	.231
	45.00	.495	.456	.436	.401	.377	.130	.543	.511	.486	.441	.419	.168
	50.00	.436	.416	.395	.359	.337	.095	.478	.467	.444	.402	.381	.135
	55.00	.388	.382	.364	.331	.287	.062	.433	.431	.408	.375	.332	.099
	60.00	.370	.338	.321	.292	.246	.032	.412	.389	.368	.336	.288	.069
	65.00	.308	.299	.290	.252	.206	.000	.361	.347	.334	.294	.250	.035
	70.00	.258	.273	.251	.224	.188	-.032	.300	.320	.303	.269	.228	.004
	75.00	.231	.226	.218	.191	.146	-.055	.274	.270	.260	.225	.185	-.027
	80.00	.193	.193	.181	.157	.109	-.082	.233	.238	.225	.191	.147	-.057
	85.00	.138	.155	.148	.122	.070	-.082	.180	.200	.190	.151	.105	-.057
	90.00	.085	.110	.096	.072	.025	-.148	.123	.150	.133	.100	.054	-.128
	95.00	.001	.036	.026	-.018	-.049	-.168	.039	.079	.056	.002	-.024	-.150

TABLE I.- STEEL WING PRESSURE COEFFICIENT DATA FOR THE TEST RANGE OF ANGLE OF ATTACK AND MACH NUMBER - Continued

	Percent c	0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2	0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2
		Pressure coefficient, P, at:											
		M = 0.94		α = -2.02°				M = 0.94			α = 0°		
Upper surface	0.00	.041	.701	.645	.609	.549	.456	.056	.730	.678	.685	.648	.685
	1.25	.323	.238	.257	.271	.256	.290	.217	.038	.021	-.024	-.058	.060
	2.50	.266	.146	.162	.170	.174	.165	.152	-.013	-.029	-.065	-.093	-.118
	5.00	.201	.125	.108	.101	.093	.115	.088	-.016	-.027	-.074	-.115	-.120
	7.50	.167	.089	.076	.059	.046	.068	.060	-.016	-.046	-.091	-.133	-.145
	10.00	.132	.061	.040	.032	.015	.023	.035	-.025	-.067	-.111	-.142	-.161
	15.00	.090	.030	.002	.010	-.024	-.060	.015	-.049	-.086	-.109	-.156	-.197
	20.00	.040	-.003	-.026	-.023	-.051	-.136	-.004	-.077	-.115	-.133	-.164	-.233
	25.00	.014	-.022	-.046	-.065	-.088	-.239	-.053	-.089	-.128	-.154	-.189	-.298
	30.00	-.041	-.038	-.060	-.089	-.101	-.268	-.131	-.106	-.136	-.171	-.199	-.298
	35.00	-.046	-.066	-.084	-.110	-.134	-.285	-.104	-.135	-.144	-.189	-.222	-.294
	40.00	-.069	-.076	-.108	-.137	-.155	-.353	-.122	-.133	-.172	-.208	-.241	-.340
	45.00	-.083	-.093	-.134	-.163	-.188	-.346	-.142	-.147	-.187	-.229	-.265	-.334
	50.00	-.073	-.115	-.162	-.186	-.219	-.366	-.124	-.161	-.214	-.250	-.293	-.358
	55.00	-.092	-.152	-.190	-.217	-.252	-.368	-.124	-.198	-.240	-.274	-.315	-.368
	60.00	-.158	-.173	-.193	-.226	-.278	-.348	-.206	-.228	-.259	-.296	-.339	-.353
	65.00	-.154	-.166	-.196	-.230	-.278	-.367	-.207	-.223	-.284	-.312	-.344	-.371
	70.00	-.165	-.170	-.193	-.230	-.298	-.326	-.229	-.230	-.259	-.309	-.373	-.331
	75.00	-.149	-.170	-.186	-.210	-.281	-.241	-.207	-.230	-.259	-.284	-.340	-.234
	80.00	-.129	-.133	-.132	-.137	-.183	-.125	-.162	-.180	-.179	-.045	-.179	-.091
	85.00	-.093	-.060	-.052	-.044	-.036	-.027	-.098	-.062	-.040	-.019	.009	.018
	90.00	-.022	-.019	.005	.027	.049	.031	-.011	.000	.029	.068	.093	.078
	95.00	.022	.016	.043	.072	.071	.046	.044	.041	.073	.108	.118	.100
Lower surface	1.25	.063	-.248	-.401	-.727	-.874	-.936	.203	.066	.006	-.037	-.038	-.101
	2.50	.003	-.195	-.263	-.551	-.742	-.875	.138	.005	-.026	-.056	-.068	-.100
	5.00	-.024	-.131	-.206	-.261	-.421	-.794	.101	.019	-.026	-.057	-.085	-.114
	7.50	-.045	-.138	-.213	-.268	-.297	-.717	.071	-.010	-.048	-.084	-.103	-.136
	10.00	-.083	-.143	-.204	-.281	-.322	-.600	.031	-.027	-.064	-.099	-.120	-.146
	15.00	-.096	-.146	-.201	-.265	-.315	-.415	.005	-.040	-.075	-.111	-.139	-.178
	20.00	-.123	-.145	-.202	-.284	-.332	-.406	-.028	-.060	-.102	-.137	-.163	-.240
	25.00	-.144	-.149	-.199	-.265	-.314	-.394	-.049	-.049	-.103	-.146	-.174	-.280
	30.00	-.125	-.167	-.217	-.273	-.317	-.335	-.035	-.079	-.119	-.157	-.183	-.293
	35.00	-.148	-.202	-.239	-.291	-.329	-.311	-.067	-.109	-.137	-.183	-.213	-.265
	40.00	-.198	-.202	-.239	-.292	-.329	-.346	-.104	-.101	-.158	-.200	-.227	-.308
	45.00	-.169	-.208	-.255	-.317	-.345	-.373	-.083	-.132	-.187	-.232	-.252	-.337
	50.00	-.219	-.246	-.286	-.343	-.376	-.394	-.146	-.170	-.214	-.258	-.281	-.359
	55.00	-.239	-.266	-.298	-.361	-.407	-.411	-.167	-.189	-.233	-.269	-.315	-.374
	60.00	-.270	-.290	-.321	-.379	-.430	-.412	-.201	-.217	-.251	-.292	-.336	-.374
	65.00	-.284	-.295	-.331	-.397	-.450	-.419	-.209	-.220	-.256	-.316	-.359	-.376
	70.00	-.292	-.287	-.329	-.383	-.438	-.391	-.214	-.211	-.250	-.291	-.355	-.327
	75.00	-.339	-.320	-.335	-.385	-.411	-.328	-.257	-.238	-.247	-.291	-.336	-.232
	80.00	-.305	-.309	-.320	-.319	-.254	-.224	-.205	-.202	-.220	-.217	-.190	-.106
	85.00	-.213	-.195	-.171	-.113	-.065	-.057	-.115	-.089	-.073	-.042	-.010	.036
	90.00	-.075	-.058	-.010	.027	.034	-.001	-.020	-.006	.040	.060	.077	.066
	95.00	.009	.009	.055	.073	.083	.058	.037	.037	.075	.097	.120	.111
		M = 0.94		α = 1.92°				M = 0.94		α = 3.88°			
Upper surface	0.00	.039	.658	.559	.569	.547	.643	.044	.542	.414	.388	.359	.598
	1.25	.078	-.222	-.341	-.757	-.867	-.716	-.058	-.945	-1.079	-1.158	-1.176	-.923
	2.50	.018	-.229	-.291	-.458	-.764	-.916	-.122	-.510	-.954	-1.042	-1.099	-1.163
	5.00	-.078	-.191	-.223	-.310	-.423	-.846	-.247	-.401	-.757	-.961	-1.013	-1.099
	7.50	-.089	-.151	-.218	-.300	-.370	-.768	-.239	-.286	-.315	-.901	-.953	-1.027
	10.00	-.100	-.151	-.224	-.296	-.361	-.702	-.230	-.269	-.325	-.842	-.936	-.984
	15.00	-.085	-.155	-.225	-.280	-.353	-.503	-.175	-.248	-.321	-.390	-.885	-.966
	20.00	-.089	-.187	-.234	-.297	-.347	-.407	-.166	-.269	-.344	-.402	-.787	-.938
	25.00	-.154	-.177	-.236	-.290	-.338	-.409	-.234	-.262	-.325	-.380	-.485	-.927
	30.00	-.237	-.202	-.251	-.300	-.344	-.348	-.310	-.281	-.333	-.384	-.417	-.879
	35.00	-.196	-.232	-.264	-.312	-.361	-.336	-.267	-.303	-.346	-.392	-.426	-.803
	40.00	-.222	-.235	-.285	-.337	-.376	-.394	-.299	-.310	-.365	-.419	-.445	-.816
	45.00	-.245	-.246	-.290	-.339	-.394	-.394	-.319	-.322	-.375	-.430	-.460	-.774
	50.00	-.224	-.252	-.298	-.356	-.410	-.419	-.297	-.322	-.382	-.446	-.485	-.722
	55.00	-.203	-.280	-.320	-.371	-.427	-.428	-.269	-.346	-.400	-.463	-.508	-.605
	60.00	-.284	-.308	-.340	-.383	-.438	-.410	-.353	-.374	-.415	-.468	-.527	-.499
	65.00	-.292	-.305	-.348	-.398	-.434	-.436	-.360	-.374	-.421	-.478	-.521	-.480
	70.00	-.306	-.309	-.344	-.401	-.452	-.411	-.373	-.374	-.416	-.479	-.538	-.458
	75.00	-.301	-.321	-.353	-.395	-.424	-.348	-.366	-.386	-.424	-.475	-.536	-.439
	80.00	-.288	-.313	-.331	-.272	-.268	-.216	-.387	-.403	-.425	-.414	-.505	-.429
	85.00	-.215	-.190	-.168	-.118	-.064	-.083	-.361	-.357	-.388	-.358	-.234	-.356
	90.00	-.075	-.058	-.018	.014	.038	.006	-.184	-.173	-.148	-.091	-.043	-.190
	95.00	.015	.009	.052	.076	.081	.044	-.025	-.038	.003	.024	.030	-.077
Lower surface	1.25	.314	.262	.246	.263	.315	.231	.436	.431	.416	.438	.486	.384
	2.50	.248	.166	.170	.188	.217	.167	.382	.323	.328	.342	.374	.311
	5.00	.204	.133	.121	.128	.126	.114	.328	.262	.253	.252	.270	.247
	7.50	.165	.090	.074	.081	.090	.055	.277	.209	.197	.198	.214	.182
	10.00	.123	.061	.046	.045	.046	.016	.228	.173	.164	.154	.159	.127
	15.00	.083	.035	.019	.003	-.005	-.057	.183	.138	.126	.109	.104	.037
	20.00	.045	.005	-.017	-.040	-.043	-.140	.134	.137	.077	.054	.064	-.047
	25.00	.024	-.008	-.027	-.061	-.077	-.221	.108	.091	.051	.023	.022	-.134
	30.00	.020	-.028	-.045	-.082	-.097	-.264	.095	.053	.025	-.005	-.003	-.178
	35.00	-.021	-.055	-.069	-.113	-.125	-.258	.037	.022	-.002	-.036	-.044	-.178
	40.00	-.045	-.067	-.092	-.138	-.148	-.319	.021	.002	-.030	-.064	-.066	-.254
	45.00	-.040	-.094	-.125	-.168	-.180	-.359	.021	-.027	-.059	-.092	-.100	-.308
	50.00	-.105	-.131	-.156	-.194	-.207	-.379	-.040	-.058	-.088	-.122	-.129	-.338
	55.00	-.124	-.150	-.175	-.212	-.250	-.384	-.067	-.083	-.104	-.137	-.173	-.342
	60.00	-.157	-.171	-.193	-.229	-.276	-.376	-.086	-.099	-.119	-.157	-.195	-.369
	65.00	-.157	-.177	-.195	-.244	-.300	-.382	-.097	-.109	-.136	-.179	-.224	-.392
	70.00	-.161	-.166	-.188	-.229	-.292	-.343	-.097	-.106	-.134	-.171	-.224	-.385
	75.00	-.206	-.192	-.199	-.236	-.294	-.275	-.150	-.133	-.148	-.193	-.251	-.383
	80.00	-.165	-.161	-.159	-.191	-.223	-.182	-.124	-.116	-.124	-.167	-.243	-.373
	85.00	-.108	-.090	-.066	-.073	-.083	-.039	-.092	-.075	-.071	-.102	-.166	-.268
	90.00	-.036	-.037	-.006	.008	.023	.009	-.043	-.038	-.024	-.028	-.047	-.176
	95.00	.011	.002	.041	.049	.070	.058	.006	-.005	.015	.013	.016	-.052

TABLE I. - STEEL WING PRESSURE COEFFICIENT DATA FOR THE TEST RANGE OF ANGLE OF ATTACK AND MACH NUMBER - Continued

		Pressure coefficient, P, at:											
	Percent c	0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2	0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2
		M = 0.94		α = 5.91°				M = 0.94		α = 7.80°			
Upper surface	0.00	.052	.411	.262	.191	.114	.479	.032	.237	.073	-.052	-.196	.293
	1.25	-.190	-1.156	-1.251	-1.305	-1.324	-1.088	-.334	-1.338	-1.385	-1.389	-1.404	-1.255
	2.50	-.260	-1.059	-1.119	-1.200	-1.260	-1.320	-.412	-1.237	-1.273	-1.339	-1.364	-1.422
	5.00	-.407	-.750	-1.030	-1.120	-1.168	-1.240	-.568	-.954	-1.204	-1.263	-1.305	-1.368
	7.50	-.384	-.408	-.966	-1.063	-1.103	-1.179	-.536	-.669	-1.130	-1.210	-1.244	-1.320
	10.00	-.361	-.383	-.898	-1.020	-1.091	-1.144	-.503	-.579	-1.095	-1.165	-1.225	-1.287
	15.00	-.278	-.346	-.440	-.931	-1.040	-1.111	-.412	-.498	-.961	-1.091	-1.181	-1.254
	20.00	-.257	-.347	-.406	-.905	-.983	-1.076	-.378	-.460	-.612	-1.064	-1.133	-1.218
	25.00	-.313	-.338	-.400	-.672	-.947	-1.061	-.407	-.440	-.543	-1.027	-1.097	-1.200
	30.00	-.372	-.349	-.400	-.515	-.922	-1.022	-.455	-.440	-.491	-.995	-1.076	-1.169
	35.00	-.339	-.368	-.411	-.474	-.907	-.939	-.431	-.458	-.492	-.783	-1.062	-1.077
	40.00	-.373	-.375	-.428	-.470	-.825	-.948	-.463	-.462	-.501	-.688	-1.053	-1.080
	45.00	-.373	-.382	-.440	-.480	-.678	-.911	-.437	-.470	-.519	-.653	-1.047	-1.047
	50.00	-.364	-.386	-.452	-.499	-.600	-.911	-.445	-.470	-.527	-.635	-1.040	-1.053
	55.00	-.330	-.408	-.462	-.521	-.585	-.910	-.415	-.487	-.539	-.636	-.999	-1.052
	60.00	-.414	-.432	-.476	-.528	-.584	-.895	-.501	-.512	-.546	-.616	-.874	-1.044
	65.00	-.422	-.434	-.484	-.543	-.577	-.908	-.501	-.512	-.555	-.626	-.795	-1.055
	70.00	-.428	-.436	-.477	-.540	-.588	-.909	-.512	-.522	-.548	-.623	-.771	-1.057
	75.00	-.417	-.449	-.486	-.538	-.592	-.904	-.491	-.532	-.580	-.612	-.745	-1.057
	80.00	-.445	-.463	-.485	-.536	-.592	-.886	-.520	-.548	-.561	-.612	-.711	-1.057
	85.00	-.459	-.454	-.485	-.534	-.495	-.767	-.545	-.545	-.561	-.612	-.629	-1.058
	90.00	-.349	-.357	-.394	-.359	-.236	-.578	-.500	-.521	-.546	-.598	-.488	-1.020
	95.00	-.110	-.142	-.110	-.098	-.109	-.529	-.267	-.332	-.369	-.393	-.373	-.950
Lower surface	1.25	.565	.566	.535	.555	.585	.474	.667	.660	.629	.627	.636	.526
	2.50	.514	.456	.451	.464	.478	.408	.624	.556	.549	.547	.546	.467
	5.00	.450	.377	.367	.366	.373	.342	.556	.488	.457	.445	.454	.405
	7.50	.394	.317	.306	.305	.317	.268	.487	.403	.395	.388	.395	.330
	10.00	.339	.278	.270	.259	.263	.212	.425	.367	.360	.336	.336	.273
	15.00	.277	.231	.216	.203	.202	.117	.358	.303	.292	.278	.270	.179
	20.00	.224	.213	.162	.146	.150	.032	.296	.277	.239	.218	.215	.088
	25.00	.191	.165	.133	.110	.106	-.049	.252	.226	.198	.182	.168	.019
	30.00	.168	.127	.102	.083	.075	-.108	.225	.185	.167	.145	.132	-.045
	35.00	.096	.093	.073	.046	.035	-.118	.145	.148	.129	.109	.091	-.069
	40.00	.090	.069	.040	.015	.009	-.191	.143	.121	.099	.077	.061	-.144
	45.00	.086	.040	.014	-.017	-.029	-.247	.134	.091	.072	.040	.025	-.200
	50.00	.019	.007	-.019	-.051	-.062	-.279	.063	.057	.040	.011	-.011	-.235
	55.00	-.006	-.015	-.040	-.065	-.104	-.303	.034	.029	.014	-.011	-.051	-.261
	60.00	-.017	-.036	-.060	-.089	-.128	-.315	.028	.009	-.009	-.037	-.084	-.279
	65.00	-.033	-.052	-.070	-.113	-.163	-.348	.005	-.012	-.023	-.063	-.120	-.317
	70.00	-.038	-.052	-.079	-.115	-.166	-.348	-.002	-.018	-.038	-.072	-.124	-.306
	75.00	-.095	-.079	-.096	-.139	-.197	-.344	-.053	-.047	-.059	-.093	-.152	-.316
	80.00	-.083	-.075	-.090	-.130	-.196	-.351	-.058	-.054	-.063	-.096	-.157	-.327
	85.00	-.075	-.061	-.064	-.105	-.183	-.301	-.068	-.054	-.060	-.095	-.166	-.278
	90.00	-.047	-.042	-.044	-.077	-.132	-.335	-.065	-.060	-.063	-.093	-.152	-.321
	95.00	-.017	-.028	-.018	-.049	-.088	-.285	-.069	-.075	-.071	-.112	-.157	-.283
		M = 0.94		α = 9.91°				M = 0.94		α = 13.64°			
Upper surface	0.00	.039	.064	-.125	-.305	-.514	.042	.032	-.336	-.596	-.836	-.871	-.431
	1.25	-.440	-1.375	-1.385	-1.356	-1.367	-1.318	-.685	-1.417	-1.320	-1.150	-1.019	-.841
	2.50	-.532	-1.267	-1.325	-1.352	-1.344	-1.402	-.801	-1.381	-1.297	-1.125	-1.039	-.828
	5.00	-.688	-1.090	-1.274	-1.293	-1.309	-1.377	-.943	-1.328	-1.320	-1.127	-1.033	-.819
	7.50	-.650	-.890	-1.204	-1.252	-1.281	-1.347	-.894	-1.266	-1.266	-1.114	-1.020	-.807
	10.00	-.611	-.810	-1.175	-1.234	-1.244	-1.326	-.844	-1.213	-1.256	-1.099	-1.046	-.803
	15.00	-.513	-.725	-1.108	-1.167	-1.232	-1.296	-.719	-1.132	-1.188	-1.091	-1.028	-.802
	20.00	-.460	-.629	-.791	-1.130	-1.204	-1.280	-.643	-1.015	-1.134	-1.086	-1.012	-.796
	25.00	-.464	-.541	-.739	-1.114	-1.167	-1.253	-.598	-.682	-1.095	-1.048	-1.022	-.805
	30.00	-.508	-.491	-.666	-1.088	-1.154	-1.214	-.618	-.523	-1.055	-1.032	-.991	-.825
	35.00	-.481	-.496	-.604	-1.043	-1.144	-1.123	-.591	-.563	-1.038	-1.010	-.955	-.834
	40.00	-.512	-.496	-.568	-.853	-1.129	-1.140	-.607	-.576	-1.015	-.986	-.931	-.812
	45.00	-.479	-.504	-.572	-.814	-1.129	-1.110	-.570	-.586	-.983	-.954	-.917	-.823
	50.00	-.490	-.509	-.577	-.802	-1.120	-1.118	-.589	-.598	-.946	-.924	-.910	-.819
	55.00	-.471	-.531	-.594	-.795	-1.120	-1.119	-.609	-.616	-.893	-.899	-.897	-.813
	60.00	-.547	-.556	-.602	-.776	-1.120	-1.112	-.644	-.637	-.827	-.861	-.888	-.813
	65.00	-.547	-.561	-.606	-.755	-1.034	-1.112	-.641	-.634	-.747	-.845	-.874	-.820
	70.00	-.547	-.566	-.606	-.725	-.926	-1.116	-.627	-.640	-.676	-.828	-.863	-.810
	75.00	-.535	-.577	-.613	-.706	-.858	-1.137	-.609	-.643	-.632	-.784	-.850	-.800
	80.00	-.560	-.597	-.614	-.688	-.789	-1.132	-.632	-.643	-.592	-.759	-.829	-.801
	85.00	-.577	-.596	-.616	-.675	-.721	-1.118	-.619	-.614	-.570	-.741	-.802	-.814
	90.00	-.561	-.590	-.603	-.658	-.649	-1.113	-.564	-.555	-.540	-.712	-.767	-.804
	95.00	-.387	-.489	-.588	-.606	-.592	-1.068	-.386	-.446	-.520	-.604	-.764	-.794
Lower surface	1.25	.771	.745	.691	.679	.675	.568	.902	.837	.776	.734	.694	.602
	2.50	.743	.647	.620	.617	.607	.518	.909	.769	.735	.706	.682	.568
	5.00	.666	.558	.536	.519	.524	.463	.819	.679	.657	.631	.628	.530
	7.50	.590	.492	.477	.459	.466	.398	.730	.621	.596	.576	.568	.477
	10.00	.522	.454	.436	.422	.411	.342	.656	.574	.560	.536	.527	.430
	15.00	.442	.384	.362	.354	.343	.244	.559	.509	.486	.467	.449	.338
	20.00	.372	.352	.311	.286	.284	.158	.485	.467	.428	.403	.395	.260
	25.00	.330	.298	.267	.246	.239	.088	.436	.407	.378	.359	.350	.192
	30.00	.295	.254	.230	.210	.202	.025	.395	.354	.342	.323	.306	.126
	35.00	.211	.215	.193	.167	.158	-.006	.309	.318	.301	.279	.260	.094
	40.00	.211	.188	.164	.139	.130	-.074	.311	.280	.269	.241	.229	.028
	45.00	.199	.154	.130	.103	.089	-.135	.293	.244	.230	.204	.188	-.032
	50.00	.131	.117	.098	.067	.050	-.174	.218	.203	.196	.166	.155	-.068
	55.00	.096	.091	.074	.047	.011	-.198	.184	.177	.171	.145	.108	-.090
	60.00	.092	.063	.045	.017	-.025	-.222	.181	.149	.139	.117	.071	-.115
	65.00	.065	.043	.028	-.010	-.057	-.259	.137	.120	.116	.084	.037	-.139
	70.00	.040	.034	.011	-.023	-.064	-.251	.104	.107	.095	.067	.032	-.146
	75.00	.022	.007	-.007	-.042	-.089	-.267	.083	.075	.068	.042	.004	-.152
	80.00	.002	-.005	-.020	-.049	-.096	-.273	.058	.063	.053	.030	-.012	-.164
	85.00	-.022	-.015	-.025	-.057	-.109	-.232	.036	.043	.041	.015	-.027	-.136
	90.00	-.035	-.031	-.043	-.065	-.109	-.274	.019	.021	.010	-.003	-.045	-.183
	95.00	-.073	-.063	-.067	-.102	-.127	-.246	-.012	-.025	-.024	-.059	-.085	-.170

TABLE I. - STEEL WING PRESSURE COEFFICIENT DATA FOR THE TEST RANGE OF ANGLE OF ATTACK AND MACH NUMBER - Continued

	Percent c	Pressure coefficient, P, at: 0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2	0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2
		M = 0.94		α = 15.66°				M = 0.94		α = 17.76°			
Upper surface	0.00	.039	-.569	-.790	-1.037	-1.104	-.640	.044	-.744	-.962	-.948	-.821	-.733
	1.25	-.819	-1.430	-1.181	-.998	-.843	-.724	-.916	-1.243	-1.007	-.875	-.799	-.723
	2.50	-.925	-1.428	-1.161	-.971	-.850	-.718	-1.015	-1.233	-1.007	-.857	-.799	-.719
	5.00	-1.049	-1.390	-1.181	-.973	-.849	-.718	-1.162	-1.219	-1.029	-.841	-.792	-.719
	7.50	-1.005	-1.347	-1.168	-.973	-.839	-.716	-1.107	-1.210	-1.011	-.839	-.787	-.719
	10.00	-.961	-1.297	-1.185	-.972	-.860	-.716	-1.051	-1.188	-1.017	-.839	-.786	-.719
	15.00	-.816	-1.230	-1.098	-.972	-.849	-.716	-.846	-1.168	-.995	-.839	-.785	-.719
	20.00	-.685	-1.176	-1.064	-.975	-.846	-.713	-.746	-1.121	-.949	-.836	-.782	-.719
	25.00	-.642	-1.025	-1.041	-.959	-.852	-.716	-.692	-1.058	-.929	-.837	-.777	-.719
	30.00	-.655	-.722	-1.002	-.950	-.836	-.720	-.656	-.975	-.931	-.832	-.775	-.718
	35.00	-.629	-.635	-.989	-.923	-.819	-.725	-.612	-.862	-.928	-.824	-.774	-.716
	40.00	-.636	-.609	-.970	-.886	-.810	-.718	-.615	-.769	-.918	-.821	-.771	-.717
	45.00	-.607	-.616	-.956	-.852	-.804	-.724	-.600	-.722	-.904	-.812	-.770	-.718
	50.00	-.619	-.624	-.938	-.835	-.803	-.722	-.638	-.688	-.887	-.804	-.767	-.719
	55.00	-.632	-.635	-.902	-.826	-.796	-.720	-.620	-.681	-.863	-.796	-.765	-.723
	60.00	-.671	-.654	-.863	-.803	-.795	-.719	-.657	-.688	-.839	-.778	-.763	-.723
	65.00	-.654	-.641	-.826	-.808	-.792	-.723	-.633	-.677	-.817	-.775	-.760	-.723
	70.00	-.627	-.635	-.791	-.807	-.791	-.724	-.598	-.672	-.808	-.774	-.757	-.723
	75.00	-.589	-.634	-.756	-.778	-.787	-.715	-.584	-.670	-.796	-.768	-.756	-.723
	80.00	-.599	-.632	-.727	-.768	-.782	-.718	-.629	-.670	-.786	-.764	-.753	-.723
	85.00	-.589	-.604	-.693	-.757	-.764	-.724	-.620	-.652	-.772	-.766	-.743	-.723
	90.00	-.545	-.560	-.659	-.742	-.744	-.717	-.605	-.639	-.757	-.760	-.734	-.723
	95.00	-.391	-.471	-.620	-.728	-.757	-.714	-.513	-.606	-.728	-.760	-.751	-.723
Lower surface	1.25	.957	.880	.803	.748	.699	.609	.986	.907	.815	.750	.683	.612
	2.50	.989	.828	.783	.743	.707	.588	1.038	.871	.815	.765	.723	.603
	5.00	.895	.750	.718	.685	.663	.565	.953	.807	.762	.724	.698	.590
	7.50	.793	.682	.661	.637	.618	.518	.864	.745	.715	.683	.660	.551
	10.00	.722	.632	.622	.601	.571	.472	.788	.699	.671	.645	.622	.509
	15.00	.624	.570	.549	.533	.509	.392	.691	.630	.608	.580	.560	.435
	20.00	.552	.527	.495	.469	.452	.312	.617	.580	.553	.517	.499	.365
	25.00	.503	.464	.447	.422	.406	.245	.566	.534	.500	.476	.456	.297
	30.00	.458	.422	.406	.386	.367	.187	.524	.478	.460	.438	.415	.230
	35.00	.368	.375	.365	.341	.321	.151	.436	.438	.416	.392	.368	.200
	40.00	.366	.342	.330	.306	.286	.084	.430	.397	.385	.355	.336	.135
	45.00	.345	.307	.294	.266	.251	.026	.403	.364	.338	.316	.297	.076
	50.00	.275	.261	.258	.226	.212	-.003	.340	.321	.308	.281	.260	.041
	55.00	.237	.231	.230	.206	.170	-.030	.297	.290	.278	.253	.213	.015
	60.00	.233	.202	.196	.173	.130	-.057	.292	.258	.244	.220	.178	-.013
	65.00	.186	.172	.172	.140	.094	-.080	.245	.223	.220	.184	.141	-.038
	70.00	.149	.154	.151	.120	.086	-.097	.195	.203	.195	.166	.127	-.057
	75.00	.128	.115	.120	.096	.056	-.107	.175	.163	.159	.137	.096	-.074
	80.00	.102	.102	.099	.076	.034	-.127	.147	.146	.137	.114	.072	-.092
	85.00	.066	.076	.082	.058	.012	-.097	.110	.117	.113	.091	.046	-.071
	90.00	.043	.047	.052	.035	-.008	-.153	.075	.089	.081	.058	.021	-.128
	95.00	.022	-.001	.005	-.028	-.055	-.144	.045	.029	.030	-.007	-.030	-.131
		M = 0.94		α = 19.85°				M = 0.94		α = 21.85°			
Upper surface	0.00	.023	-.931	-1.104	-.949	-.848	-.806	-.022	-1.003	-.964	-.936	-.894	-.860
	1.25	-1.002	-1.118	-.978	-.932	-.835	-.778	-.939	-.952	-.945	-.931	-.885	-.832
	2.50	-1.019	-1.110	-.976	-.889	-.834	-.776	-.924	-.953	-.937	-.916	-.883	-.832
	5.00	-1.066	-1.109	-1.006	-.872	-.831	-.773	-.924	-.955	-.943	-.906	-.883	-.829
	7.50	-1.068	-1.107	-.997	-.868	-.827	-.774	-.929	-.956	-.931	-.905	-.878	-.829
	10.00	-1.069	-1.097	-.999	-.867	-.825	-.774	-.933	-.955	-.939	-.905	-.874	-.829
	15.00	-.960	-1.093	-.982	-.863	-.827	-.774	-.939	-.962	-.942	-.904	-.879	-.831
	20.00	-.870	-1.077	-.972	-.859	-.824	-.774	-.938	-.958	-.942	-.897	-.881	-.830
	25.00	-.788	-1.043	-.961	-.859	-.822	-.773	-.902	-.952	-.937	-.901	-.876	-.828
	30.00	-.734	-1.005	-.957	-.853	-.818	-.771	-.880	-.949	-.937	-.897	-.876	-.825
	35.00	-.673	-.968	-.952	-.849	-.817	-.770	-.784	-.944	-.937	-.894	-.874	-.824
	40.00	-.661	-.914	-.945	-.847	-.814	-.772	-.780	-.932	-.937	-.896	-.873	-.829
	45.00	-.631	-.868	-.935	-.840	-.815	-.773	-.754	-.925	-.933	-.894	-.873	-.827
	50.00	-.649	-.822	-.920	-.838	-.813	-.773	-.735	-.912	-.929	-.892	-.872	-.829
	55.00	-.642	-.796	-.903	-.831	-.809	-.774	-.711	-.896	-.928	-.888	-.870	-.831
	60.00	-.707	-.779	-.886	-.818	-.810	-.775	-.794	-.891	-.920	-.875	-.868	-.831
	65.00	-.691	-.753	-.873	-.817	-.807	-.770	-.778	-.872	-.917	-.875	-.867	-.826
	70.00	-.657	-.740	-.859	-.815	-.803	-.773	-.733	-.858	-.912	-.875	-.861	-.828
	75.00	-.638	-.731	-.851	-.812	-.803	-.774	-.733	-.840	-.904	-.874	-.860	-.832
	80.00	-.674	-.730	-.842	-.810	-.802	-.774	-.783	-.834	-.899	-.875	-.859	-.829
	85.00	-.681	-.713	-.834	-.810	-.790	-.770	-.774	-.810	-.890	-.875	-.848	-.825
	90.00	-.673	-.704	-.821	-.806	-.782	-.773	-.756	-.800	-.877	-.872	-.838	-.827
	95.00	-.583	-.681	-.795	-.806	-.799	-.770	-.660	-.768	-.862	-.871	-.855	-.827
Lower surface	1.25	.991	.921	.817	.740	.663	.591	.965	.922	.813	.719	.631	.570
	2.50	1.073	.914	.835	.773	.724	.599	1.073	.938	.848	.773	.721	.586
	5.00	.995	.860	.799	.750	.715	.599	1.017	.892	.828	.770	.731	.601
	7.50	.902	.802	.756	.715	.686	.569	.941	.841	.789	.741	.759	.579
	10.00	.836	.758	.715	.681	.649	.533	.872	.804	.753	.712	.699	.550
	15.00	.743	.686	.653	.626	.591	.463	.781	.733	.697	.658	.628	.486
	20.00	.667	.624	.598	.565	.546	.398	.710	.670	.641	.605	.580	.423
	25.00	.620	.587	.549	.523	.499	.334	.661	.627	.597	.561	.537	.364
	30.00	.574	.536	.507	.483	.457	.271	.614	.580	.551	.529	.494	.301
	35.00	.489	.493	.466	.434	.410	.235	.532	.534	.509	.477	.451	.265
	40.00	.479	.454	.429	.398	.380	.177	.521	.496	.469	.436	.418	.207
	45.00	.453	.414	.390	.362	.341	.113	.494	.455	.429	.401	.381	.144
	50.00	.389	.375	.352	.325	.303	.080	.427	.411	.393	.362	.342	.112
	55.00	.346	.336	.323	.299	.256	.050	.384	.378	.363	.334	.293	.080
	60.00	.331	.305	.285	.262	.217	.023	.367	.342	.324	.296	.254	.048
	65.00	.284	.269	.258	.225	.181	-.004	.321	.302	.295	.259	.217	.022
	70.00	.236	.247	.230	.201	.167	-.028	.267	.278	.265	.235	.204	-.006
	75.00	.212	.208	.199	.175	.132	-.047	.241	.239	.230	.203	.165	-.027
	80.00	.178	.183	.169	.147	.103	-.069	.205	.209	.195	.171	.134	-.053
	85.00	.137	.151	.146	.119	.073	-.059	.158	.172	.169	.142	.099	-.046
	90.00	.099	.114	.105	.088	.041	-.114	.113	.133	.123	.102	.063	-.107
	95.00	.047	.051	.048	.011	-.018	-.118	.050	.061	.063	.023	-.001	-.116

TABLE I. - STEEL WING PRESSURE COEFFICIENT DATA FOR THE TEST RANGE OF ANGLE OF ATTACK AND MACH NUMBER - Continued

		Pressure coefficient, P, at:											
		0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2	0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2
	Percent c	M = 0.94		α = 23.90°				M = 0.94		α = 25.95°			
Upper surface	0.30	-.090	-1.009	-.986	-.964	-.946	-.906	-.312	-1.140	-1.120	-1.100	-1.061	-1.002
	1.25	-.986	-.994	-.982	-.961	-.939	-.887	-1.104	-1.123	-1.110	-1.089	-1.050	-.973
	2.50	-.976	-.995	-.973	-.957	-.938	-.883	-1.092	-1.116	-1.102	-1.078	-1.044	-.971
	5.00	-.973	-.996	-.979	-.953	-.936	-.880	-1.091	-1.122	-1.115	-1.070	-1.043	-.967
	7.50	-.978	-.999	-.969	-.952	-.930	-.881	-1.101	-1.119	-1.095	-1.070	-1.037	-.967
	10.00	-.982	-1.001	-.979	-.952	-.931	-.883	-1.109	-1.116	-1.103	-1.073	-1.031	-.964
	15.00	-.980	-1.006	-.981	-.950	-.932	-.881	-1.094	-1.126	-1.108	-1.068	-1.037	-.966
	20.00	-.982	-.999	-.981	-.945	-.931	-.881	-1.017	-1.126	-1.110	-1.056	-1.041	-.966
	25.00	-.926	-.997	-.981	-.947	-.929	-.879	-.869	-1.111	-1.101	-1.065	-1.034	-.960
	30.00	-.892	-.995	-.980	-.945	-.928	-.879	-.843	-1.108	-1.105	-1.056	-1.033	-.955
	35.00	-.817	-.990	-.980	-.943	-.928	-.879	-.846	-1.091	-1.104	-1.055	-1.031	-.950
	40.00	-.817	-.976	-.978	-.943	-.925	-.882	-.824	-1.046	-1.105	-1.054	-1.026	-.960
	45.00	-.786	-.965	-.977	-.941	-.925	-.882	-.797	-.997	-1.097	-1.050	-1.027	-.955
	50.00	-.779	-.946	-.974	-.940	-.924	-.883	-.768	-.928	-1.087	-1.048	-1.023	-.955
	55.00	-.744	-.929	-.971	-.938	-.921	-.885	-.745	-.875	-1.081	-1.044	-1.020	-.957
	60.00	-.819	-.921	-.969	-.920	-.921	-.885	-.741	-.848	-1.070	-1.027	-1.019	-.956
	65.00	-.826	-.897	-.964	-.924	-.918	-.882	-.720	-.798	-1.056	-1.024	-1.012	-.945
	70.00	-.780	-.882	-.960	-.927	-.915	-.883	-.688	-.757	-1.043	-1.020	-1.003	-.946
	75.00	-.777	-.872	-.956	-.923	-.915	-.885	-.669	-.724	-1.015	-1.023	-.999	-.952
	80.00	-.822	-.870	-.948	-.923	-.915	-.883	-.644	-.704	-.993	-1.018	-.996	-.946
	85.00	-.818	-.848	-.938	-.924	-.900	-.879	-.654	-.683	-.963	-1.010	-.978	-.935
	90.00	-.805	-.836	-.924	-.922	-.892	-.880	-.644	-.673	-.930	-.991	-.962	-.938
	95.00	-.695	-.812	-.895	-.918	-.910	-.878	-.524	-.623	-.877	-.985	-.971	-.938
Lower surface	1.25	.934	.926	.806	.700	.603	.554	.913	.925	.794	.680	.571	.528
	2.50	1.070	.960	.856	.774	.716	.581	1.071	.978	.867	.766	.706	.564
	5.00	1.038	.930	.857	.789	.744	.606	1.061	.960	.874	.802	.756	.604
	7.50	.971	.886	.825	.771	.776	.598	1.005	.924	.858	.794	.752	.606
	10.00	.914	.847	.794	.750	.712	.575	.950	.890	.831	.777	.734	.588
	15.00	.823	.784	.740	.696	.659	.518	.863	.828	.780	.735	.690	.540
	20.00	.753	.727	.688	.648	.617	.459	.803	.775	.730	.686	.651	.486
	25.00	.706	.678	.641	.604	.578	.405	.758	.729	.691	.646	.613	.434
	30.00	.663	.630	.597	.569	.538	.345	.711	.682	.646	.607	.574	.376
	35.00	.581	.584	.558	.524	.492	.308	.635	.635	.605	.566	.531	.342
	40.00	.566	.543	.518	.480	.461	.247	.618	.596	.568	.527	.501	.286
	45.00	.536	.502	.476	.444	.424	.185	.590	.561	.528	.488	.454	.224
	50.00	.473	.458	.439	.409	.387	.153	.527	.513	.491	.451	.425	.187
	55.00	.432	.422	.409	.377	.338	.120	.484	.475	.456	.421	.378	.154
	60.00	.413	.385	.367	.339	.297	.090	.465	.437	.421	.383	.335	.122
	65.00	.362	.347	.337	.299	.260	.058	.413	.392	.386	.349	.298	.094
	70.00	.307	.317	.299	.276	.241	.028	.356	.371	.352	.317	.280	.061
	75.00	.276	.275	.264	.242	.202	.003	.329	.324	.310	.282	.236	.031
	80.00	.240	.246	.236	.209	.167	-.024	.288	.295	.278	.247	.202	.003
	85.00	.191	.210	.202	.177	.192	-.020	.238	.254	.242	.211	.161	.002
	90.00	.137	.164	.152	.132	.088	-.085	.182	.204	.187	.164	.114	-.063
	95.00	.054	.087	.086	.046	.021	-.100	.080	.122	.119	.072	.041	-.086

TABLE I.- STEEL WING PRESSURE COEFFICIENT DATA FOR THE TEST RANGE OF ANGLE OF ATTACK AND MACH NUMBER - Continued

		Pressure coefficient, P, at:											
	Percent c	0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2	0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2
		M = 0.98		α = -2.00°				M = 0.98		α = -0.08°			
Upper surface	0.00	.029	.718	.668	.630	.570	.485	.061	.761	.714	.711	.673	.708
	1.25	.345	.259	.273	.286	.271	.307	.268	.107	.088	.062	.024	.120
	2.50	.288	.179	.175	.186	.191	.181	.205	.060	.038	.009	-.011	-.055
	5.00	.226	.149	.130	.116	.113	.140	.147	.052	.035	-.004	-.041	-.067
	7.50	.205	.115	.095	.079	.063	.089	.119	.047	.015	-.025	-.070	-.084
	10.00	.159	.089	.063	.056	.035	.046	.091	.033	-.006	-.045	-.069	-.101
	15.00	.115	.052	.022	.026	-.003	-.029	.065	.006	-.032	-.057	-.092	-.130
	20.00	.052	.017	-.015	-.013	-.035	-.102	.036	-.024	-.061	-.054	-.115	-.165
	25.00	.037	-.003	-.024	-.048	-.066	-.205	-.005	-.035	-.068	-.111	-.131	-.242
	30.00	-.005	-.021	-.041	-.063	-.084	-.240	-.072	-.057	-.089	-.119	-.139	-.242
	35.00	-.030	-.046	-.064	-.082	-.112	-.259	-.060	-.084	-.108	-.124	-.164	-.240
	40.00	-.052	-.051	-.089	-.120	-.134	-.327	-.086	-.101	-.121	-.153	-.184	-.292
	45.00	-.061	-.069	-.113	-.143	-.164	-.322	-.103	-.103	-.132	-.169	-.207	-.281
	50.00	-.049	-.092	-.145	-.169	-.194	-.350	-.078	-.106	-.153	-.194	-.231	-.304
	55.00	-.072	-.132	-.176	-.199	-.225	-.354	-.083	-.141	-.184	-.218	-.259	-.316
	60.00	-.143	-.163	-.186	-.216	-.256	-.338	-.158	-.176	-.205	-.239	-.275	-.303
	65.00	-.147	-.159	-.187	-.227	-.263	-.364	-.161	-.176	-.209	-.250	-.280	-.330
	70.00	-.160	-.170	-.198	-.228	-.292	-.366	-.178	-.183	-.216	-.258	-.306	-.334
	75.00	-.166	-.181	-.211	-.232	-.307	-.357	-.183	-.193	-.223	-.259	-.321	-.325
	80.00	-.186	-.200	-.211	-.241	-.315	-.346	-.198	-.214	-.226	-.254	-.323	-.315
	85.00	-.209	-.190	-.211	-.245	-.307	-.321	-.225	-.208	-.226	-.253	-.304	-.295
	90.00	-.177	-.182	-.196	-.240	-.283	-.301	-.195	-.203	-.217	-.253	-.275	-.277
	95.00	-.147	-.178	-.195	-.230	-.277	-.309	-.145	-.188	-.204	-.239	-.272	-.285
Lower surface	1.25	.091	-.206	-.377	-.682	-.859	-.915	.235	.090	.019	-.033	-.078	-.138
	2.50	.029	-.167	-.225	-.554	-.732	-.844	.171	.042	-.002	-.056	-.105	-.122
	5.00	.006	-.094	-.179	-.241	-.434	-.766	.137	.055	.005	-.034	-.073	-.127
	7.50	-.016	-.108	-.184	-.253	-.268	-.693	.104	.026	-.019	-.060	-.077	-.136
	10.00	-.054	-.116	-.183	-.262	-.294	-.601	.070	.008	-.040	-.075	-.089	-.132
	15.00	-.070	-.120	-.181	-.243	-.289	-.397	.041	-.005	-.042	-.085	-.101	-.146
	20.00	-.099	-.120	-.187	-.263	-.309	-.377	.001	-.019	-.067	-.116	-.126	-.193
	25.00	-.124	-.130	-.178	-.242	-.288	-.377	-.017	-.019	-.075	-.111	-.134	-.231
	30.00	-.107	-.142	-.190	-.250	-.293	-.316	-.006	-.045	-.089	-.115	-.142	-.244
	35.00	-.126	-.179	-.217	-.277	-.313	-.291	-.036	-.085	-.096	-.139	-.169	-.218
	40.00	-.182	-.186	-.226	-.288	-.320	-.329	-.082	-.086	-.100	-.157	-.179	-.259
	45.00	-.151	-.190	-.236	-.300	-.341	-.361	-.051	-.095	-.133	-.194	-.207	-.289
	50.00	-.200	-.224	-.268	-.326	-.360	-.385	-.112	-.129	-.167	-.223	-.235	-.311
	55.00	-.220	-.245	-.284	-.339	-.381	-.401	-.131	-.152	-.180	-.233	-.271	-.329
	60.00	-.253	-.272	-.303	-.360	-.399	-.407	-.166	-.182	-.208	-.253	-.291	-.334
	65.00	-.268	-.279	-.316	-.377	-.422	-.411	-.175	-.192	-.217	-.275	-.308	-.340
	70.00	-.275	-.267	-.314	-.365	-.414	-.405	-.183	-.180	-.217	-.265	-.307	-.340
	75.00	-.324	-.302	-.321	-.375	-.426	-.390	-.238	-.215	-.223	-.270	-.328	-.325
	80.00	-.328	-.323	-.330	-.382	-.405	-.394	-.240	-.233	-.244	-.280	-.328	-.334
	85.00	-.342	-.326	-.331	-.369	-.398	-.337	-.255	-.234	-.244	-.271	-.332	-.277
	90.00	-.319	-.321	-.331	-.354	-.382	-.376	-.234	-.239	-.244	-.269	-.317	-.312
	95.00	-.225	-.274	-.293	-.313	-.334	-.343	-.170	-.211	-.232	-.277	-.289	-.273
		M = 0.98		α = 1.96°				M = 0.98		α = 3.88°			
Upper surface	0.00	.064	.713	.606	.616	.600	.680	.069	.610	.488	.465	.438	.643
	1.25	.150	-.133	-.260	-.614	-.701	-.584	.021	-.779	-.887	-.952	-.982	-.782
	2.50	.087	-.145	-.224	-.372	-.612	-.764	-.041	-.386	-.795	-.861	-.904	-.998
	5.00	.010	-.107	-.170	-.246	-.342	-.699	-.162	-.303	-.641	-.772	-.830	-.942
	7.50	-.009	-.077	-.160	-.248	-.284	-.637	-.158	-.206	-.226	-.733	-.794	-.885
	10.00	-.027	-.081	-.165	-.245	-.275	-.589	-.153	-.186	-.240	-.694	-.754	-.848
	15.00	-.022	-.086	-.175	-.220	-.279	-.430	-.110	-.181	-.246	-.302	-.738	-.830
	20.00	-.024	-.119	-.185	-.240	-.285	-.333	-.099	-.201	-.271	-.307	-.688	-.808
	25.00	-.086	-.115	-.187	-.237	-.274	-.347	-.168	-.195	-.255	-.307	-.427	-.792
	30.00	-.165	-.139	-.199	-.247	-.280	-.292	-.246	-.211	-.262	-.311	-.341	-.740
	35.00	-.133	-.166	-.218	-.263	-.297	-.275	-.207	-.238	-.276	-.319	-.353	-.664
	40.00	-.163	-.174	-.235	-.290	-.313	-.328	-.233	-.248	-.297	-.346	-.369	-.700
	45.00	-.181	-.188	-.250	-.298	-.331	-.329	-.258	-.261	-.308	-.359	-.390	-.656
	50.00	-.160	-.187	-.256	-.318	-.352	-.354	-.237	-.261	-.316	-.374	-.414	-.646
	55.00	-.143	-.214	-.278	-.331	-.372	-.368	-.209	-.282	-.334	-.386	-.438	-.580
	60.00	-.224	-.248	-.297	-.339	-.385	-.353	-.294	-.312	-.347	-.399	-.455	-.465
	65.00	-.230	-.248	-.300	-.350	-.374	-.381	-.302	-.314	-.347	-.406	-.451	-.419
	70.00	-.245	-.250	-.303	-.352	-.389	-.388	-.311	-.315	-.353	-.407	-.462	-.388
	75.00	-.245	-.264	-.312	-.355	-.391	-.383	-.311	-.324	-.359	-.407	-.465	-.384
	80.00	-.269	-.282	-.316	-.350	-.371	-.368	-.336	-.342	-.361	-.404	-.463	-.381
	85.00	-.295	-.282	-.316	-.350	-.348	-.349	-.359	-.342	-.361	-.404	-.418	-.383
	90.00	-.266	-.280	-.305	-.338	-.317	-.330	-.335	-.342	-.350	-.393	-.362	-.384
	95.00	-.181	-.249	-.287	-.290	-.300	-.335	-.237	-.312	-.341	-.363	-.338	-.401
Lower surface	1.25	.351	.306	.267	.284	.337	.261	.469	.459	.445	.463	.504	.411
	2.50	.297	.218	.202	.213	.241	.204	.416	.357	.359	.375	.393	.341
	5.00	.253	.184	.149	.150	.163	.153	.364	.298	.295	.289	.300	.280
	7.50	.217	.139	.104	.113	.126	.102	.314	.246	.238	.234	.247	.215
	10.00	.176	.112	.074	.067	.086	.066	.269	.210	.202	.194	.196	.168
	15.00	.136	.086	.045	.036	.036	.000	.221	.176	.170	.153	.141	.085
	20.00	.098	.055	.017	-.010	-.001	-.080	.174	.178	.128	.093	.104	.002
	25.00	.076	.044	.009	-.028	-.027	-.154	.150	.131	.095	.065	.064	-.081
	30.00	.067	.021	-.011	-.045	-.051	-.201	.133	.094	.070	.043	.035	-.131
	35.00	.025	-.005	-.040	-.078	-.079	-.196	.075	.063	.041	.010	-.001	-.158
	40.00	.004	-.020	-.058	-.100	-.098	-.260	.061	.040	.015	-.018	-.025	-.201
	45.00	.010	-.040	-.091	-.136	-.131	-.300	.063	.011	-.019	-.053	-.055	-.258
	50.00	-.052	-.076	-.122	-.165	-.159	-.321	-.005	-.019	-.044	-.081	-.090	-.286
	55.00	-.074	-.098	-.142	-.179	-.195	-.334	-.025	-.047	-.066	-.093	-.129	-.312
	60.00	-.108	-.124	-.166	-.199	-.224	-.327	-.052	-.068	-.085	-.117	-.155	-.325
	65.00	-.115	-.137	-.172	-.221	-.252	-.335	-.060	-.080	-.094	-.135	-.183	-.340
	70.00	-.117	-.126	-.175	-.210	-.253	-.329	-.066	-.079	-.099	-.136	-.187	-.343
	75.00	-.179	-.162	-.191	-.228	-.277	-.321	-.130	-.116	-.127	-.157	-.214	-.343
	80.00	-.179	-.178	-.207	-.240	-.293	-.327	-.135	-.127	-.139	-.180	-.232	-.348
	85.00	-.195	-.178	-.207	-.244	-.308	-.275	-.149	-.127	-.137	-.185	-.249	-.297
	90.00	-.172	-.181	-.207	-.247	-.300	-.314	-.132	-.132	-.141	-.185	-.249	-.345
	95.00	-.130	-.159	-.198	-.259	-.277	-.273	-.108	-.122	-.137	-.198	-.239	-.304

TABLE I.- STEEL WING PRESSURE COEFFICIENT DATA FOR THE TEST RANGE OF ANGLE OF ATTACK AND MACH NUMBER - Continued

		Pressure coefficient, P, at:											
		0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2	0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2
	Percent c	M = 0.98		α = 5.93°				M = 0.98		α = 7.92°			
Upper surface	0.00	.071	.472	.327	.267	.208	.536	.075	.319	.162	.057	-.059	.377
	1.25	-.122	-1.010	-1.097	-1.142	-1.144	-.944	-.243	-1.171	-1.234	-1.269	-1.276	-1.100
	2.50	-.188	-.907	-.992	-1.058	-1.079	-1.158	-.320	-1.084	-1.132	-1.193	-1.223	-1.281
	5.00	-.333	-.638	-.919	-.979	-1.010	-1.105	-.473	-.813	-1.063	-1.118	-1.154	-1.227
	7.50	-.316	-.331	-.848	-.931	-.962	-1.034	-.445	-.547	-.993	-1.070	-1.101	-1.169
	10.00	-.298	-.312	-.795	-.902	-.936	-1.008	-.416	-.463	-.964	-1.036	-1.081	-1.159
	15.00	-.216	-.286	-.383	-.821	-.908	-.978	-.325	-.405	-.877	-.964	-1.043	-1.106
	20.00	-.197	-.285	-.349	-.788	-.870	-.950	-.296	-.376	-.534	-.937	-1.002	-1.082
	25.00	-.257	-.280	-.340	-.661	-.827	-.938	-.327	-.363	-.452	-.911	-.961	-1.062
	30.00	-.322	-.291	-.345	-.459	-.808	-.897	-.381	-.367	-.420	-.883	-.948	-1.032
	35.00	-.281	-.309	-.358	-.415	-.801	-.819	-.356	-.384	-.420	-.739	-.935	-.947
	40.00	-.317	-.324	-.375	-.415	-.774	-.839	-.384	-.393	-.435	-.613	-.929	-.954
	45.00	-.324	-.328	-.389	-.423	-.670	-.805	-.368	-.399	-.449	-.575	-.928	-.921
	50.00	-.310	-.330	-.399	-.441	-.548	-.809	-.375	-.403	-.463	-.558	-.923	-.930
	55.00	-.280	-.350	-.411	-.461	-.523	-.808	-.349	-.419	-.474	-.547	-.917	-.930
	60.00	-.362	-.377	-.424	-.468	-.519	-.793	-.429	-.444	-.484	-.540	-.839	-.923
	65.00	-.368	-.382	-.422	-.481	-.506	-.796	-.432	-.447	-.483	-.546	-.717	-.929
	70.00	-.379	-.381	-.426	-.482	-.516	-.800	-.436	-.451	-.490	-.545	-.683	-.934
	75.00	-.364	-.390	-.433	-.482	-.521	-.807	-.416	-.458	-.497	-.542	-.663	-.937
	80.00	-.396	-.408	-.433	-.482	-.527	-.801	-.453	-.477	-.497	-.545	-.644	-.937
	85.00	-.417	-.408	-.435	-.482	-.509	-.783	-.473	-.477	-.499	-.547	-.599	-.936
	90.00	-.398	-.410	-.425	-.476	-.450	-.787	-.456	-.477	-.488	-.540	-.510	-.940
	95.00	-.313	-.384	-.422	-.455	-.403	-.775	-.373	-.454	-.488	-.524	-.414	-.950
Lower surface	1.25	.583	.589	.557	.571	.600	.501	.692	.683	.642	.642	.657	.554
	2.50	.539	.482	.473	.482	.497	.431	.652	.583	.567	.568	.571	.492
	5.00	.480	.406	.389	.390	.403	.372	.586	.493	.474	.465	.475	.425
	7.50	.420	.347	.330	.326	.344	.303	.519	.431	.419	.404	.416	.365
	10.00	.367	.313	.292	.284	.289	.250	.459	.398	.382	.363	.363	.311
	15.00	.307	.265	.245	.230	.233	.160	.391	.337	.321	.307	.299	.217
	20.00	.255	.245	.197	.172	.181	.074	.332	.310	.263	.243	.249	.134
	25.00	.223	.196	.161	.133	.140	-.007	.293	.260	.226	.203	.204	.059
	30.00	.199	.160	.129	.109	.108	-.059	.263	.222	.197	.173	.170	-.003
	35.00	.126	.127	.097	.072	.070	-.070	.184	.186	.164	.133	.127	-.020
	40.00	.121	.105	.071	.043	.043	-.144	.184	.158	.128	.101	.098	-.098
	45.00	.117	.072	.039	.012	.010	-.201	.174	.124	.097	.070	.064	-.155
	50.00	.050	.039	.008	-.020	-.024	-.235	.105	.090	.070	.038	.029	-.189
	55.00	.026	.011	-.011	-.039	-.065	-.260	.074	.063	.049	.020	-.013	-.217
	60.00	.010	-.009	-.035	-.063	-.094	-.273	.066	.041	.019	-.007	-.042	-.237
	65.00	-.007	-.024	-.047	-.084	-.126	-.301	.047	.023	.004	-.032	-.078	-.274
	70.00	-.015	-.027	-.059	-.091	-.129	-.308	.039	.017	-.009	-.043	-.085	-.274
	75.00	-.079	-.062	-.087	-.123	-.166	-.310	-.017	-.015	-.034	-.069	-.118	-.276
	80.00	-.082	-.073	-.095	-.137	-.182	-.318	-.027	-.024	-.042	-.078	-.150	-.290
	85.00	-.096	-.076	-.094	-.137	-.200	-.272	-.037	-.031	-.040	-.079	-.140	-.248
	90.00	-.083	-.077	-.097	-.135	-.194	-.323	-.037	-.039	-.046	-.074	-.132	-.297
	95.00	-.088	-.085	-.097	-.144	-.183	-.290	-.059	-.055	-.058	-.097	-.129	-.262
		M = 0.98		α = 9.91°				M = 0.98		α = 13.68°			
Upper surface	0.00	.070	.147	-.027	-.195	-.372	.154	.040	-.229	-.481	-.706	-.880	-.402
	1.25	-.357	-1.255	-1.287	-1.304	-1.311	-1.194	-.584	-1.288	-1.257	-1.241	-1.278	-1.282
	2.50	-.447	-1.179	-1.210	-1.266	-1.280	-1.320	-.691	-1.254	-1.232	-1.242	-1.273	-1.319
	5.00	-.601	-.989	-1.158	-1.193	-1.224	-1.278	-.828	-1.187	-1.267	-1.239	-1.277	-1.311
	7.50	-.566	-.760	-1.088	-1.149	-1.175	-1.244	-.787	-1.142	-1.201	-1.235	-1.247	-1.305
	10.00	-.531	-.679	-1.066	-1.120	-1.152	-1.217	-.745	-1.098	-1.171	-1.217	-1.234	-1.292
	15.00	-.440	-.592	-1.007	-1.056	-1.122	-1.183	-.634	-1.008	-1.089	-1.170	-1.220	-1.277
	20.00	-.391	-.540	-.728	-1.026	-1.092	-1.161	-.559	-.923	-1.024	-1.125	-1.193	-1.260
	25.00	-.394	-.484	-.645	-1.009	-1.053	-1.142	-.516	-.651	-.982	-1.097	-1.165	-1.245
	30.00	-.440	-.444	-.589	-.984	-1.038	-1.112	-.546	-.448	-.942	-1.061	-1.140	-1.232
	35.00	-.420	-.442	-.537	-.966	-1.026	-1.024	-.520	-.505	-.924	-1.030	-1.119	-1.174
	40.00	-.446	-.440	-.503	-.826	-1.016	-1.030	-.540	-.526	-.906	-.999	-1.101	-1.160
	45.00	-.420	-.442	-.501	-.734	-1.016	-1.003	-.513	-.532	-.887	-.981	-1.082	-1.141
	50.00	-.434	-.449	-.509	-.721	-1.012	-1.013	-.540	-.543	-.861	-.969	-1.061	-1.132
	55.00	-.448	-.469	-.525	-.710	-1.011	-1.011	-.560	-.565	-.823	-.960	-1.048	-1.117
	60.00	-.487	-.494	-.534	-.692	-1.008	-1.007	-.592	-.593	-.782	-.940	-1.017	-1.107
	65.00	-.490	-.499	-.534	-.676	-.980	-1.011	-.596	-.600	-.716	-.935	-.968	-1.088
	70.00	-.490	-.506	-.538	-.655	-.892	-1.016	-.588	-.608	-.655	-.931	-.920	-1.064
	75.00	-.475	-.516	-.549	-.639	-.803	-1.025	-.586	-.619	-.616	-.916	-.873	-1.045
	80.00	-.510	-.536	-.551	-.629	-.738	-1.025	-.619	-.641	-.602	-.895	-.825	-1.032
	85.00	-.526	-.537	-.554	-.613	-.658	-1.024	-.625	-.647	-.608	-.878	-.775	-1.046
	90.00	-.514	-.537	-.546	-.599	-.592	-1.031	-.619	-.644	-.604	-.757	-.723	-1.069
	95.00	-.416	-.502	-.545	-.575	-.539	-1.053	-.534	-.602	-.613	-.640	-.701	-1.074
Lower surface	1.25	.785	.761	.712	.702	.699	.599	.929	.874	.805	.768	.726	.639
	2.50	.759	.668	.645	.637	.632	.547	.939	.808	.766	.734	.712	.605
	5.00	.686	.580	.568	.542	.547	.491	.858	.724	.689	.666	.652	.572
	7.50	.611	.514	.501	.483	.490	.428	.768	.654	.631	.611	.600	.521
	10.00	.546	.478	.459	.436	.440	.379	.694	.614	.590	.566	.550	.475
	15.00	.468	.411	.388	.382	.375	.283	.604	.538	.520	.506	.489	.384
	20.00	.402	.378	.342	.312	.320	.201	.530	.504	.462	.442	.432	.307
	25.00	.359	.324	.296	.277	.271	.129	.479	.448	.418	.399	.387	.240
	30.00	.325	.281	.265	.245	.235	.064	.441	.392	.379	.364	.346	.170
	35.00	.240	.245	.226	.205	.192	.043	.352	.356	.340	.321	.303	.139
	40.00	.240	.217	.195	.169	.165	-.032	.352	.325	.308	.286	.271	.076
	45.00	.232	.183	.160	.132	.129	-.092	.337	.291	.271	.247	.234	.007
	50.00	.162	.147	.123	.101	.094	-.123	.267	.254	.238	.214	.199	-.026
	55.00	.127	.120	.110	.080	.052	-.152	.228	.216	.214	.194	.160	-.051
	60.00	.122	.096	.080	.052	.021	-.178	.229	.197	.183	.166	.117	-.076
	65.00	.091	.074	.062	.027	-.013	-.213	.188	.164	.163	.133	.087	-.107
	70.00	.075	.063	.048	.012	-.021	-.209	.150	.154	.141	.119	.081	-.119
	75.00	.053	.036	.024	-.015	-.049	-.221	.134	.122	.118	.094	.053	-.130
	80.00	.037	.026	.015	-.020	-.059	-.235	.114	.110	.100	.081	.035	-.142
	85.00	.014	.016	.014	-.020	-.070	-.188	.085	.093	.092	.071	.019	-.114
	90.00	.003	.004	.000	-.024	-.066	-.234	.062	.068	.065	.054	.008	-.150
	95.00	-.033	-.022	-.021	-.059	-.075	-.205	.047	.032	.036	.007	-.022	-.137

TABLE I.- STEEL WING PRESSURE COEFFICIENT DATA FOR THE TEST RANGE OF ANGLE OF ATTACK AND MACH NUMBER - Continued

	Percent c	Pressure coefficient, P, at: 0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2	0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2
		M = 0.98	α = 15.84°					M = 0.98	α = 17.97°				
Upper surface	0.00	.024	-.474	-.696	-.943	-1.065	-.716	.011	-.660	-.872	-1.078	-1.163	-.904
	1.25	-.712	-1.346	-1.293	-1.231	-1.123	-1.011	-.839	-1.326	-1.276	-1.133	-1.031	-.914
	2.50	-.823	-1.345	-1.263	-1.228	-1.128	-1.004	-.949	-1.306	-1.254	-1.122	-1.031	-.909
	5.00	-.946	-1.292	-1.293	-1.224	-1.124	-1.000	-1.045	-1.300	-1.275	-1.114	-1.038	-.904
	7.50	-.906	-1.253	-1.266	-1.224	-1.115	-.994	-1.012	-1.291	-1.245	-1.114	-1.041	-.903
	10.00	-.866	-1.203	-1.266	-1.231	-1.123	-.994	-.979	-1.264	-1.264	-1.114	-1.049	-.902
	15.00	-.744	-1.150	-1.207	-1.222	-1.120	-.994	-.833	-1.223	-1.251	-1.109	-1.066	-.901
	20.00	-.632	-1.105	-1.160	-1.199	-1.110	-.994	-.691	-1.184	-1.229	-1.091	-1.075	-.897
	25.00	-.589	-.901	-1.121	-1.189	-1.105	-.994	-.660	-1.117	-1.200	-1.087	-1.072	-.894
	30.00	-.604	-.527	-1.085	-1.164	-1.094	-.995	-.643	-.674	-1.188	-1.065	-1.056	-.892
	35.00	-.580	-.559	-1.069	-1.148	-1.081	-.997	-.636	-.618	-1.177	-1.049	-1.047	-.888
	40.00	-.593	-.579	-1.056	-1.132	-1.067	-.996	-.646	-.618	-1.167	-1.031	-1.037	-.890
	45.00	-.571	-.591	-1.044	-1.117	-1.056	-.990	-.617	-.630	-1.156	-1.000	-1.033	-.886
	50.00	-.595	-.606	-1.035	-1.102	-1.040	-.983	-.640	-.650	-1.144	-.979	-1.020	-.883
	55.00	-.626	-.629	-1.007	-1.083	-1.020	-.971	-.657	-.671	-1.118	-.984	-1.006	-.881
	60.00	-.658	-.653	-.949	-1.045	-1.002	-.963	-.700	-.699	-1.033	-.948	-.996	-.877
	65.00	-.663	-.663	-.836	-1.026	-.981	-.956	-.701	-.710	-.891	-.943	-.975	-.871
	70.00	-.645	-.672	-.686	-1.009	-.966	-.950	-.673	-.721	-.780	-.934	-.957	-.872
	75.00	-.633	-.682	-.614	-.968	-.950	-.949	-.667	-.708	-.723	-.922	-.943	-.871
	80.00	-.679	-.698	-.603	-.942	-.932	-.949	-.709	-.684	-.703	-.918	-.929	-.868
	85.00	-.674	-.696	-.617	-.911	-.898	-.952	-.675	-.660	-.707	-.914	-.904	-.865
	90.00	-.661	-.688	-.596	-.867	-.864	-.958	-.604	-.656	-.707	-.901	-.880	-.869
	95.00	-.568	-.584	-.571	-.821	-.869	-.954	-.512	-.611	-.707	-.890	-.894	-.866
Lower surface	1.25	.977	.906	.828	.775	.719	.635	1.015	.936	.846	.778	.715	.641
	2.50	1.010	.859	.806	.767	.732	.618	1.073	.909	.843	.796	.756	.636
	5.00	.919	.780	.741	.712	.689	.597	.981	.841	.793	.758	.732	.627
	7.50	.830	.716	.688	.662	.647	.553	.887	.784	.743	.716	.698	.591
	10.00	.754	.671	.650	.624	.602	.512	.819	.739	.703	.680	.660	.553
	15.00	.663	.601	.579	.563	.540	.427	.725	.669	.636	.614	.593	.477
	20.00	.589	.567	.525	.496	.484	.351	.652	.617	.583	.557	.542	.407
	25.00	.542	.502	.475	.455	.443	.290	.601	.572	.536	.513	.499	.343
	30.00	.493	.457	.435	.420	.401	.225	.557	.523	.495	.475	.458	.274
	35.00	.407	.413	.396	.375	.358	.192	.468	.476	.456	.431	.411	.244
	40.00	.407	.379	.362	.338	.325	.128	.464	.442	.420	.393	.383	.180
	45.00	.384	.341	.327	.298	.289	.064	.440	.407	.383	.355	.343	.118
	50.00	.322	.306	.291	.263	.253	.032	.380	.365	.349	.319	.308	.085
	55.00	.281	.271	.263	.243	.209	-.002	.338	.331	.321	.300	.263	.055
	60.00	.276	.244	.230	.209	.173	-.027	.330	.299	.284	.268	.226	.029
	65.00	.232	.210	.208	.181	.137	-.054	.284	.267	.260	.237	.195	.004
	70.00	.194	.198	.185	.162	.130	-.076	.241	.252	.238	.217	.183	-.017
	75.00	.173	.165	.159	.138	.101	-.085	.222	.215	.207	.187	.147	-.033
	80.00	.150	.148	.140	.120	.080	-.101	.188	.191	.183	.164	.125	-.053
	85.00	.117	.126	.125	.101	.056	-.071	.154	.165	.166	.144	.099	-.037
	90.00	.095	.100	.097	.080	.041	-.123	.118	.135	.131	.115	.076	-.084
	95.00	.062	.056	.057	.023	-.001	-.110	.075	.084	.083	.051	.029	-.084
		M = 0.98	α = 20.13°					M = 0.98	α = 22.24°				
Upper surface	0.00	-.027	-.850	-1.047	-1.159	-1.047	-.931	-.104	-.987	-1.141	-1.055	-.988	-.974
	1.25	-.971	-1.333	-1.288	-1.106	-1.004	-.908	-1.052	-1.265	-1.190	-1.043	-.982	-.945
	2.50	-1.072	-1.315	-1.265	-1.108	-1.004	-.908	-1.133	-1.236	-1.195	-1.044	-.977	-.945
	5.00	-1.143	-1.314	-1.288	-1.102	-1.003	-.907	-1.182	-1.232	-1.195	-1.034	-.980	-.941
	7.50	-1.107	-1.309	-1.259	-1.101	-.996	-.908	-1.159	-1.229	-1.168	-1.036	-.975	-.943
	10.00	-1.070	-1.296	-1.281	-1.099	-.997	-.908	-1.136	-1.223	-1.172	-1.039	-.972	-.942
	15.00	-.922	-1.274	-1.281	-1.089	-.999	-.909	-.991	-1.234	-1.173	-1.035	-.983	-.941
	20.00	-.752	-1.236	-1.270	-1.075	-1.000	-.908	-.786	-1.206	-1.168	-1.020	-.993	-.941
	25.00	-.709	-1.227	-1.231	-1.075	-.993	-.907	-.739	-1.189	-1.128	-1.031	-.989	-.937
	30.00	-.700	-.922	-1.212	-1.062	-.990	-.906	-.724	-1.076	-1.128	-1.021	-.992	-.930
	35.00	-.686	-.738	-1.181	-1.060	-.986	-.904	-.686	-.891	-1.122	-1.019	-.989	-.927
	40.00	-.705	-.699	-1.153	-1.048	-.981	-.907	-.694	-.806	-1.122	-1.021	-.984	-.935
	45.00	-.675	-.695	-1.117	-1.034	-.977	-.905	-.659	-.794	-1.104	-1.021	-.987	-.929
	50.00	-.673	-.689	-1.089	-1.026	-.974	-.904	-.669	-.805	-1.085	-1.023	-.981	-.929
	55.00	-.678	-.677	-1.055	-1.014	-.967	-.906	-.681	-.815	-1.077	-1.023	-.980	-.932
	60.00	-.725	-.686	-1.035	-.995	-.965	-.906	-.760	-.835	-1.063	-1.008	-.978	-.931
	65.00	-.695	-.708	-1.009	-.991	-.958	-.902	-.786	-.838	-1.046	-1.002	-.973	-.920
	70.00	-.645	-.728	-.986	-.987	-.948	-.903	-.751	-.828	-1.030	-.998	-.963	-.923
	75.00	-.635	-.746	-.952	-.981	-.946	-.906	-.756	-.822	-1.007	-1.000	-.956	-.929
	80.00	-.703	-.777	-.916	-.978	-.942	-.904	-.817	-.827	-.991	-.996	-.951	-.924
	85.00	-.721	-.782	-.888	-.977	-.924	-.900	-.790	-.814	-.975	-.992	-.933	-.915
	90.00	-.729	-.782	-.857	-.967	-.909	-.902	-.780	-.805	-.954	-.979	-.919	-.921
	95.00	-.704	-.769	-.836	-.961	-.930	-.902	-.717	-.790	-.935	-.972	-.936	-.921
Lower surface	1.25	1.021	.947	.846	.764	.687	.621	1.002	.957	.844	.749	.667	.604
	2.50	1.101	.942	.865	.802	.754	.630	1.106	.971	.882	.804	.755	.625
	5.00	1.025	.888	.827	.780	.748	.633	1.053	.932	.861	.804	.769	.639
	7.50	.939	.832	.785	.746	.722	.606	.977	.884	.826	.773	.805	.625
	10.00	.870	.793	.747	.715	.688	.572	.913	.840	.791	.752	.747	.597
	15.00	.779	.722	.688	.658	.629	.504	.824	.774	.734	.704	.671	.534
	20.00	.712	.656	.633	.602	.583	.439	.756	.714	.681	.647	.626	.475
	25.00	.655	.622	.586	.561	.542	.379	.705	.669	.633	.602	.586	.418
	30.00	.610	.575	.540	.524	.499	.316	.662	.623	.592	.568	.543	.356
	35.00	.530	.531	.501	.478	.455	.280	.584	.585	.552	.523	.500	.319
	40.00	.522	.492	.466	.443	.426	.222	.571	.544	.518	.489	.471	.263
	45.00	.496	.457	.429	.403	.388	.159	.542	.508	.478	.451	.434	.201
	50.00	.428	.414	.393	.369	.349	.127	.484	.465	.441	.409	.395	.170
	55.00	.391	.380	.363	.343	.305	.093	.438	.427	.410	.386	.350	.137
	60.00	.382	.346	.326	.306	.266	.066	.425	.396	.372	.350	.311	.108
	65.00	.332	.309	.300	.270	.231	.040	.377	.359	.343	.314	.275	.081
	70.00	.283	.292	.273	.253	.219	.014	.323	.336	.316	.291	.263	.050
	75.00	.262	.253	.241	.223	.182	-.005	.299	.297	.282	.258	.226	.030
	80.00	.230	.230	.218	.195	.156	-.028	.263	.268	.251	.231	.194	.005
	85.00	.185	.201	.192	.172	.127	-.016	.218	.236	.226	.203	.163	.011
	90.00	.146	.163	.155	.138	.097	-.066	.175	.199	.183	.168	.132	-.043
	95.00	.093	.105	.099	.069	.043	-.071	.104	.129	.121	.090	.071	-.052

TABLE I.- STEEL WING PRESSURE COEFFICIENT DATA FOR THE TEST RANGE OF ANGLE OF ATTACK AND MACH NUMBER - Continued

	Percent c	0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2	0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2
		Pressure coefficient, P, at:											
		M = 0.98		α = 24.37°				M = 0.98		α = 26.48°			
Upper surface	0.00	-.159	-1.095	-1.225	-1.100	-1.072	-1.001	-.227	-1.164	-1.189	-1.132	-1.107	-1.058
	1.25	-1.131	-1.226	-1.171	-1.090	-1.055	-.974	-1.169	-1.195	-1.167	-1.123	-1.095	-1.031
	2.50	-1.169	-1.208	-1.153	-1.089	-1.053	-.973	-1.171	-1.184	-1.152	-1.122	-1.091	-1.030
	5.00	-1.256	-1.204	-1.170	-1.082	-1.050	-.970	-1.214	-1.185	-1.169	-1.115	-1.089	-1.026
	7.50	-1.233	-1.205	-1.146	-1.082	-1.044	-.970	-1.206	-1.181	-1.146	-1.115	-1.082	-1.025
	10.00	-1.210	-1.202	-1.157	-1.082	-1.042	-.968	-1.197	-1.180	-1.157	-1.116	-1.078	-1.025
	15.00	-1.074	-1.207	-1.155	-1.077	-1.044	-.968	-1.129	-1.189	-1.161	-1.111	-1.084	-1.026
	20.00	-.889	-1.196	-1.153	-1.067	-1.045	-.966	-.951	-1.185	-1.163	-1.099	-1.089	-1.025
	25.00	-.818	-1.179	-1.141	-1.072	-1.040	-.963	-.854	-1.173	-1.150	-1.109	-1.081	-1.020
	30.00	-.810	-1.157	-1.141	-1.065	-1.037	-.961	-.860	-1.161	-1.155	-1.099	-1.081	-1.015
	35.00	-.779	-1.126	-1.139	-1.065	-1.034	-.959	-.865	-1.140	-1.152	-1.099	-1.077	-1.011
	40.00	-.777	-1.045	-1.140	-1.065	-1.028	-.960	-.869	-1.080	-1.154	-1.098	-1.073	-1.017
	45.00	-.743	-.974	-1.127	-1.064	-1.025	-.957	-.838	-1.015	-1.148	-1.095	-1.073	-1.013
	50.00	-.750	-.923	-1.113	-1.061	-1.021	-.955	-.829	-.971	-1.134	-1.092	-1.068	-1.011
	55.00	-.740	-.891	-1.104	-1.054	-1.017	-.955	-.808	-.936	-1.130	-1.088	-1.067	-1.011
	60.00	-.805	-.885	-1.095	-1.032	-1.014	-.952	-.850	-.935	-1.126	-1.063	-1.065	-1.008
	65.00	-.826	-.869	-1.085	-1.032	-1.007	-.945	-.864	-.919	-1.117	-1.065	-1.059	-1.000
	70.00	-.786	-.857	-1.079	-1.031	-1.001	-.947	-.826	-.899	-1.111	-1.064	-1.052	-1.000
	75.00	-.785	-.852	-1.059	-1.025	-.997	-.947	-.828	-.886	-1.095	-1.064	-1.047	-1.005
	80.00	-.843	-.857	-1.049	-1.022	-.991	-.944	-.854	-.889	-1.086	-1.064	-1.042	-1.000
	85.00	-.832	-.854	-1.033	-1.022	-.973	-.940	-.862	-.886	-1.067	-1.063	-1.021	-.992
	90.00	-.824	-.860	-1.011	-1.014	-.956	-.941	-.868	-.899	-1.046	-1.052	-1.003	-.993
	95.00	-.798	-.845	-.993	-1.002	-.976	-.938	-.820	-.884	-1.025	-1.029	-1.017	-.994
Lower surface	1.25	.968	.956	.839	.738	.639	.592	.932	.956	.828	.715	.608	.566
	2.50	1.100	.991	.894	.811	.751	.622	1.090	1.007	.895	.803	.739	.604
	5.00	1.070	.965	.890	.831	.785	.649	1.084	.990	.907	.840	.791	.643
	7.50	1.007	.919	.864	.812	.824	.642	1.031	.957	.889	.831	.789	.646
	10.00	.948	.885	.834	.790	.752	.622	.979	.926	.869	.816	.772	.631
	15.00	.863	.821	.783	.740	.705	.569	.897	.862	.818	.775	.733	.585
	20.00	.796	.766	.729	.692	.664	.511	.834	.813	.769	.729	.695	.532
	25.00	.749	.718	.686	.651	.625	.455	.789	.765	.727	.688	.656	.483
	30.00	.708	.673	.643	.614	.586	.398	.746	.720	.686	.656	.617	.428
	35.00	.632	.635	.602	.571	.542	.364	.673	.678	.646	.613	.577	.393
	40.00	.615	.589	.568	.535	.513	.307	.655	.639	.609	.576	.547	.335
	45.00	.587	.554	.530	.495	.478	.245	.629	.600	.573	.536	.511	.277
	50.00	.526	.513	.491	.460	.439	.213	.567	.555	.534	.498	.476	.243
	55.00	.482	.474	.460	.430	.396	.179	.527	.520	.503	.471	.428	.210
	60.00	.460	.439	.424	.393	.354	.147	.503	.483	.463	.435	.388	.178
	65.00	.416	.402	.389	.357	.320	.120	.456	.443	.432	.396	.353	.149
	70.00	.361	.376	.363	.333	.302	.088	.399	.417	.400	.372	.336	.114
	75.00	.336	.334	.325	.296	.263	.063	.366	.371	.359	.339	.294	.087
	80.00	.297	.304	.293	.268	.232	.038	.332	.339	.327	.302	.257	.058
	85.00	.252	.266	.264	.257	.196	.041	.280	.302	.294	.270	.219	.059
	90.00	.206	.224	.216	.196	.158	-.017	.228	.252	.243	.225	.178	-.003
	95.00	.119	.157	.154	.116	.091	-.034	.131	.180	.176	.135	.107	-.021

TABLE I.- STEEL WING PRESSURE COEFFICIENT DATA FOR THE TEST RANGE OF ANGLE OF ATTACK AND MACH NUMBER - Continued

		Pressure coefficient, P, at:											
		0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2	0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2
	Percent c	M = 1.00		α = -2.00°				M = 1.00		α = -0.04°			
Upper surface	0.00	.074	.758	.708	.675	.616	.539	.067	.768	.722	.722	.687	.724
	1.25	.374	.297	.322	.328	.306	.343	.260	.088	.079	.015	-.022	.103
	2.50	.325	.220	.229	.220	.221	.217	.205	.042	.033	-.013	-.058	-.082
	5.00	.273	.192	.179	.170	.150	.194	.151	.041	.033	-.022	-.068	-.089
	7.50	.242	.162	.147	.131	.106	.130	.126	.041	.015	-.035	-.084	-.099
	10.00	.211	.139	.114	.106	.076	.091	.100	.032	-.008	-.051	-.094	-.116
	15.00	.168	.107	.080	.080	.042	.014	.072	.014	-.029	-.059	-.105	-.140
	20.00	.105	.074	.048	.041	.014	-.054	.044	-.016	-.055	-.087	-.113	-.165
	25.00	.092	.057	.029	.011	-.017	-.151	.005	-.030	-.065	-.100	-.130	-.232
	30.00	.058	.041	.019	-.010	-.030	-.182	-.055	-.045	-.086	-.118	-.138	-.236
	35.00	.030	.012	-.008	-.030	-.060	-.204	-.045	-.074	-.102	-.123	-.164	-.233
	40.00	.006	.003	-.030	-.066	-.083	-.265	-.072	-.089	-.113	-.154	-.184	-.285
	45.00	-.001	-.013	-.051	-.087	-.112	-.262	-.087	-.097	-.126	-.165	-.202	-.272
	50.00	.009	-.034	-.081	-.110	-.144	-.286	-.063	-.097	-.148	-.189	-.230	-.295
	55.00	-.029	-.072	-.113	-.140	-.171	-.294	-.088	-.133	-.178	-.211	-.251	-.306
	60.00	-.083	-.103	-.128	-.156	-.199	-.283	-.144	-.169	-.198	-.228	-.270	-.293
	65.00	-.086	-.103	-.136	-.171	-.205	-.306	-.149	-.169	-.207	-.250	-.275	-.322
	70.00	-.101	-.113	-.136	-.176	-.238	-.311	-.168	-.178	-.207	-.259	-.302	-.329
	75.00	-.106	-.125	-.154	-.176	-.250	-.298	-.169	-.192	-.222	-.255	-.317	-.317
	80.00	-.129	-.148	-.154	-.182	-.261	-.291	-.190	-.211	-.222	-.251	-.317	-.305
	85.00	-.155	-.136	-.154	-.189	-.254	-.272	-.220	-.203	-.225	-.248	-.297	-.287
	90.00	-.125	-.131	-.143	-.188	-.233	-.254	-.188	-.200	-.213	-.248	-.268	-.269
	95.00	-.100	-.131	-.143	-.181	-.243	-.264	-.139	-.189	-.205	-.233	-.268	-.276
Lower surface	1.25	.146	-.129	-.301	-.567	-.707	-.745	.251	.127	.069	.031	.033	-.033
	2.50	.091	-.102	-.154	-.432	-.588	-.709	.199	.067	.041	.008	.001	-.035
	5.00	.066	-.033	-.104	-.162	-.320	-.636	.164	.077	.036	.008	-.026	-.052
	7.50	.048	-.044	-.113	-.174	-.204	-.569	.136	.053	.015	-.021	-.039	-.074
	10.00	.012	-.052	-.109	-.183	-.224	-.467	.102	.034	.001	-.044	-.054	-.084
	15.00	-.002	-.052	-.111	-.167	-.220	-.315	.072	.020	-.014	-.054	-.074	-.112
	20.00	-.031	-.053	-.115	-.187	-.241	-.310	.043	.004	-.040	-.077	-.100	-.164
	25.00	-.049	-.062	-.106	-.168	-.221	-.310	.020	-.003	-.047	-.080	-.108	-.213
	30.00	-.036	-.072	-.120	-.177	-.227	-.257	.025	-.023	-.064	-.056	-.120	-.227
	35.00	-.054	-.110	-.149	-.204	-.246	-.233	-.011	-.057	-.075	-.115	-.149	-.199
	40.00	-.107	-.121	-.154	-.215	-.253	-.269	-.047	-.058	-.091	-.137	-.160	-.243
	45.00	-.077	-.118	-.167	-.227	-.272	-.299	-.023	-.072	-.118	-.167	-.187	-.276
	50.00	-.125	-.155	-.198	-.251	-.287	-.322	-.082	-.107	-.153	-.196	-.214	-.296
	55.00	-.150	-.176	-.212	-.266	-.314	-.341	-.104	-.128	-.172	-.207	-.248	-.313
	60.00	-.180	-.202	-.233	-.288	-.334	-.344	-.138	-.156	-.189	-.229	-.269	-.318
	65.00	-.193	-.211	-.244	-.305	-.354	-.344	-.147	-.165	-.199	-.248	-.291	-.322
	70.00	-.200	-.201	-.244	-.292	-.347	-.340	-.155	-.158	-.199	-.235	-.287	-.322
	75.00	-.253	-.234	-.253	-.306	-.358	-.327	-.210	-.194	-.205	-.246	-.308	-.310
	80.00	-.257	-.254	-.264	-.314	-.342	-.338	-.215	-.215	-.217	-.256	-.314	-.320
	85.00	-.271	-.262	-.266	-.305	-.341	-.280	-.228	-.218	-.222	-.248	-.318	-.262
	90.00	-.258	-.266	-.271	-.295	-.327	-.319	-.214	-.221	-.222	-.248	-.303	-.296
	95.00	-.179	-.232	-.253	-.278	-.291	-.288	-.157	-.201	-.206	-.261	-.285	-.256
		M = 1.00		α = 1.90°				M = 1.00		α = 3.93°			
Upper surface	0.00	.093	.734	.646	.653	.630	.713	.089	.623		.482	.463	.668
	1.25	.179	-.097	-.183	-.572	-.678	-.536	.050	-.784	-.899	-.973	-.994	-.746
	2.50	.118	-.109	-.168	-.293	-.589	-.723	-.008	-.417	-.784	-.871	-.923	-.981
	5.00	.054	-.072	-.105	-.206	-.287	-.656	-.121	-.305	-.639	-.795	-.848	-.919
	7.50	.036	-.043	-.100	-.196	-.254	-.584	-.120	-.176	-.205	-.741	-.797	-.858
	10.00	.017	-.040	-.107	-.189	-.264	-.518	-.118	-.162	-.215	-.699	-.785	-.821
	15.00	.020	-.043	-.112	-.159	-.249	-.352	-.072	-.146	-.208	-.288	-.745	-.810
	20.00	.015	-.073	-.122	-.183	-.242	-.293	-.065	-.161	-.232	-.290	-.684	-.780
	25.00	-.047	-.068	-.124	-.174	-.227	-.311	-.128	-.156	-.216	-.270	-.417	-.779
	30.00	-.116	-.092	-.138	-.188	-.233	-.256	-.197	-.176	-.222	-.277	-.310	-.735
	35.00	-.089	-.120	-.156	-.199	-.251	-.248	-.158	-.197	-.237	-.284	-.322	-.669
	40.00	-.115	-.126	-.171	-.227	-.268	-.292	-.191	-.206	-.258	-.311	-.340	-.682
	45.00	-.136	-.142	-.187	-.238	-.288	-.298	-.208	-.218	-.269	-.322	-.356	-.647
	50.00	-.115	-.143	-.194	-.250	-.307	-.321	-.191	-.218	-.279	-.339	-.380	-.637
	55.00	-.126	-.173	-.218	-.264	-.329	-.331	-.198	-.242	-.293	-.355	-.401	-.572
	60.00	-.180	-.204	-.232	-.275	-.337	-.314	-.251	-.271	-.309	-.362	-.419	-.451
	65.00	-.185	-.204	-.247	-.292	-.331	-.344	-.257	-.272	-.318	-.373	-.417	-.405
	70.00	-.203	-.208	-.241	-.292	-.344	-.346	-.268	-.277	-.312	-.375	-.428	-.367
	75.00	-.203	-.222	-.251	-.293	-.347	-.340	-.263	-.289	-.325	-.370	-.431	-.353
	80.00	-.229	-.242	-.254	-.291	-.330	-.325	-.290	-.306	-.325	-.374	-.431	-.353
	85.00	-.257	-.238	-.254	-.289	-.307	-.306	-.316	-.306	-.325	-.371	-.386	-.365
	90.00	-.227	-.237	-.246	-.276	-.281	-.289	-.296	-.306	-.316	-.341	-.335	-.360
	95.00	-.151	-.215	-.237	-.242	-.265	-.294	-.207	-.279	-.310	-.333	-.312	-.373
Lower surface	1.25	.377	.340	.318	.330	.371	.293	.492	.502	.482	.499	.531	.438
	2.50	.329	.249	.248	.260	.278	.234	.451	.394	.401	.409	.425	.373
	5.00	.289	.216	.201	.204	.194	.189	.408	.337	.331	.321	.329	.313
	7.50	.253	.179	.161	.161	.160	.135	.359	.286		.265	.275	.246
	10.00	.213	.153	.131	.124	.122	.100	.314	.293	.241	.230	.226	.201
	15.00	.176	.127	.102	.091	.073	.038	.265	.218	.205	.189	.170	.118
	20.00	.141	.099	.069	.047	.035	-.043	.223	.216	.163	.134	.132	.057
	25.00	.120	.086	.061	.026	.006	-.122	.195	.172	.132	.102	.095	-.045
	30.00	.110	.062	.045	.012	-.012	-.166	.178	.134	.110	.080	.064	-.094
	35.00	.066	.037	.020	-.020	-.049	-.158	.114	.102	.077	.045	.029	-.125
	40.00	.042	.025	-.001	-.043	-.064	-.218	.105	.085	.054	.017	.006	-.163
	45.00	.051	.001	-.032	-.078	-.095	-.257	.105	.058	.020	-.017	-.026	-.218
	50.00	-.012	-.032	-.062	-.103	-.120	-.277	.039	.022	-.010	-.042	-.053	-.248
	55.00	-.034	-.055	-.082	-.120	-.157	-.290	.013	-.004	-.028	-.057	-.099	-.272
	60.00	-.064	-.080	-.109	-.140	-.184	-.290	-.008	-.023	-.048	-.080	-.124	-.288
	65.00	-.074	-.089	-.112	-.159	-.216	-.291	-.018	-.037	-.059	-.101	-.153	-.304
	70.00	-.076	-.084	-.118	-.149	-.212	-.291	-.022	-.037	-.067	-.102	-.156	-.308
	75.00	-.137	-.122	-.135	-.169	-.238	-.280	-.085	-.075	-.090	-.124	-.184	-.309
	80.00	-.141	-.141	-.148	-.180	-.255	-.289	-.091	-.086	-.104	-.143	-.203	-.314
	85.00	-.156	-.139	-.146	-.183	-.267	-.235	-.104	-.086	-.102	-.149	-.217	-.261
	90.00	-.135	-.139	-.146	-.187	-.263	-.272	-.088	-.092	-.108	-.151	-.217	-.310
	95.00	-.097	-.123	-.145	-.201	-.244	-.234	-.070	-.083	-.103	-.160	-.211	-.272

TABLE I.- STEEL WING PRESSURE COEFFICIENT DATA FOR THE TEST RANGE OF ANGLE OF ATTACK AND MACH NUMBER - Continued

	Pressure coefficient, P, at:												
	Percent c	0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2	0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2
		M = 1.00		α = 5.90°				M = 1.00		α = 7.90°			
Upper surface	0.00	.088	.504		.308	.253	.577	.062	.339	.191	.086	-.036	.400
	1.25	-.073	-.976	-1.062	-1.117	-1.137	-.897	-.210	-1.149	-1.197	-1.229	-1.255	-1.065
	2.50	-.139	-.886	-.943	-1.022	-1.073	-1.127	-.287	-1.070	-1.090	-1.168	-1.218	-1.256
	5.00	-.272	-.588	-.868	-.939	-.988	-1.062	-.436	-.798	-1.019	-1.078	-1.137	-1.204
	7.50	-.256	-.282	-.808	-.885	-.930	-.996	-.407	-.536	-.957	-1.028	-1.074	-1.143
	10.00	-.240	-.265	-.761	-.853	-.917	-.963	-.377	-.444	-.928	-.995	-1.064	-1.110
	15.00	-.161	-.237	-.318	-.785	-.878	-.935	-.285	-.373	-.851	-.925	-1.019	-1.074
	20.00	-.144	-.237	-.297	-.760	-.839	-.907	-.255	-.346	-.489	-.897	-.972	-1.049
	25.00	-.202	-.291	-.292	-.596	-.798	-.899	-.287	-.331	-.422	-.866	-.935	-1.034
	30.00	-.262	-.243	-.291	-.408	-.778	-.864	-.343	-.331	-.379	-.846	-.916	-1.007
	35.00	-.229	-.266	-.304	-.367	-.767	-.789	-.313	-.348	-.379	-.717	-.907	-.922
	40.00	-.263	-.277	-.325	-.364	-.739	-.801	-.347	-.357	-.389	-.581	-.898	-.924
	45.00	-.272	-.284	-.336	-.373	-.618	-.769	-.336	-.363	-.403	-.539	-.895	-.895
	50.00	-.254	-.284	-.347	-.391	-.505	-.772	-.336	-.366	-.410	-.519	-.893	-.905
	55.00	-.264	-.307	-.358	-.410	-.482	-.772	-.346	-.385	-.428	-.511	-.888	-.901
	60.00	-.315	-.332	-.370	-.416	-.477	-.760	-.392	-.410	-.437	-.499	-.827	-.892
	65.00	-.320	-.335	-.377	-.431	-.465	-.772	-.398	-.414	-.445	-.506	-.708	-.902
	70.00	-.324	-.336	-.375	-.431	-.477	-.776	-.402	-.419	-.441	-.506	-.670	-.909
	75.00	-.311	-.347	-.384	-.430	-.482	-.773	-.381	-.427	-.455	-.496	-.644	-.905
	80.00	-.347	-.367	-.385	-.434	-.487	-.770	-.420	-.445	-.453	-.498	-.618	-.908
	85.00	-.372	-.367	-.387	-.430	-.469	-.767	-.444	-.442	-.454	-.500	-.573	-.913
	90.00	-.353	-.367	-.376	-.422	-.411	-.760	-.424	-.444	-.444	-.495	-.485	-.914
	95.00	-.271	-.350	-.376	-.405	-.370	-.742	-.344	-.428	-.444	-.476	-.391	-.929
Lower surface	1.25	.611	.617	.589	.603	.625	.520	.713	.713	.677	.673	.679	.577
	2.50	.570	.509	.508	.515	.526	.457	.680	.607	.603	.597	.593	.516
	5.00	.516	.438	.425	.422	.425	.394	.618	.525	.514	.503	.502	.457
	7.50	.460	.379		.362	.368	.328	.551	.464	.453	.446	.442	.389
	10.00	.406	.346	.331	.322	.315	.277	.492	.431	.420	.401	.390	.336
	15.00	.347	.298	.282	.269	.258	.186	.426	.368	.356	.347	.323	.245
	20.00	.294	.276	.235	.212	.208	.106	.366	.342	.306	.283	.273	.160
	25.00	.261	.230	.199	.176	.169	.028	.324	.291	.266	.244	.228	.090
	30.00	.238	.194	.172	.152	.134	-.028	.296	.250	.237	.213	.198	.028
	35.00	.162	.161	.138	.113	.097	-.036	.212	.215	.202	.173	.154	.009
	40.00	.161	.137	.111	.084	.072	-.109	.215	.189	.172	.145	.127	-.063
	45.00	.158	.107	.078	.052	.038	-.169	.207	.159	.140	.111	.095	-.121
	50.00	.092	.073	.049	.024	.006	-.197	.137	.121	.107	.078	.061	-.154
	55.00	.060	.045	.030	.006	-.035	-.227	.104	.094	.088	.065	.017	-.182
	60.00	.047	.026	.005	-.017	-.062	-.242	.103	.075	.063	.035	-.013	-.202
	65.00	.028	.008	-.010	-.040	-.093	-.267	.073	.053	.047	.011	-.045	-.235
	70.00	.028	.005	-.019	-.044	-.098	-.277	.059	.046	.033	.001	-.092	-.239
	75.00	-.036	-.033	-.047	-.075	-.132	-.278	.036	.014	.006	-.031	-.089	-.241
	80.00	-.044	-.041	-.057	-.091	-.153	-.286	.008	.007	.001	-.034	-.099	-.258
	85.00	-.053	-.044	-.056	-.091	-.166	-.258	-.006	.002	.000	-.035	-.113	-.219
	90.00	-.048	-.047	-.061	-.089	-.163	-.288	-.008	-.004	-.006	-.035	-.104	-.263
	95.00	-.048	-.053	-.059	-.102	-.154	-.257	-.020	-.020	-.015	-.053	-.102	-.230
		M = 1.00		α = 9.87°				M = 1.00		α = 13.74°			
Upper surface	0.00	.049	.179	.009	-.153	-.528	.191	-.029	-.191	-.431	-.652	-.823	-.356
	1.25	-.321	-1.212	-1.255	-1.259	-1.268	-1.153	-.528	-1.219	-1.190	-1.171	-1.208	-1.232
	2.50	-.407	-1.132	-1.161	-1.226	-1.249	-1.282	-.645	-1.182	-1.170	-1.172	-1.199	-1.266
	5.00	-.559	-.940	-1.112	-1.153	-1.191	-1.241	-.773	-1.136	-1.198	-1.166	-1.203	-1.259
	7.50	-.524	-.723	-1.046	-1.112	-1.157	-1.204	-.734	-1.080	-1.137	-1.161	-1.177	-1.249
	10.00	-.489	-.643	-1.024	-1.076	-1.120	-1.175	-.695	-1.028	-1.109	-1.149	-1.160	-1.246
	15.00	-.391	-.550	-.955	-1.016	-1.085	-1.140	-.574	-.949	-1.029	-1.105	-1.152	-1.223
	20.00	-.338	-.496	-.675	-.982	-1.049	-1.116	-.499	-.864	-.968	-1.062	-1.132	-1.209
	25.00	-.353	-.440	-.604	-.960	-1.006	-1.103	-.480	-.603	-.923	-1.038	-1.108	-1.193
	30.00	-.401	-.400	-.549	-.943	-.993	-1.074	-.504	-.402	-.889	-1.002	-1.084	-1.169
	35.00	-.373	-.397	-.429	-.922	-.985	-.992	-.473	-.450	-.870	-.970	-1.063	-1.115
	40.00	-.405	-.398	-.456	-.788	-.976	-.988	-.492	-.475	-.852	-.940	-1.043	-1.108
	45.00	-.387	-.402	-.458	-.692	-.973	-.961	-.485	-.497	-.834	-.924	-1.035	-1.088
	50.00	-.392	-.408	-.448	-.676	-.970	-.973	-.504	-.503	-.808	-.914	-1.011	-1.081
	55.00	-.410	-.428	-.481	-.670	-.969	-.972	-.521	-.523	-.777	-.902	-.998	-1.071
	60.00	-.450	-.455	-.491	-.649	-.967	-.965	-.547	-.546	-.734	-.882	-.972	-1.064
	65.00	-.454	-.459	-.489	-.636	-.949	-.972	-.550	-.557	-.674	-.877	-.926	-1.043
	70.00	-.460	-.465	-.495	-.617	-.862	-.977	-.552	-.562	-.616	-.876	-.871	-1.022
	75.00	-.439	-.476	-.507	-.596	-.772	-.985	-.545	-.570	-.577	-.863	-.821	-1.007
	80.00	-.474	-.494	-.510	-.583	-.702	-.985	-.579	-.590	-.563	-.848	-.781	-.990
	85.00	-.492	-.494	-.512	-.571	-.619	-.986	-.589	-.596	-.568	-.834	-.726	-.996
	90.00	-.482	-.497	-.503	-.555	-.555	-.993	-.583	-.595	-.562	-.714	-.677	-1.019
	95.00	-.387	-.466	-.505	-.534	-.510	-1.014	-.496	-.549	-.567	-.600	-.652	-1.026
Lower surface	1.25	.799	.786	.736	.724	.719	.617	.944	.890	.824	.785	.747	.657
	2.50	.774	.691	.672	.658	.657	.566	.957	.823	.784	.758	.731	.624
	5.00	.705	.608	.587	.571	.570	.516	.874	.740	.720	.685	.670	.589
	7.50	.630	.540	.525	.504	.512	.453	.787	.671	.653	.631	.624	.535
	10.00	.570	.505	.487	.468	.459	.396	.714	.632	.617	.593	.574	.484
	15.00	.492	.440	.422	.406	.397	.305	.625	.567	.544	.529	.510	.407
	20.00	.426	.405	.367	.340	.345	.229	.555	.527	.488	.464	.456	.330
	25.00	.382	.352	.327	.302	.300	.158	.500	.469	.444	.423	.411	.264
	30.00	.349	.312	.290	.271	.264	.088	.465	.425	.405	.386	.368	.198
	35.00	.263	.273	.254	.233	.221	.071	.374	.385	.368	.345	.324	.164
	40.00	.266	.244	.226	.197	.192	-.005	.380	.353	.335	.311	.297	.102
	45.00	.255	.212	.190	.166	.159	-.062	.363	.318	.300	.277	.261	.037
	50.00	.183	.175	.162	.134	.125	-.098	.290	.281	.270	.241	.227	.001
	55.00	.151	.147	.154	.114	.083	-.125	.255	.251	.242	.222	.183	-.025
	60.00	.150	.124	.111	.086	.045	-.146	.255	.221	.211	.191	.149	-.052
	65.00	.119	.102	.093	.061	.013	-.184	.214	.194	.192	.159	.112	-.081
	70.00	.091	.093	.076	.047	.007	-.179	.175	.182	.174	.146	.120	-.093
	75.00	.073	.063	.055	.025	-.022	-.195	.160	.151	.147	.125	.078	-.102
	80.00	.057	.056	.049	.015	-.032	-.207	.140	.138	.134	.110	.065	-.115
	85.00	.040	.049	.043	.015	-.040	-.160	.108	.123	.126	.100	.047	-.084
	90.00	.029	.036	.034	.011	-.037	-.206	.087	.101	.099	.085	.041	-.126
	95.00	.014	.009	.015	-.019	-.048	-.176	.067	.062	.069	.036	.007	-.108

TABLE I.- STEEL WING PRESSURE COEFFICIENT DATA FOR THE TEST RANGE OF ANGLE OF ATTACK AND MACH NUMBER - Continued

		Pressure coefficient, P, at											
	Percent c	0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2	0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2
		M = 1.00		α = 15.83°				M = 1.00		α = 18.03°			
Upper surface	0.00	-.054	-.423	-.645	-.888	-1.008	-.663	-.072	-.613	-.826	-1.042	-1.124	-.879
	1.25	-.657	-1.280	-1.231	-1.171	-1.089	-.983	-.787	-1.299	-1.254	-1.111	-1.012	-.905
	2.50	-.778	-1.271	-1.202	-1.172	-1.088	-.979	-.908	-1.287	-1.225	-1.105	-1.013	-.902
	5.00	-.898	-1.232	-1.228	-1.172	-1.087	-.969	-1.006	-1.275	-1.252	-1.098	-1.019	-.895
	7.50	-.861	-1.186	-1.204	-1.172	-1.078	-.964	-.974	-1.263	-1.229	-1.096	-1.020	-.896
	10.00	-.823	-1.145	-1.203	-1.181	-1.080	-.964	-.942	-1.236	-1.246	-1.094	-1.029	-.895
	15.00	-.691	-1.079	-1.149	-1.170	-1.078	-.964	-.802	-1.185	-1.224	-1.087	-1.043	-.892
	20.00	-.597	-1.039	-1.102	-1.145	-1.072	-.964	-.655	-1.138	-1.195	-1.075	-1.052	-.890
	25.00	-.547	-.851	-1.064	-1.135	-1.064	-.963	-.617	-1.085	-1.165	-1.070	-1.047	-.885
	30.00	-.566	-.477	-1.028	-1.113	-1.057	-.966	-.608	-.634	-1.146	-1.056	-1.037	-.882
	35.00	-.539	-.515	-1.012	-1.094	-1.047	-.963	-.599	-.586	-1.133	-1.042	-1.026	-.877
	40.00	-.546	-.537	-.998	-1.077	-1.033	-.962	-.615	-.593	-1.120	-1.027	-1.017	-.882
	45.00	-.542	-.548	-.991	-1.063	-1.024	-.957	-.596	-.608	-1.110	-1.006	-1.012	-.876
	50.00	-.572	-.562	-.984	-1.049	-1.011	-.951	-.618	-.620	-1.106	-.989	-1.000	-.873
	55.00	-.593	-.587	-.956	-1.030	-.997	-.942	-.645	-.642	-1.091	-.974	-.988	-.873
	60.00	-.609	-.608	-.904	-1.002	-.977	-.930	-.665	-.664	-1.047	-.953	-.975	-.869
	65.00	-.617	-.617	-.802	-.988	-.958	-.922	-.673	-.673	-.932	-.941	-.953	-.861
	70.00	-.604	-.625	-.657	-.970	-.940	-.918	-.641	-.681	-.725	-.930	-.937	-.860
	75.00	-.593	-.633	-.582	-.939	-.921	-.918	-.637	-.686	-.591	-.915	-.922	-.862
	80.00	-.641	-.652	-.572	-.903	-.901	-.918	-.682	-.681	-.582	-.907	-.905	-.859
	85.00	-.645	-.649	-.587	-.875	-.868	-.921	-.676	-.643	-.617	-.896	-.881	-.856
	90.00	-.626	-.645	-.578	-.820	-.835	-.926	-.635	-.619	-.635	-.881	-.853	-.859
	95.00	-.553	-.552	-.571	-.765	-.836	-.924	-.466	-.557	-.655	-.871	-.864	-.859
Lower surface	1.25	1.001	.927	.848	.791	.738	.659	1.027	.949	.859	.791	.731	.658
	2.50	1.034	.875	.826	.783	.750	.640	1.082	.923	.858	.806	.771	.654
	5.00	.948	.808	.763	.726	.705	.619	.991	.858	.805	.766	.744	.644
	7.50	.848	.735	.708	.679	.665	.577	.903	.796	.758	.721	.713	.610
	10.00	.780	.693	.672	.643	.622	.534	.830	.751	.718	.689	.674	.570
	15.00	.684	.625	.603	.579	.560	.453	.741	.681	.657	.626	.610	.497
	20.00	.614	.590	.546	.519	.507	.383	.666	.629	.598	.569	.558	.427
	25.00	.567	.529	.502	.474	.462	.312	.619	.586	.553	.525	.517	.364
	30.00	.523	.481	.460	.439	.421	.254	.575	.540	.514	.489	.475	.299
	35.00	.433	.442	.418	.397	.377	.221	.488	.493	.473	.444	.432	.267
	40.00	.433	.407	.386	.361	.349	.154	.483	.458	.439	.410	.402	.203
	45.00	.411	.371	.350	.322	.310	.092	.462	.422	.402	.371	.365	.140
	50.00	.347	.332	.316	.291	.275	.060	.398	.382	.367	.336	.328	.108
	55.00	.303	.299	.292	.267	.229	.032	.354	.351	.339	.312	.284	.078
	60.00	.306	.274	.256	.234	.196	.004	.348	.319	.306	.278	.247	.052
	65.00	.259	.246	.233	.204	.161	-.021	.302	.288	.278	.246	.214	.028
	70.00	.216	.230	.214	.187	.155	-.042	.255	.272	.255	.230	.204	.004
	75.00	.198	.201	.184	.159	.128	-.053	.237	.234	.226	.201	.172	-.009
	80.00	.174	.180	.165	.142	.107	-.071	.209	.214	.205	.181	.148	-.028
	85.00	.144	.155	.154	.130	.084	-.037	.169	.190	.186	.161	.124	-.013
	90.00	.118	.130	.126	.109	.068	-.084	.139	.160	.154	.134	.099	-.058
	95.00	.081	.087	.088	.053	.033	-.078	.095	.109	.109	.071	.055	-.057
		M = 1.00		α = 20.15°				M = 1.00		α = 22.28°			
Upper surface	0.00	-.121	-.786	-.980	-1.113	-1.091	-.909	-.211	-.935	-1.112	-1.084	-.988	-.932
	1.25	-.902	-1.289	-1.248	-1.080	-.989	-.880	-1.009	-1.284	-1.233	-1.042	-.983	-.915
	2.50	-1.011	-1.276	-1.214	-1.083	-.991	-.878	-1.102	-1.267	-1.214	-1.063	-.979	-.916
	5.00	-1.086	-1.273	-1.239	-1.081	-.995	-.874	-1.158	-1.264	-1.238	-1.058	-.980	-.914
	7.50	-1.053	-1.270	-1.204	-1.081	-.992	-.875	-1.128	-1.261	-1.202	-1.056	-.972	-.914
	10.00	-1.019	-1.257	-1.238	-1.080	-.995	-.875	-1.097	-1.255	-1.217	-1.056	-.973	-.915
	15.00	-.872	-1.222	-1.232	-1.073	-1.004	-.875	-.953	-1.241	-1.209	-1.051	-.976	-.916
	20.00	-.704	-1.177	-1.220	-1.058	-1.006	-.873	-.755	-1.216	-1.195	-1.044	-.976	-.916
	25.00	-.657	-1.182	-1.194	-1.061	-.995	-.870	-.705	-1.211	-1.167	-1.050	-.974	-.914
	30.00	-.642	-.877	-1.178	-1.044	-.992	-.868	-.689	-1.018	-1.148	-1.042	-.971	-.914
	35.00	-.635	-.683	-1.163	-1.031	-.987	-.866	-.678	-.805	-1.129	-1.039	-.969	-.913
	40.00	-.652	-.650	-1.146	-1.017	-.979	-.868	-.686	-.703	-1.111	-1.039	-.964	-.915
	45.00	-.626	-.654	-1.129	-.999	-.975	-.865	-.647	-.682	-1.093	-1.036	-.964	-.914
	50.00	-.635	-.664	-1.099	-.983	-.967	-.864	-.639	-.693	-1.073	-1.035	-.959	-.915
	55.00	-.643	-.680	-1.046	-.969	-.960	-.865	-.628	-.724	-1.058	-1.031	-.957	-.915
	60.00	-.688	-.707	-.997	-.947	-.954	-.864	-.673	-.763	-1.042	-1.006	-.956	-.909
	65.00	-.705	-.712	-.962	-.944	-.942	-.859	-.700	-.778	-1.024	-1.006	-.952	-.911
	70.00	-.675	-.698	-.931	-.942	-.929	-.860	-.703	-.789	-1.008	-1.003	-.948	-.914
	75.00	-.668	-.677	-.891	-.931	-.923	-.861	-.718	-.797	-.988	-.994	-.944	-.911
	80.00	-.685	-.687	-.849	-.929	-.916	-.860	-.786	-.807	-.968	-.992	-.942	-.908
	85.00	-.643	-.697	-.818	-.929	-.896	-.858	-.776	-.800	-.944	-.991	-.924	-.910
	90.00	-.625	-.703	-.780	-.917	-.877	-.858	-.765	-.791	-.918	-.983	-.908	-.910
	95.00	-.638	-.707	-.754	-.907	-.898	-.858	-.728	-.778	-.898	-.971	-.929	-.910
Lower surface	1.25	1.037	.971	.868	.786	.710	.645	1.020	.973	.862	.767	.685	.629
	2.50	1.114	.963	.885	.822	.776	.655	1.119	.987	.896	.822	.774	.648
	5.00	1.038	.909	.850	.801	.769	.658	1.065	.944	.879	.821	.787	.665
	7.50	.954	.854	.808	.768	.742	.631	.990	.895	.844	.792	.829	.648
	10.00	.888	.814	.771	.738	.708	.600	.929	.858	.814	.766	.770	.621
	15.00	.798	.742	.710	.683	.653	.532	.839	.790	.754	.714	.689	.561
	20.00	.729	.683	.657	.628	.609	.469	.773	.728	.701	.662	.645	.501
	25.00	.676	.642	.614	.587	.567	.408	.723	.688	.656	.619	.605	.446
	30.00	.633	.601	.571	.550	.525	.348	.678	.647	.612	.584	.564	.384
	35.00	.551	.556	.532	.506	.483	.310	.603	.602	.572	.541	.522	.351
	40.00	.543	.521	.497	.470	.453	.255	.588	.566	.538	.504	.493	.291
	45.00	.519	.485	.460	.431	.415	.192	.562	.525	.500	.465	.456	.230
	50.00	.454	.444	.425	.395	.381	.161	.500	.485	.462	.428	.419	.198
	55.00	.413	.409	.396	.369	.334	.130	.457	.451	.432	.406	.374	.164
	60.00	.404	.377	.360	.335	.296	.103	.445	.416	.395	.370	.337	.136
	65.00	.357	.342	.332	.302	.262	.078	.398	.385	.368	.330	.302	.109
	70.00	.307	.323	.307	.284	.251	.049	.345	.361	.342	.313	.288	.079
	75.00	.289	.289	.276	.256	.217	.032	.321	.319	.308	.281	.251	.060
	80.00	.254	.264	.249	.227	.189	.009	.293	.293	.279	.252	.224	.036
	85.00	.212	.232	.228	.206	.163	.021	.242	.266	.253	.227	.191	.042
	90.00	.179	.201	.194	.172	.133	-.029	.198	.228	.209	.187	.156	-.010
	95.00	.117	.141	.139	.106	.082	-.032	.128	.159	.155	.116	.100	-.020

TABLE I.- STEEL WING PRESSURE COEFFICIENT DATA FOR THE TEST RANGE OF ANGLE OF ATTACK AND MACH NUMBER - Continued

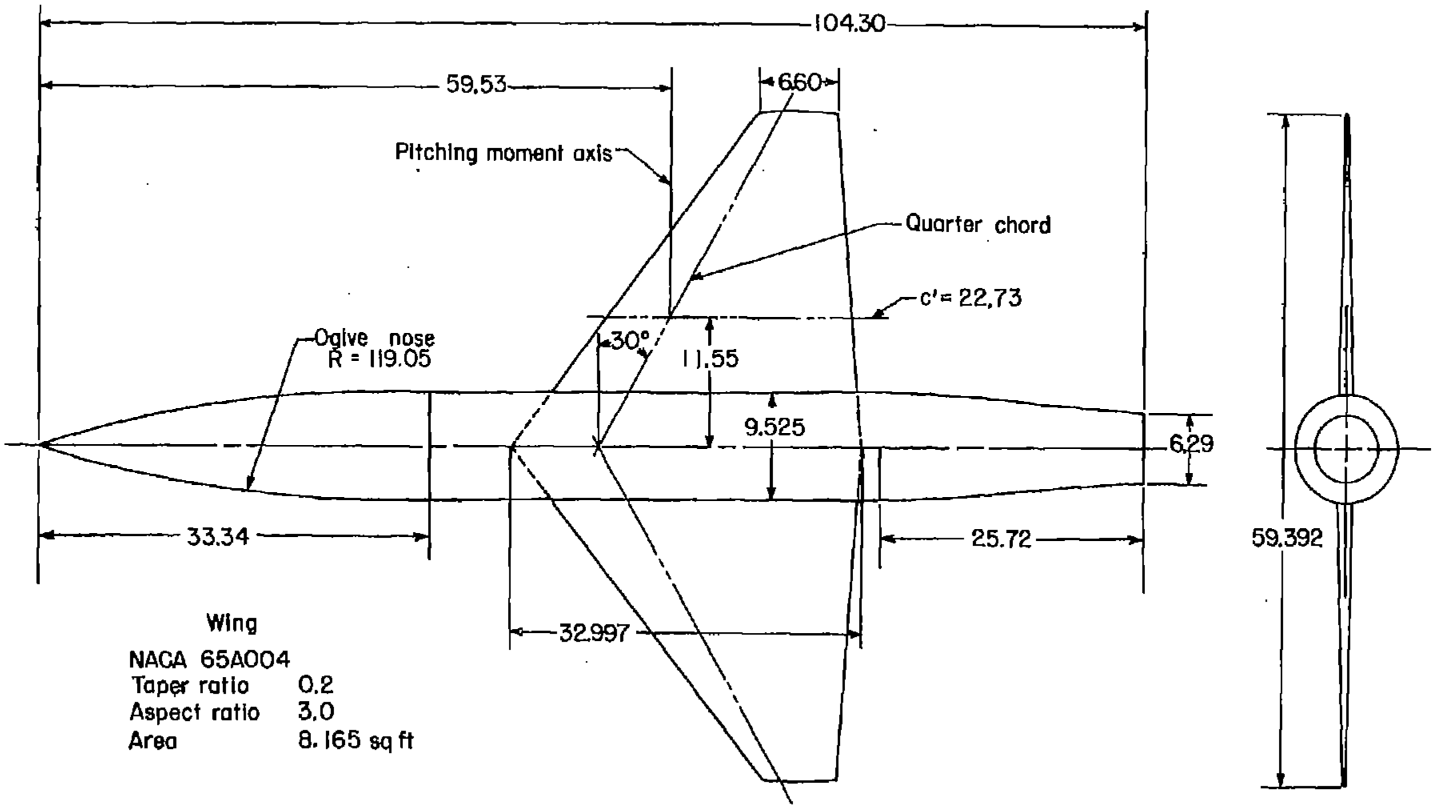
	Percent c	Pressure coefficient, P, at:											
		0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2	0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2
		M = 1.03	α = -2.02°					M = 1.03	α = -0.06°				
Upper surface	0.00	-.019	.737	.695	.667	.616	.560	-.001	.758	.722	.725	.694	.739
	1.25	.312	.245	.281	.303	.296	.357	.222	.065	.088	.064	.039	.147
	2.50	.262	.173	.190	.204	.220	.234	.164	.025	.029	.013	-.001	-.043
	5.00	.212	.148	.143	.147	.146	.195	.108	.025	.034	.005	-.026	-.051
	7.50	.185	.120	.107	.103	.102	.153	.088	.025	.012	-.016	-.052	-.059
	10.00	.158	.095	.078	.077	.072	.112	.067	.015	-.005	-.038	-.053	-.078
	15.00	.121	.066	.046	.045	.039	.044	.050	.000	-.030	-.044	-.076	-.100
	20.00	.056	.037	.015	.013	.011	-.023	.007	-.027	-.059	-.068	-.095	-.114
	25.00	.055	.027	.003	-.007	-.009	-.122	-.008	-.042	-.057	-.085	-.111	-.195
	30.00	.023	.011	-.012	-.025	-.023	-.153	-.031	-.051	-.073	-.100	-.121	-.201
	35.00	-.001	-.022	-.032	-.036	-.051	-.176	-.055	-.076	-.094	-.116	-.145	-.201
	40.00	-.039	-.033	-.049	-.063	-.067	-.239	-.086	-.098	-.117	-.146	-.163	-.251
	45.00	-.016	-.033	-.058	-.080	-.095	-.239	-.086	-.109	-.132	-.158	-.185	-.239
	50.00	-.027	-.042	-.081	-.098	-.127	-.269	-.065	-.097	-.137	-.177	-.210	-.265
	55.00	-.042	-.082	-.109	-.127	-.150	-.279	-.089	-.124	-.162	-.189	-.231	-.276
	60.00	-.088	-.108	-.125	-.144	-.174	-.263	-.132	-.159	-.180	-.202	-.246	-.264
	65.00	-.090	-.103	-.128	-.158	-.178	-.285	-.138	-.155	-.187	-.216	-.237	-.288
	70.00	-.092	-.109	-.129	-.159	-.212	-.288	-.144	-.161	-.188	-.222	-.255	-.296
	75.00	-.098	-.116	-.140	-.159	-.225	-.273	-.152	-.168	-.194	-.227	-.267	-.285
	80.00	-.129	-.128	-.137	-.160	-.236	-.263	-.180	-.184	-.195	-.224	-.280	-.269
	85.00	-.138	-.116	-.134	-.162	-.230	-.243	-.201	-.182	-.195	-.219	-.263	-.247
	90.00	-.105	-.109	-.118	-.161	-.209	-.227	-.165	-.179	-.185	-.210	-.242	-.230
	95.00	-.077	-.107	-.112	-.152	-.220	-.236	-.130	-.174	-.180	-.201	-.249	-.239
Lower surface	1.25	.086	-.180	-.428	-.587	-.741	-.787	.192	.067	.017	-.008	-.027	-.080
	2.50	.019	-.156	-.189	-.492	-.618	-.722	.131	.012	.002	-.031	-.060	-.073
	5.00	-.007	-.093	-.147	-.183	-.454	-.646	.098	.033	.008	-.014	-.042	-.082
	7.50	-.022	-.101	-.158	-.205	-.263	-.593	.074	.007	-.017	-.038	-.052	-.097
	10.00	-.050	-.111	-.156	-.223	-.250	-.540	.044	-.009	-.036	-.056	-.063	-.096
	15.00	-.066	-.107	-.150	-.210	-.239	-.419	.024	-.009	-.046	-.063	-.078	-.109
	20.00	-.083	-.112	-.169	-.234	-.269	-.319	-.001	-.040	-.061	-.096	-.107	-.139
	25.00	-.105	-.111	-.149	-.198	-.243	-.306	-.013	-.030	-.057	-.091	-.111	-.184
	30.00	-.085	-.115	-.152	-.196	-.233	-.272	-.003	-.040	-.070	-.101	-.120	-.201
	35.00	-.100	-.141	-.172	-.214	-.247	-.213	-.036	-.067	-.095	-.126	-.144	-.172
	40.00	-.147	-.169	-.187	-.233	-.234	-.237	-.073	-.095	-.114	-.145	-.157	-.213
	45.00	-.119	-.157	-.201	-.254	-.274	-.268	-.046	-.084	-.122	-.160	-.178	-.241
	50.00	-.158	-.185	-.218	-.275	-.298	-.286	-.093	-.114	-.149	-.182	-.193	-.263
	55.00	-.184	-.206	-.232	-.281	-.328	-.307	-.114	-.131	-.167	-.196	-.223	-.279
	60.00	-.209	-.226	-.256	-.295	-.343	-.319	-.144	-.161	-.182	-.219	-.242	-.283
	65.00	-.220	-.239	-.260	-.315	-.357	-.323	-.149	-.169	-.191	-.256	-.262	-.283
	70.00	-.220	-.224	-.263	-.303	-.339	-.330	-.149	-.157	-.195	-.228	-.254	-.284
	75.00	-.268	-.252	-.269	-.309	-.352	-.317	-.200	-.182	-.197	-.237	-.272	-.269
	80.00	-.268	-.266	-.273	-.316	-.342	-.328	-.203	-.199	-.203	-.246	-.287	-.278
	85.00	-.279	-.266	-.275	-.311	-.328	-.271	-.213	-.201	-.206	-.237	-.289	-.221
	90.00	-.262	-.269	-.272	-.297	-.315	-.315	-.203	-.206	-.203	-.223	-.280	-.262
	95.00	-.166	-.229	-.248	-.277	-.282	-.282	-.151	-.192	-.198	-.229	-.258	-.225
		M = 1.03	α = 1.96°					M = 1.03	α = 3.88°				
Upper surface	0.00	-.009	.722	.645	.666	.654	.727	-.025	.652	.540	.523	.514	.692
	1.25	.119	-.108	-.256	-.473	-.555	-.459	.036	-.685	-.779	-.841	-.877	-.685
	2.50	.058	-.158	-.184	-.329	-.476	-.628	-.032	-.421	-.692	-.764	-.806	-.901
	5.00	-.008	-.125	-.138	-.190	-.321	-.575	-.142	-.314	-.607	-.693	-.750	-.853
	7.50	-.025	-.090	-.130	-.192	-.241	-.524	-.149	-.205	-.286	-.653	-.709	-.789
	10.00	-.041	-.084	-.136	-.190	-.234	-.490	-.155	-.192	-.226	-.627	-.681	-.761
	15.00	-.040	-.091	-.135	-.174	-.232	-.410	-.113	-.176	-.225	-.423	-.663	-.746
	20.00	-.047	-.110	-.158	-.200	-.250	-.293	-.106	-.183	-.242	-.258	-.646	-.725
	25.00	-.088	-.108	-.150	-.195	-.239	-.293	-.160	-.183	-.237	-.270	-.554	-.711
	30.00	-.134	-.122	-.156	-.193	-.229	-.270	-.215	-.193	-.233	-.275	-.344	-.664
	35.00	-.121	-.147	-.173	-.203	-.245	-.234	-.185	-.214	-.244	-.277	-.316	-.601
	40.00	-.154	-.161	-.198	-.232	-.262	-.271	-.219	-.228	-.266	-.303	-.322	-.624
	45.00	-.164	-.174	-.209	-.243	-.279	-.260	-.229	-.238	-.278	-.309	-.337	-.587
	50.00	-.138	-.169	-.219	-.258	-.300	-.287	-.201	-.233	-.288	-.327	-.359	-.590
	55.00	-.152	-.194	-.234	-.276	-.320	-.300	-.214	-.257	-.297	-.347	-.380	-.590
	60.00	-.198	-.223	-.248	-.282	-.336	-.285	-.260	-.278	-.311	-.352	-.399	-.547
	65.00	-.201	-.222	-.253	-.288	-.330	-.316	-.263	-.280	-.316	-.361	-.393	-.488
	70.00	-.211	-.221	-.252	-.290	-.341	-.325	-.272	-.281	-.313	-.358	-.409	-.396
	75.00	-.215	-.228	-.257	-.294	-.342	-.319	-.270	-.285	-.320	-.357	-.413	-.354
	80.00	-.245	-.243	-.257	-.291	-.336	-.308	-.298	-.299	-.316	-.356	-.414	-.331
	85.00	-.264	-.244	-.257	-.290	-.303	-.291	-.314	-.299	-.316	-.356	-.383	-.323
	90.00	-.241	-.243	-.248	-.275	-.277	-.280	-.293	-.295	-.302	-.344	-.325	-.317
	95.00	-.165	-.233	-.243	-.254	-.276	-.289	-.202	-.275	-.299	-.319	-.307	-.338
Lower surface	1.25	.291	.261	.262	.291	.332	.295	.420	.439	.440	.478	.526	.455
	2.50	.245	.173	.195	.223	.240	.236	.386	.339	.361	.394	.422	.390
	5.00	.208	.153	.155	.171	.172	.187	.347	.288	.296	.311	.330	.332
	7.50	.174	.117	.112	.124	.141	.143	.306	.237	.246	.259	.279	.270
	10.00	.140	.092	.082	.098	.108	.110	.263	.207	.210	.224	.231	.222
	15.00	.112	.074	.062	.058	.061	.052	.222	.184	.181	.187	.181	.145
	20.00	.081	.042	.038	.018	.027	-.019	.180	.183	.146	.153	.145	.065
	25.00	.066	.045	.025	.004	.004	-.088	.162	.147	.121	.109	.112	-.006
	30.00	.066	.023	.006	.003	-.012	-.141	.153	.116	.102	.090	.082	-.062
	35.00	.008	-.010	-.012	-.026	-.043	-.134	.080	.089	.075	.056	.053	-.092
	40.00	-.004	-.010	-.012	-.047	-.057	-.192	.090	.076	.056	.034	.031	-.130
	45.00	.018	-.018	-.046	-.081	-.086	-.235	.096	.050	.025	-.001	-.002	-.184
	50.00	-.037	-.053	-.075	-.110	-.117	-.257	.028	.019	-.003	-.023	-.027	-.213
	55.00	-.065	-.075	-.090	-.122	-.147	-.277	-.003	-.008	-.017	-.036	-.071	-.239
	60.00	-.080	-.097	-.117	-.143	-.168	-.279	.002	-.022	-.034	-.053	-.091	-.255
	65.00	-.090	-.108	-.125	-.163	-.192	-.278	-.015	-.031	-.038	-.072	-.120	-.266
	70.00	-.083	-.097	-.123	-.155	-.193	-.279	-.028	-.032	-.044	-.071	-.122	-.274
	75.00	-.135	-.122	-.131	-.162	-.217	-.266	-.042	-.052	-.062	-.092	-.148	-.271
	80.00	-.138	-.135	-.140	-.171	-.233	-.269	-.051	-.058	-.070	-.105	-.170	-.279
	85.00	-.151	-.134	-.137	-.166	-.245	-.213	-.066	-.059	-.067	-.109	-.183	-.228
	90.00	-.133	-.134	-.131	-.163	-.242	-.251	-.058	-.065	-.066	-.106	-.178	-.271
	95.00	-.089	-.115	-.125	-.175	-.229	-.213	-.049	-.061	-.066	-.116	-.172	-.239

TABLE I.- STEEL WING PRESSURE COEFFICIENT DATA FOR THE TEST RANGE OF ANGLE OF ATTACK AND MACH NUMBER - Continued

	Percent c	Pressure coefficient, P, at: 0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2	0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2
		M = 1.03		α = 5.95°				M = 1.03		α = 7.92°			
Upper surface	0.00	-.093	.495	.399	.342	.290	.601	-.058	.386	.236	.128	.031	.448
	1.25	-.100	-.948	-.966	-1.008	-1.036	-.834	-.136	-1.030	-1.081	-1.113	-1.123	-.963
	2.50	-.182	-.881	-.873	-.932	-.976	-1.049	-.222	-.946	-.994	-1.051	-1.072	-1.134
	5.00	-.313	-.625	-.808	-.855	-.908	-.993	-.367	-.716	-.932	-.977	-1.012	-1.079
	7.50	-.314	-.352	-.746	-.813	-.882	-.927	-.357	-.519	-.871	-.933	-.966	-1.031
	10.00	-.315	-.338	-.716	-.791	-.838	-.900	-.347	-.386	-.846	-.906	-.942	-1.005
	15.00	-.241	-.314	-.350	-.730	-.814	-.874	-.253	-.338	-.777	-.852	-.916	-.977
	20.00	-.223	-.304	-.296	-.697	-.782	-.851	-.217	-.310	-.488	-.818	-.885	-.953
	25.00	-.279	-.303	-.290	-.673	-.747	-.845	-.265	-.299	-.401	-.799	-.847	-.935
	30.00	-.336	-.306	-.287	-.453	-.728	-.799	-.318	-.299	-.355	-.767	-.832	-.900
	35.00	-.296	-.334	-.294	-.367	-.721	-.723	-.277	-.316	-.346	-.734	-.824	-.823
	40.00	-.331	-.344	-.310	-.349	-.714	-.744	-.315	-.329	-.357	-.565	-.815	-.838
	45.00	-.338	-.347	-.325	-.349	-.704	-.709	-.314	-.328	-.370	-.509	-.815	-.805
	50.00	-.311	-.348	-.331	-.361	-.578	-.713	-.290	-.330	-.380	-.487	-.809	-.813
	55.00	-.323	-.369	-.342	-.380	-.497	-.716	-.303	-.345	-.389	-.473	-.807	-.814
	60.00	-.370	-.394	-.354	-.387	-.464	-.705	-.346	-.361	-.400	-.460	-.803	-.807
	65.00	-.371	-.389	-.352	-.399	-.444	-.714	-.345	-.366	-.399	-.456	-.706	-.811
	70.00	-.376	-.385	-.354	-.397	-.445	-.714	-.345	-.364	-.399	-.453	-.625	-.819
	75.00	-.373	-.391	-.361	-.397	-.445	-.716	-.342	-.367	-.405	-.452	-.587	-.823
	80.00	-.401	-.402	-.356	-.395	-.449	-.712	-.366	-.375	-.401	-.451	-.566	-.821
	85.00	-.414	-.403	-.353	-.395	-.435	-.701	-.377	-.375	-.400	-.451	-.513	-.815
	90.00	-.385	-.398	-.340	-.386	-.381	-.704	-.355	-.375	-.385	-.443	-.425	-.822
	95.00	-.271	-.369	-.334	-.367	-.324	-.710	-.256	-.364	-.383	-.426	-.333	-.833
Lower surface	1.25	.535	.562	.593	.619	.642	.555	.705	.714	.687	.693	.703	.611
	2.50	.518	.451	.511	.532	.545	.490	.696	.614	.613	.618	.621	.548
	5.00	.472	.383	.435	.440	.453	.433	.638	.535	.528	.525	.532	.489
	7.50	.413	.327	.377	.384	.395	.363	.575	.476	.471	.466	.475	.427
	10.00	.361	.298	.345	.347	.345	.316	.511	.442	.436	.426	.421	.375
	15.00	.300	.249	.294	.297	.286	.225	.441	.381	.373	.368	.359	.284
	20.00	.250	.229	.249	.236	.240	.145	.383	.360	.325	.306	.309	.204
	25.00	.216	.187	.221	.205	.203	.078	.352	.313	.291	.270	.270	.141
	30.00	.195	.152	.194	.180	.172	.016	.317	.276	.260	.242	.235	.074
	35.00	.109	.124	.159	.144	.133	.010	.227	.245	.225	.203	.193	.059
	40.00	.121	.105	.139	.119	.110	-.066	.243	.221	.199	.172	.171	-.015
	45.00	.124	.072	.106	.087	.076	-.124	.235	.186	.166	.136	.131	-.075
	50.00	.057	.037	.080	.057	.044	-.153	.170	.149	.135	.109	.100	-.105
	55.00	.021	.012	.063	.041	.004	-.182	.136	.127	.118	.091	.061	-.136
	60.00	.025	-.006	.041	.018	-.022	-.197	.138	.106	.091	.066	.028	-.154
	65.00	-.004	-.023	.026	-.002	-.051	-.215	.106	.085	.078	.043	-.003	-.186
	70.00	-.025	-.027	.021	-.006	-.052	-.230	.075	.079	.064	.032	-.009	-.190
	75.00	-.039	-.056	-.004	-.034	-.089	-.228	.065	.052	.041	.004	-.042	-.190
	80.00	-.054	-.063	-.013	-.050	-.109	-.238	.048	.041	.031	-.004	-.056	-.208
	85.00	-.075	-.067	-.013	-.050	-.120	-.192	.028	.036	.031	-.004	-.068	-.165
	90.00	-.082	-.073	-.014	-.048	-.120	-.238	.022	.030	.031	-.002	-.061	-.212
	95.00	-.066	-.081	-.019	-.062	-.111	-.210	.024	.017	.018	-.022	-.057	-.182
		M = 1.03		c = 9.89°				M = 1.03		α = 13.68°			
Upper surface	0.00	-.091	.206	.044	-.108	-.270	.239	-.121	-.159	-.406	-.598	-.760	-.290
	1.25	-.260	-1.136	-1.163	-1.165	-1.183	-1.074	-.483	-1.133	-1.087	-1.046	-1.065	-1.102
	2.50	-.358	-1.068	-1.084	-1.136	-1.155	-1.198	-.594	-1.092	-1.073	-1.052	-1.044	-1.120
	5.00	-.513	-.888	-1.036	-1.072	-1.109	-1.161	-.721	-1.053	-1.096	-1.036	-1.048	-1.114
	7.50	-.496	-.696	-.974	-1.028	-1.058	-1.128	-.680	-.996	-1.044	-1.040	-1.043	-1.101
	10.00	-.479	-.611	-.958	-1.003	-1.038	-1.101	-.640	-.951	-1.021	-1.038	-1.008	-1.101
	15.00	-.359	-.518	-.898	-.946	-1.012	-1.068	-.496	-.880	-.963	-1.008	-1.024	-1.091
	20.00	-.302	-.463	-.703	-.911	-.982	-1.046	-.421	-.805	-.907	-.962	-1.020	-1.080
	25.00	-.332	-.426	-.578	-.896	-.943	-1.028	-.415	-.537	-.860	-.958	-.997	-1.056
	30.00	-.373	-.387	-.532	-.872	-.929	-1.001	-.438	-.344	-.828	-.922	-.978	-1.028
	35.00	-.333	-.371	-.478	-.857	-.921	-.922	-.412	-.395	-.810	-.902	-.960	-.975
	40.00	-.364	-.371	-.430	-.808	-.912	-.924	-.427	-.420	-.794	-.882	-.941	-.991
	45.00	-.369	-.371	-.414	-.661	-.912	-.899	-.435	-.435	-.774	-.868	-.934	-.959
	50.00	-.355	-.373	-.419	-.626	-.909	-.909	-.444	-.447	-.748	-.858	-.918	-.950
	55.00	-.366	-.391	-.431	-.620	-.909	-.908	-.456	-.465	-.715	-.846	-.902	-.944
	60.00	-.403	-.419	-.442	-.603	-.910	-.903	-.483	-.494	-.672	-.827	-.883	-.931
	65.00	-.407	-.417	-.440	-.590	-.897	-.908	-.488	-.500	-.607	-.812	-.839	-.903
	70.00	-.407	-.418	-.445	-.577	-.861	-.914	-.498	-.504	-.543	-.803	-.798	-.089
	75.00	-.399	-.424	-.454	-.557	-.745	-.922	-.487	-.508	-.508	-.808	-.752	-.895
	80.00	-.424	-.437	-.454	-.540	-.644	-.922	-.521	-.525	-.500	-.793	-.724	-.885
	85.00	-.441	-.438	-.454	-.527	-.559	-.922	-.530	-.530	-.505	-.777	-.666	-.877
	90.00	-.424	-.438	-.443	-.505	-.507	-.927	-.518	-.530	-.504	-.657	-.618	-.907
	95.00	-.325	-.426	-.443	-.472	-.467	-.946	-.443	-.499	-.510	-.548	-.597	-.908
Lower surface	1.25	.809	.803	.757	.749	.741	.646	.958	.914	.844	.807	.771	.685
	2.50	.803	.716	.694	.689	.678	.595	.979	.850	.808	.777	.754	.657
	5.00	.742	.622	.611	.596	.594	.541	.902	.767	.735	.708	.693	.618
	7.50	.663	.562	.553	.542	.540	.481	.818	.704	.677	.657	.646	.567
	10.00	.598	.526	.511	.499	.490	.427	.741	.663	.640	.614	.602	.520
	15.00	.516	.463	.448	.438	.427	.335	.657	.595	.570	.556	.541	.438
	20.00	.449	.429	.394	.376	.373	.260	.586	.557	.515	.492	.488	.366
	25.00	.410	.378	.359	.340	.333	.195	.538	.501	.475	.455	.445	.301
	30.00	.376	.338	.323	.308	.292	.125	.496	.458	.435	.417	.405	.236
	35.00	.281	.303	.287	.268	.252	.112	.403	.416	.397	.378	.360	.204
	40.00	.292	.272	.258	.237	.227	.037	.409	.386	.365	.345	.331	.139
	45.00	.284	.241	.222	.195	.187	-.024	.393	.353	.329	.305	.294	.078
	50.00	.218	.203	.193	.170	.156	-.057	.327	.310	.299	.273	.258	.045
	55.00	.176	.178	.173	.151	.113	-.086	.294	.279	.275	.252	.217	.022
	60.00	.185	.158	.146	.123	.084	-.109	.291	.257	.240	.223	.181	-.004
	65.00	.147	.133	.128	.099	.048	-.141	.250	.227	.221	.192	.149	-.037
	70.00	.113	.126	.114	.086	.047	-.141	.205	.218	.204	.181	.144	-.044
	75.00	.102	.096	.090	.062	.014	-.149	.194	.186	.180	.157	.117	-.056
	80.00	.083	.085	.080	.055	.001	-.167	.173	.172	.166	.146	.103	-.069
	85.00	.062	.078	.080	.054	-.005	-.125	.140	.158	.155	.134	.086	-.041
	90.00	.050	.066	.068	.050	-.005	-.164	.120	.136	.134	.123	.075	-.079
	95.00	.041	.044	.049	.022	-.012	-.137	.089	.097	.104	.075	.046	-.065

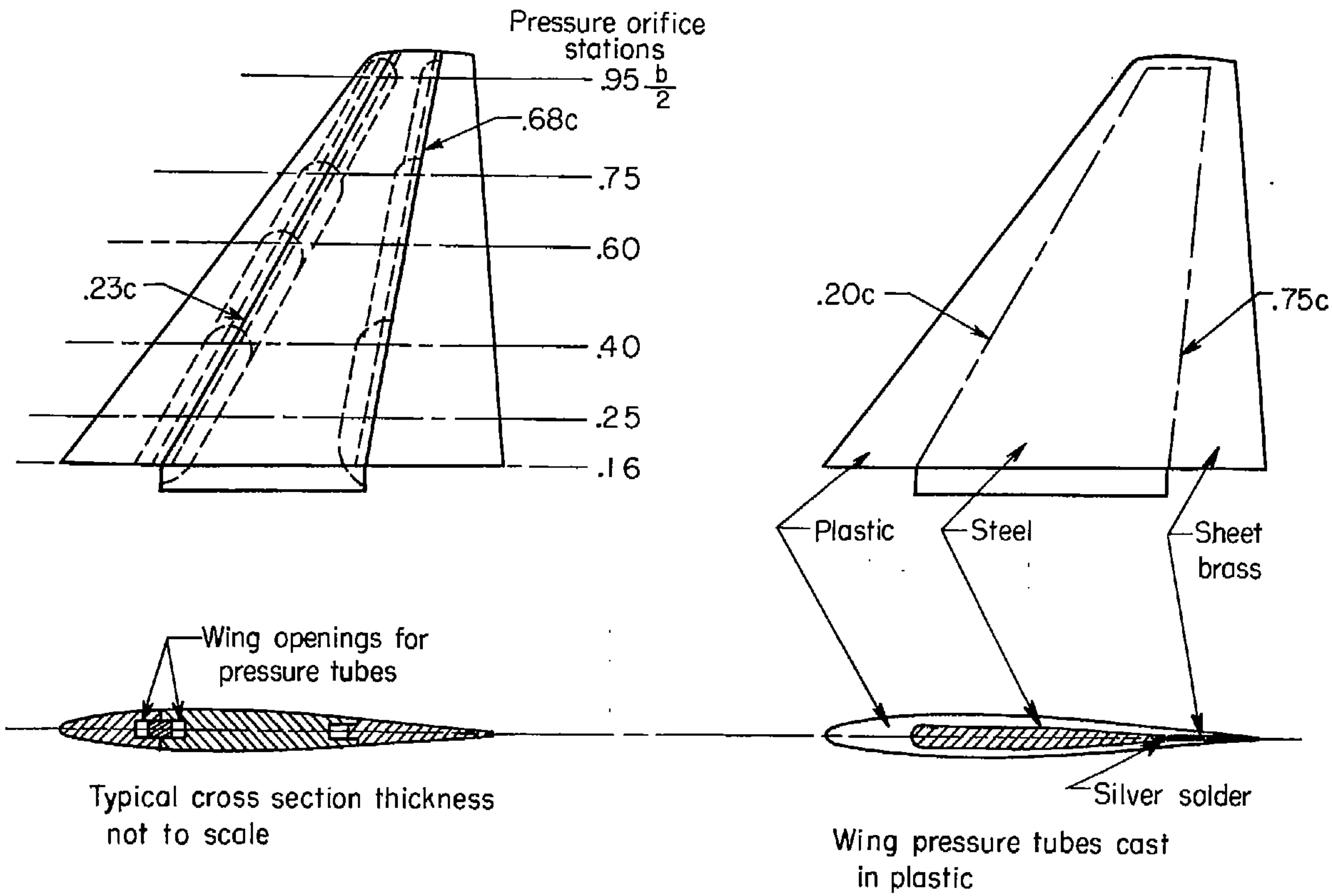
TABLE I.- STEEL WING PRESSURE COEFFICIENT DATA FOR THE TEST RANGE OF ANGLE OF ATTACK AND MACH NUMBER - Concluded

		Pressure coefficient, P, at:											
		0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2	0.16b/2	0.25b/2	0.40b/2	0.60b/2	0.75b/2	0.95b/2
	Percent c	M = 1.03		α = 15.85°				M = 1.03		α = 17.97°			
Upper surface	0.00	-.152	-.378	-.588	-.818	-.932	-.599	-.172	-.559	-.765	-.965	-1.062	-.830
	1.25	-.604	-1.188	-1.129	-1.072	-1.003	-.918	-.709	-1.245	-1.193	-1.047	-.973	-.870
	2.50	-.715	-1.179	-1.108	-1.076	-1.003	-.909	-.835	-1.233	-1.166	-1.042	-.973	-.866
	5.00	-.827	-1.144	-1.132	-1.068	-1.002	-.901	-.936	-1.213	-1.196	-1.037	-.979	-.861
	7.50	-.794	-1.107	-1.107	-1.072	-.990	-.894	-.952	-1.203	-1.162	-1.036	-.977	-.861
	10.00	-.761	-1.064	-1.103	-1.080	-.985	-.894	-.868	-1.176	-1.190	-1.033	-.989	-.860
	15.00	-.602	-1.016	-1.064	-1.073	-.992	-.894	-.733	-1.115	-1.158	-1.026	-1.002	-.859
	20.00	-.517	-.977	-1.023	-1.049	-.990	-.894	-.618	-1.073	-1.125	-1.018	-1.007	-.856
	25.00	-.493	-.743	-.984	-1.049	-.981	-.891	-.561	-1.021	-1.098	-1.015	-1.004	-.853
	30.00	-.506	-.419	-.956	-1.022	-.976	-.889	-.565	-.573	-1.074	-1.003	-.994	-.851
	35.00	-.476	-.454	-.939	-1.005	-.970	-.884	-.545	-.529	-1.061	-.995	-.984	-.850
	40.00	-.486	-.478	-.929	-.992	-.960	-.888	-.552	-.544	-1.048	-.983	-.977	-.849
	45.00	-.497	-.493	-.920	-.977	-.953	-.879	-.553	-.560	-1.039	-.965	-.971	-.844
	50.00	-.513	-.505	-.909	-.966	-.940	-.872	-.574	-.571	-1.035	-.950	-.959	-.841
	55.00	-.528	-.527	-.883	-.953	-.926	-.866	-.587	-.592	-1.025	-.935	-.948	-.837
	60.00	-.551	-.552	-.830	-.927	-.914	-.858	-.607	-.615	-.999	-.913	-.934	-.835
	65.00	-.555	-.561	-.721	-.910	-.896	-.847	-.613	-.620	-.911	-.899	-.914	-.828
	70.00	-.553	-.566	-.580	-.894	-.872	-.843	-.601	-.624	-.703	-.886	-.896	-.828
	75.00	-.542	-.573	-.517	-.873	-.855	-.843	-.594	-.632	-.509	-.866	-.880	-.828
	80.00	-.587	-.585	-.513	-.844	-.839	-.843	-.625	-.643	-.466	-.855	-.866	-.825
	85.00	-.593	-.589	-.526	-.812	-.804	-.843	-.641	-.621	-.517	-.843	-.839	-.824
	90.00	-.579	-.587	-.522	-.754	-.772	-.852	-.618	-.575	-.549	-.827	-.811	-.826
	95.00	-.507	-.509	-.518	-.696	-.773	-.851	-.431	-.484	-.580	-.813	-.820	-.826
Lower surface	1.25	1.013	.950	.876	.822	.769	.688	1.047	.974	.882	.818	.755	.682
	2.50	1.048	.908	.855	.815	.781	.674	1.103	.943	.882	.836	.792	.678
	5.00	.967	.830	.795	.761	.741	.650	1.019	.885	.832	.798	.768	.670
	7.50	.877	.767	.742	.713	.698	.609	.928	.823	.785	.754	.735	.635
	10.00	.804	.727	.706	.677	.655	.566	.861	.778	.746	.718	.699	.596
	15.00	.720	.660	.638	.613	.599	.487	.772	.710	.685	.657	.634	.525
	20.00	.645	.619	.581	.596	.543	.418	.697	.660	.626	.599	.584	.456
	25.00	.595	.559	.537	.515	.500	.395	.647	.617	.577	.559	.541	.396
	30.00	.557	.514	.497	.480	.463	.292	.604	.567	.543	.525	.501	.329
	35.00	.462	.475	.457	.437	.417	.259	.515	.526	.503	.483	.459	.299
	40.00	.465	.444	.425	.402	.388	.195	.515	.489	.469	.443	.429	.236
	45.00	.448	.404	.390	.363	.352	.136	.494	.455	.434	.407	.393	.176
	50.00	.380	.366	.354	.330	.318	.102	.431	.412	.398	.374	.358	.142
	55.00	.342	.334	.331	.308	.278	.074	.388	.382	.370	.352	.314	.114
	60.00	.339	.309	.297	.279	.242	.047	.382	.352	.337	.320	.277	.089
	65.00	.295	.280	.278	.247	.205	.022	.339	.319	.313	.286	.245	.062
	70.00	.250	.264	.255	.252	.200	.003	.289	.306	.291	.271	.236	.037
	75.00	.237	.233	.229	.210	.170	-.009	.271	.269	.264	.242	.204	.025
	80.00	.209	.214	.210	.191	.155	-.022	.246	.248	.241	.222	.179	.006
	85.00	.175	.197	.197	.179	.133	.004	.207	.229	.225	.205	.156	.023
	90.00	.151	.173	.170	.156	.115	-.041	.178	.199	.192	.178	.134	-.021
	95.00	.110	.126	.135	.105	.081	-.032	.125	.150	.150	.118	.092	-.019



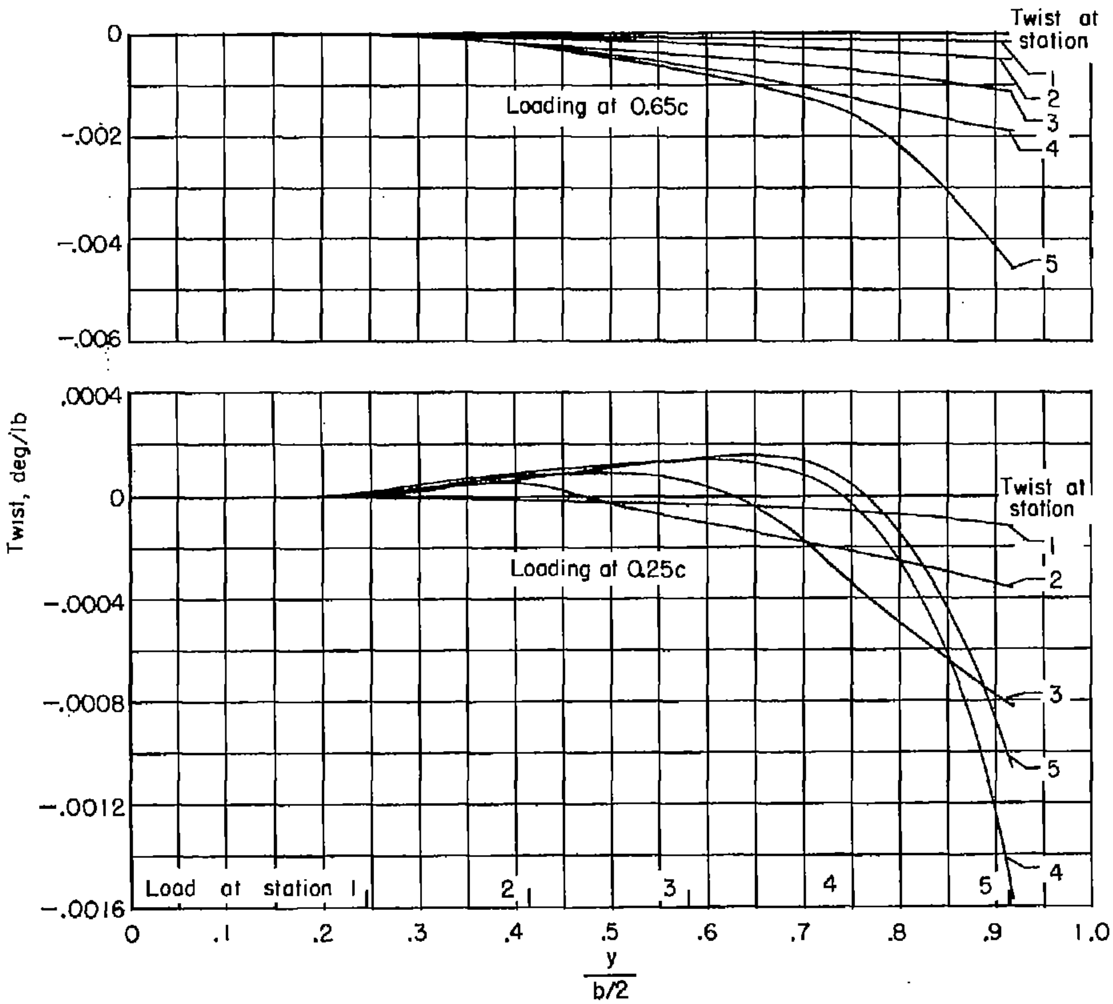
(a) Complete model.

Figure 1.- General model arrangement. All dimensions in inches.



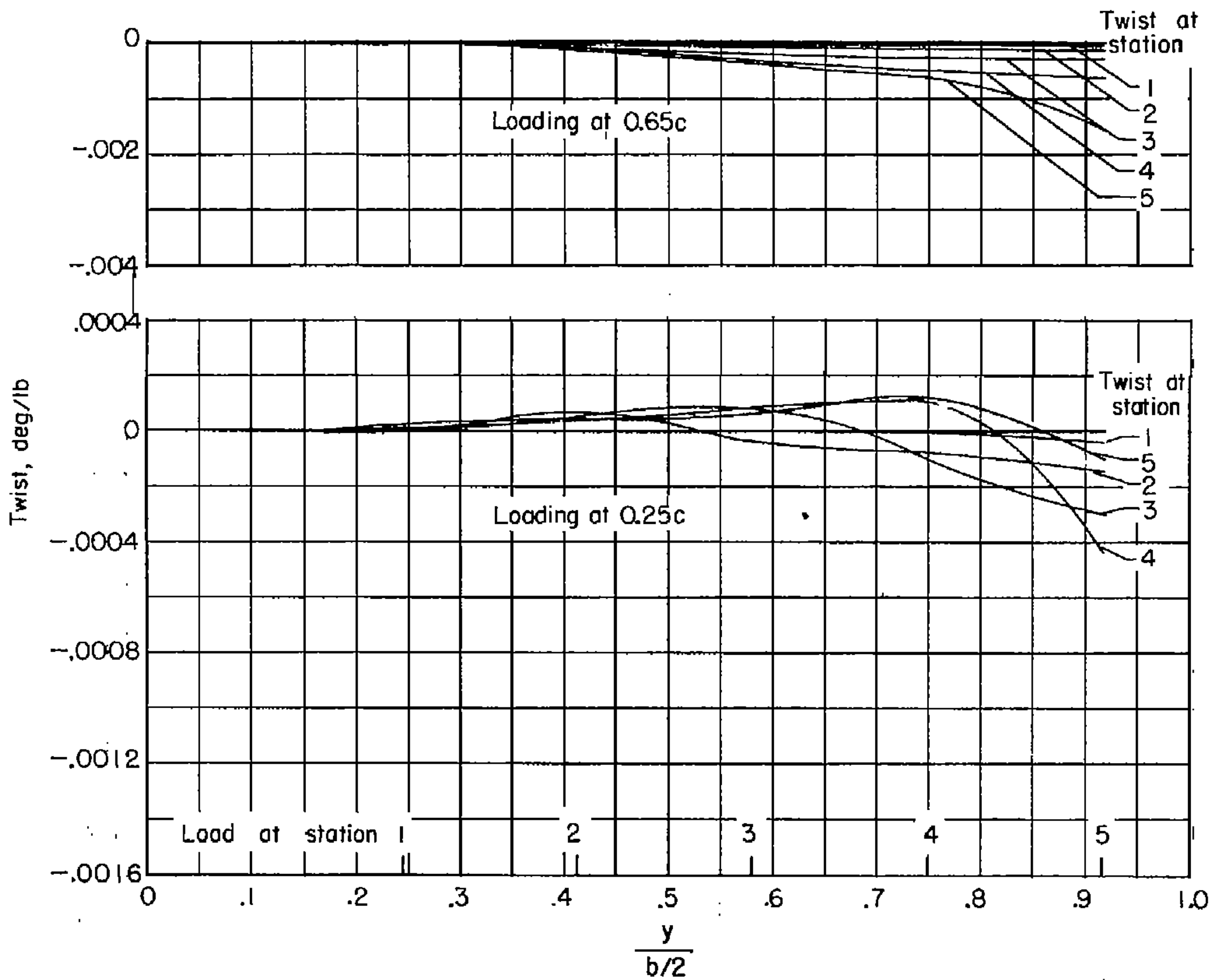
(b) Wings.

Figure 1.- Concluded.



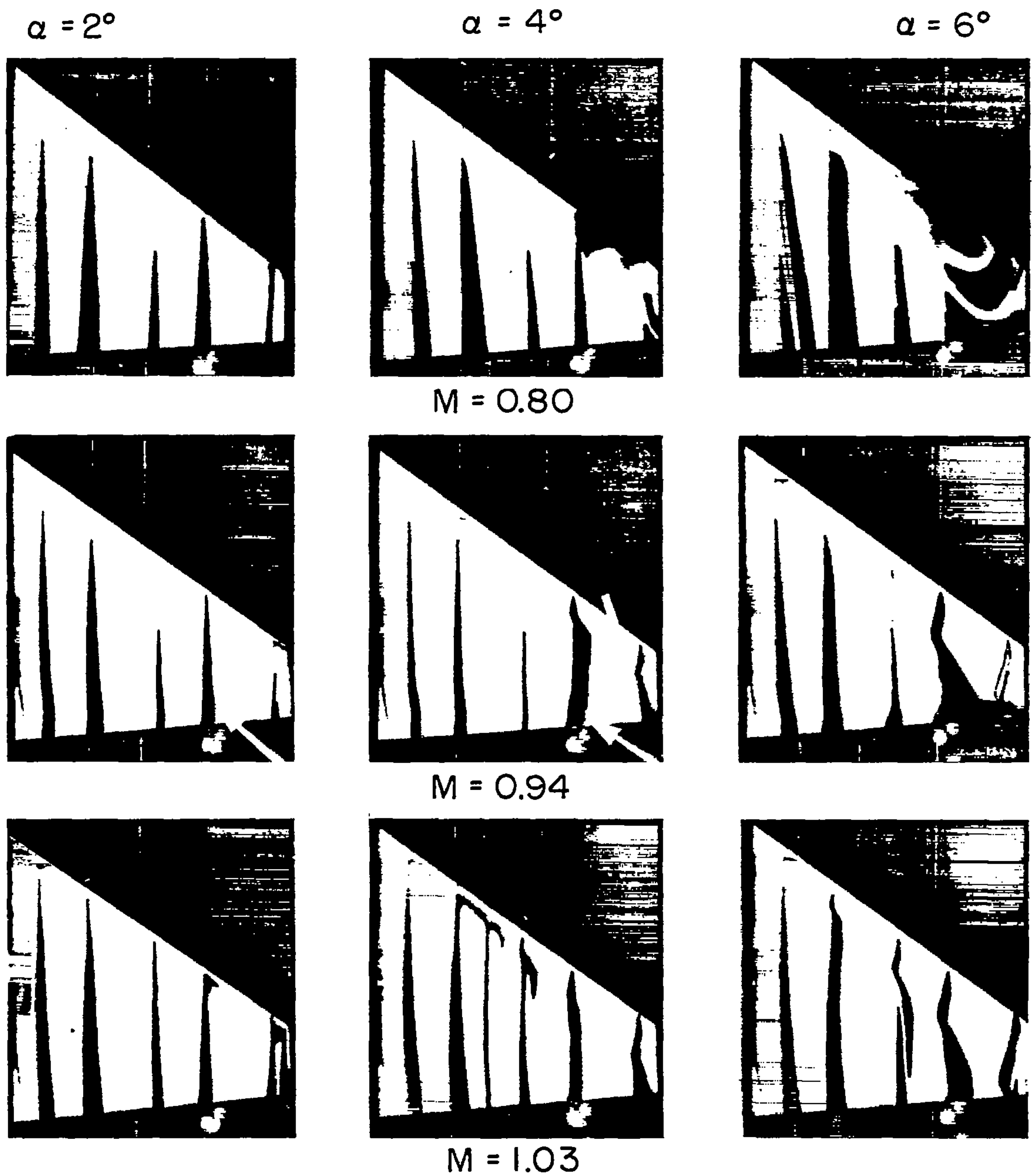
(a) Plastic wing.

Figure 2.- Wing elastic characteristics obtained experimentally, from which the influence coefficients were determined for twist in the angle-of-attack plane about 0.25c.



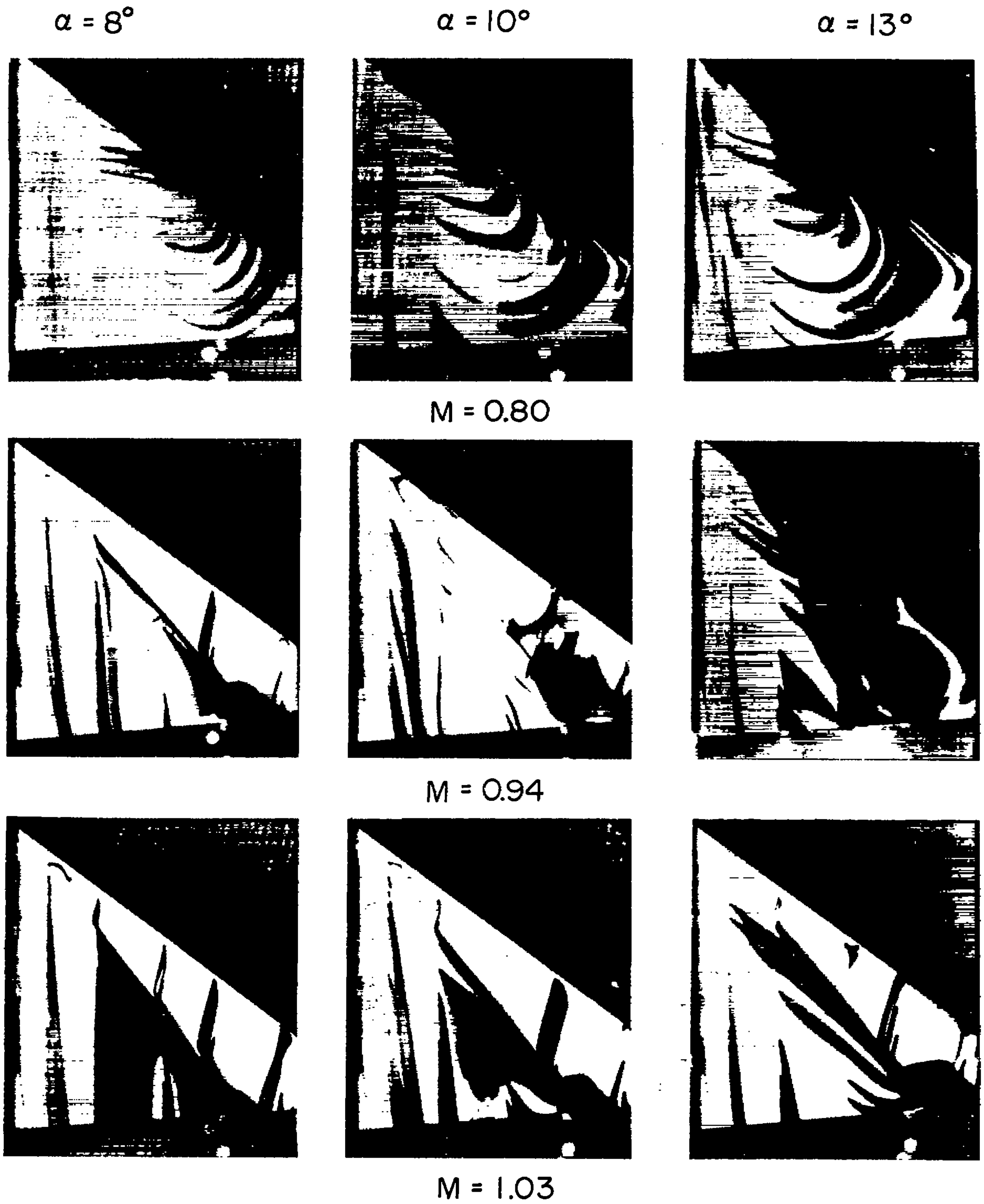
(b) Steel wing.

Figure 2.- Concluded.



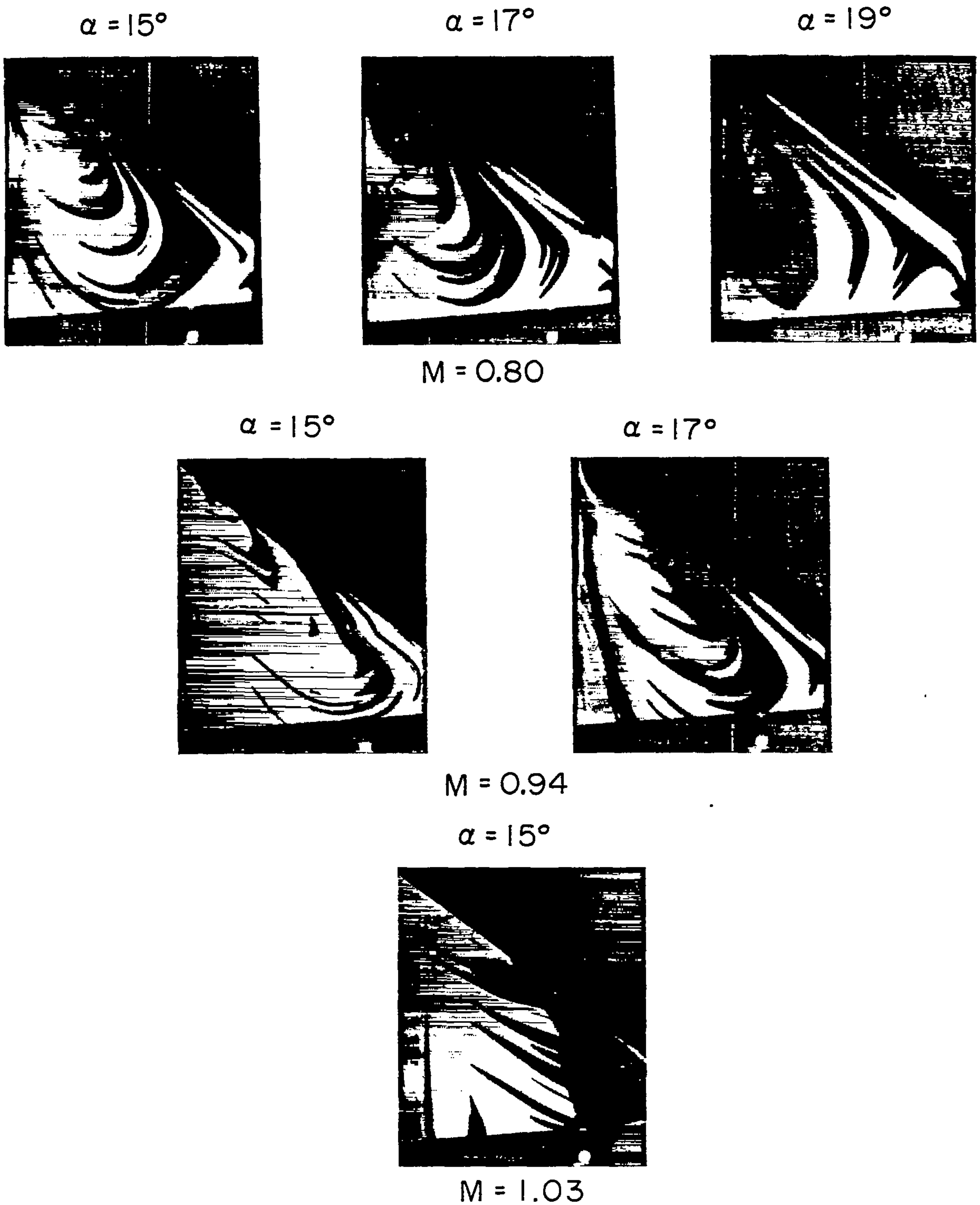
(a) $\alpha = 2^{\circ}$ to 6°. L-57-1639

Figure 3.- Typical flow study photographs for a range of Mach number and angle of attack, plastic wing.



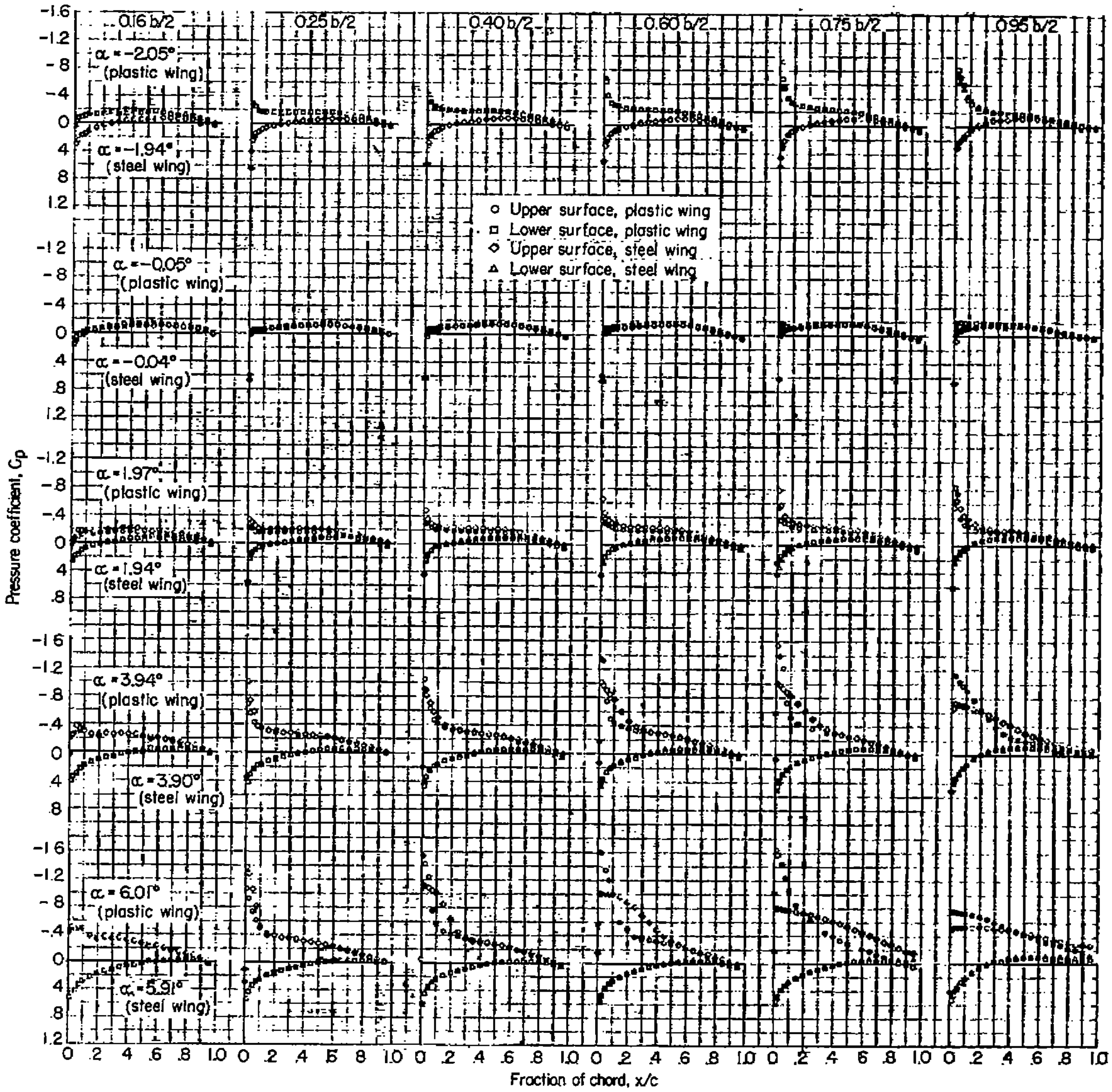
(b) $\alpha = 8^{\circ}$ to 13°. L-57-1640

Figure 3.- Continued.



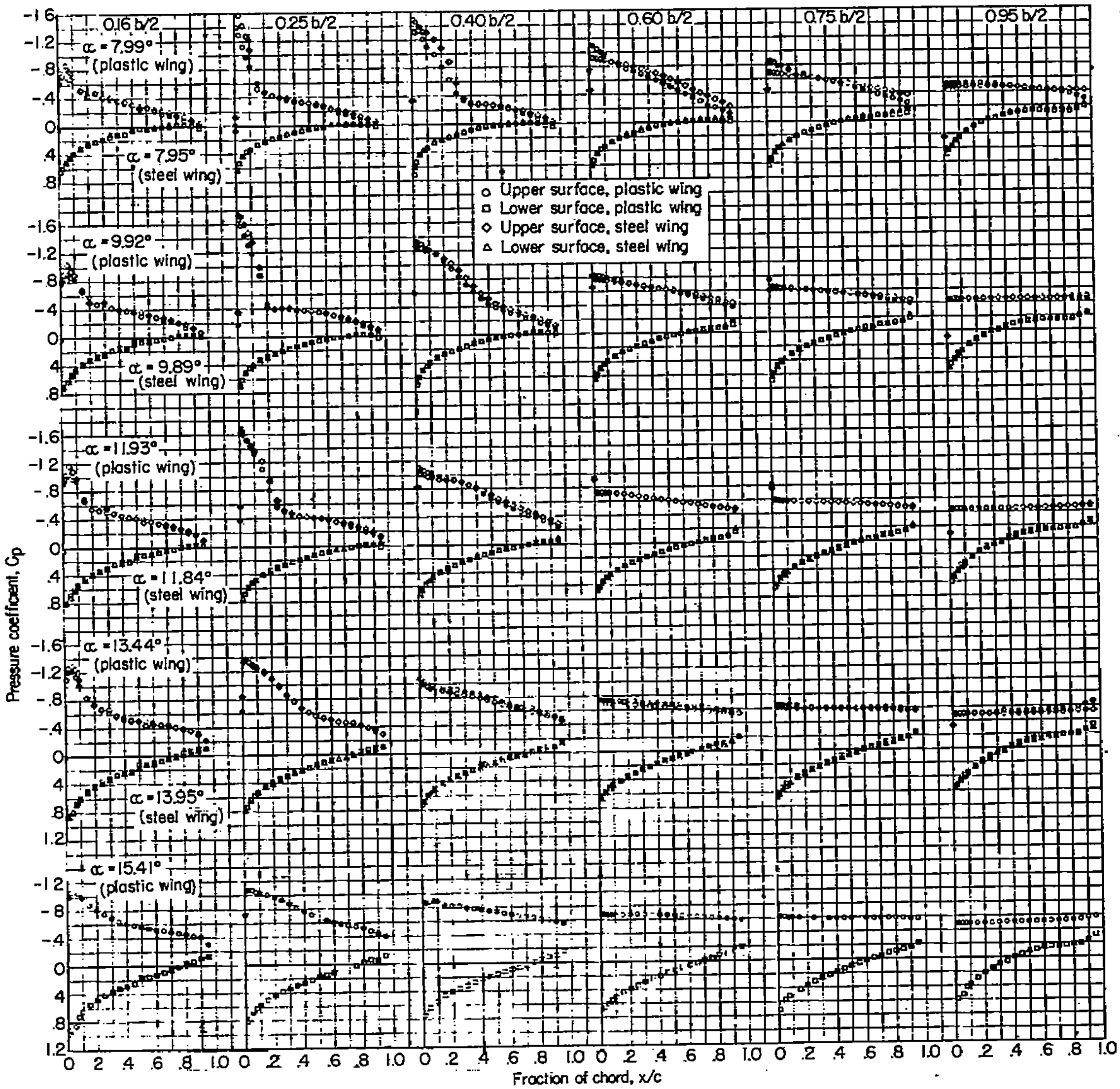
(c) α = 15° to 19°. L-57-1641

Figure 3.- Concluded.



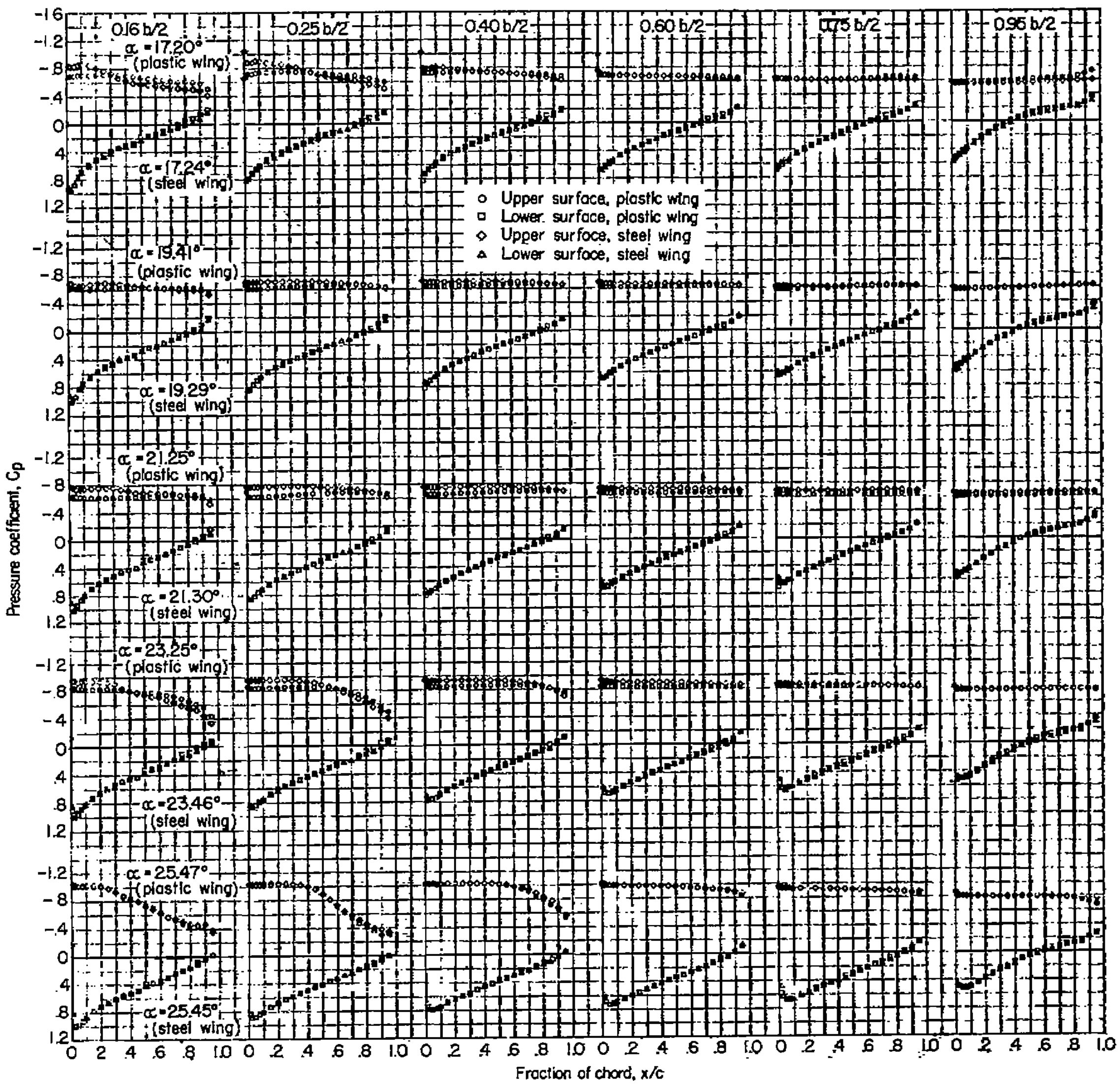
(a) M = 0.80.

Figure 4.- Comparison of chordwise pressure distributions for steel and plastic wings.



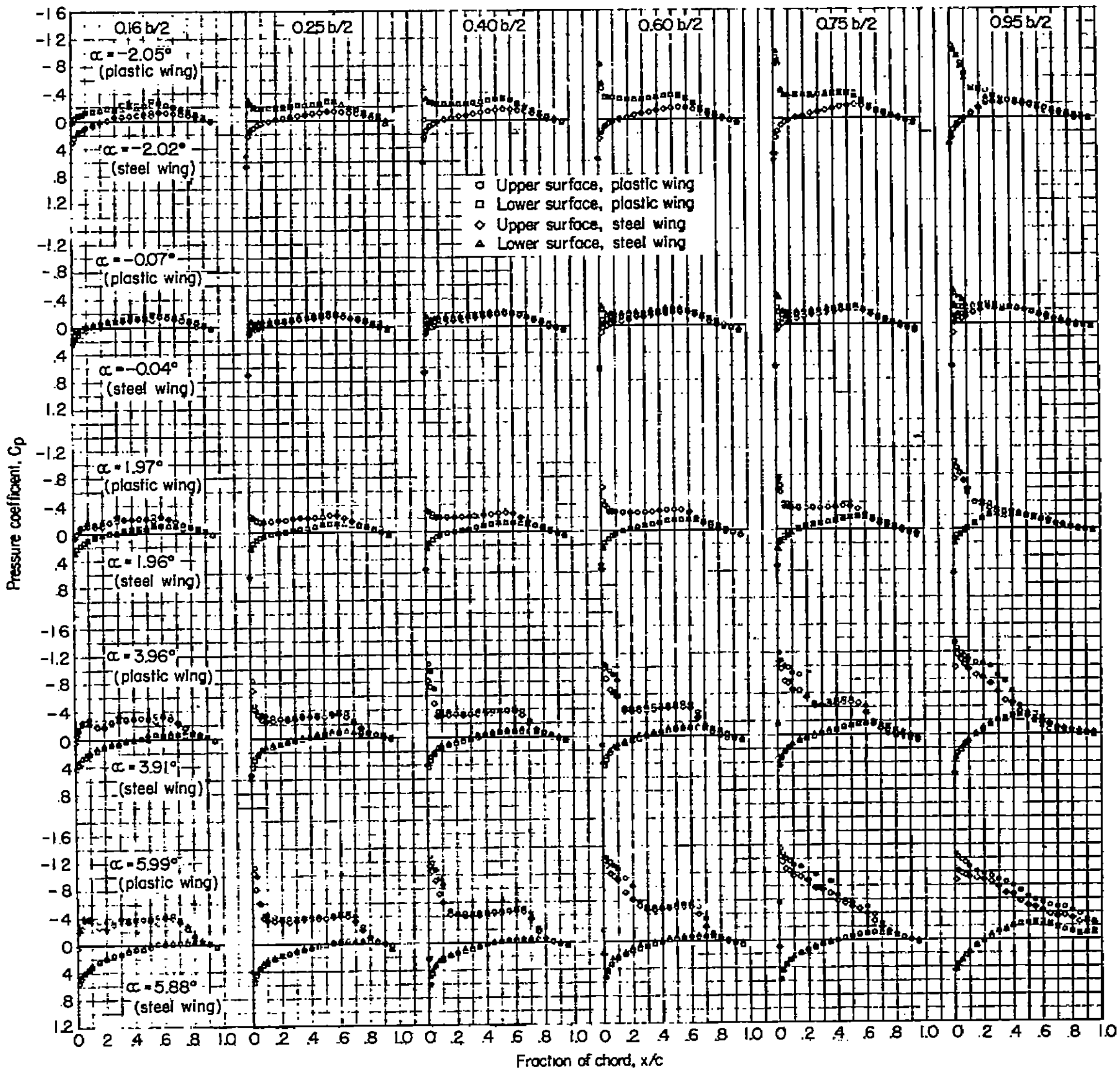
(a) M = 0.80, continued.

Figure 4.- Continued.



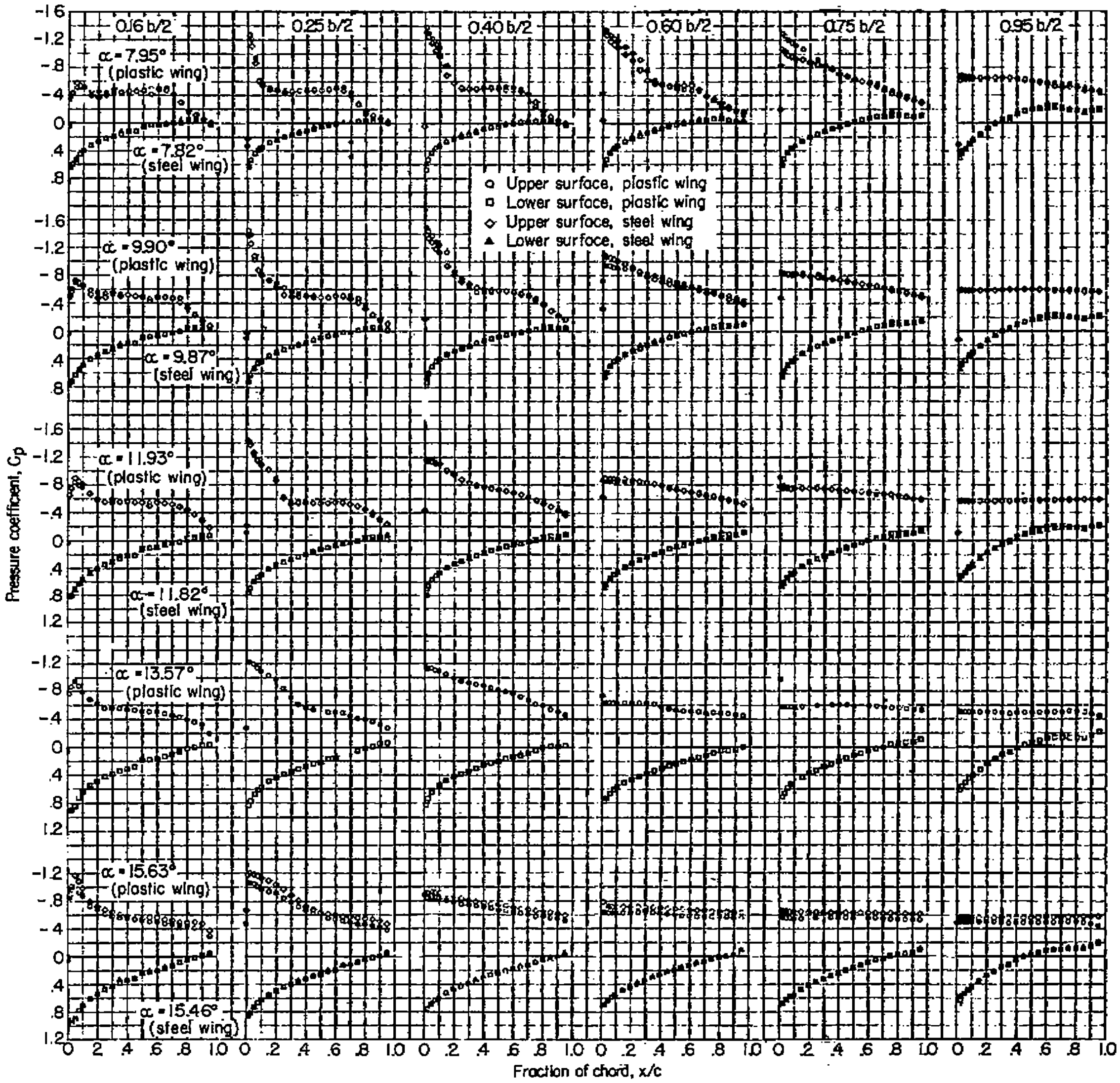
(a) M = 0.80, concluded.

Figure 4.- Continued.



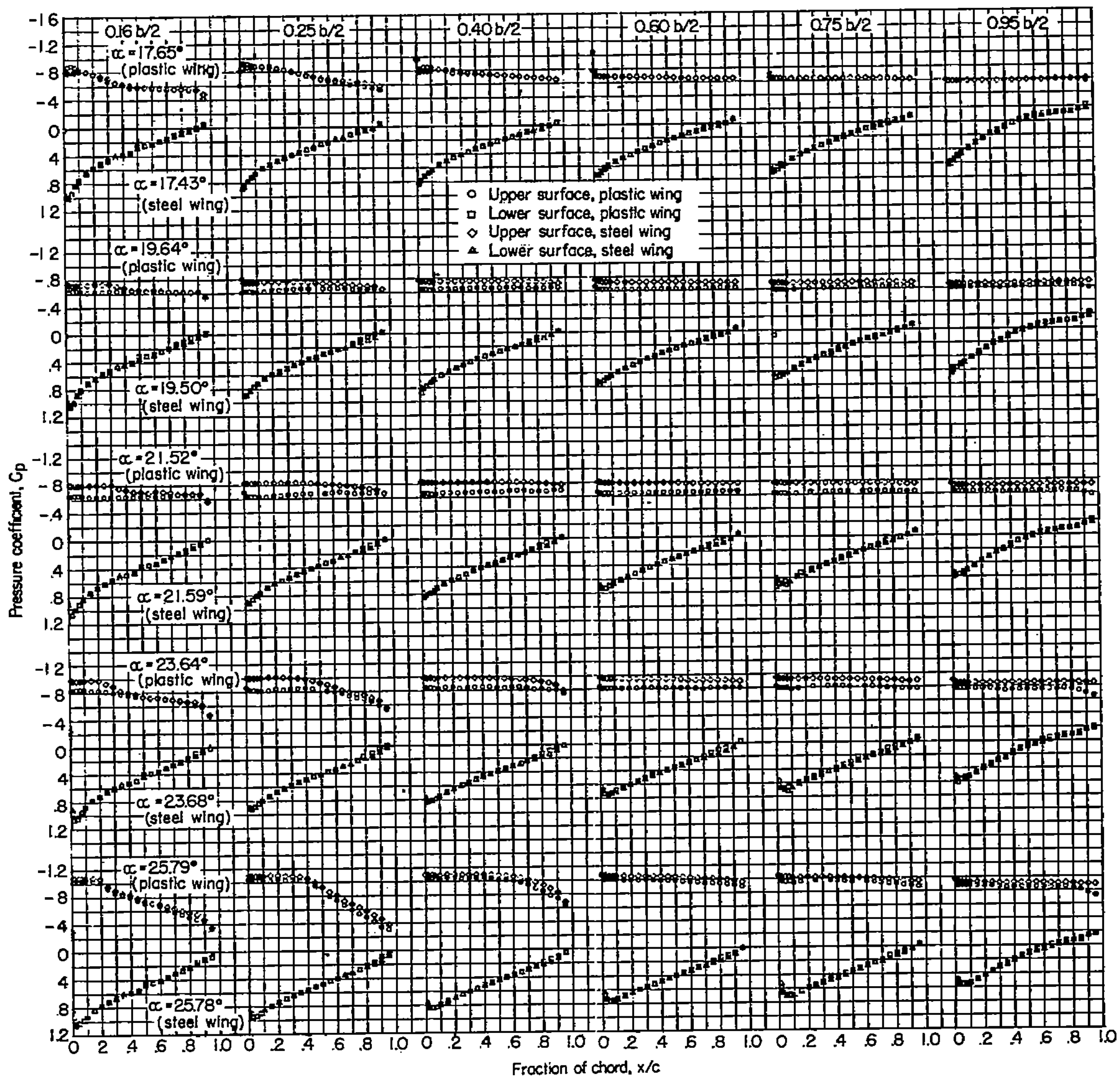
(b) M = 0.90.

Figure 4.- Continued.



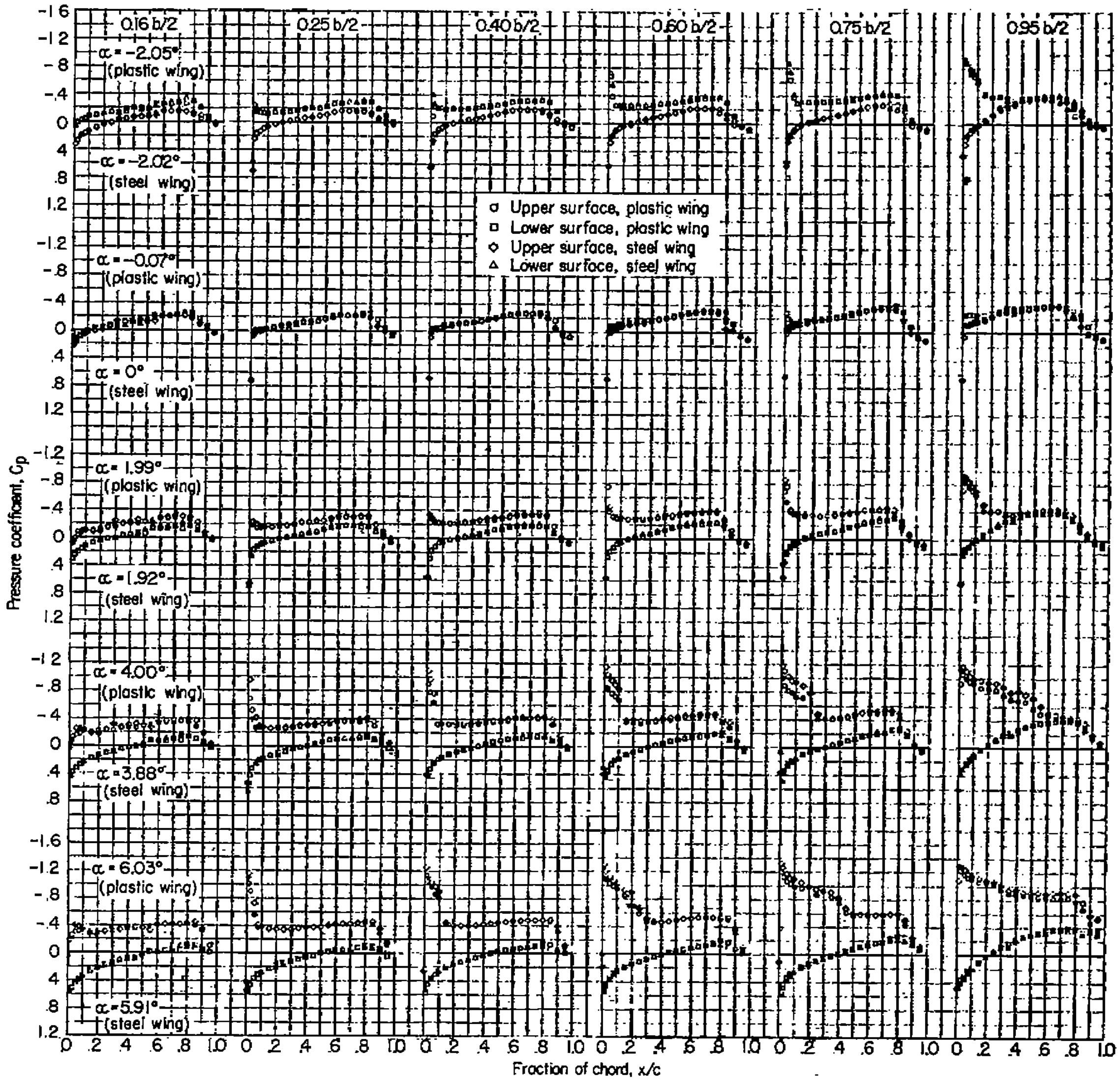
(b) M = 0.90, continued.

Figure 4.- Continued.



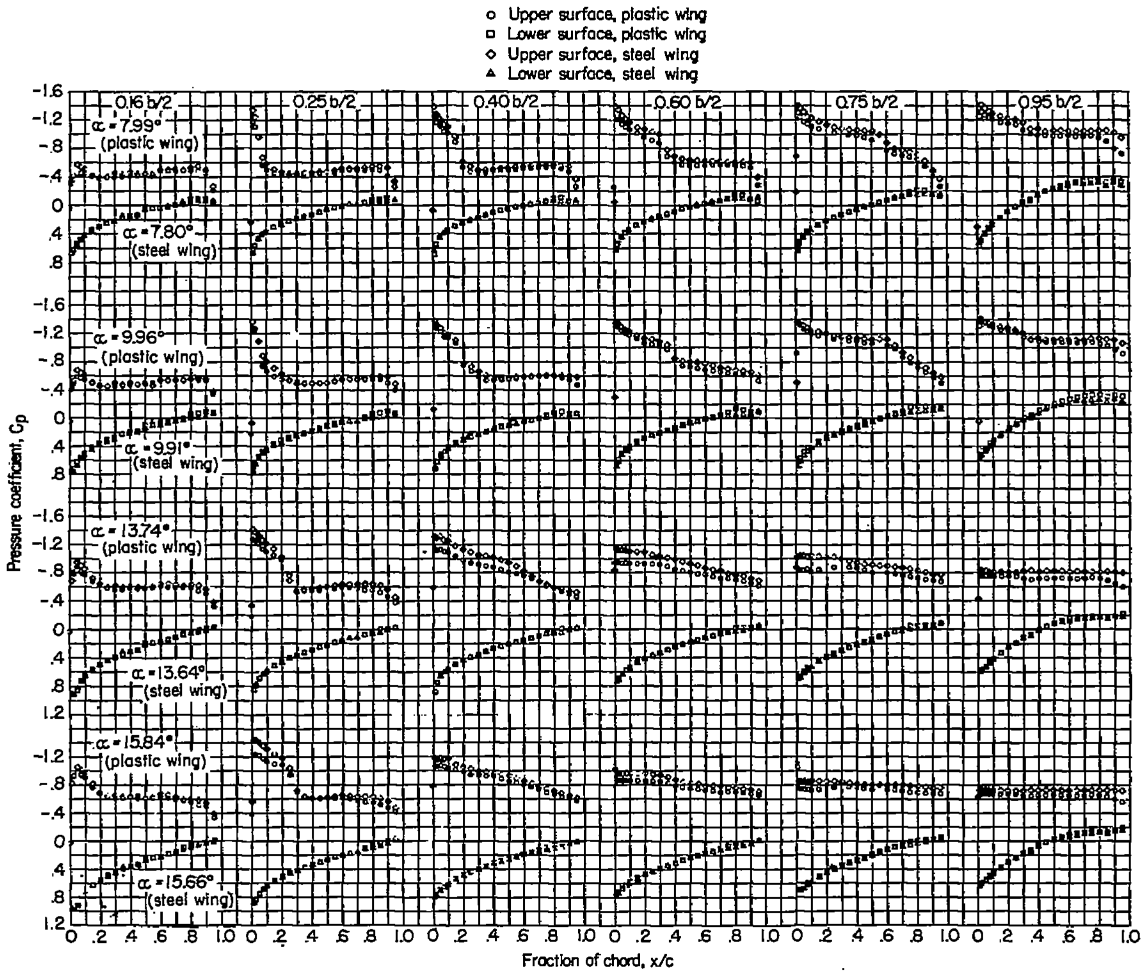
(b) M = 0.90, concluded.

Figure 4.- Continued.



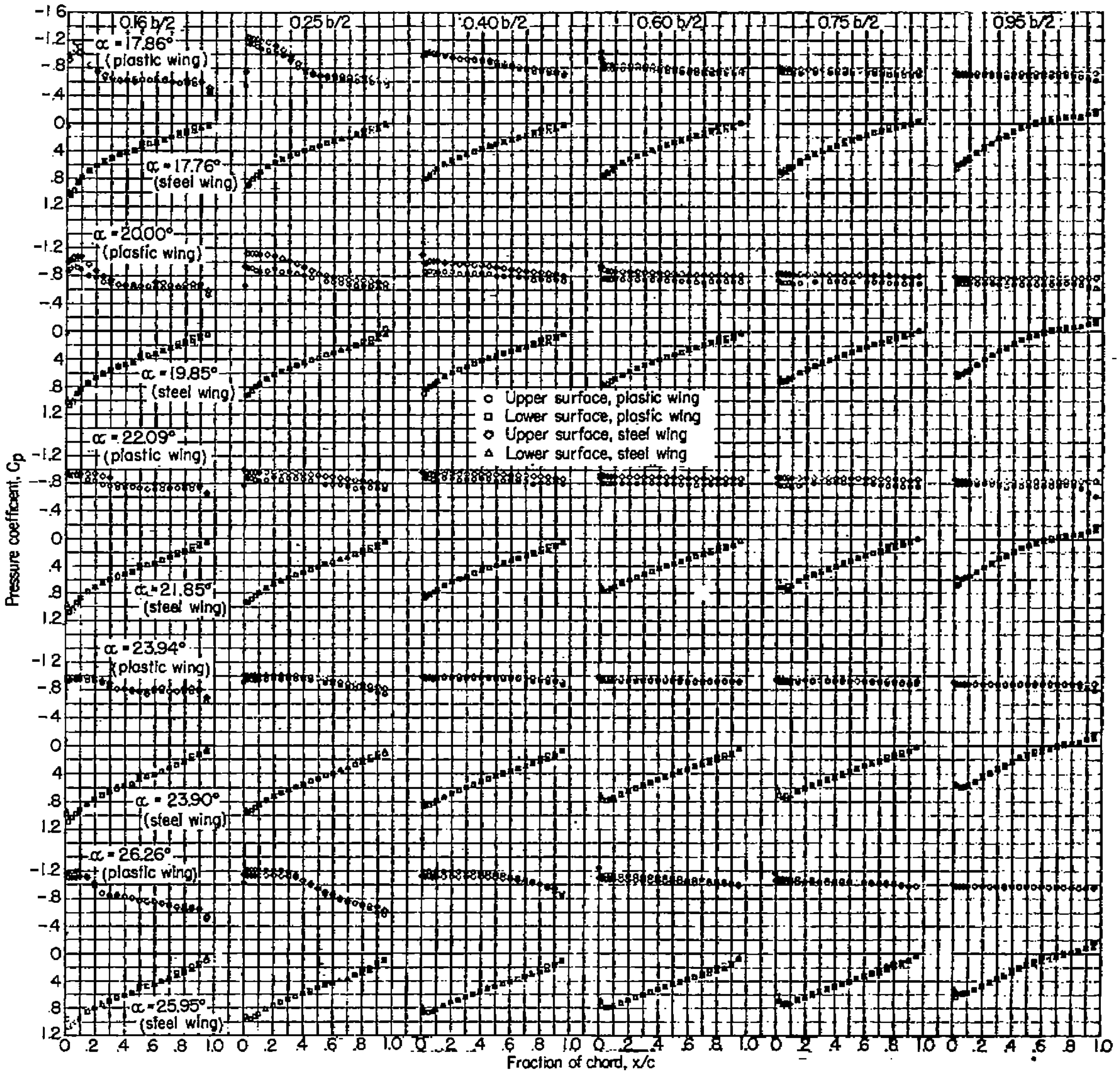
(c) M = 0.94.

Figure 4.- Continued.



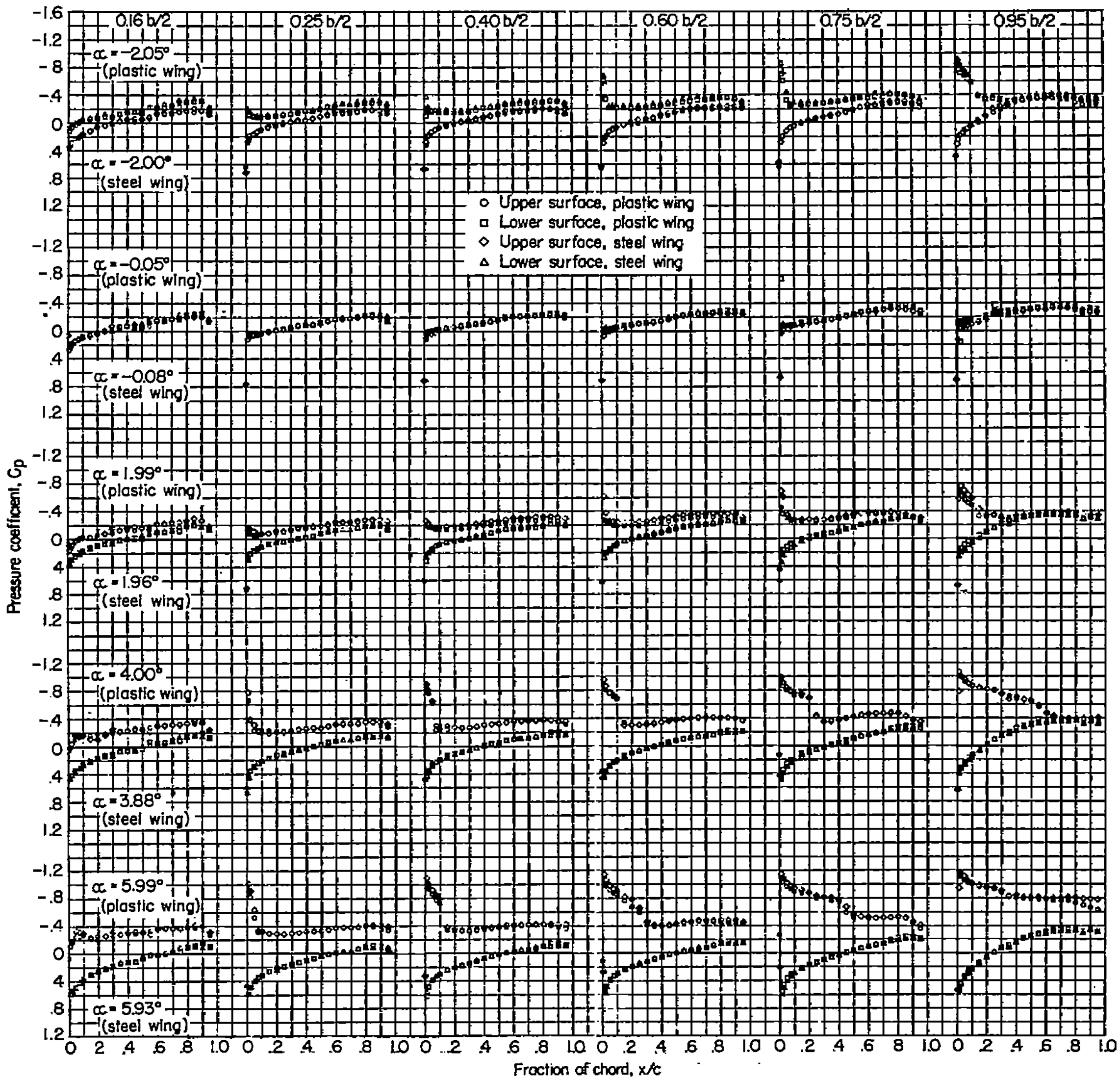
(c) M = 0.94, continued.

Figure 4.- Continued.



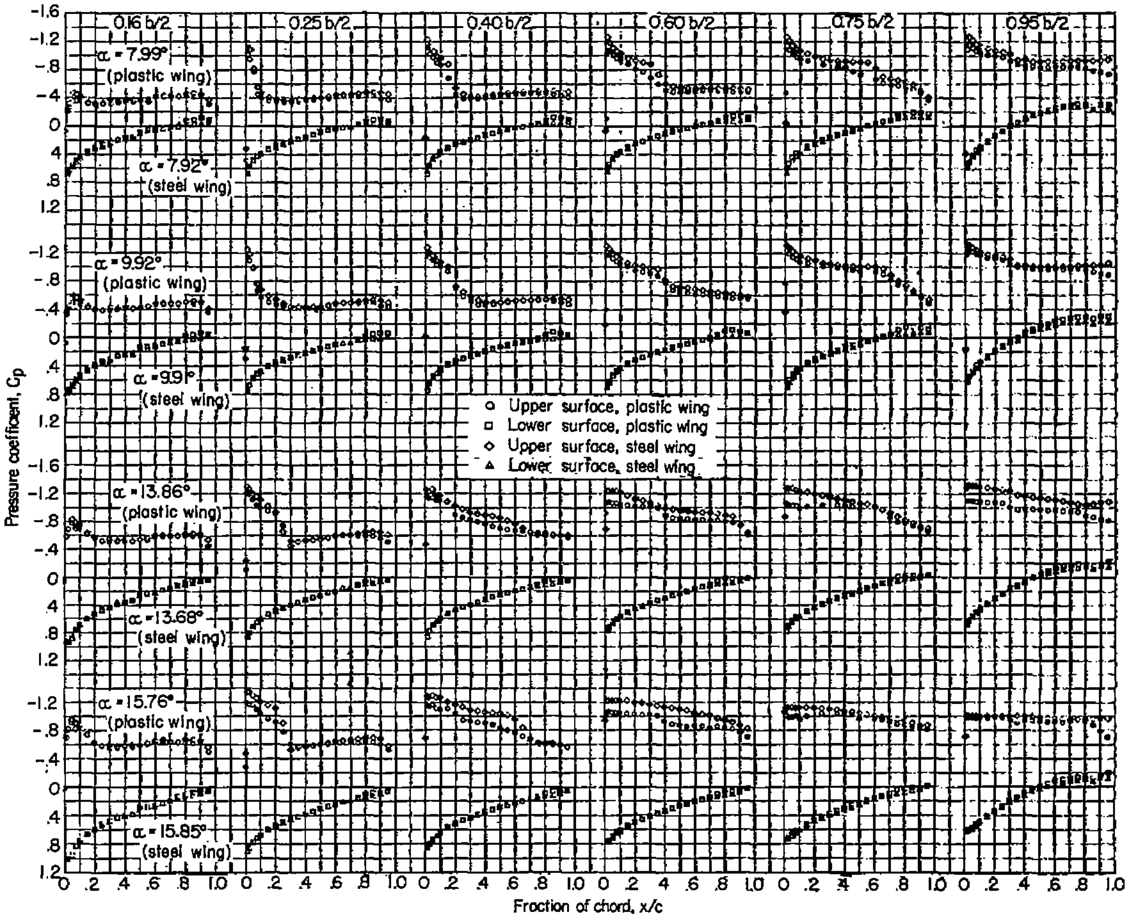
(c) M = 0.94, concluded.

Figure 4.- Continued.



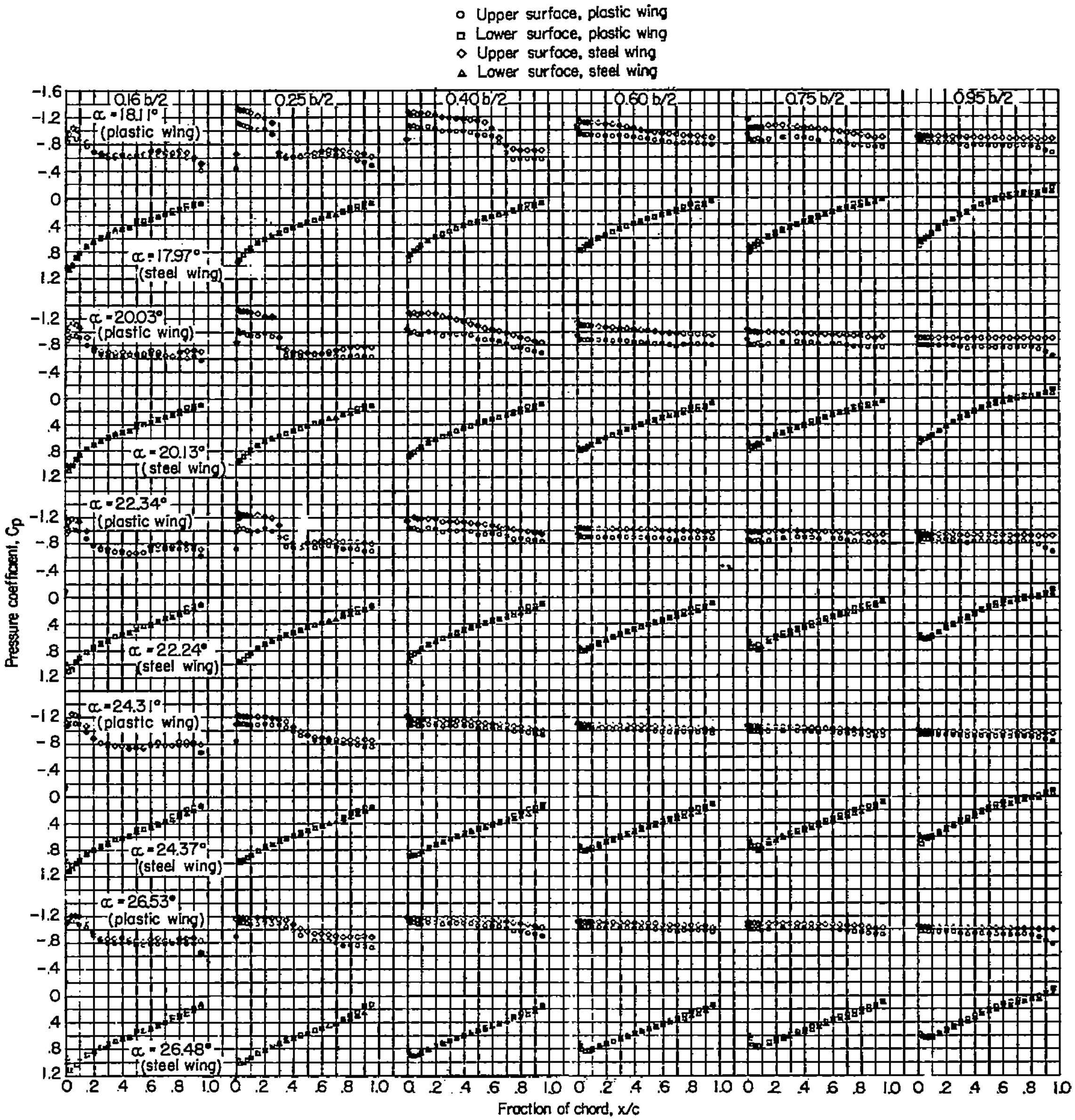
(d) $M = 0.98$.

Figure 4.- Continued.



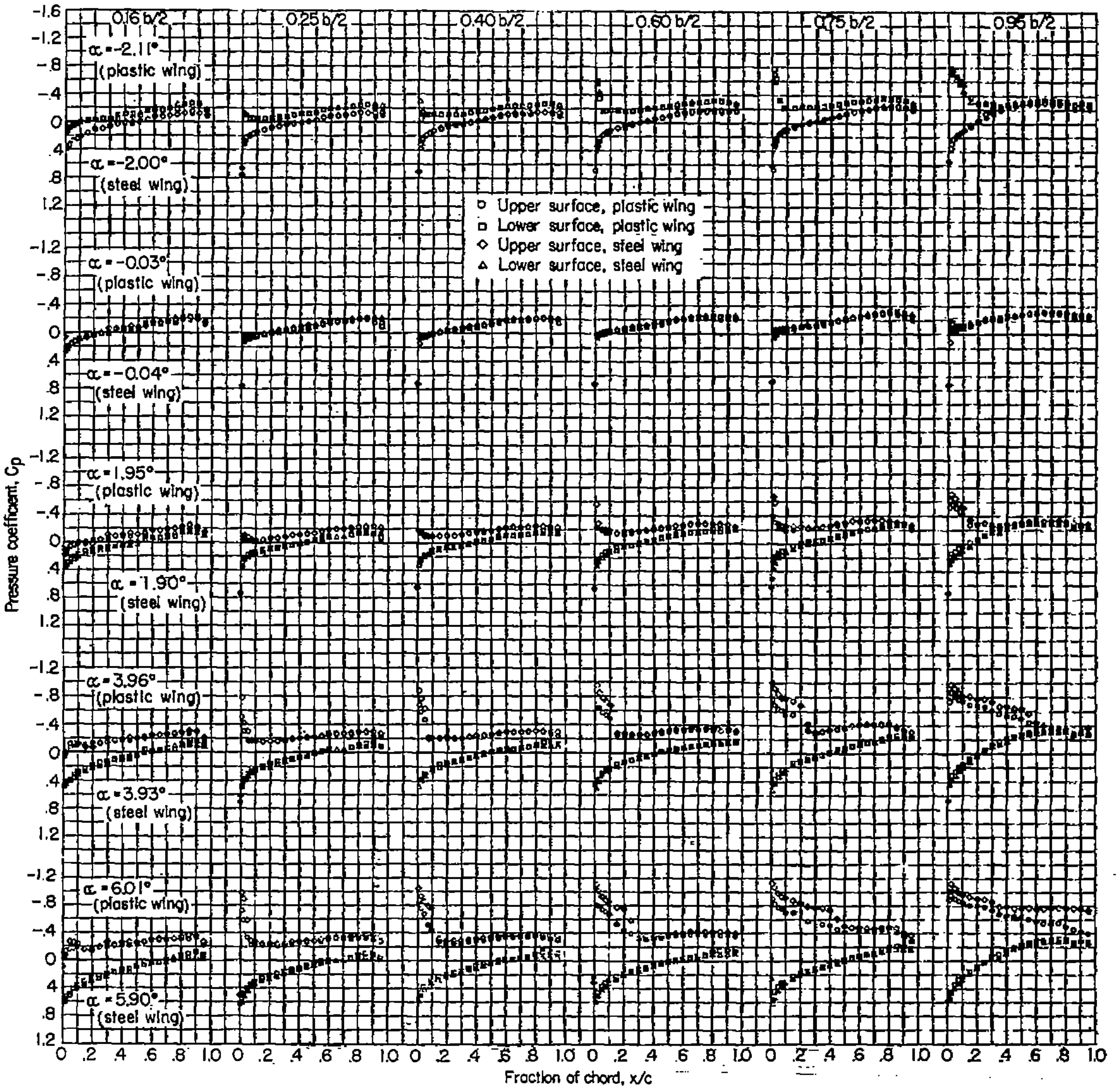
(d) M = 0.98, continued.

Figure 4.- Continued.



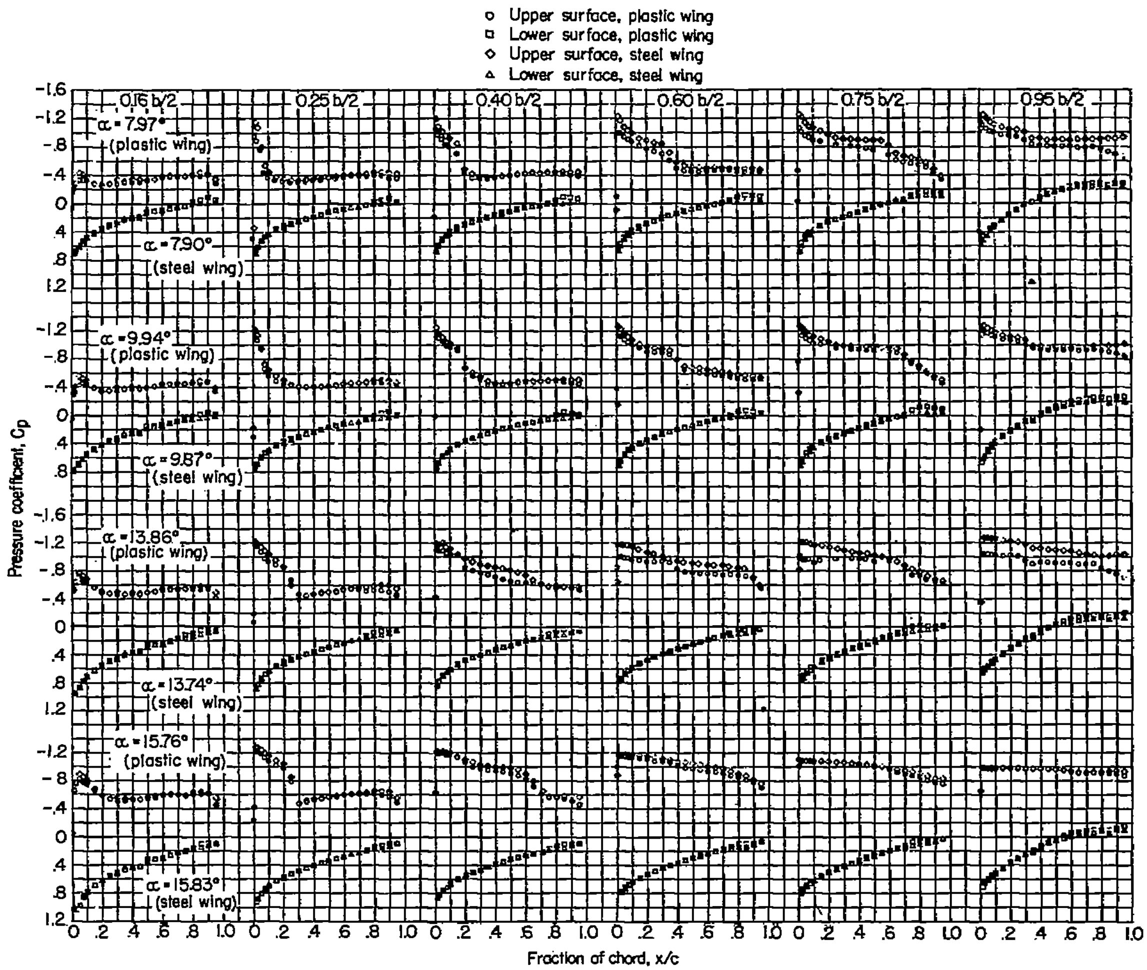
(d) M = 0.98, concluded.

Figure 4.- Continued.



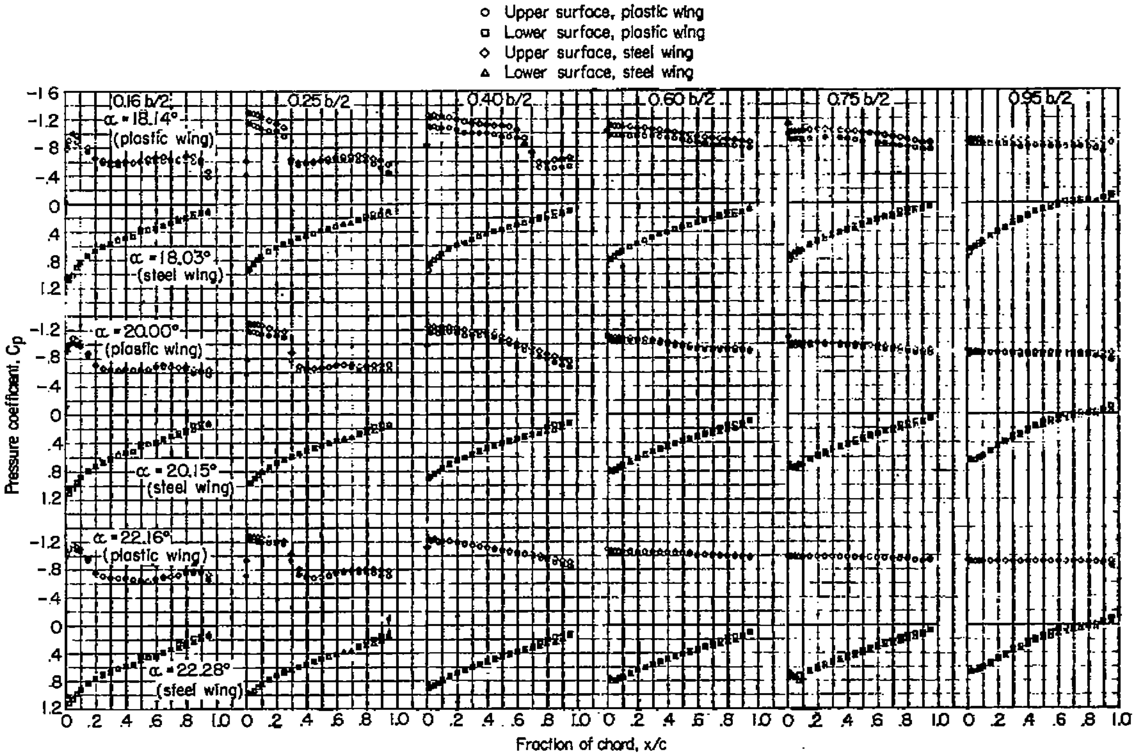
(e) M = 1.00.

Figure 4.- Continued.



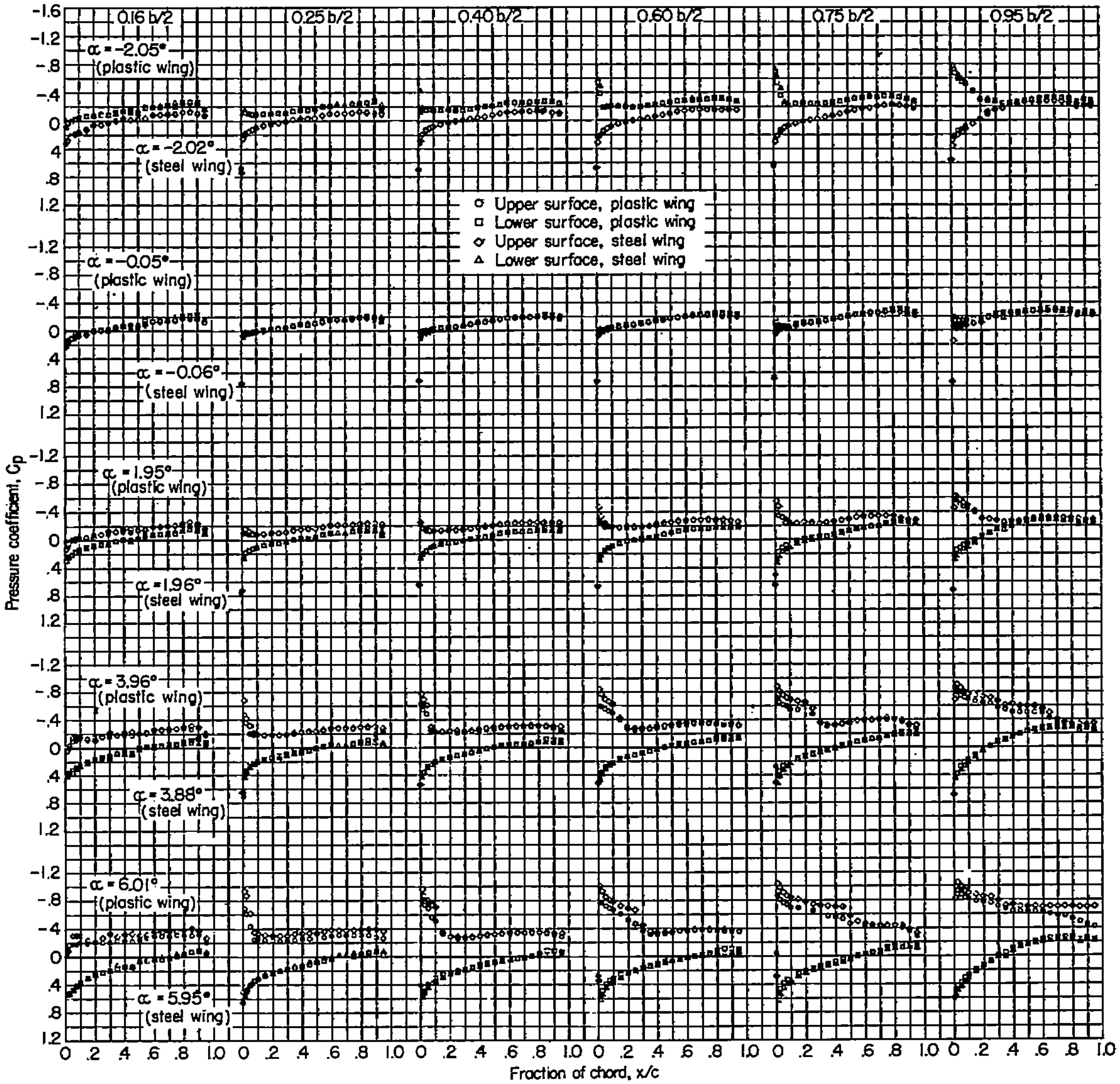
(e) M = 1.00, continued.

Figure 4.- Continued.



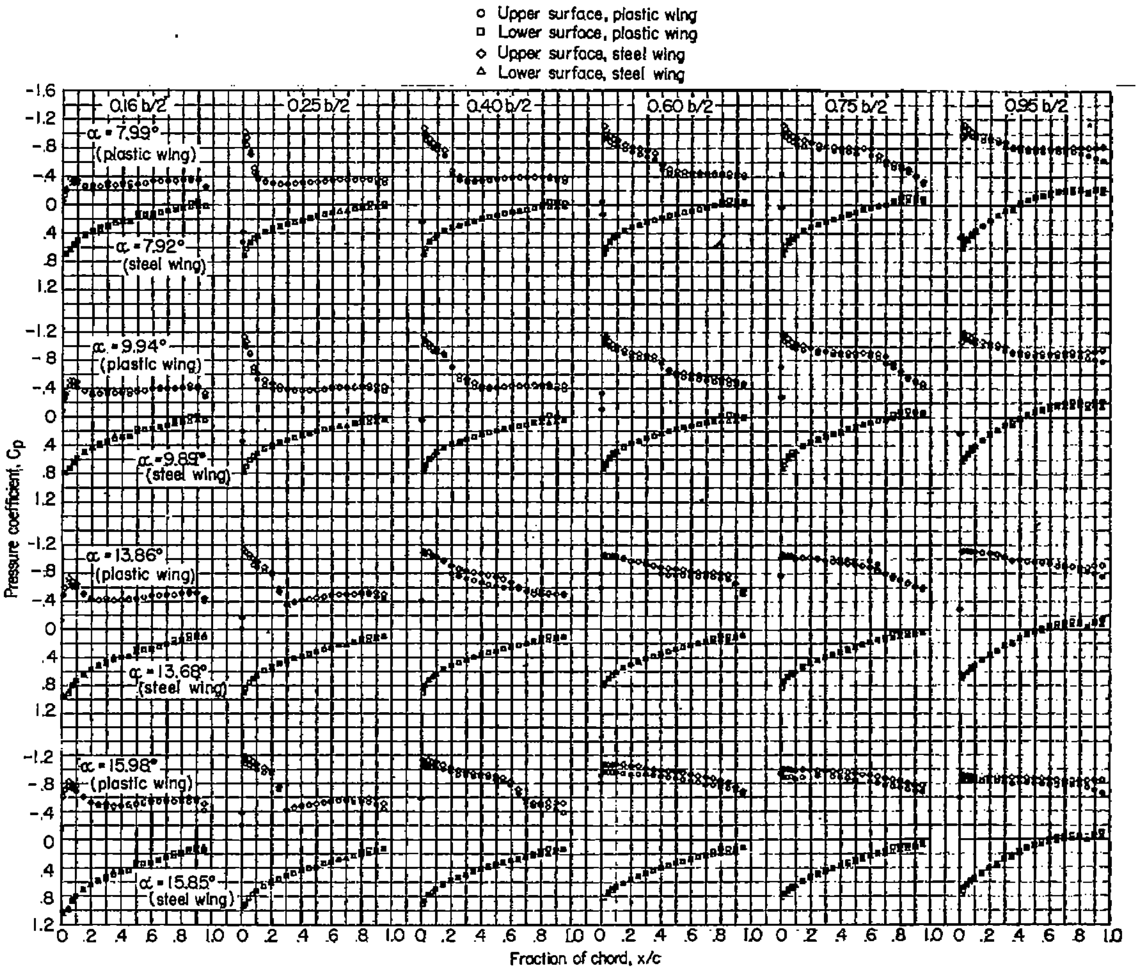
(e) M = 1.00, concluded.

Figure 4.- Continued.



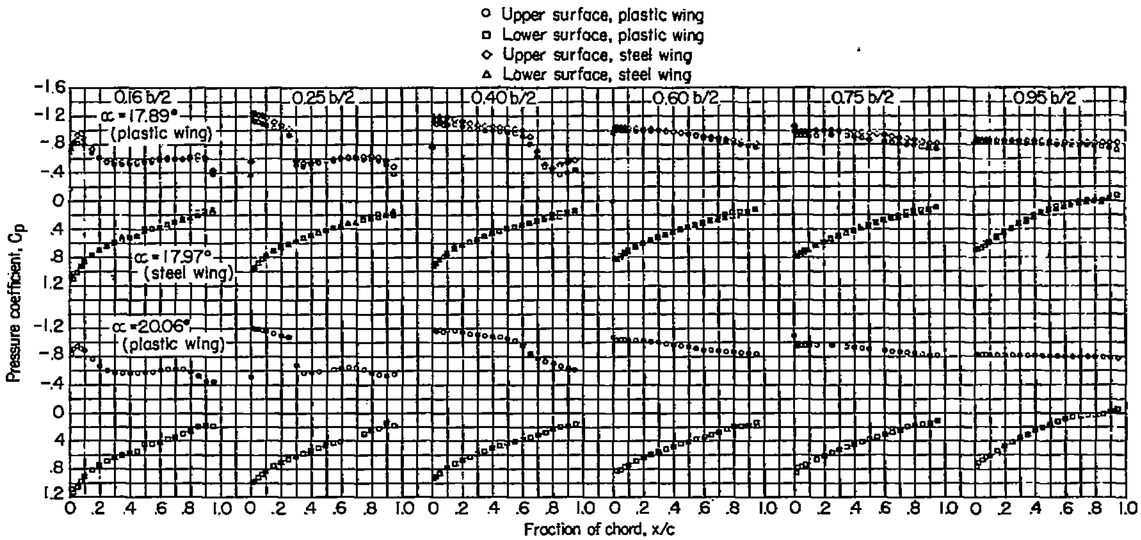
(f) $M = 1.03$.

Figure 4.- Continued.



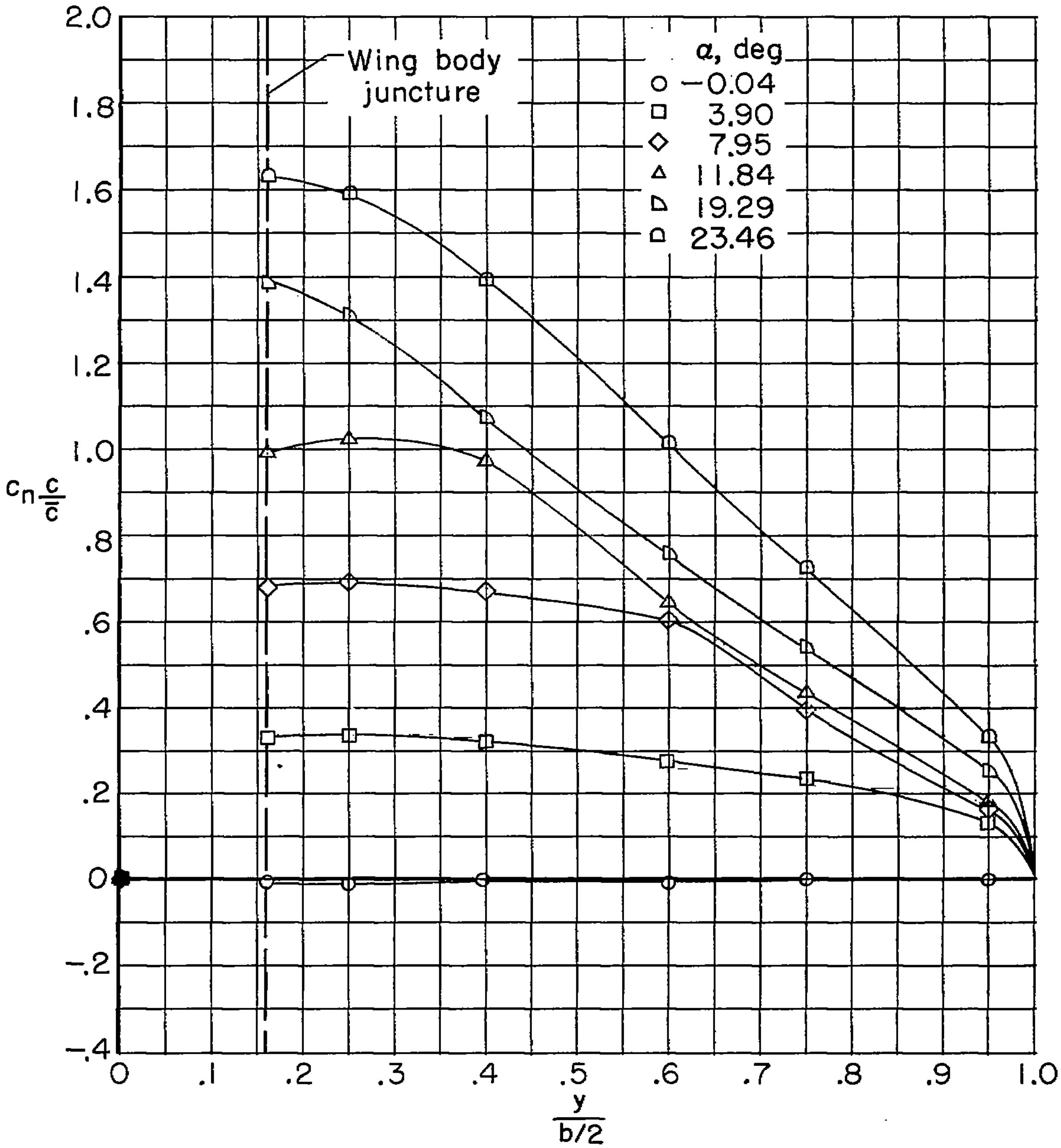
(f) M = 1.03, continued.

Figure 4.- Continued.



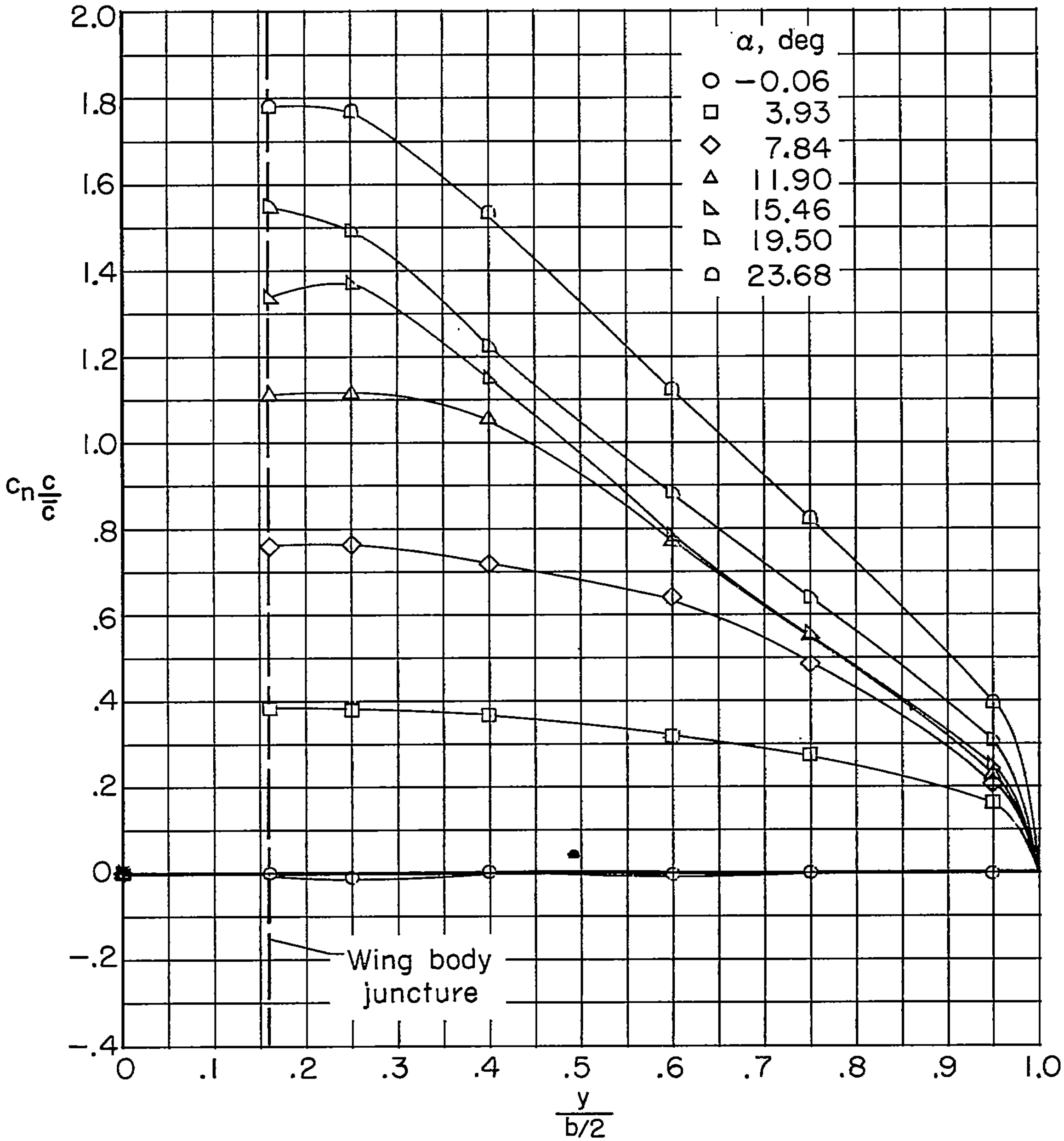
(f) M = 1.03, concluded.

Figure 4.- Concluded.



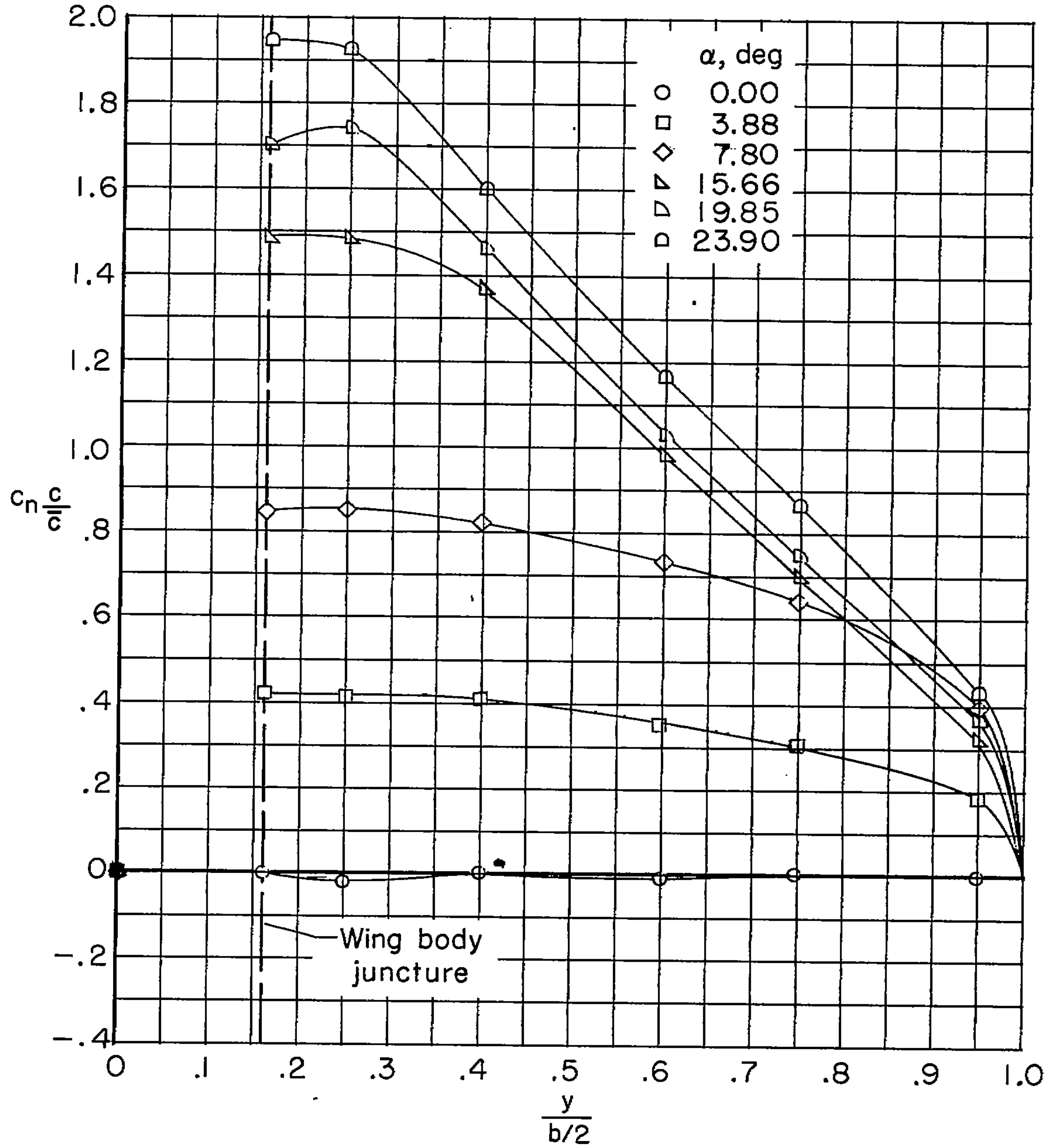
(a) M = 0.80.

Figure 5.- Spanwise variation of normal-load parameter for steel wing at various angles of attack and Mach numbers.



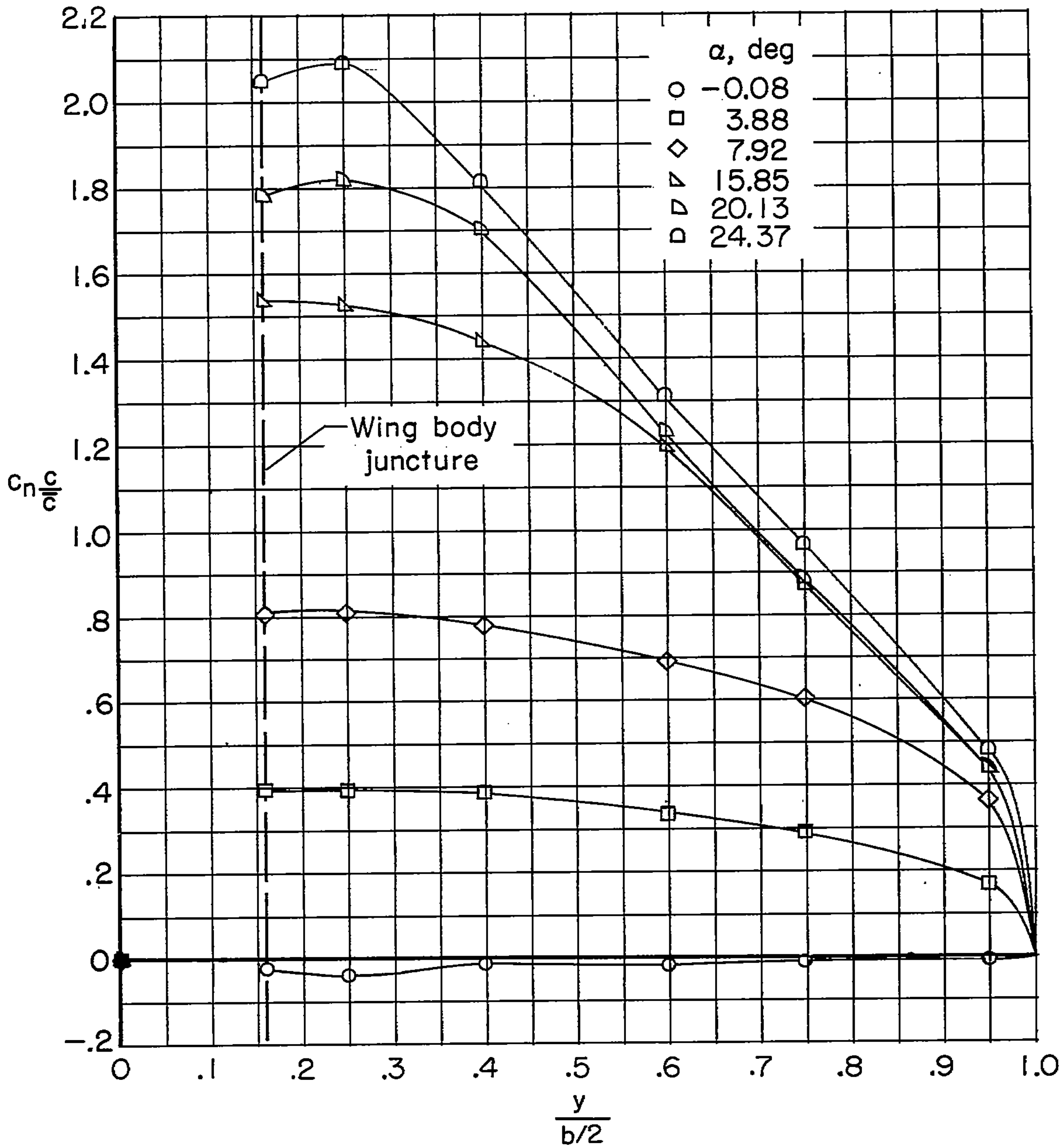
(b) M = 0.90.

Figure 5.- Continued.



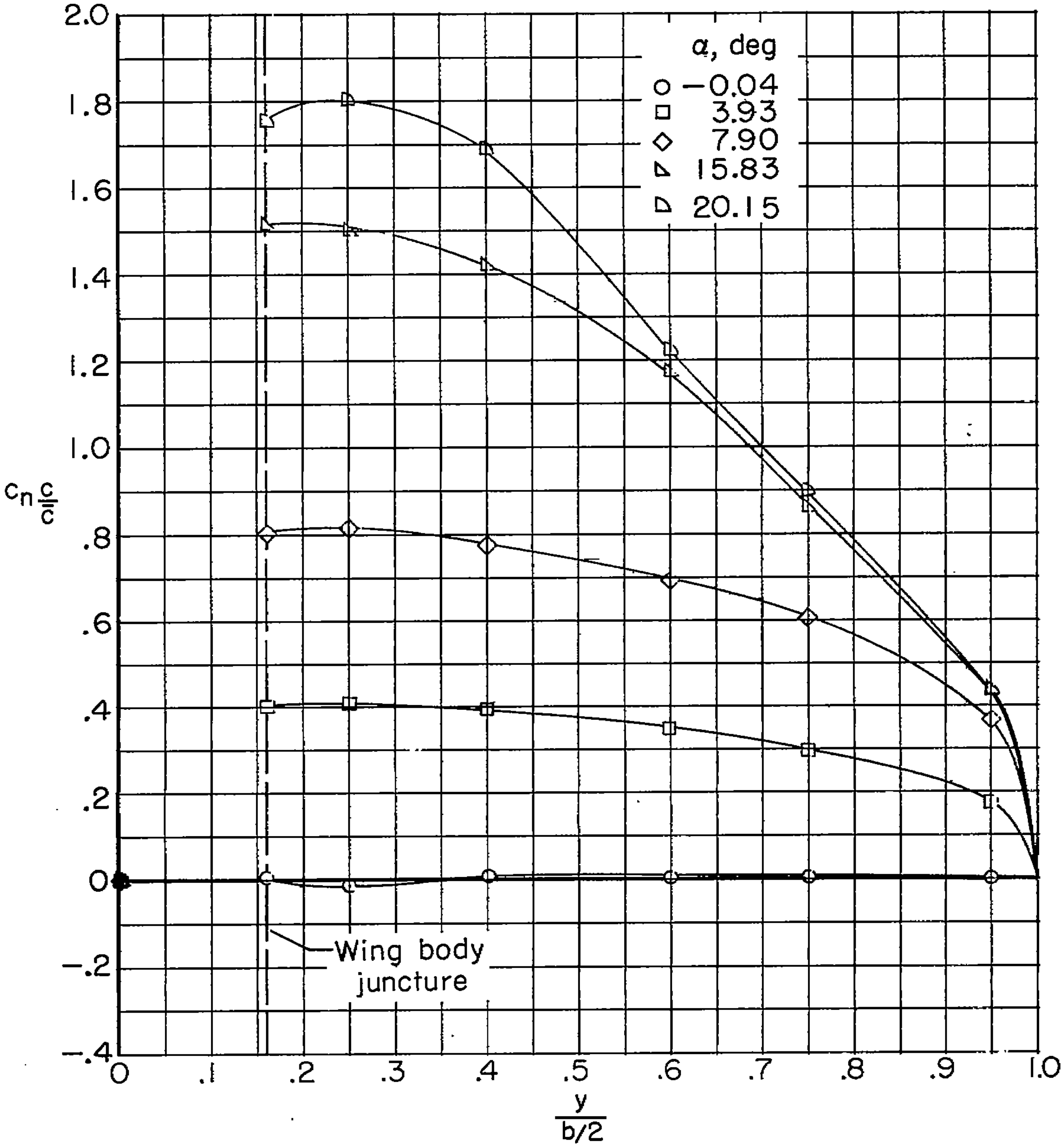
(c) M = 0.94.

Figure 5.- Continued.



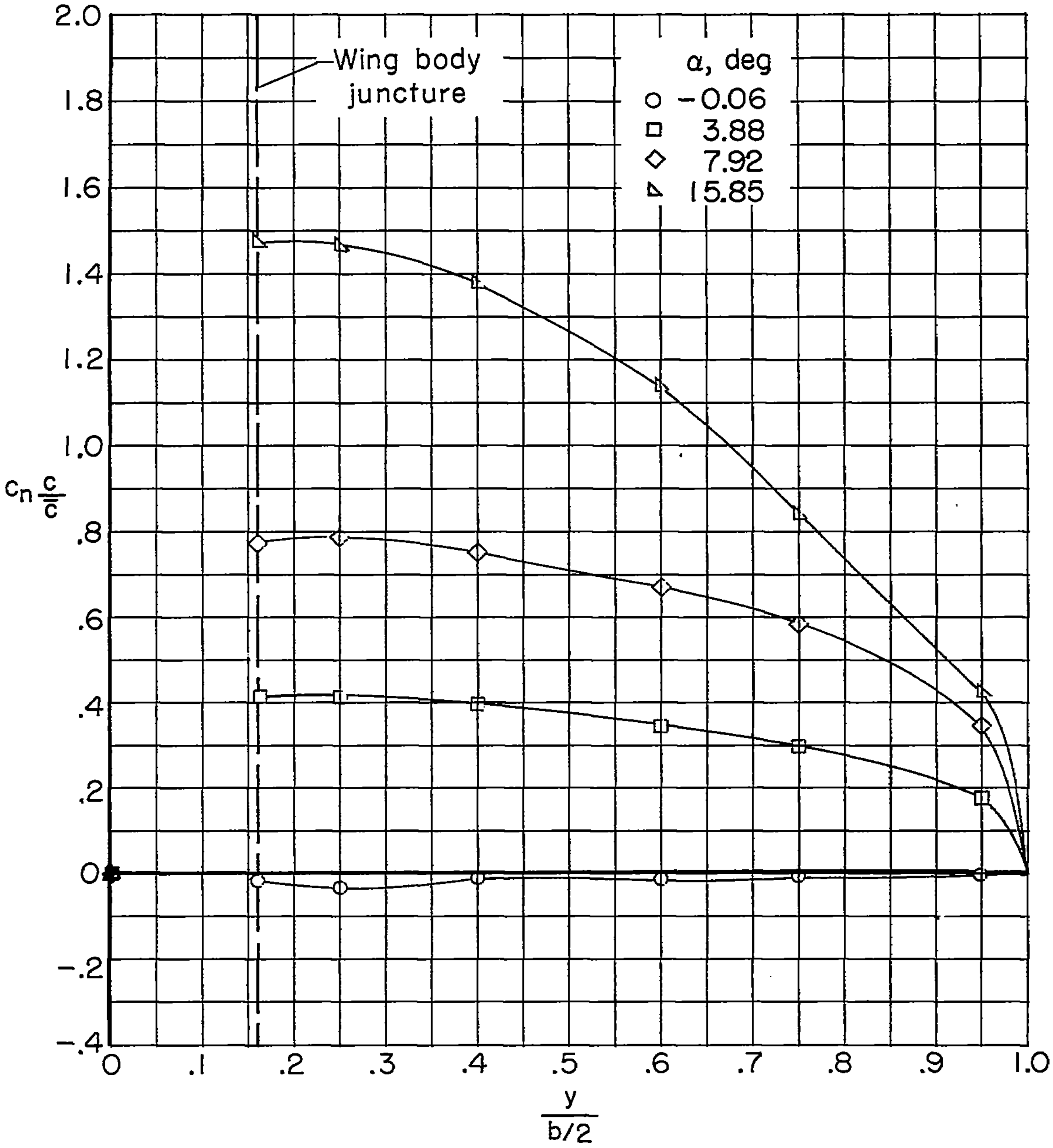
(d) M = 0.98.

Figure 5.- Continued.



(e) M = 1.00.

Figure 5.- Continued.



(f) M = 1.03.

Figure 5.- Concluded.

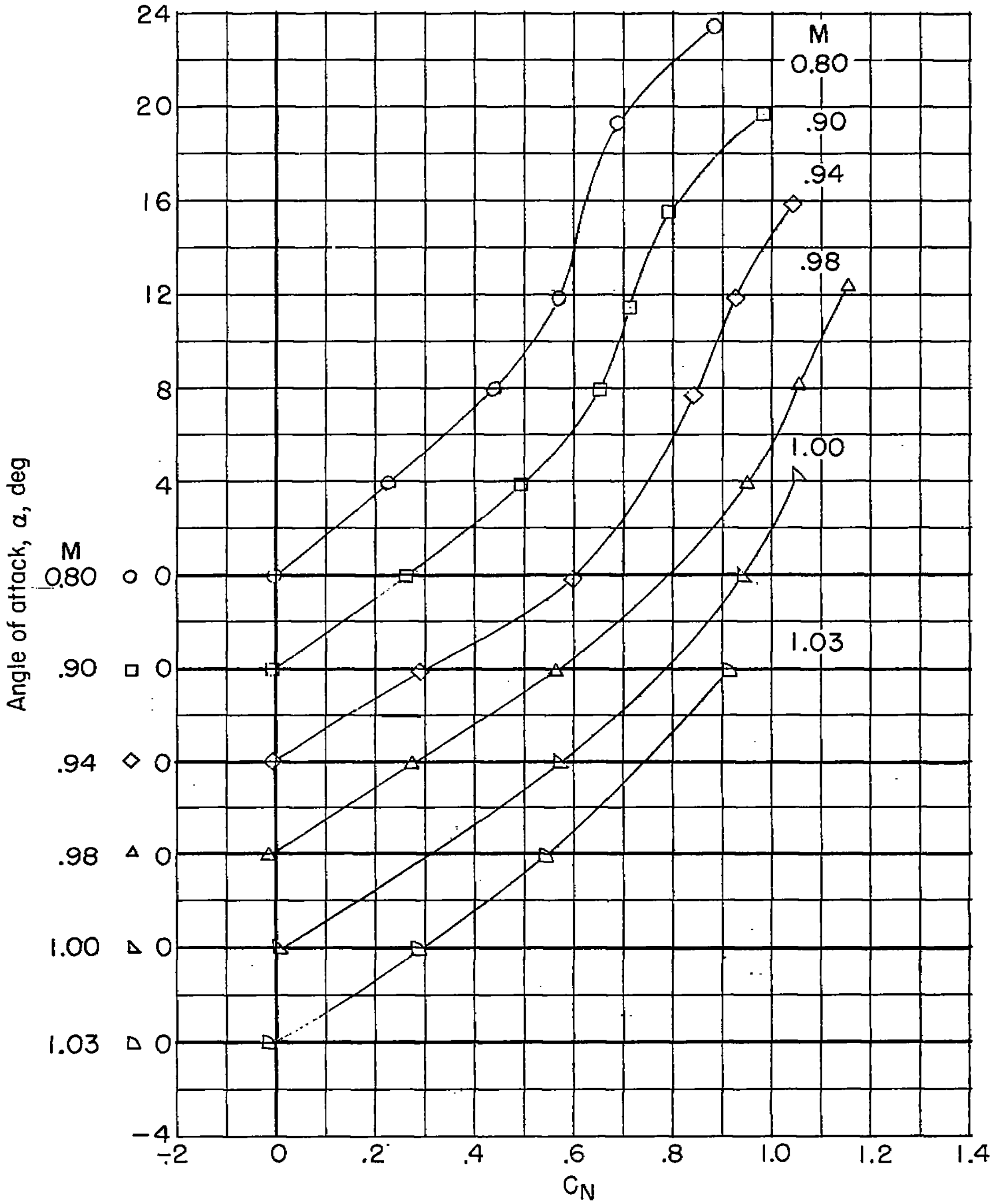


Figure 6.- Variation of angle of attack with wing normal-force coefficient for several Mach numbers. Steel wing.

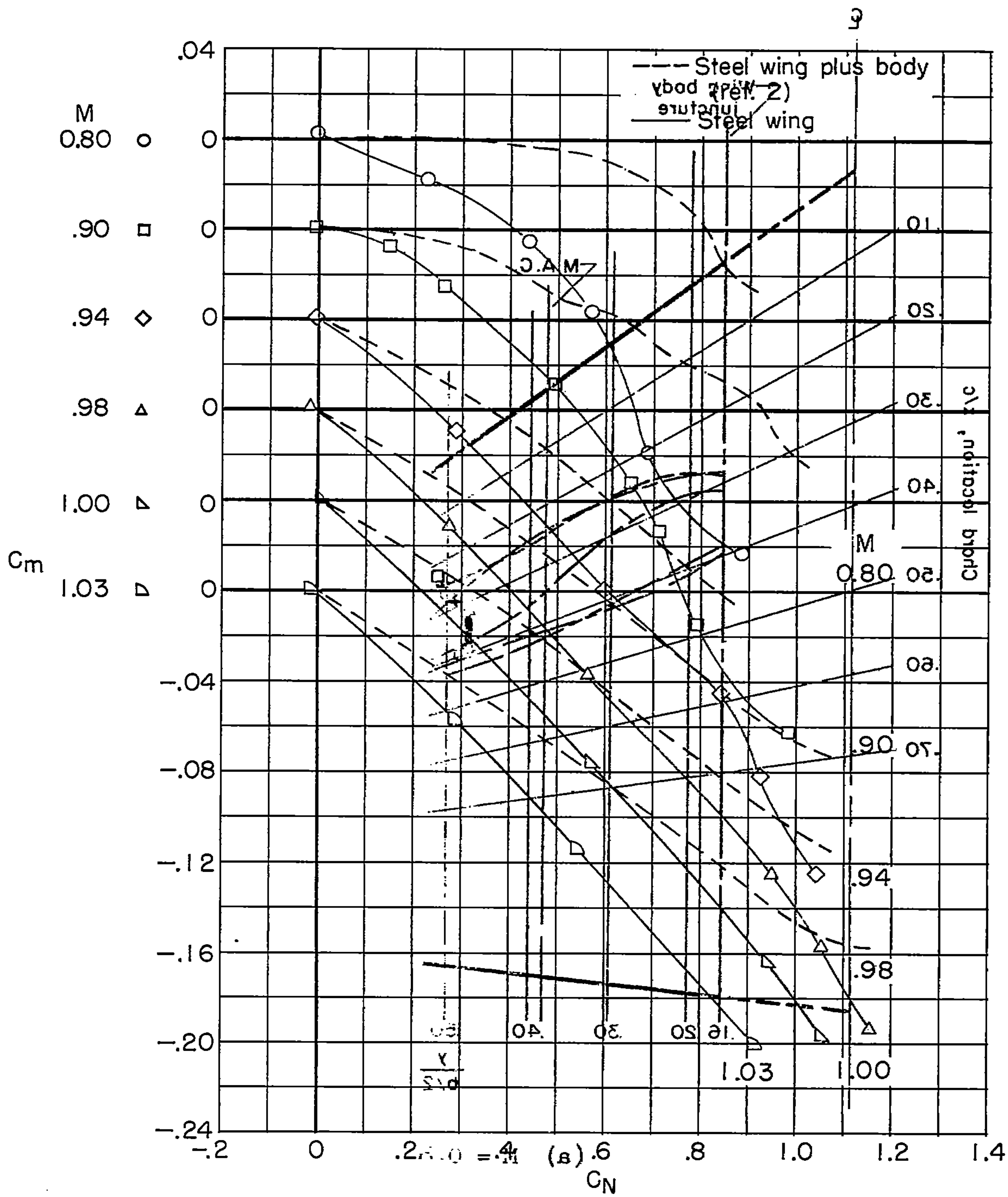
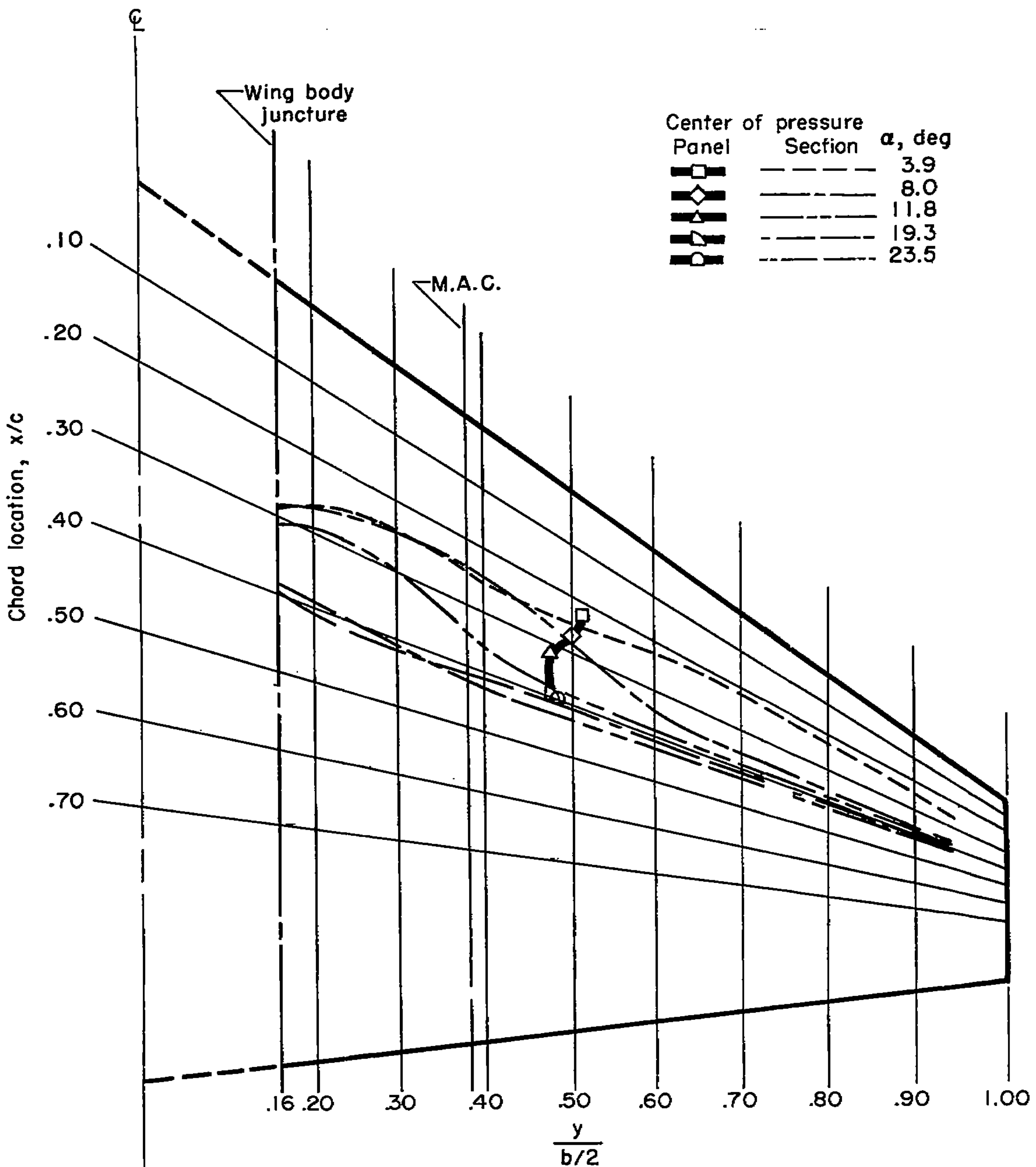
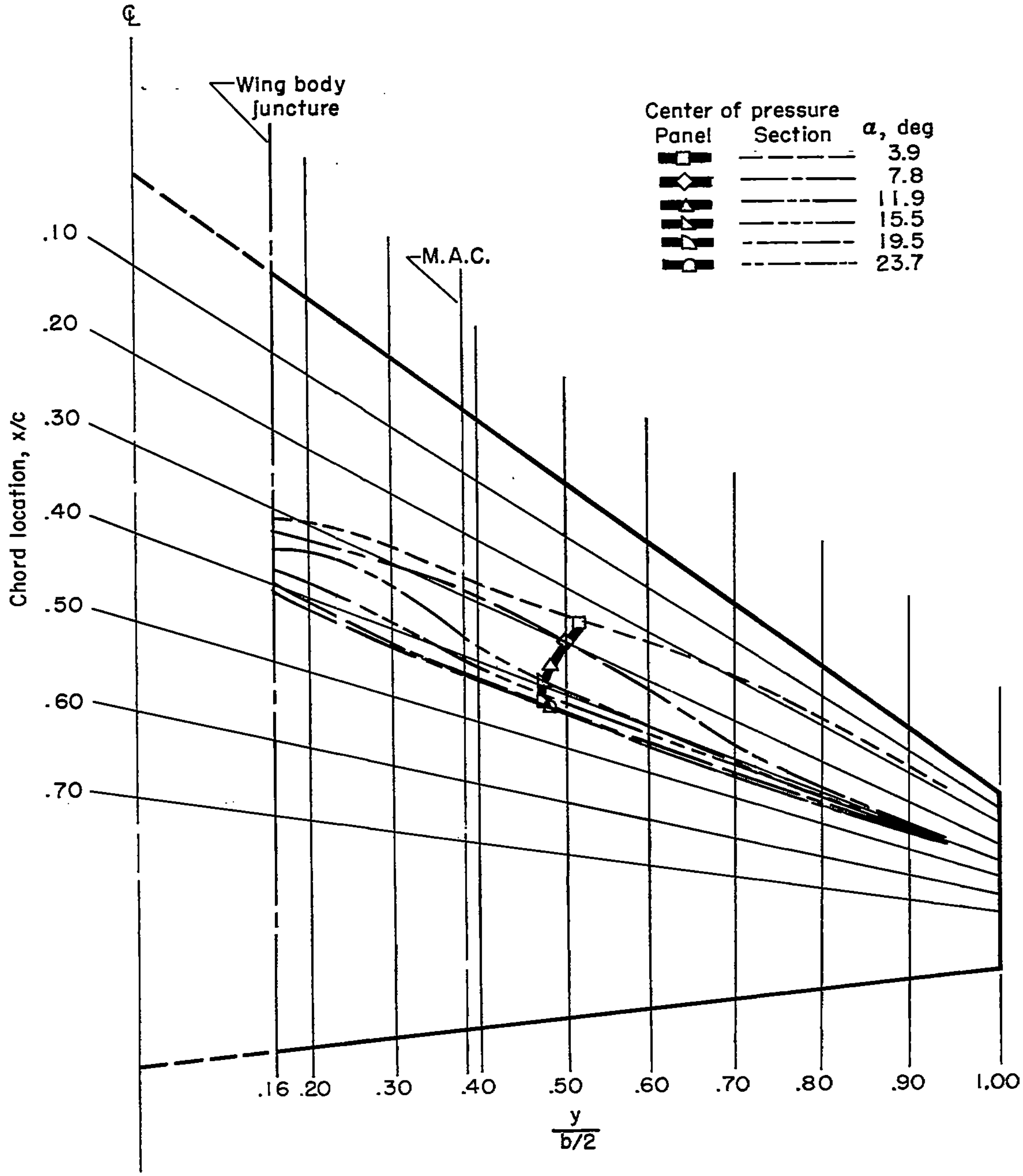


Figure 7.- Variation of wing pitching-moment coefficient with wing normal-force coefficient for several Mach numbers. Steel wing.



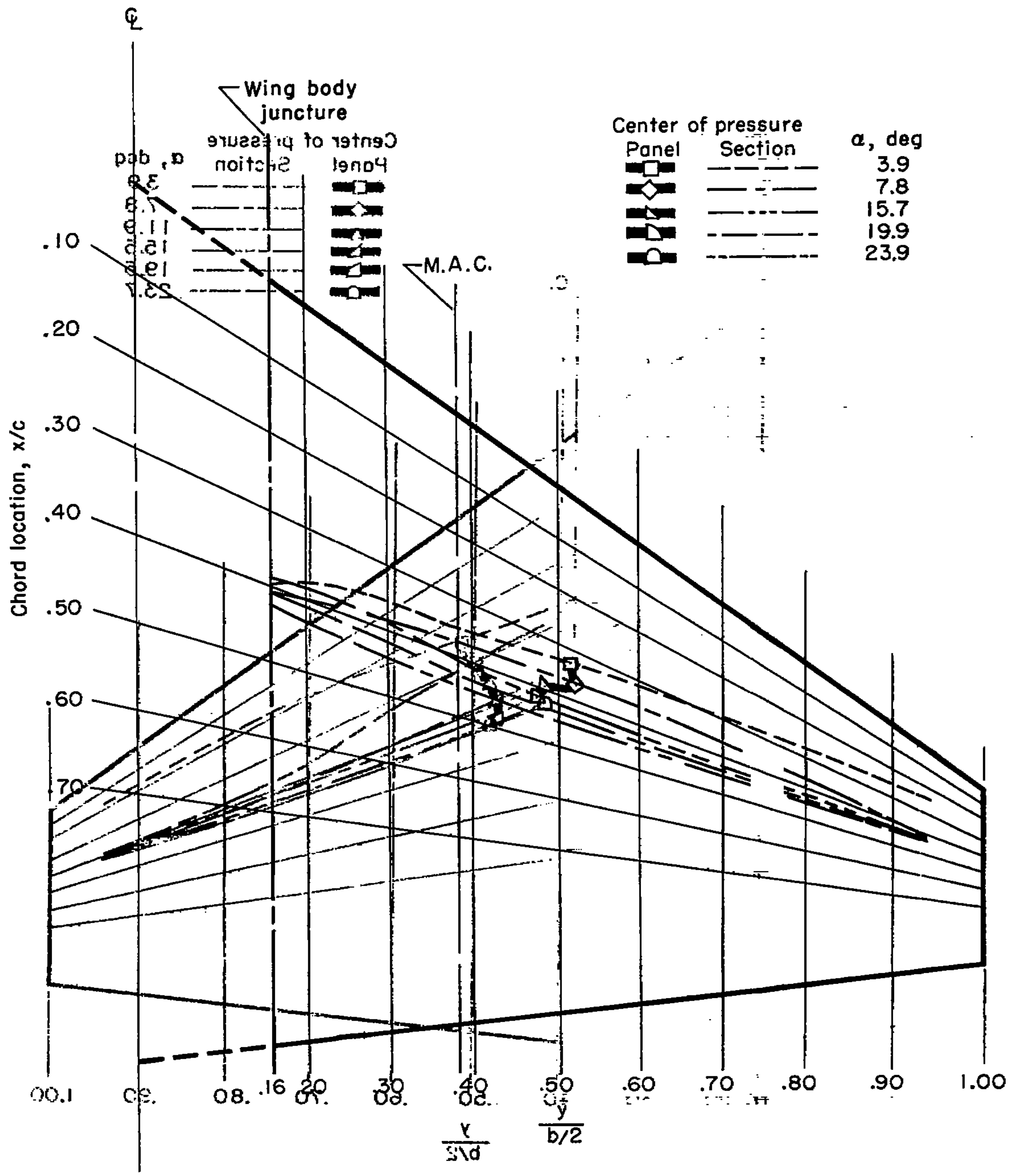
(a) M = 0.80.

Figure 8.- Variation of center-of-pressure location for wing panel and for local sections with angle of attack for several Mach numbers. Steel wing.



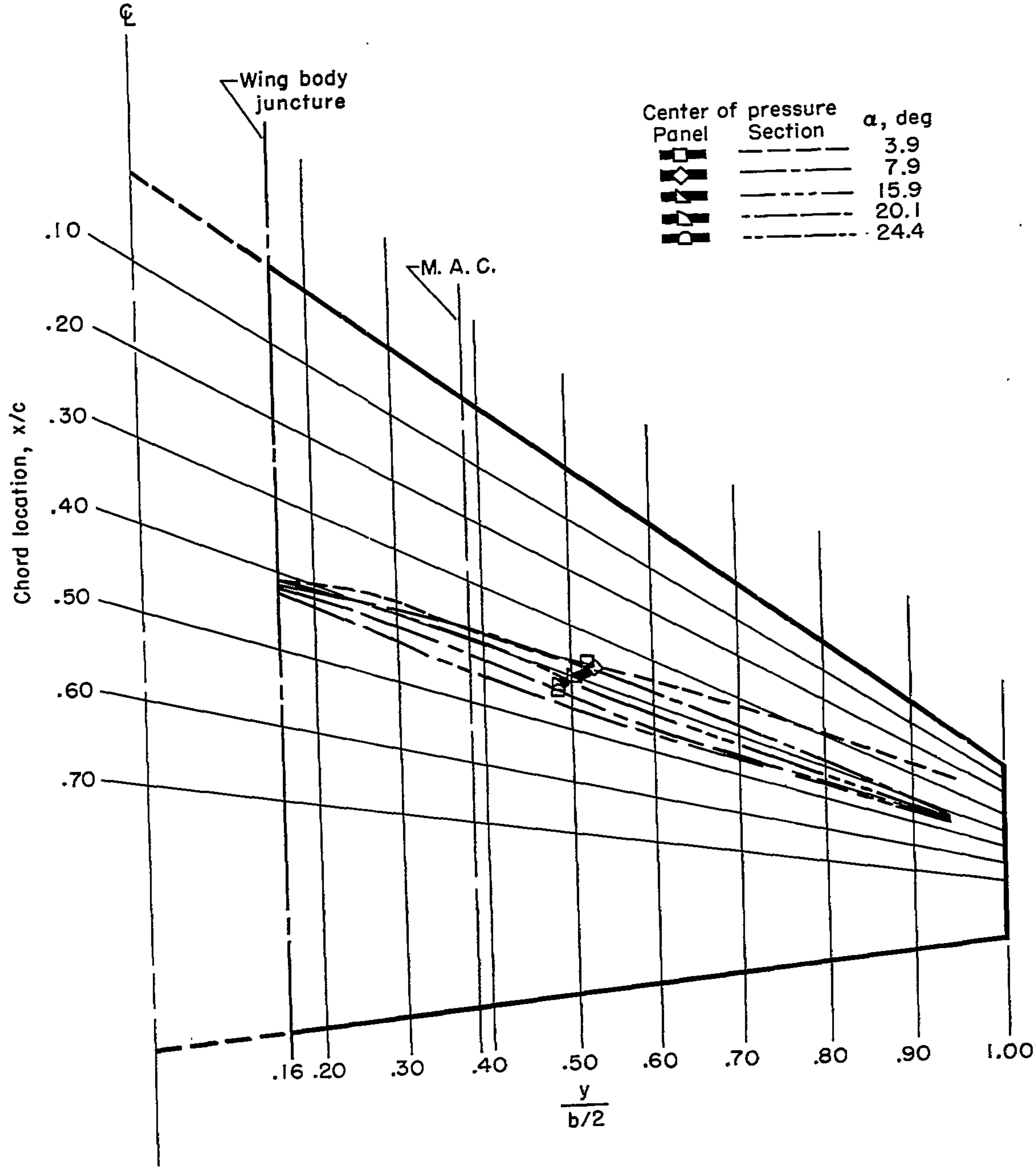
(b) M = 0.90.

Figure 8.- Continued.



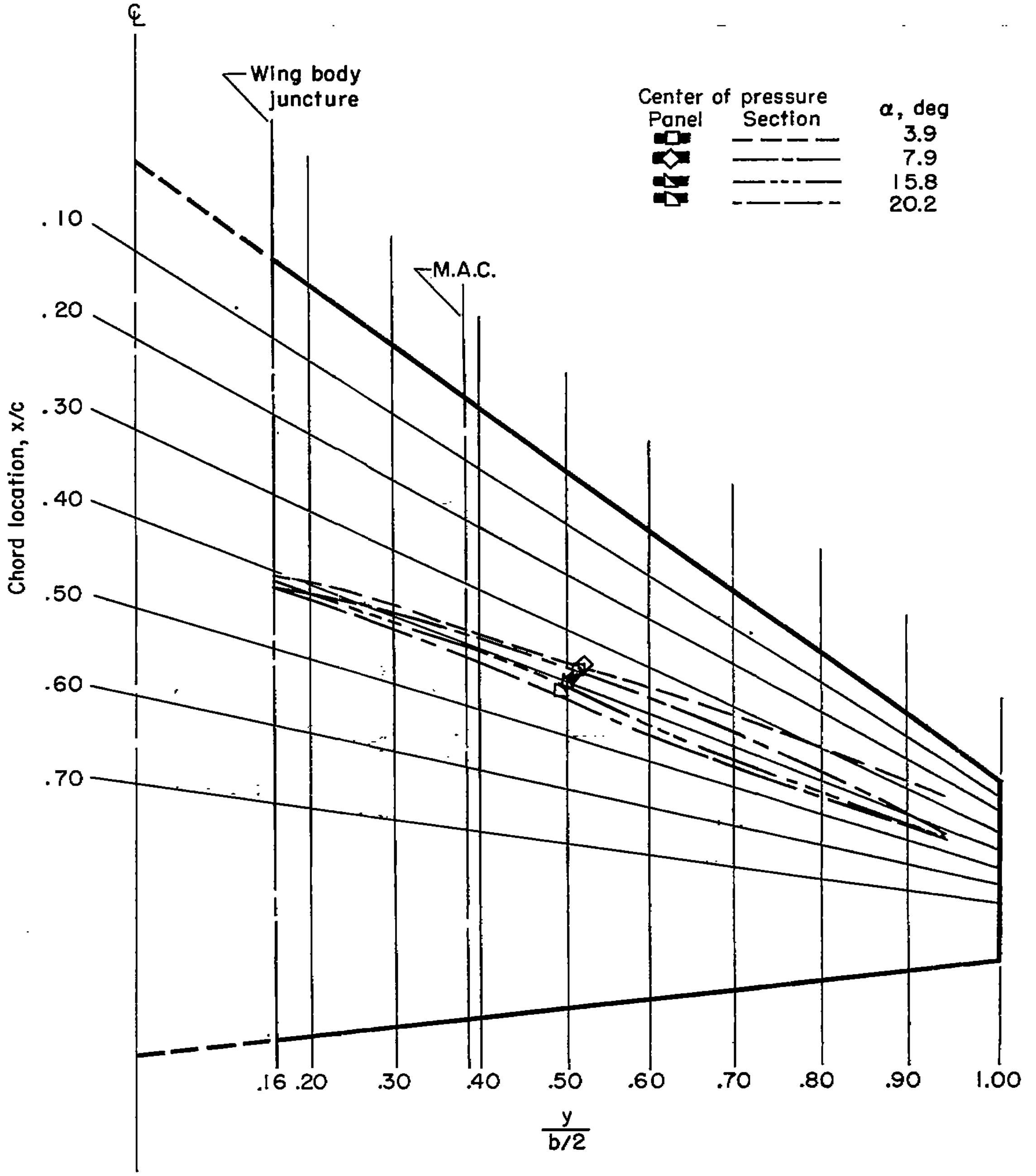
(c) M = 0.94.

Figure 8.- Continued.



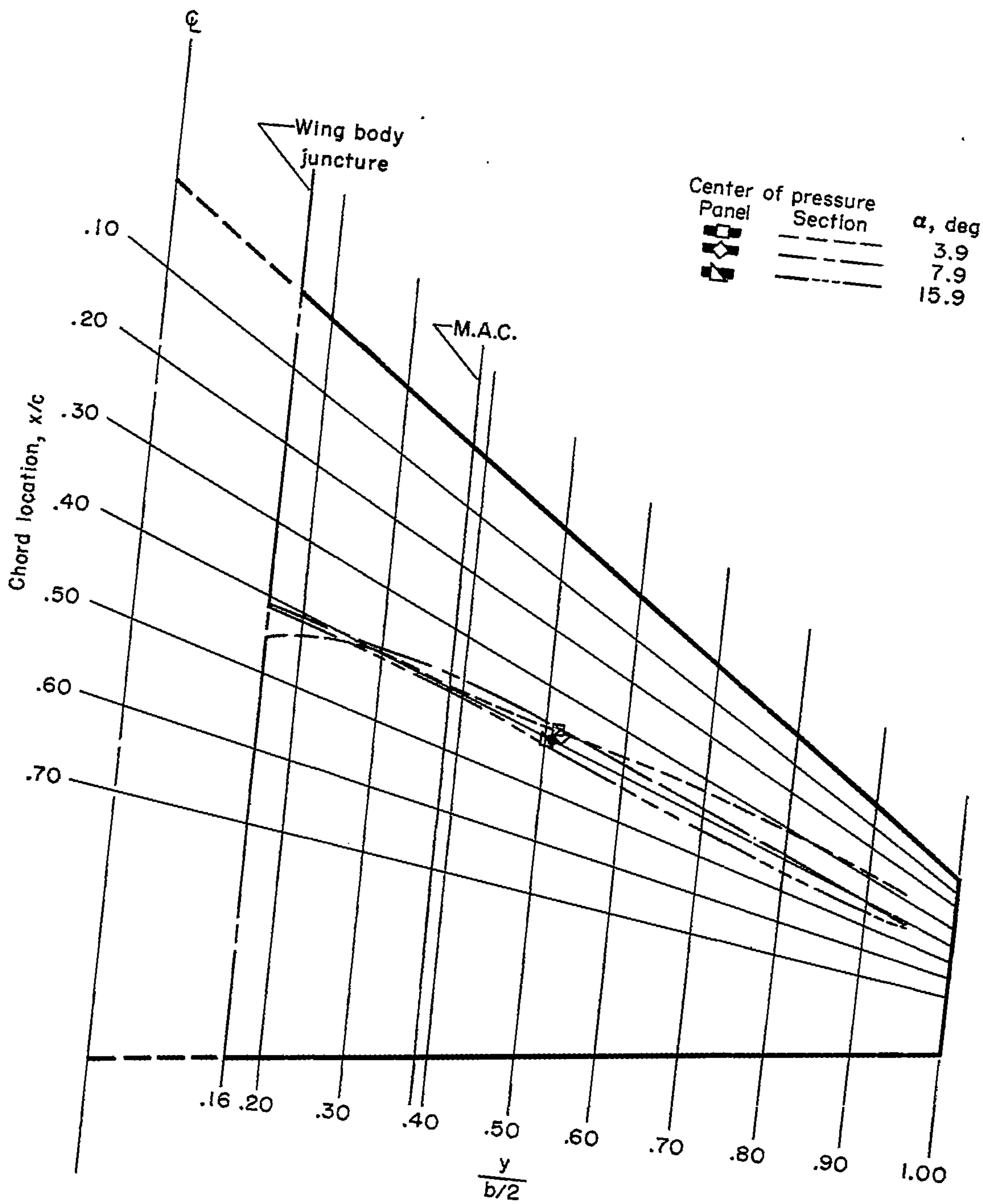
(d) M = 0.98.

Figure 8.- Continued.



(e) M = 1.00.

Figure 8.- Continued.



(f) M = 1.03.

Figure 8.- Concluded.

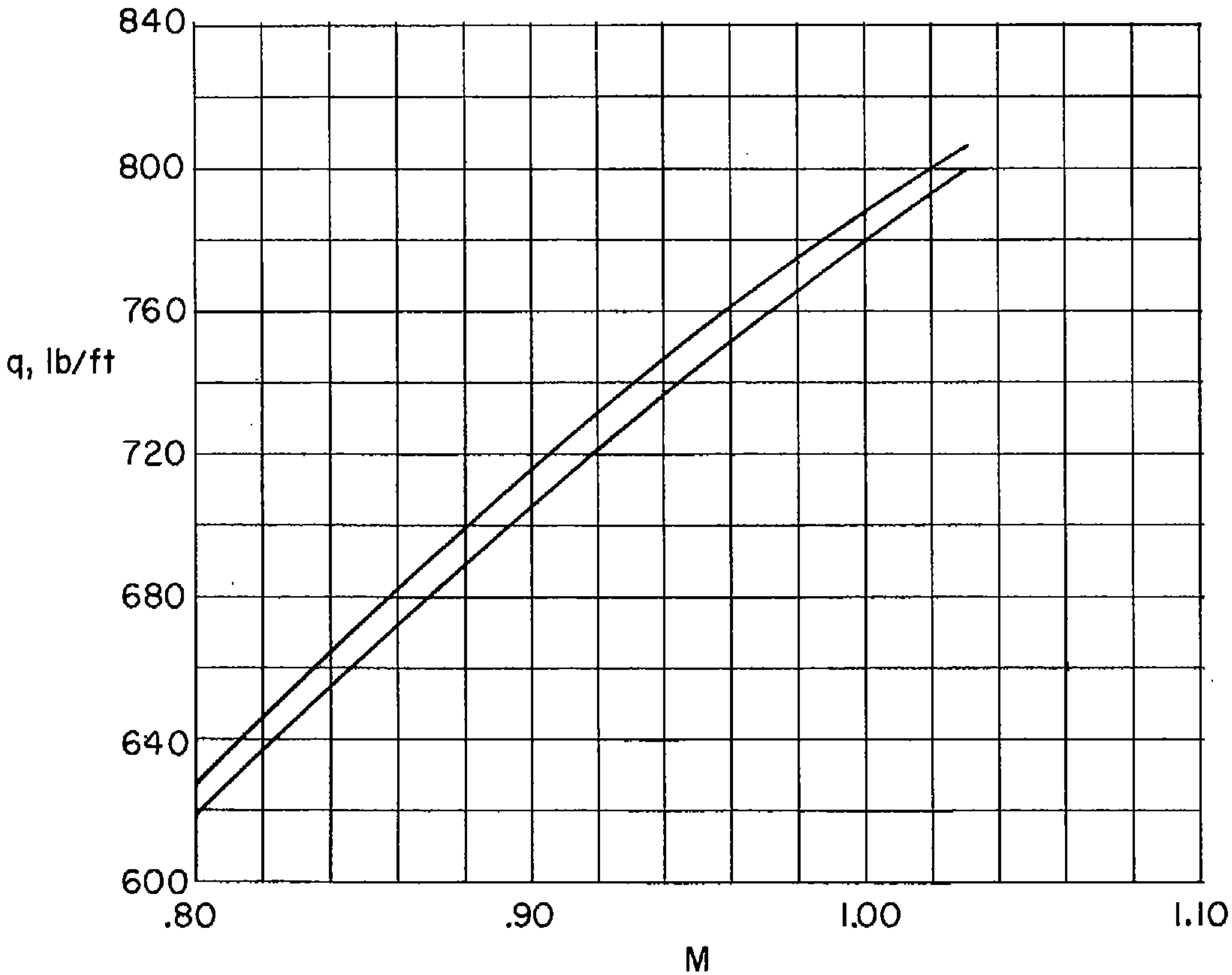
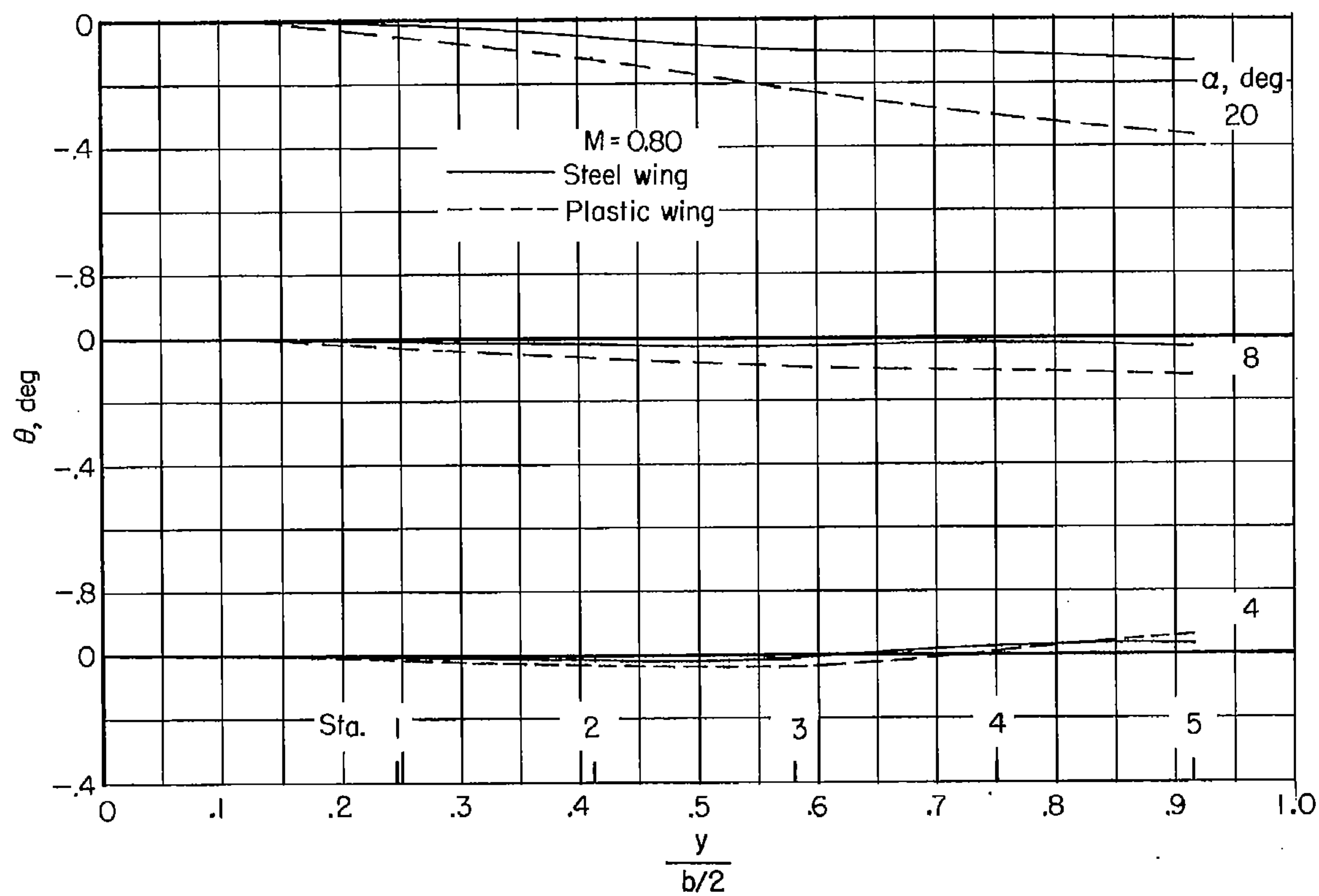
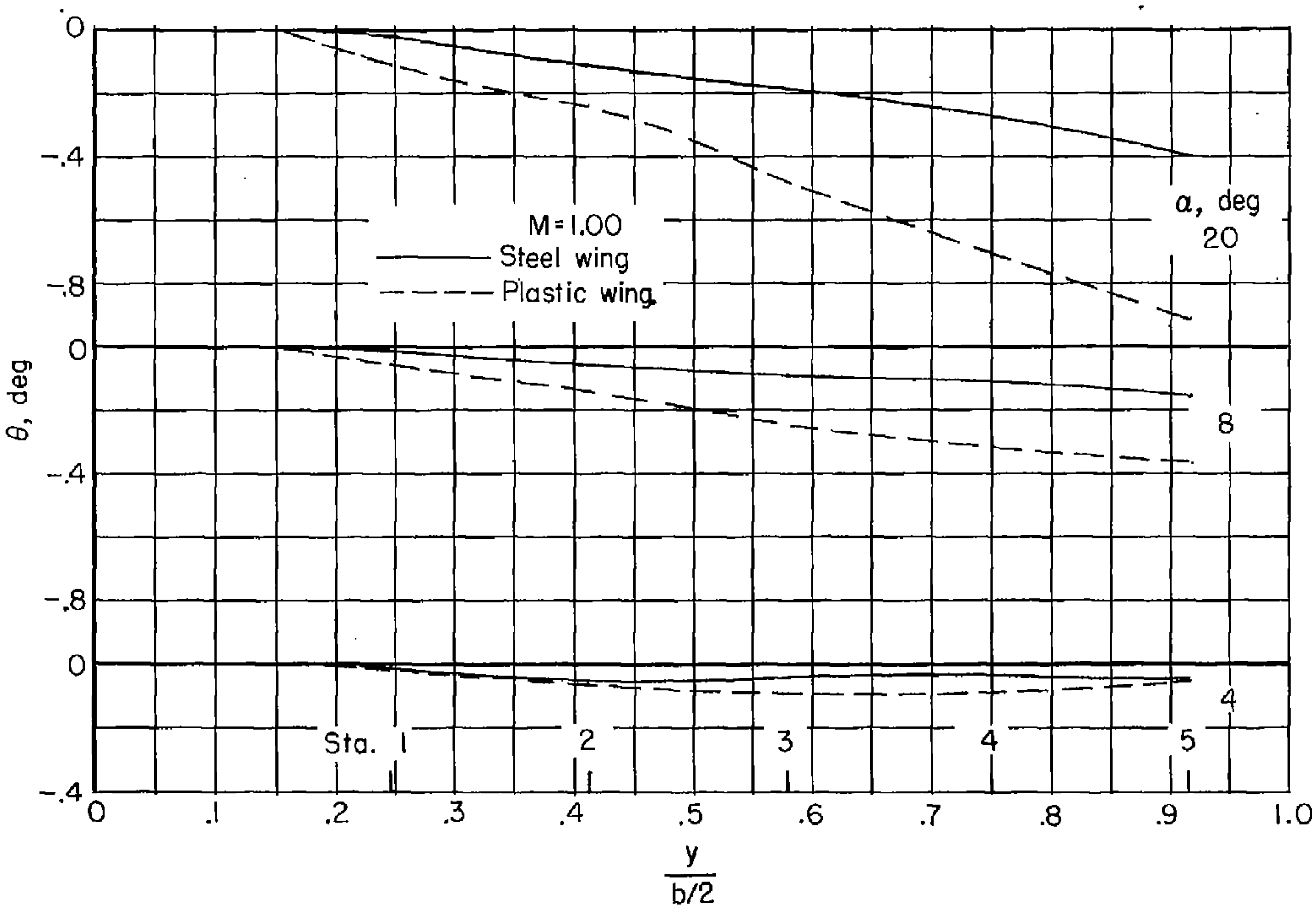


Figure 9.- Range of dynamic pressures for both steel and plastic wings for these tests.



(a) M = 0.80.

Figure 10.- Comparison of the calculated twist distribution due to experimental aerodynamic forces and moments, measured parallel to the angle-of-attack plane.



(b) M = 1.00.

Figure 10.- Concluded.

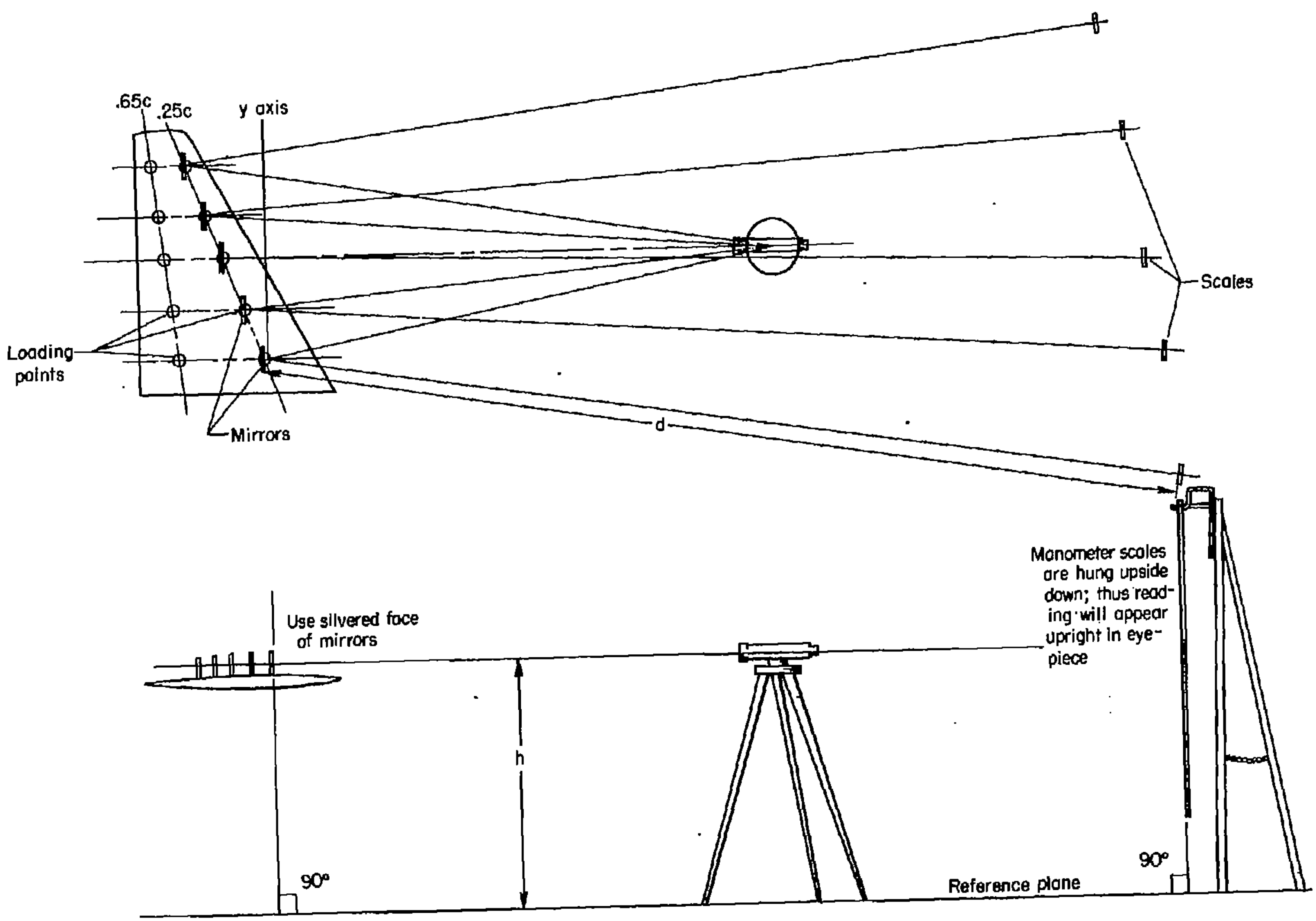


Figure 11.- Typical setup for measuring twist with mirrors.