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DESIGN GUIDE FOR BOLTED JOINTS IN COMPOSITE STRUCTURES

R. L. RAMKUMAR

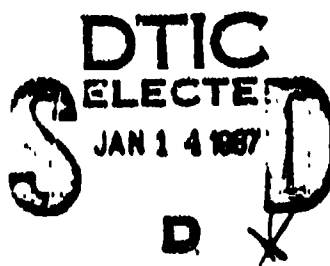
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March 1986

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19. ABSTRACT (Continue on reverse if necessary and identify by block number) A design guide was developed for bolted composite structural joints. The guide includes general design guidelines for the various joint parameters, an analytical design methodology, a description of the analytical design tools, an illustration of the use of corresponding computer codes (SASCJ and SAMCJ), and a listing of the computer codes. The proposed design procedure is purely analytical, and enables the user to rapidly evaluate many different joint concepts for a selected application. When the bolted structure is fabricated using existing (fully characterized) materials, the design requires no complementary test results. Presented analytical design tools are currently restricted to primarily uniaxially loaded joints and fastener arrangements that are currently used in aircraft structures. Also, the bolted joint is assumed to be strength-critical. However, sample fatigue test results are presented to illustrate a durability check on the joint, assuming a simplified fatigue analysis and assuming that fatigue failure is induced by excessive hole elongation. Despite its current restrictions, this guide is the first government document that provides guidance and analytical tools for the design of bolted composite structures.				
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PREFACE

This report was prepared under Contract F33615-82-C-3217, titled "Bolted Joints in Composite Structures: Design, Analysis and Verification," and administered by the Air Force Wright Aeronautical Laboratories. Dr. V. B. Venkayya was the Air Force project engineer, and was assisted by Capt. M. Sobota and Lt. D. L. Graves as program co-monitors. The program manager and principal investigator at Northrop was Dr. R. L. Ramkumar.

This report is a guide for the design of bolted joints in composite structures, and was prepared under Task 4 in the referenced program (Project 2401).

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SECTION 1

INTRODUCTION

Bolted joints are a prime means of load transfer between structural parts in aircraft. Compared to other joining methods (bonding, welding, etc.), mechanical fastening is more reliable, with a potential for improved structural efficiency, maintainability and cost effectiveness. However, bolted joints are a source of stress concentration and could precipitate structural failures if they are designed improperly.

Prior to the initiation of this Northrop/AFWAL program, no analysis was available to be used as an exclusive design tool for bolted parts, especially if they were laminated composites. Consequently, their design has hitherto been based on extensive testing, empirical data and approximate analyses. The analysis developed in this Northrop/AFWAL program eliminates the need for extensive testing and provides a tool for the rapid evaluation of a bolted joint concept. If the structural part is to be fabricated using a characterized material, it eliminates the need for experimental information.

In the following sub-sections, the scope of this design guide is stated, sample bolted concepts are presented, criteria for the design of bolted joints in composite structures are discussed, the proposed design procedure is described, the analytical and experimental requirements for the design procedure are outlined, and its current restrictions are mentioned. In Section 2, general guidelines for the design of a bolted joint in composite structures are presented, along with summary statements on the effects of critical joint parameters. Section 3 presents the computer codes developed in this program for the strength analyses of single and multiple fastener joints in composites (SASCJ and SAMCJ, respectively). Section 4 demonstrates the use of the developed

analysis in predicting the strength of a realistic structural element.

1.1 Scope of the Design Guide

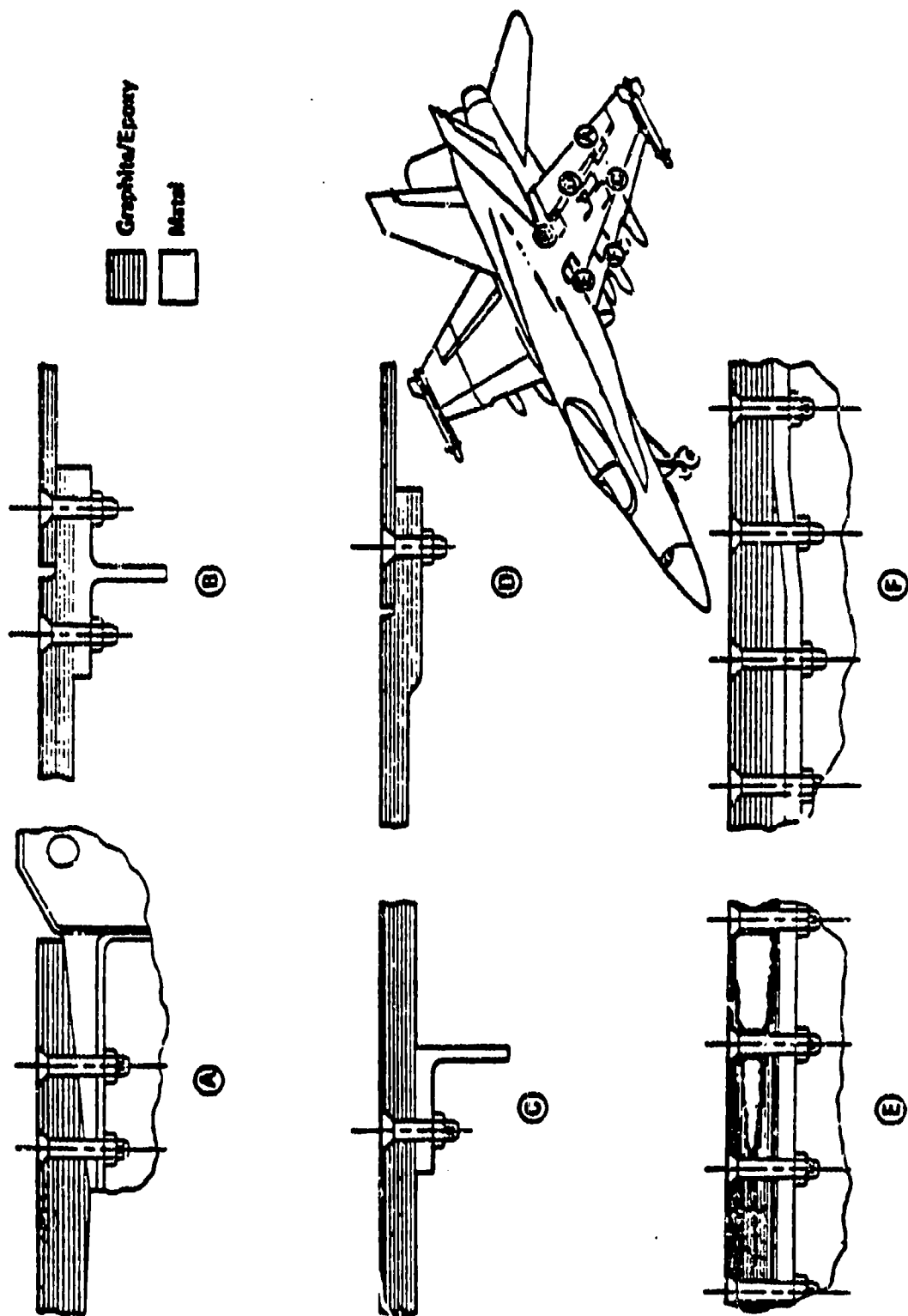
This design guide summarizes the effects of critical parameters on the strength and lifetime of bolted joints in composite structures, and presents general design guidelines. It also describes a test-independent analytical procedure for the strength evaluation of a bolted concept, based on the analyses developed in this program. The reader is familiarized with the computer codes (SASCJ and SAMCJ) that perform these analyses, and an application to a realistic structural bolted joint is demonstrated. This design guide will enable one to perform a rapid analytical evaluation of many joint configurations, and to select an efficient bolting concept. The described computer codes are currently restricted to uniaxial loading, conventionally used fastener spacing and protruding head fasteners.

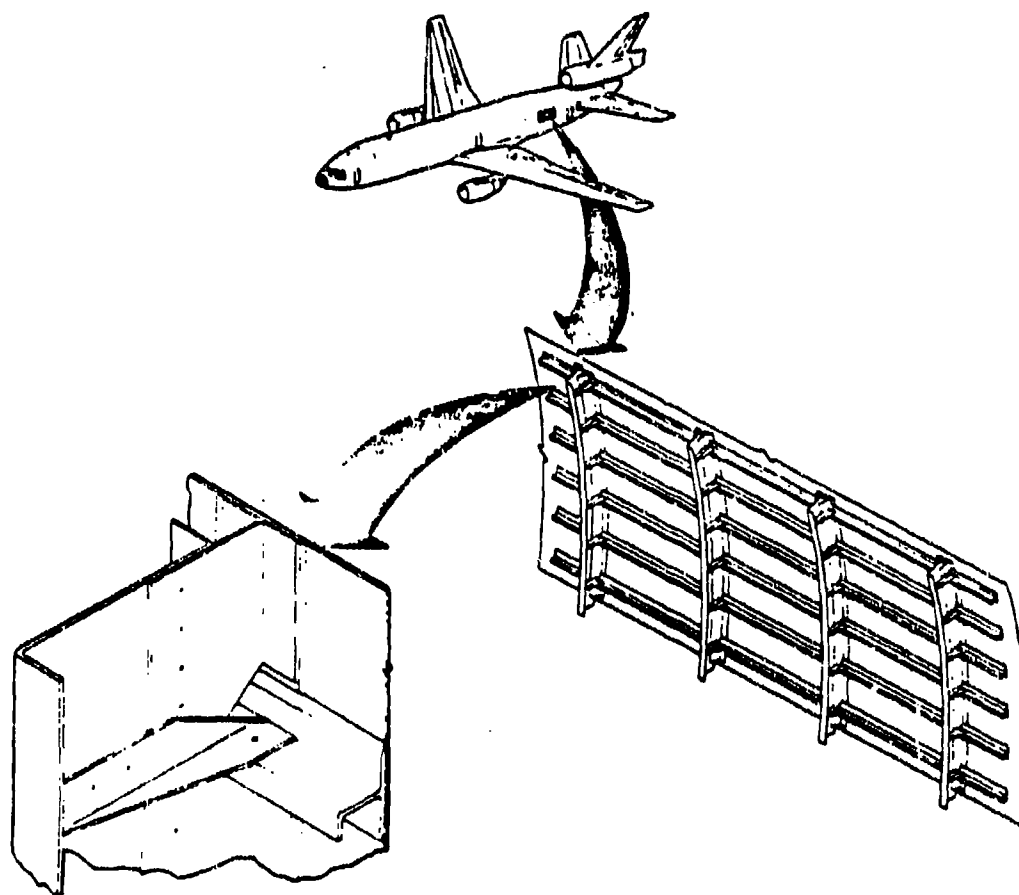
1.2 Sample Joint Configurations

Figure 1 presents six composite-to-metal bolted joint configurations used in the F/A-18A aircraft wing (Reference 1). Figures 2 and 3 present joint configurations used in a typical fuselage structure (Reference 2). A skin-to-root fitting bolted joint in the F-20 horizontal stabilizer is shown in Figure 4. Many bolted joint concepts have been studied recently as potential alternative joining concepts for the F/A-18A wing root section and the F/A-18A vertical tail root section (Figures 5 and 6, respectively). The sample bolted configurations in Figures 1 to 6 illustrate the possible variety in this joining concept.

1.3 Overview of Design Methodology

There are many variables in the design of a bolted joint in composite structures. These include the geometry and the





STIFFENER ATTACHMENT

Figure 2. A Bolted Joint Concept for a Fuselage Structure
(Reference 2).

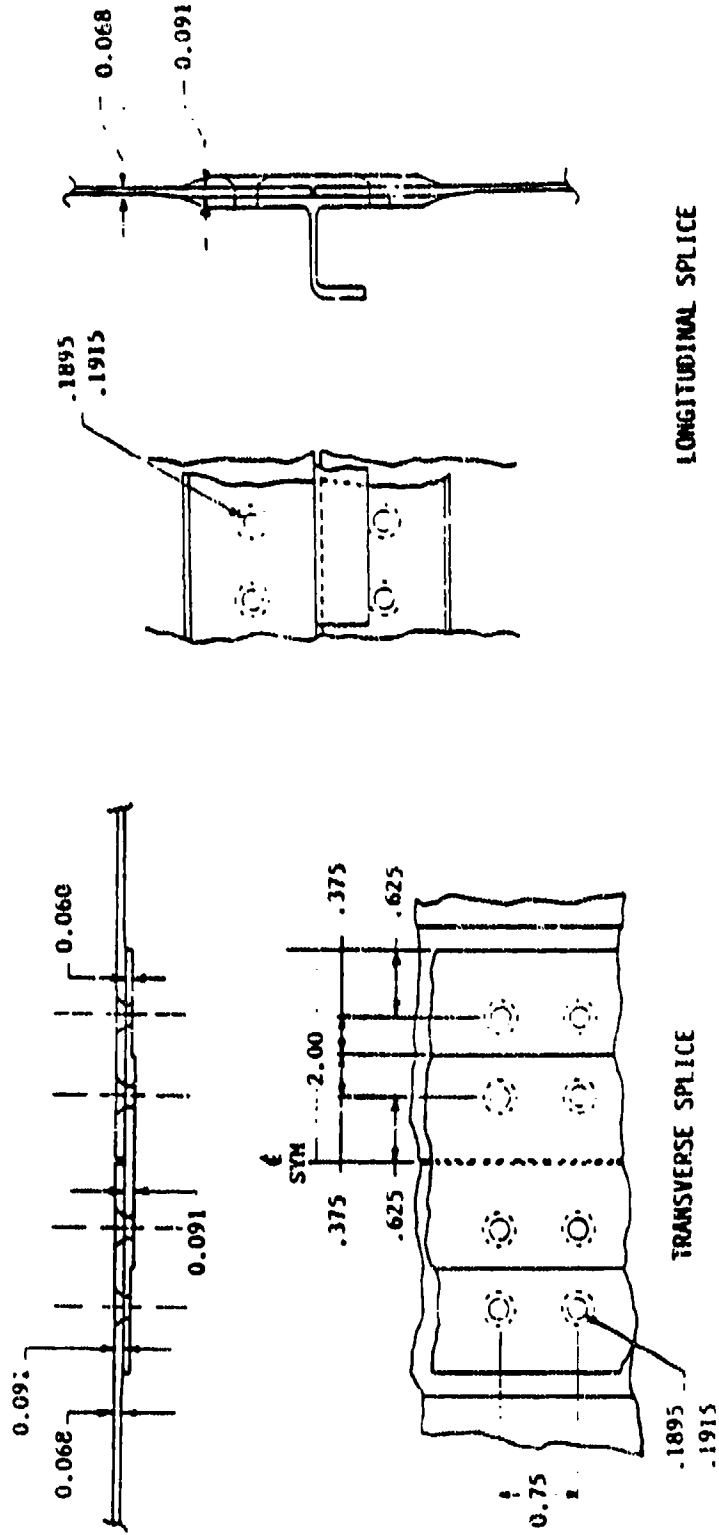


Figure 3. Bolted Joint Concepts for Composite Fuselage Structures (Reference 2).



Figure 4. Skin-to-Root Fitting Joint in the F-20 Horizontal Stabilizer.

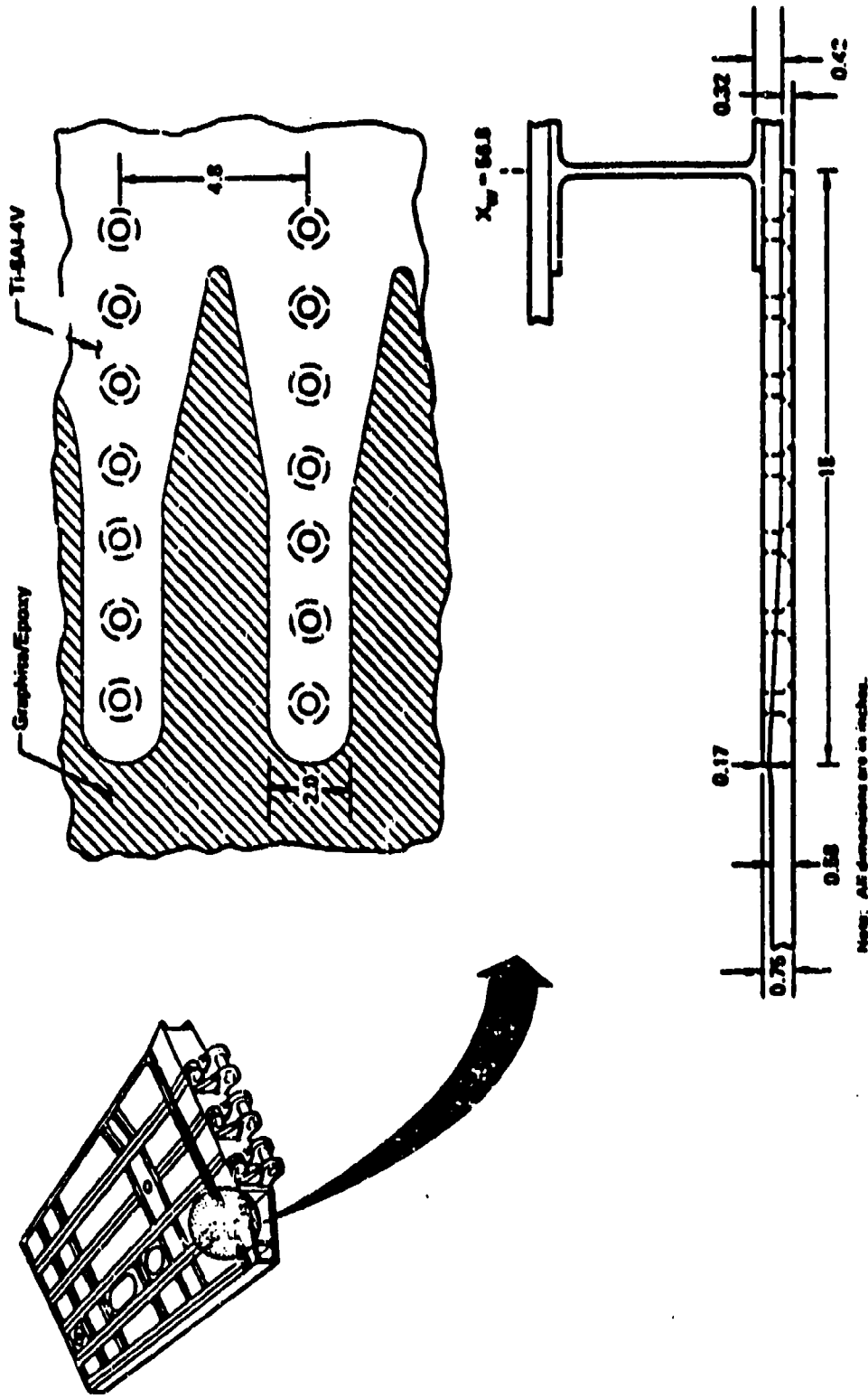
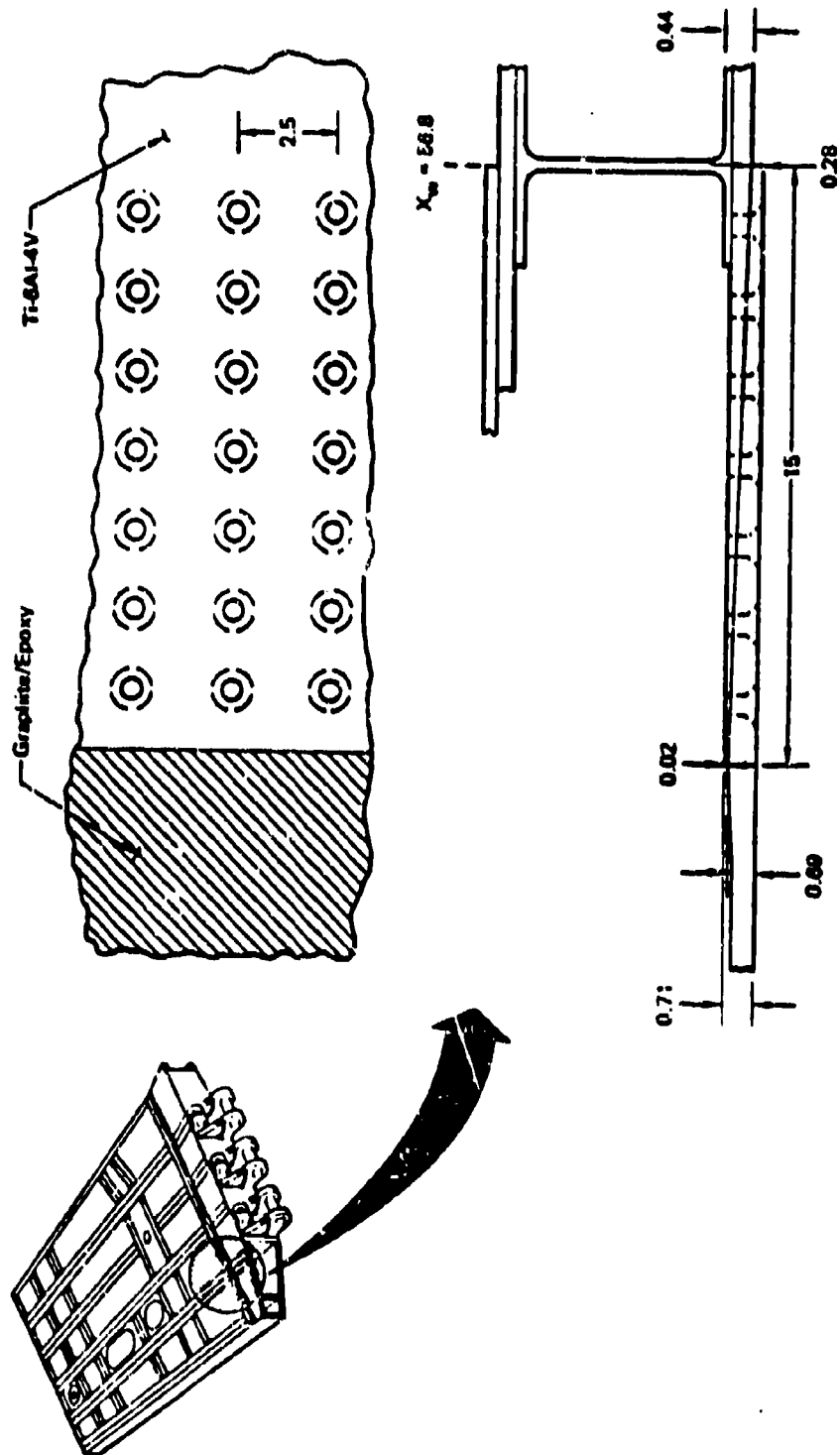


Figure 5 . Alternative Bolted Joint Concept Evaluated for the F/A-18A Wing Root Section (Reference 1).



Note: All dimensions are in inches

Figure 5 . Alternative Bolted Joint Concept Evaluated for the F/A-18A Wing Root Section.
 (Reference 1; Concluded).

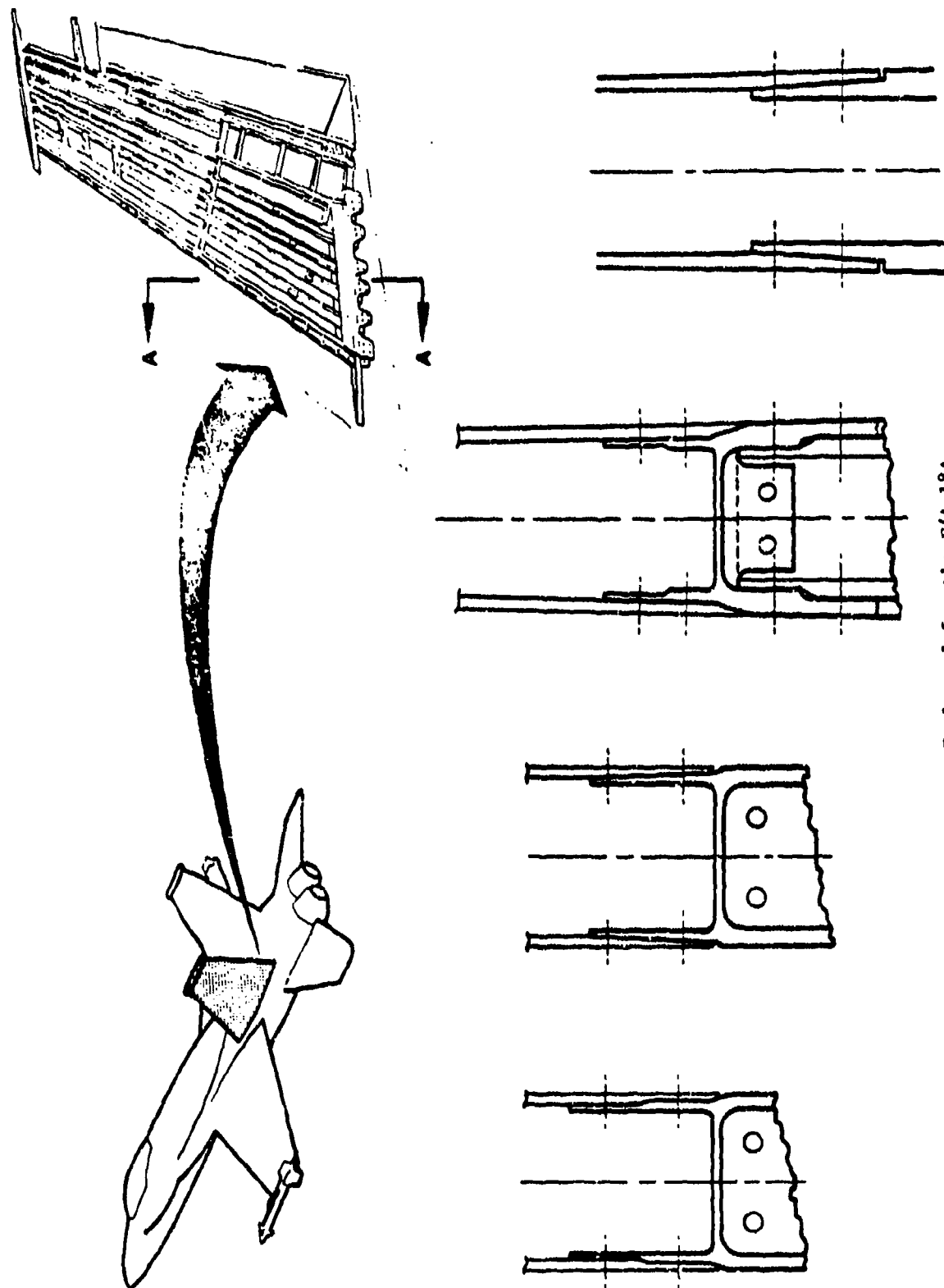


Figure 6. Alternative Bolted Joint Concepts Evaluated for the F/A-18A Vertical Tail Root Section (References 3 to 5).

material properties of the bolted parts, the size and arrangement of the fasteners, the fastener material properties and torque, applied loading and the load transfer mechanism (single versus double shear), etc. The design of a bolted joint involves a parametric study of the effects of the above variables on the joint efficiency, for a specified loading condition. A preliminary analysis of a structural component, based on conventional assumptions, yields the general biaxial loading transferred at the joint location (see Figure 7). The design procedure recommended in this guide assumes a predominantly uniaxial loading at the joint location.

The design of a uniaxially loaded joint in composite structures may be performed using the analyses developed in this Northrop/AFWAL program. Section 3 describes the use of the SASCF and the SAMCF computer codes for the strength prediction of single and multiple fastener joints in composites, respectively. The SASCF code predicts the strength of joints when a single fastener transfers the applied load between the bolted plates. This analysis accounts for material nonlinearity in the bolted plates, the non-uniform fastener load distribution in the thickness direction of the bolted plates, and the progression of ply-level failures based on a choice among a few failure criteria. The SAMCF code predicts the strength of plates bolted together by one or many fasteners. It computes the magnitude and orientation of the load at every fastener location, the applied load level for averaged stress components to reach critical levels at fastener and cut-out locations, the failure value of the applied load, the failure location and the failure mode (net section, shear-out or bearing). Failure predictions are made at the laminate level using average stress failure criteria.

The proposed design procedure involves the use of the developed analyses to evaluate the effect of joint variables on joint efficiency. If the bolted plates are fabricated using characterized materials, the joint design is tested-independent.

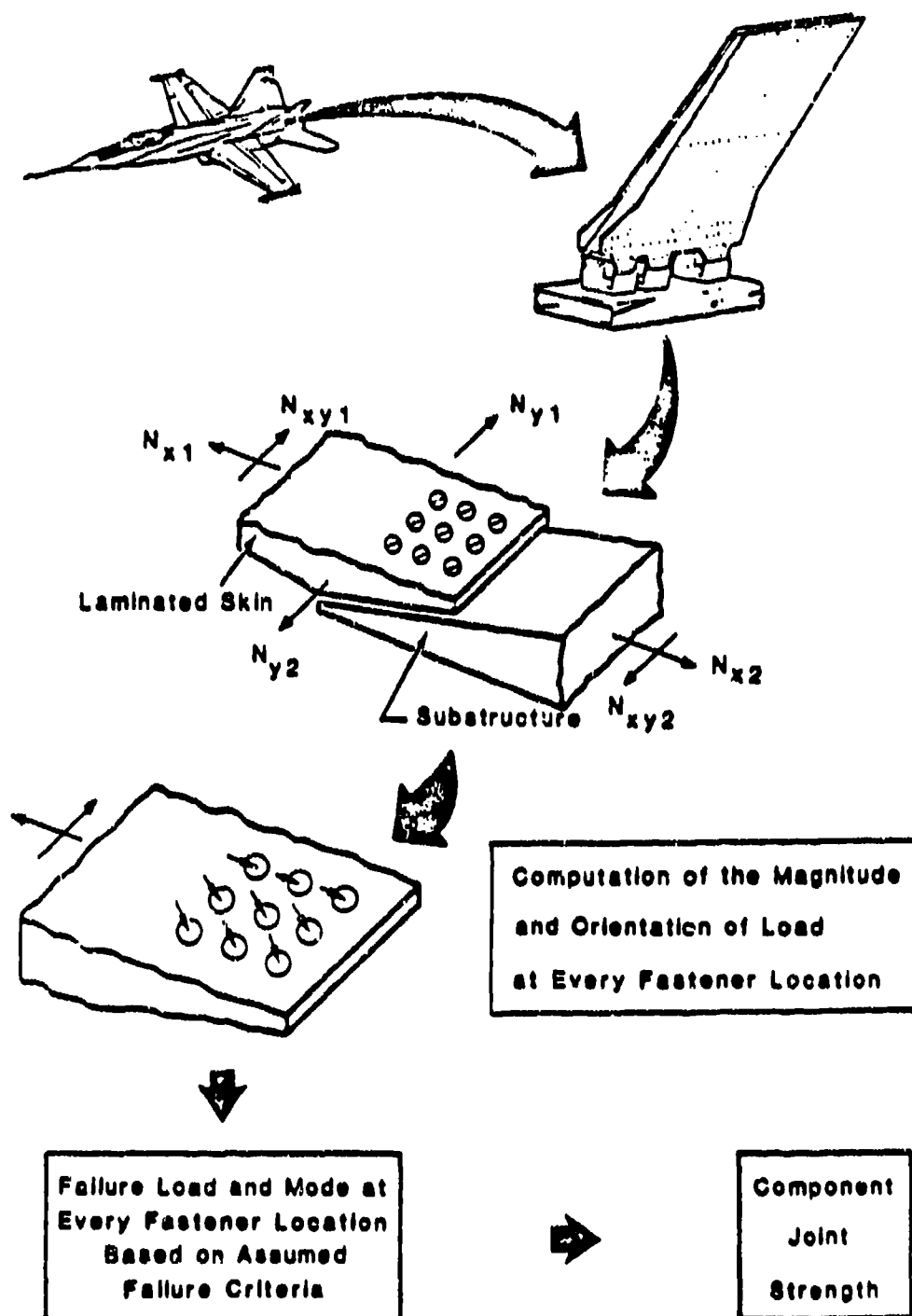


Figure 7. Overview of the Strength Analysis of Bolted Structures.

Candidate bolted joint concepts are selected following the general guidelines outlined in Section 2. The fastener size and arrangement (spacing between fasteners), the geometry of the bolted plates, the load transfer mechanism, etc. are varied without violating the constraints imposed by the structural application. The strength and durability of each bolted joint concept, along with its impact on manufacturing costs and maintenance, are evaluated to establish joint efficiency. An efficient bolted joint concept can thus be designed using a purely analytical tool on a finite number of concepts that are selected in accordance with established guidelines.

1.4 Analytical Requirements

The design of a bolted joint for composite structures requires the analyses developed in this Northrop/AFWAL program (References 6 and 7). The analysis of plates bolted together by a single fastener may be performed using the SASCF (Strength Analysis of Single Fastener Composite Joints) or the SAMCF (Strength Analysis of Multiple Fastener Composite Joints) computer code. Plates bolted together by many fasteners are analyzed using SAMCF computer code. Section 3 presents a brief description of these analyses. The reader is referred to References 6 and 7 for further details.

1.5 Test Requirements

A test-independent, purely analytical design tool has been developed to design a bolted joint for composite structures that are fabricated using characterized materials. The engineering properties (Young's moduli in the fiber and transverse directions, major Poisson's ratio and the shear modulus in the fiber coordinate system), the strengths or failure strains (under tension, compression and shear), and the failure parameters for the assumed failure criteria (characteristic distances for net section, shear-out and bearing failure predictions using the average stress

failure criteria, for example) are known for a characterized composite material (lamina). Tests required to obtain the above material properties must be performed on a new (uncharacterized) material system, prior to designing bolted joints for structural parts made from this material. When previously characterized materials are used in the bolted plates, the test requirements are nil for the design of an efficient bolted joint concept.

1.6 Current Restrictions

The design of bolted joints in composite structures is influenced by the current restrictions in the developed analytical tools. The primary restrictions are listed below:

- (1) The developed strength analyses (SASCJ and SAMCJ computer codes) do not account for countersunk fastener effects.
- (2) SASCJ and SAMCJ contain a stress analysis that approximates the fastener/plate contact problem by an assumed radial stress distribution.
- (3) SASCJ AND SAMCJ are restricted to a uniaxial applied loading, in tension or in compression.
- (4) The prediction of the durability of a joint is restricted to the incorporation of the bearing stress at critical fastener locations into experimentally obtained curves for joint life.
- (5) SAMCJ restricts the user to rectangular element geometries and currently used fastener spacing and arrangement.

Despite the above restrictions, the developed analyses and the proposed design procedure mark a significant improvement

over the state-of-the-art with respect to the design and analysis
of bolted joints in composite structures.

SECTION 2

GENERAL DESIGN GUIDELINES AND JOINT VARIABLES

The design of boted joints in composite structures involves the definition of many variables. The major design considerations are listed below:

- (a) The loads that must be transferred from one part to another.
- (b) The load transfer location in the structure.
- (c) Geometric constraints, if any, at the load transfer location.
- (d) Fastener type, size and arrangement.
- (e) The environmental range the joint will be exposed to.
- (f) The effect of the joint concept on structural efficiency and reliability.

The following sub-sections discuss the primary variables that influence the design of boited joints in composite structures. Design guidelines corresponding to the discussed joint parameters are highlighted within the sub-sections.

2.1 Joint Location in the Structure

The location of the joint in a structure influences the selection of the joint variables significantly. Design guidelines pertaining to selected joint locations are presented below:

- (a) When aerodynamic surfaces in an aircraft structure are joined to substructural parts, or segments of a surface are joined together, the requirement of a smooth outer moldline should not be

violated. The use of protruding head fasteners on such surfaces, or the presence of any other geometric discontinuity (step) at the joint location, will adversely affect the lift distribution on these surfaces and their aerodynamic performance.

On aerodynamic surfaces, fasteners must be installed to be flush with the surface, without exposed fastener heads, and joined members must retain a smooth outer moldline.

(1)

(b) When structural members are joined together in fuel-containment areas, measures must be taken to preclude leakage of the fuel and service-related hazards. The use of metallic fasteners on the outer surface, for instance, introduce the threat of arcing within the fuel cell in the event of a lightning strike. In designing joints for these locations, special consideration must be given to the mentioned sealing requirements.

In fuel containment areas, joints must be sealed to be leak-proof. Fasteners must also be sealed to prevent arcing within the fuel cell in the event of a lightning strike.

(2)

(c) When bolted joints are designed for structural regions with limited or restricted access, special fastener types have to be used.

In areas of restricted accessibility, blind fasteners must be used.

(3)

(d) When a laminated part is bolted to a metallic substructure, the threat of joint corrosion must be considered.

In composite-to-metal joint locations, corrosion barriers like fiberglass layers must be used.

(4)

2.2 Joint Configurations

Selected joint configurations are significantly influenced by their structural locations. Figures 1 to 6 present typical structural joint configurations in current aircraft. Figure 8 presents the localized structural joint configurations along with their equivalent configurations that are analyzed. The configurations that transfer loads in single shear introduce localized bending effects that could adversely affect the strength and durability of the joint. Stepped lap and scarf configurations involve thickness changes that provide an additional design variable (layup) in bolted laminates.

2.3 Joint Loading

Structural joints are designed to be effective over their design lifetime, when subjected to the anticipated design spectrum fatigue loading. The durability considerations for structural joints are discussed in Section 4. This design guide emphasizes the strength analysis of a bolted joint, and presents computer codes that perform it. The reader must supplement the joint design based on a strength analysis with a durability check, using information similar to that presented in Section 2.9. The effect of joint loading is discussed further below, at three levels -- structural, among fastener rows, and at an isolated fastener location.

2.3.1 Joint Loads at the Structural Level

Joint loads at the structural level fall into two basic categories -- inplane loads and out-of-plane or bending loads. Figure 9 presents some possible inplane load conditions in typical wing skin-to-substructure attachments. The analyses developed in this Northrop/AFWAL program, and described in Section 3, assume that the joint at each location is subjected to a predominant unidirectional load. Figure 9 illustrates that this assumption will

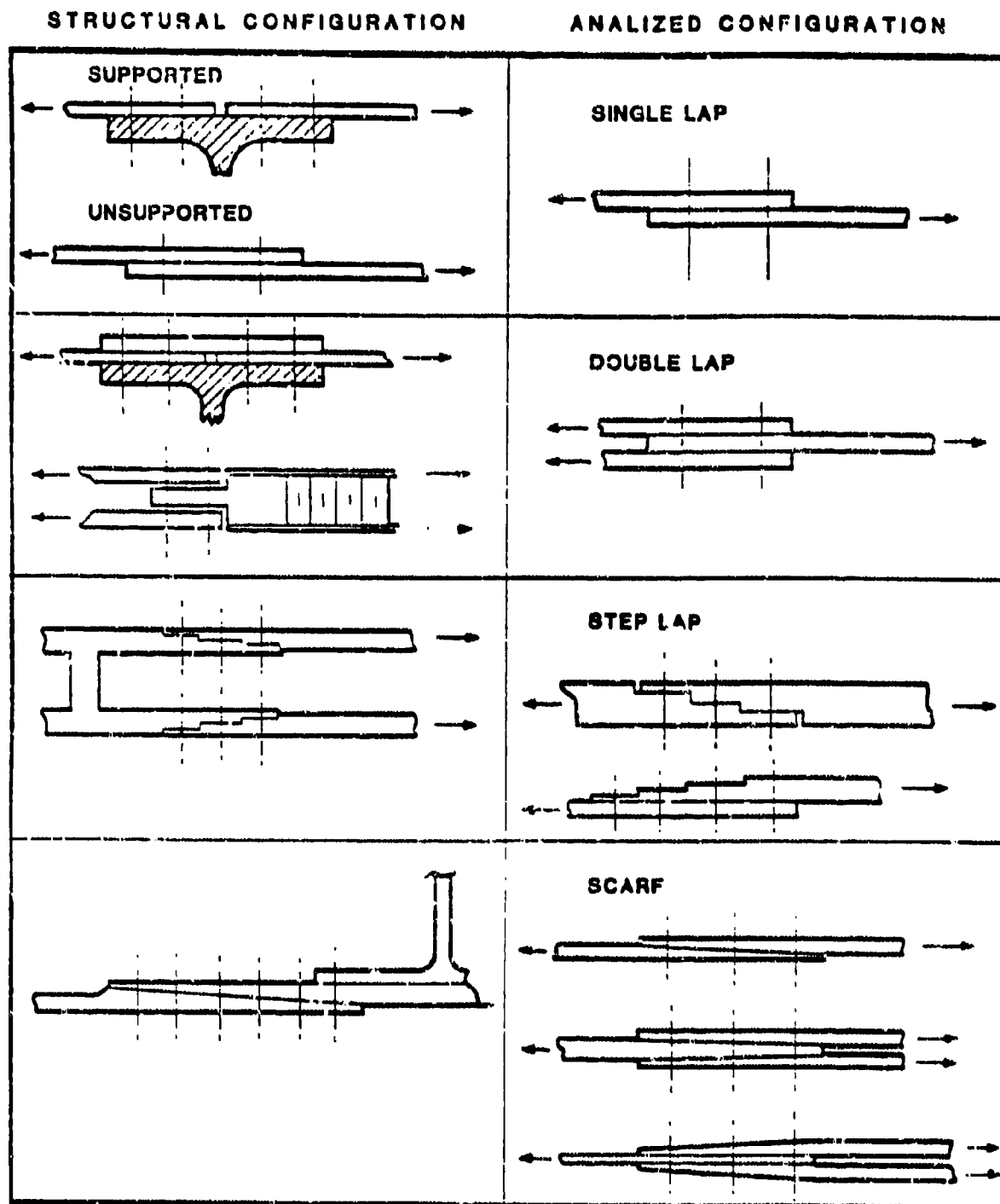


Figure 8. Structural and Analyzed Bolted Joint Configurations.

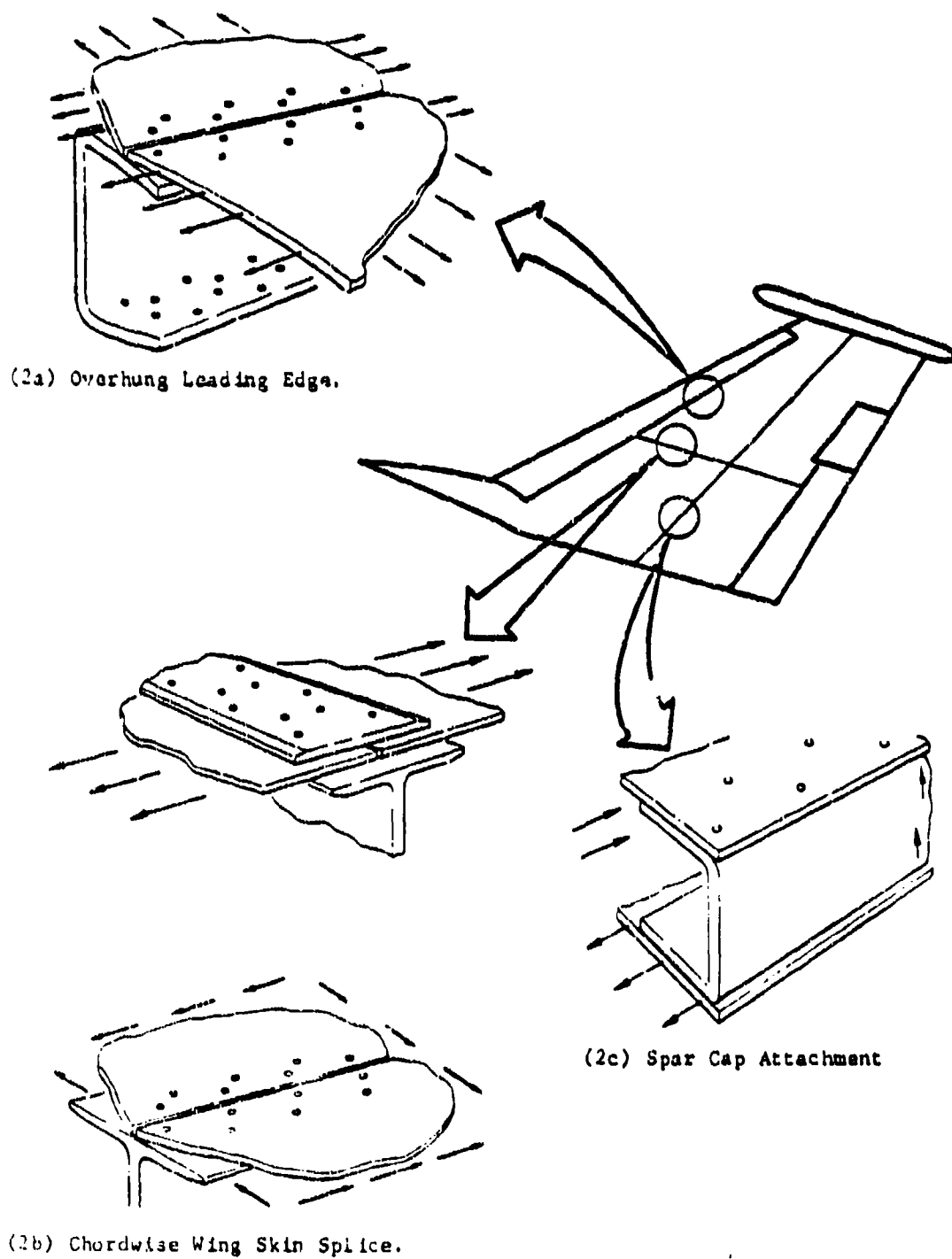


Figure 9. Inplane Loads in Typical Wing Skin-to-Substructure Attachments.

not be valid at some locations.

Figure 10 presents sample situations where considerable out-of-plane (bending) loads are introduced at the joint location. This is inherent in single shear load transfer configurations (see Figure 8), and adversely affects joint strength and durability. If one of the bolted plates is very stiff compared to the other, the deleterious effects of load eccentricity in a single shear configuration are minimized. In double-shear load transfer configurations (see Figure 8), the out-of-plane loads are reduced to a negligible level.

Single - shear load transfer joint configurations introduce out-of-plane (bending) loads that could significantly reduce the strength of the joint. When one of the bolted members is very stiff, the effect of the out-of-plane loads is minimized.

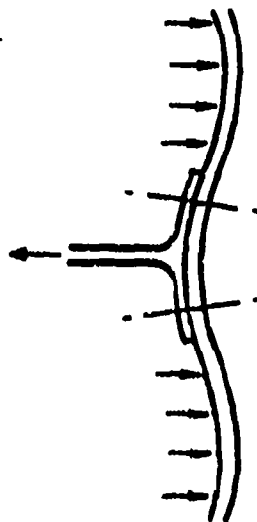
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Double - shear load transfer joint configurations essentially introduce inplane loads in the bolted plates.

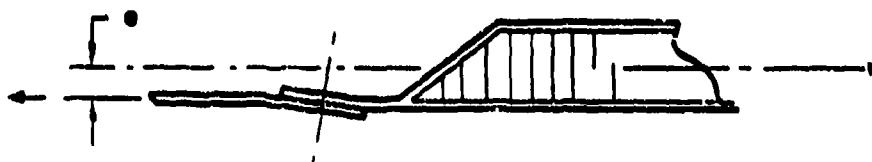
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2.3.2 Load Distribution Among Rows of Fasteners

Assuming a unidirectional applied load, the fasteners in a row are arranged perpendicular to the load direction. Joint configurations affect the distribution of the applied load among the various rows of fasteners in a joint, and the distribution of the row-wise load fraction among the fasteners in any row. Hitherto, the fasteners in a row have been assumed to carry equal loads, and only the row-wise load distribution has been analytically predicted. The SAMCJ code developed in this Northrop/AFWAL program overcomes this limitation, and predicts the two-dimensional load distribution (magnitude and orientation of fastener loads at all locations) for a selected fastener pattern.



**A. OUT-OF-PLANE JOINT LOADING DUE TO INTERNAL PRESSURE
(e.g., FUEL PRESSURE, FUSELAGE CABIN PRESSURE, ETC.)**



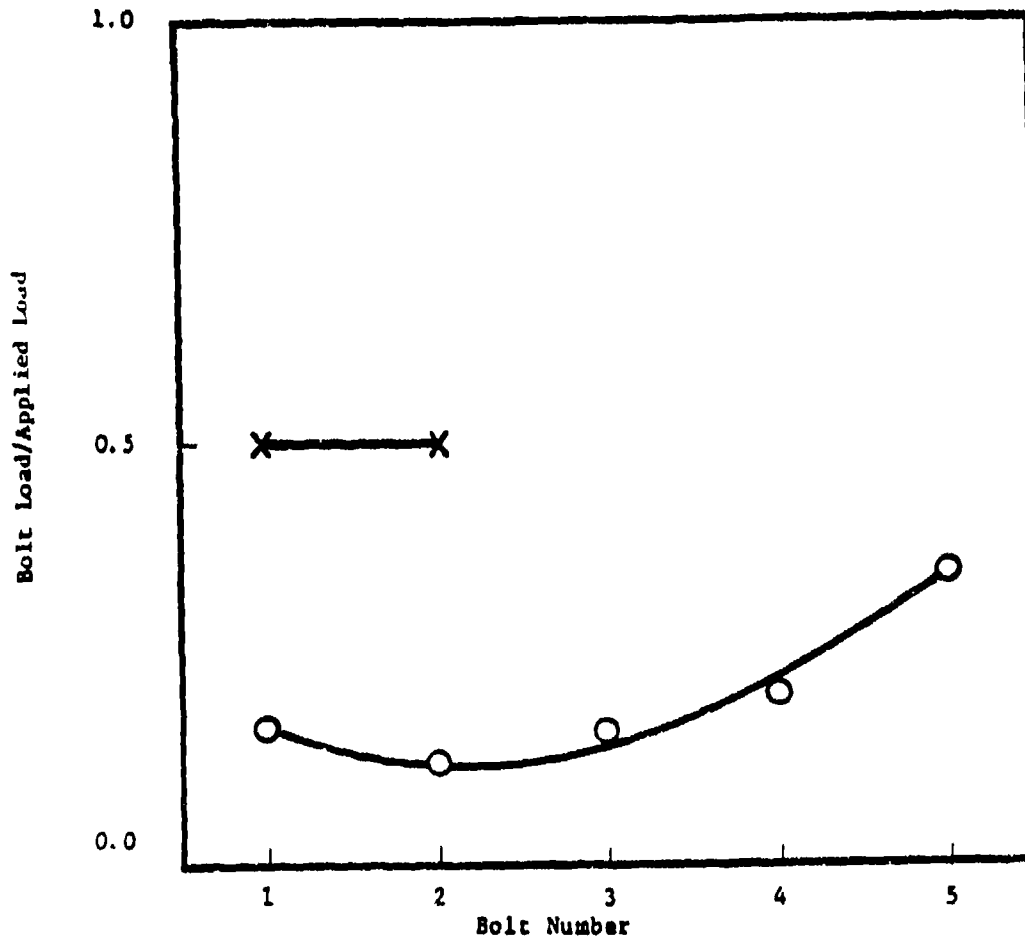
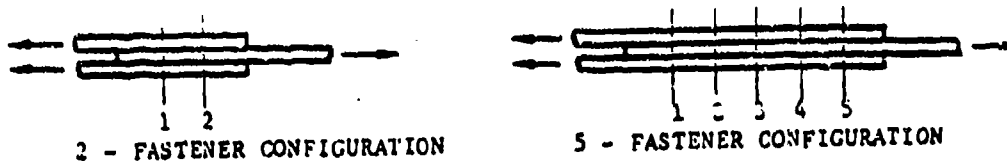
B. OUT-OF-PLANE JOINT LOAD DUE TO LOAD PATH ECCENTRICITY

**Figure 10. Sample Joint Configurations that Introduce Significant
Out-of-Plane Loads at the Joint Location.**

Figure 11 presents the load distributions for two and five fastener, double shear joint configurations tested in this program (References 7 and 8). The bolted plates in Figure 11 were uniform in thickness. Figure 12 illustrates how the load distribution among four rows of fasteners can be varied by changing the joint configuration. In the strongest configuration (4), a combination of tapering and reinforcing of the splice plates minimizes the bearing load where the by-pass load is the largest (station 1), and maximizes the bearing load where there is no by-pass load (station 4). The plate width-to-bolt diameter ratio (W/D) is 5 at station 1, and 4 at stations 2 and 3. A larger bolt is used at station 4 ($W/D=3$). This results in a reduction of the bearing stresses at stations 2 to 4, and the strongest configuration (see References 9 and 10).

In bolted metallic plates, the fastener load distribution is similar to those shown in Figure 11 for low values of the applied load. But, as the applied joint load increases toward the failure value, yielding will occur at peak fastener load locations. This causes the incremental applied load to be carried by the remaining fasteners, generally resulting in a uniform fastener load distribution near failure. For the five fastener configuration in Figure 11, for example, every fastener will carry one-fifth of the applied load at failure. However, laminated plates generally exhibit a linear elastic and brittle behavior, with negligible ductility or yielding. The non-uniform load distribution among rows of fasteners in composite laminates, therefore, remains non-uniform at failure. This reduces the failure load level if the peaks in the load distribution are not accompanied by appropriate thickness tapering and other changes in the joint configuration. Joint efficiency is determined by the overall load-carrying capability of the joint.

The load distribution among rows of fasteners in a bolted laminate generally remains non-uniform at the failure load level, in contrast to what is



Note: Double-shear load transfer between 50/40/10, AS1/3501-6 graphite/epoxy laminate and aluminum using 5/16-inch diameter, protruding head steel fasteners torqued to 100 in-lb ; static tension; RTD.

Figure 11. Fastener Load Distribution in the Laminated Plate for Two Double-Shear Configurations (References 7, 8).

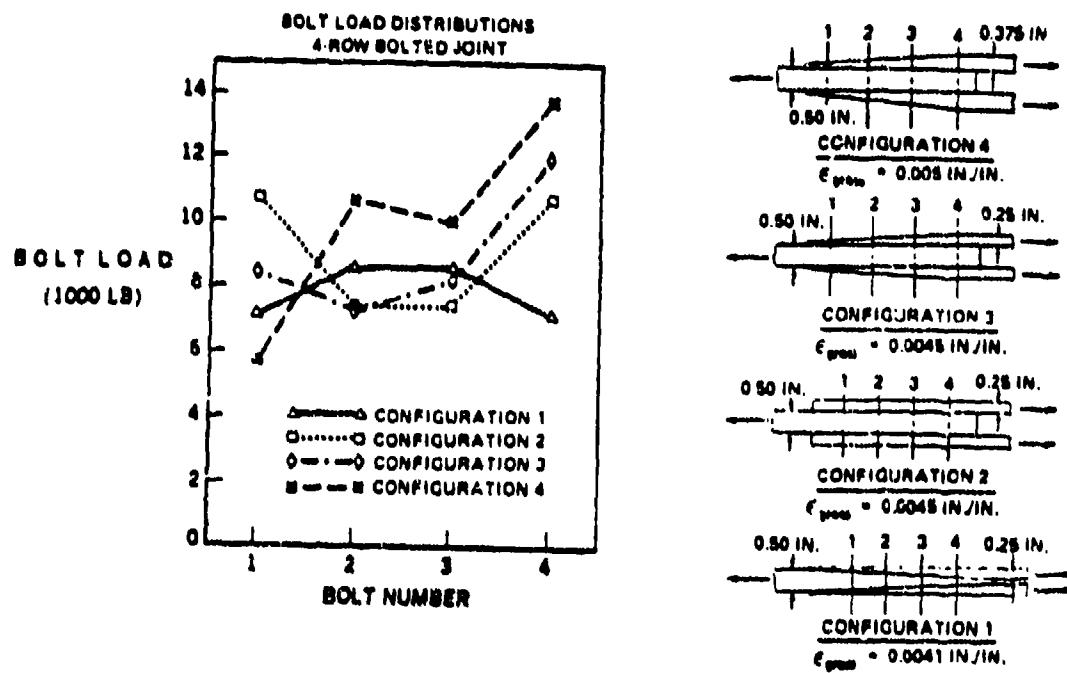


Figure 12. Effect of Joint Configuration on Fastener Load Distribution (Reference 9).

assumed in bolted ductile metals. This adversely influences the failure load for bolted laminates, unless thickness tapering or other configuration changes are introduced.

(7)

2.3.3 Bearing and By-Pass Loads at an Isolated Fastener Location

Figure 13 illustrates the bearing and by-pass loads, and the interaction between them, at an isolated fastener location in a bolted laminate. The failure of the bolted plate is generally assumed to coincide with the failure at the most critical fastener location. The identification of the most critical fastener location requires a knowledge of the load distribution among the fasteners, and an understanding of the interaction between the bearing and by-pass loads at a fastener location (Figure 13).

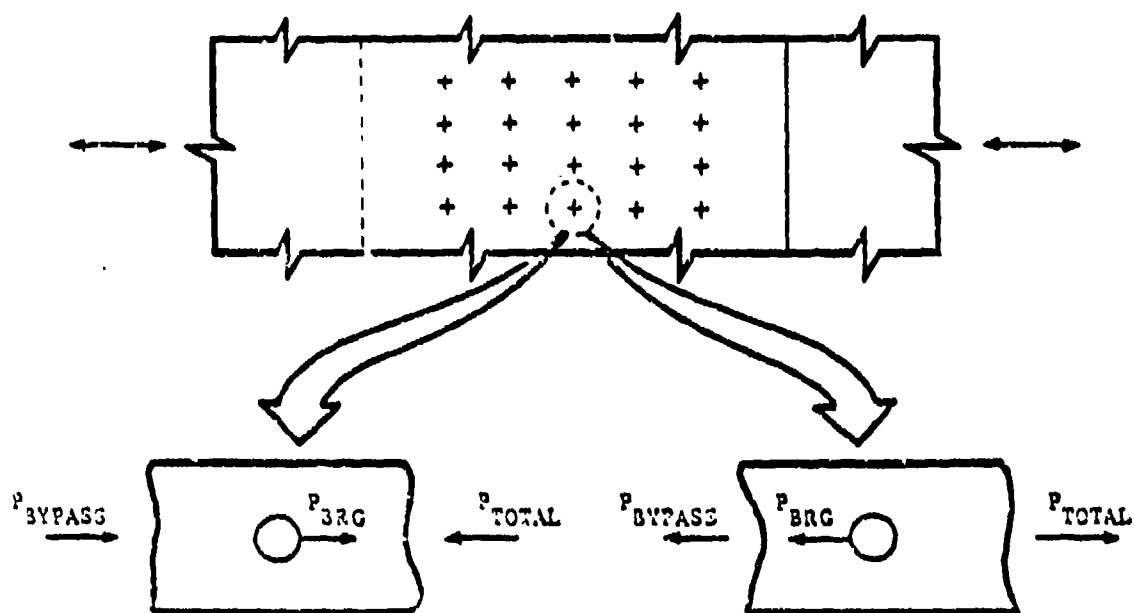
In ductile metals, minimal interaction is assumed between the bearing load and the by-pass load. However, in composites, a significant interaction has been demonstrated between the two loads under tensile loading (see Figure 13). Only a minimal interaction is observed under compression (see Figure 13). The open hole and bearing strengths of laminates (under tension and compression) are dependent on the laminate layup. The bearing stress at failure is also dependent on the edge distance (geometry) of the bolted laminate when its layup contains more than 40% of 0-degree plies.

Under tensile loading, an increase in the bearing stress reduces the by-pass stress value at failure in bolted laminates.

(8)

Under compressive loading, a minimal or negligible interaction between the bearing and by-pass loads is observed in bolted laminates

(9)



Bearing Stress at Failure

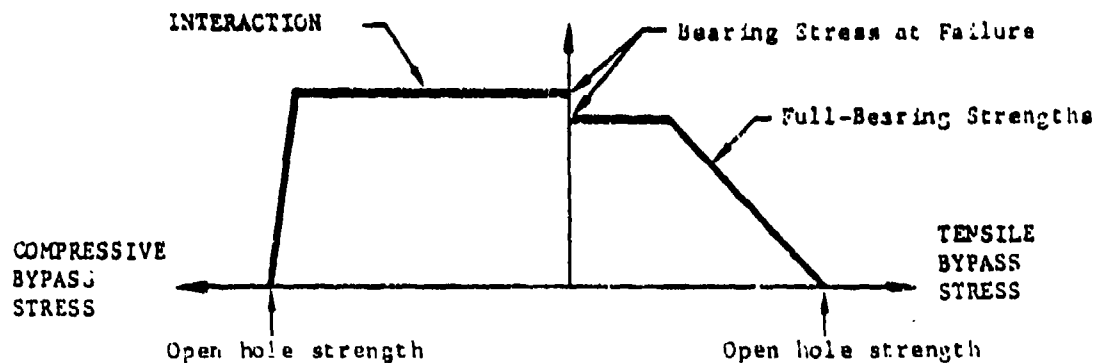
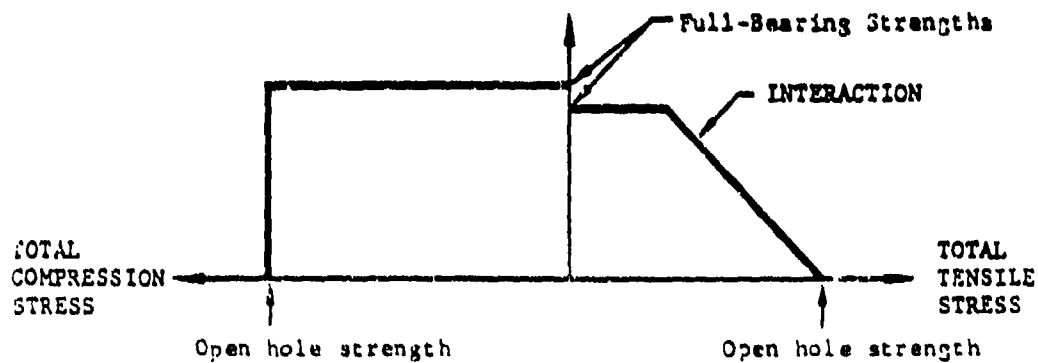


Figure 13. Interaction Between Bearing and by-Pass loads at a Fastener Location.

2.4 Failure Modes in Bolted Laminates

Bolted laminates exhibit one or more among a variety of failure modes, depending on their layup and geometry, the fastener type and the loading configuration. Figure 14 presents the basic failure modes observed in bolted laminates and possible fastener or fastener-induced failures. In the design of bolted laminates using the SAMCJ computer code, only the net section, shear-out and bearing modes of failures in the laminate are considered, and fastener-related failures are assumed to be precluded a priori. Net section and shear-out failures lead to catastrophic joint failures, while bearing failure is generally non-catastrophic. Critical, highly-loaded structural joints should, therefore, be designed to fail in a bearing mode.

Ensuring that fastener-related failures are predicted, highly-loaded structural joints must be designed to fail in a bearing mode to avoid the catastrophic failures induced by net section and shear-out modes of failure.

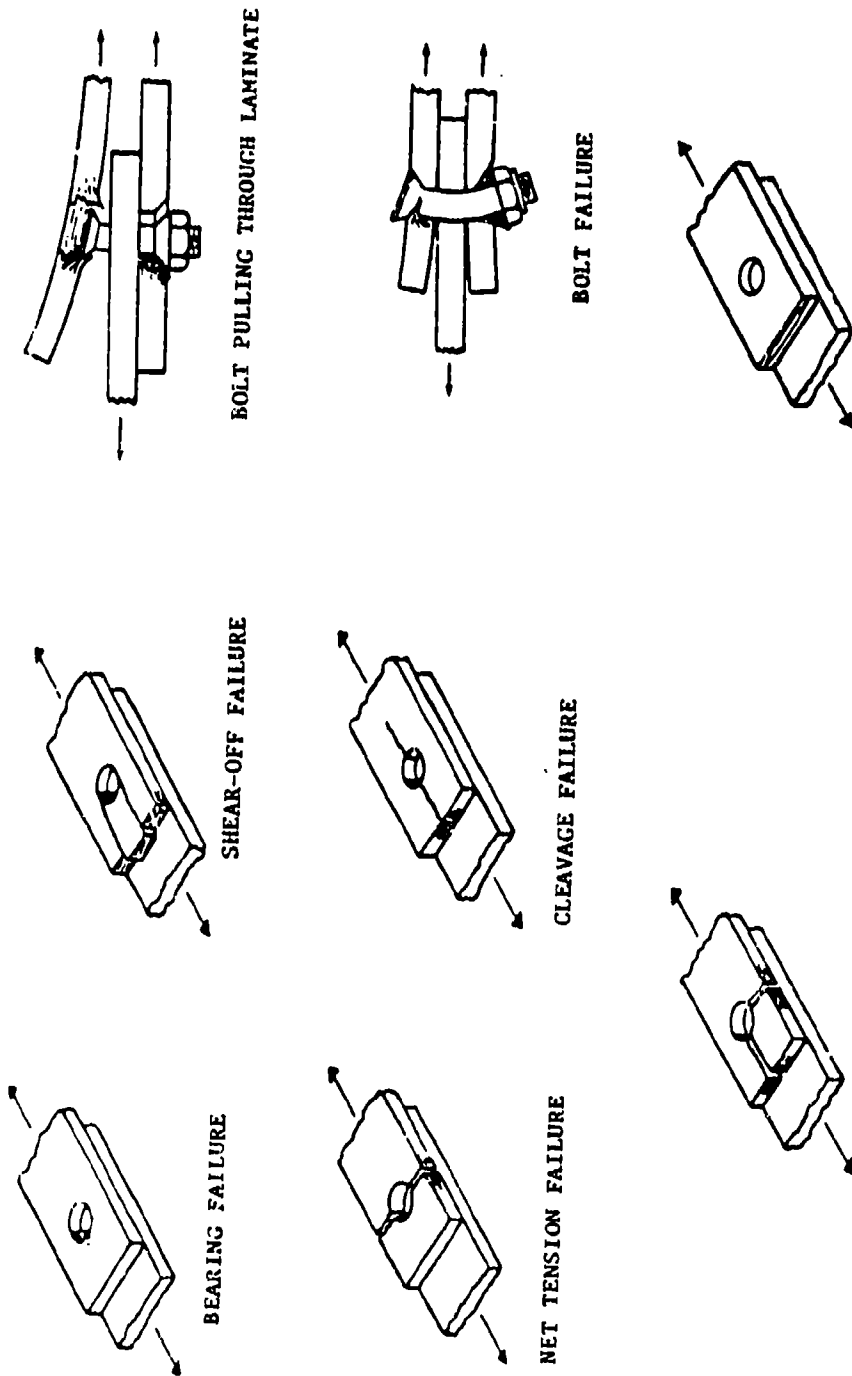
(10)

2.5 Fastener Type, Material and Installation Variables

In selecting fasteners for bolted composite structures, many variables have to be considered. These are briefly discussed below.

2.5.1 Fastener Type

Fasteners are available in different forms for different applications, and are broadly classified as protruding head fasteners or countersunk (flush head) fasteners. Countersunk fasteners generally have a 100 degree head angle, and are referred to as tension head or shear head fasteners based on the countersunk depth. Special fastener types include hi-lok, big foot, Jo-bolt, Eddie-bolt, k-Lobe, composite fasteners, etc. (Reference 11).



COMBINATION FAILURE
 Figure 14. Basic Failure Modes in Bolted Laminates and Fastener-Related Failures.

The joint location influences the selected fastener type and introduces sealing requirements (see Section 2.1). The three guidelines corresponding to this are repeated below:

Flush head (countersunk) fasteners should be used on aerodynamic surfaces to maintain contour smoothness. (1)

In fuel containment areas, the fastener locations must be sealed to be leak-proof and to prevent arcing in the fuel cell in the event of a lightning strike. (2)

In areas of restricted accessibility, blind fasteners must be used. (3)

Tension head countersunk fasteners have a larger countersunk depth than shear head countersunk fasteners. Tension head fasteners, therefore, rest over a larger area of the bolted plate, and carry the load primarily in tension along the fastener axis. Shear head fasteners have a smaller countersunk depth, and carry the load primarily in shear over the fastener cross-section. Consequently, tension head fasteners are capable of carrying larger loads than shear head fasteners. But, when the countersunk depth exceeds approximately 70% of the bolted plate thickness, the fastener effectiveness is reduced due to the local "knife edge" effect, influencing the selection of the fastener type.

Tension head fasteners are preferred over shear head fasteners when the countersunk depth is below approximately 70% of the bolted plate thickness. (11)

2.5.2 Fastener Material

The main considerations in the selection of the fastener material are its compatibility with the bolted plate material and its mechanical properties. Galvanic corrosion is a problem when steel or aluminum is used adjacent to graphite/epoxy composites, especially in a salt spray atmosphere (see Table 1, Figure 15 and Reference 12). Titanium does not corrode when it is in contact with graphite/epoxy composites. The compatibility of other materials with graphite/epoxy composites is rated in Table 1. Consequently, titanium fasteners are preferred for use in bolted composite structures. Also, a corrosion barrier is generally introduced between bolted composite and metallic parts, if the metal is steel or aluminum (see Figure 15).

Titanium fasteners are preferred for use with graphite-reinforced composites. Steel and aluminum fasteners are not recommended for use with these composites due to their corrosion susceptibility.

(12)

2.5.3 Fastener Size

The fastener size is generally selected to preclude excessive fastener bending effects that could reduce its load transfer capability and induce premature fastener failure. As a general rule, the ratio of the fastener diameter (D) to the bolted plate thickness (t) should be greater than 1 (see Figure 16).

The fastener diameter must be larger than the thickness of either bolted plate.

(13)

2.5.4 Fastener Fit and Hole Quality

Structural parts that are mechanically fastened together are drilled in accordance with established process specifications. Nevertheless, the presence of flaws at fastener locations is commonplace. These flaws include improper fastener seating,

TABLE 1. GALVANIC COMPATIBILITY OF FASTENER MATERIALS WITH COMPOSITES (REFERENCE 12).

Fastener Material	Compatibility with Graphite/Epoxy Composites
Titanium and its alloys	Very Good
MP-35N, INCO 600 (Nickel, Cobalt alloys)	Good
A286, PH13-8MO (Molybdenum alloys)	Acceptable
Monel	Marginal
Low Alloy Steel	Not Compatible
Silver Plate, Chrom. Plate	Adequate with/A286, PH13 13-8MO
Cadmium or Zinc Plate	Not Compatible
Aluminum or Magnesium Alloys	Not Compatible

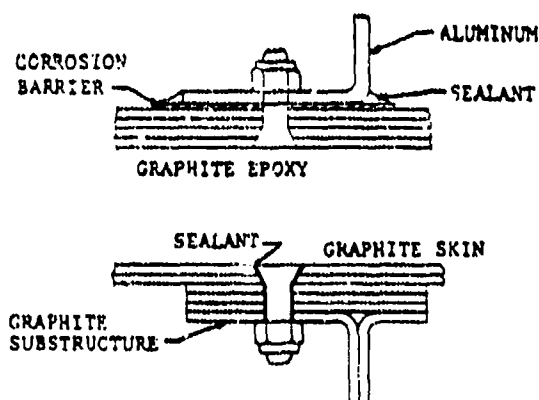


Figure 15. Galvanic Compatibility and Corrosion Prevention.

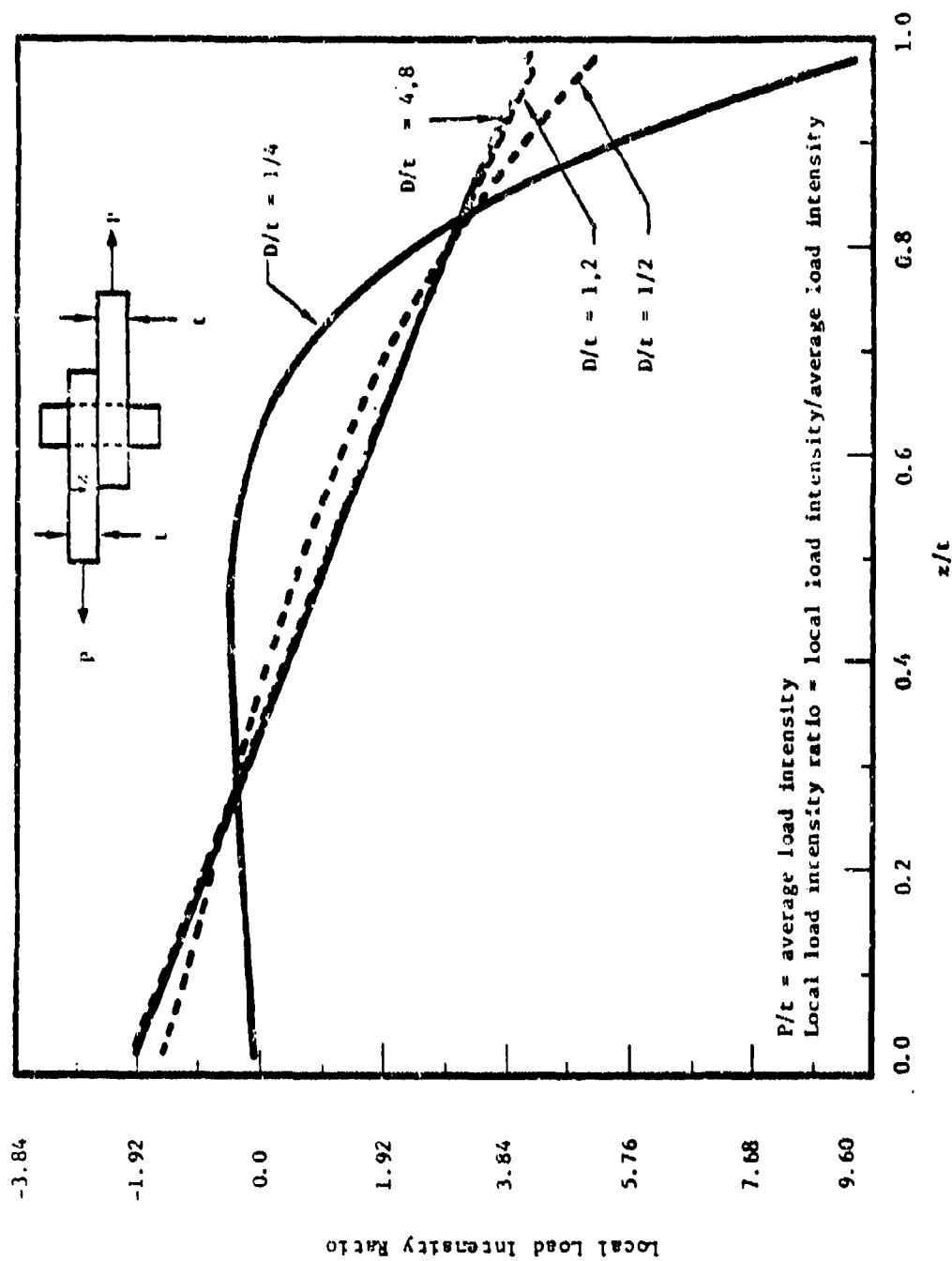


Figure 16. Effect of Fastener Size on the Load Distribution
 (Reference 6).

cratering of the hole boundary, broken and separated fibers at the drill exit side, delaminations near the exit surface, and a slight tilt (<10 degrees) in the hole axis away from the normal to the bolted plate (Reference 13). Interference fit of fasteners will also affect hole quality and influence the efficiency of the joint. The effects of interference fits and fastener hole flaws were studied in Reference 13 (see Table 2). A summary of the results is presented below:

Interference fastener fits (up to 0.008 inch of interference) induce negligible tensile strength losses. Nevertheless, they are generally not recommended due to installation problems and their effect on hole quality.

(14)

If the countersunk fastener seating (assuming 50% of the bolted plate thickness to be the nominal countersunk depth) is increased beyond 80% of the bolted plate thickness, the joint strength is decreased considerably (20 to 50%).

(15)

If the countersunk hole axis is at least 10 degrees away from the normal to the bolted plate, significant joint strength losses result (over 20% for a 10 degree tilt).

(16)

Other flaws (exit side broken fibers and delaminations, less than a moderate level of porosity in bolted laminates, holes offset by less than 0.005 inch, etc.) at fastener locations introduce negligible joint strength losses ($<10\%$).

(17)

2.5.5 Fastener Torque-Up

Static and fatigue tests on composite-to-metal joints

TABLE 2. EFFECTS OF FLAWS AT FASTENER HOLE LOCATIONS (REFERENCE 13).

	Percent Change in Strength*			
	COMPRESSION		TENS (OH)	
	RTW	250W	RTD	250W
OUT-OF-ROUND HOLES				
• 50/40/10 LAMINATE	-	-	<2.0	-
• 30/60/10 LAMINATE	-	-	-4.8	-
BROKEN FIBERS EXIT SIDE OF HOLE				
• SEVERE	-3.5	-12.2	-9.5	-
• MODERATE	-6.2	<2.0	-4.9	-
POROSITY AROUND HOLE				
• SEVERE	-12.1	-32.8	<2.0	-
• SEVERE WITH FREEZE-THAW	-13.3	-	-	-
• MODERATE	-5.4	-17.9	-	-
• MODERATE WITH FREEZE THAW	-7.9	-	-	-
IMPROPER FASTENER SEATING DEPTH (50% OF NOMINAL)				
• 80% THICKNESS	-	-	-23.2	-
• 100% THICKNESS	-	-	-56.9	-
TILTED COUNTERSINKS				
• AWAY FROM BEARING SURFACE	-	-20.0	<2.0	-
• TOWARD BEARING SURFACE	-	-22.7	-23.7	-
INTERFERENCE FIT TOLERANCES (INCH)				
• 50/40/10 @ 0.003	-	-	<2.0	+14.7
• @ 0.008	-	-	<2.0	+11.2
• 30/60/10 @ 0.003	-	-	<2.0	+2.4
• @ 0.008	-	-	<2.0	<2.0
FASTENER REMOVAL AND REINSTALLATION				
• 100 CYCLES	-	-7.4	<2.0	-

* RTD - Room Temperature, Dry; RTW - Room Temperature, Wet; 250W - 250°F, Wet;
Wet - 0.86% Moisture by Weight.

were conducted in Reference 13, varying the fastener torque-up value from 0 in-lb to 160 in-lbs. Fastener torque-up significantly improved the static strength of the joint (15 to 30%), and its fatigue life at a selected stress level. Similar results were observed in Reference 14. Under fatigue loading, the torque-up inhibits the initial growth of local failures in the joint, and the results in a more abrupt fatigue failure due to excessive hole elongation than a joint with no applied torque.

Fastener torque-up increases the static strength of a joint and its fatigue life at a selected stress level.

(18)

2.6 Bolted Laminate Properties

The basic material and its layup (stacking sequence) in bolted laminates influence the joint performance considerably. When graphite/epoxy laminates are bolted to metallic substructures, galvanic corrosion must be addressed (see Figure 15 and Table 1). For example, a corrosion barrier like a glass/epoxy layer must be used between graphite-reinforced composites and aluminum substructures.

When graphite-reinforced composites are bolted to metallic substructures, corrosion barriers must be introduced if the metal is not compatible with the composite material (see Table 1).

(19)

The bolted laminate layup is generally denoted by the percentages of plies with fiber orientations of 0, + or -45 and 90 degrees, with respect to the primary loading direction, for most structural laminates. The envelope within which a bearing failure mode and the maximum bearing strength are realized is shown in Figure 17. Within this envelope, the strength is independent of the actual stacking sequence. This assumes a laminate width-to-fastener diameter ratio (W/D) of at least 4, and an edge distance (E) of at

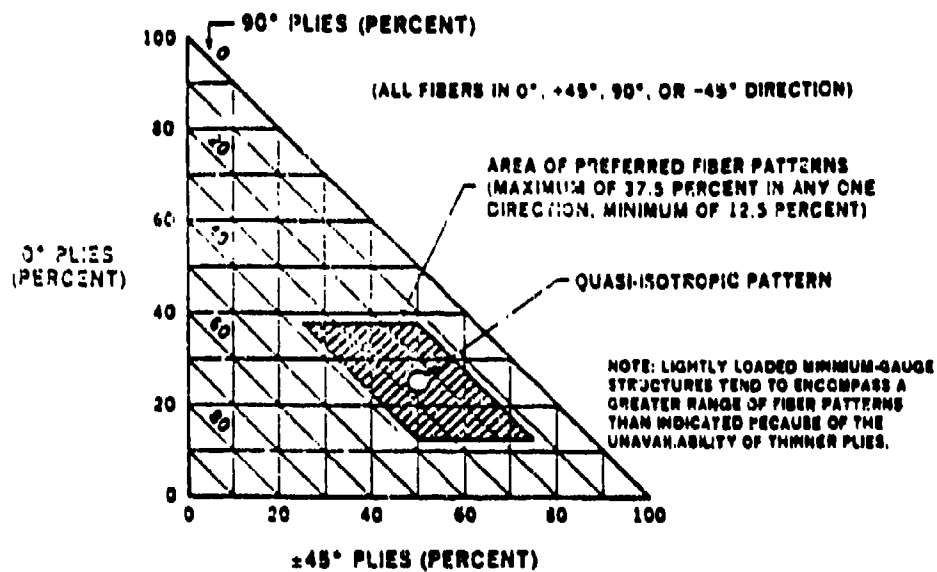


Figure 17. The Envelop of Bolted Laminate Layups for Realizing a Bearing Mode of Failure and the Maximum Bearing Strength (Reference 10).

least 3D. When the percentage of 0 degree plies exceeds 40, a shear-out mode of failure is introduced, reducing the bearing stress value at failure. Section 2.7 presents the effects of fastener spacing and the geometry of a bolted plate on its strength.

The bearing strength of a laminate is maximum when its layup contains less than 10% each of 0, + or -45 and 90 degree plies. The corresponding failure occurs in a bearing mode.

(20)

In addition, the individual plies must be arranged such that adjacent plies have different fiber orientations. If the stacking sequence contains groups of plies with identical fiber orientations, delamination-related failures will occur and reduce the joint strength.

Ply with different fiber orientations should be interspersed within the laminate, to the maximum possible extent, to minimize delamination-induced strength losses. Group of identical plies should not exceed 0.02 inch in thickness.

(21)

2.7 Fastener Spacing and Arrangement

The geometrical parameters that define the fastener spacing and the fastener arrangement in a bolted plate are illustrated in Figure 18. E is the edge distance, S_L and S_T are the fastener spacings in the loading and transverse directions, and $W = S_T$ for a single fastener joint. The effects of these geometrical parameters were studied in References 8, 13 and 14. The results are summarized below:

The bearing and net section strengths decrease when the fastener size increases (see Figure 19).

(22)

The bearing stress at failure decreases

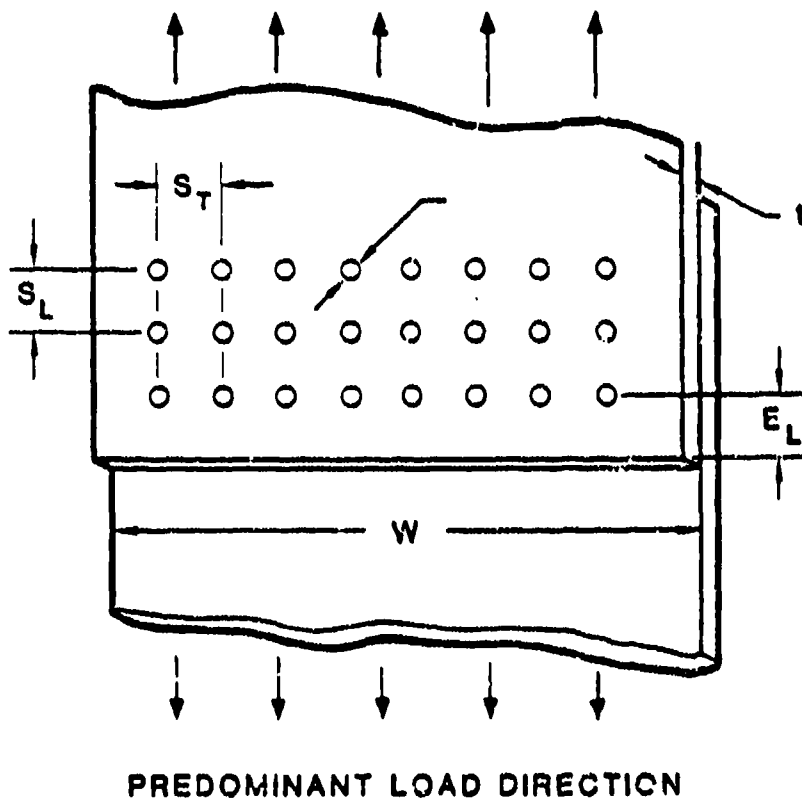
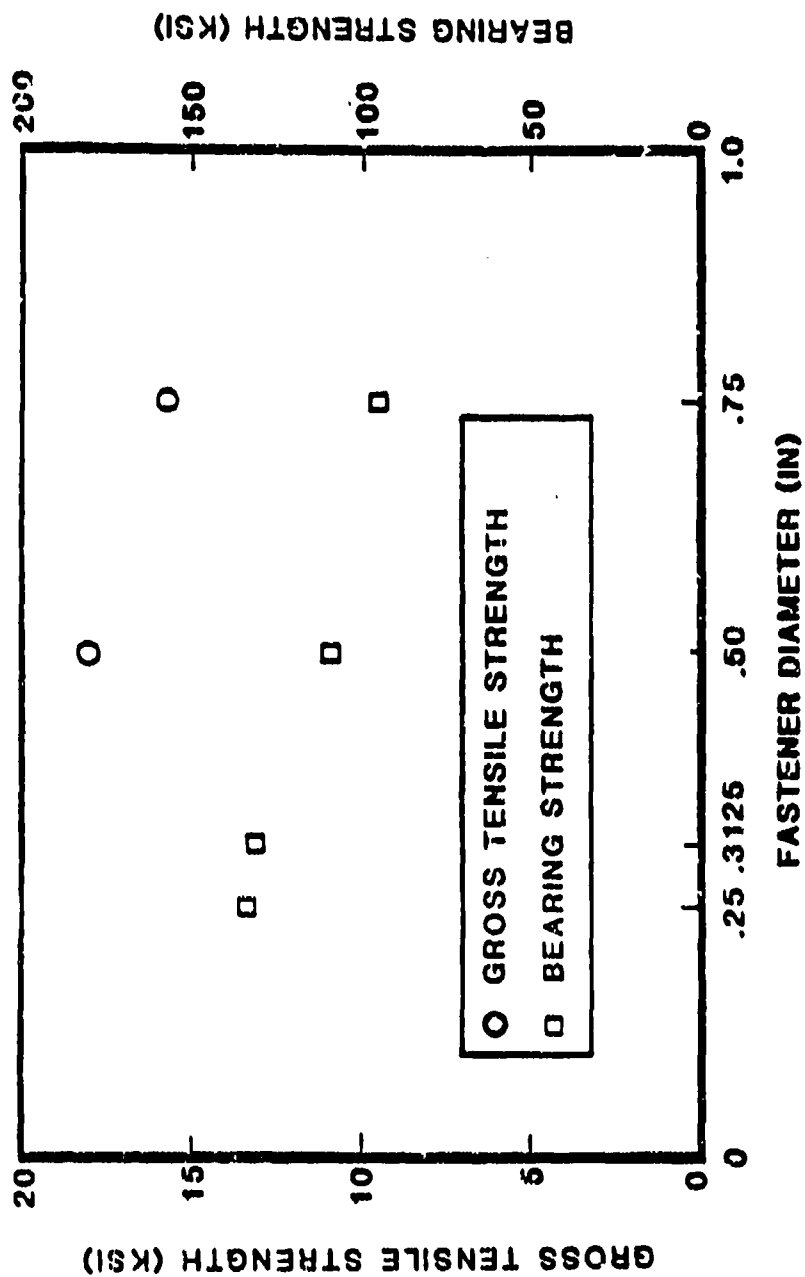


Figure 18. Geometrical Parameters for a Bolted Plate.



NOTE: 20-Ply, 50/40/10, AS1/3501-6 Layup; 0.31 in. Aluminum Plate;
 Torque = 100 in-lbs, E/D = 3, W/D = 6, Protruding Head Steel
 Fasteners; RTD; Net Section Strength = 1.2 x Gross Strength

Figure 19. Effect of Fastener Size on the Tensile Response of Composite-to-Metal Joints in Single Shear.

significantly when E/D is reduced below 3 (see Figure 20). A bearing mode of failure is observed only when $E/D > 4$, and the percentage of 0 degree plies is less than 40. A shear-out mode of failure results when $E/D < 3$, or when the percentage of 0 degree plies is > 40 .

(23)

The bearing stress at failure decreases significantly when S/D (W/D for a single-fastener joint) is reduced below 4 (see Figure 21). When $E/D > 3$, $W/D > 4$, and the percentage of 0 degree plies is below 40, a bearing mode of failure occurs. When $W/D < 4$, a net section failure occurs in the same laminate.

(24)

When the fastener spacing in the loading direction (S_L/D) is decreased below 4, the joint strength decreases due to stress concentration interaction (see Figure 22). The same effect is observed with S_T/D (see Figure 21).

(25)

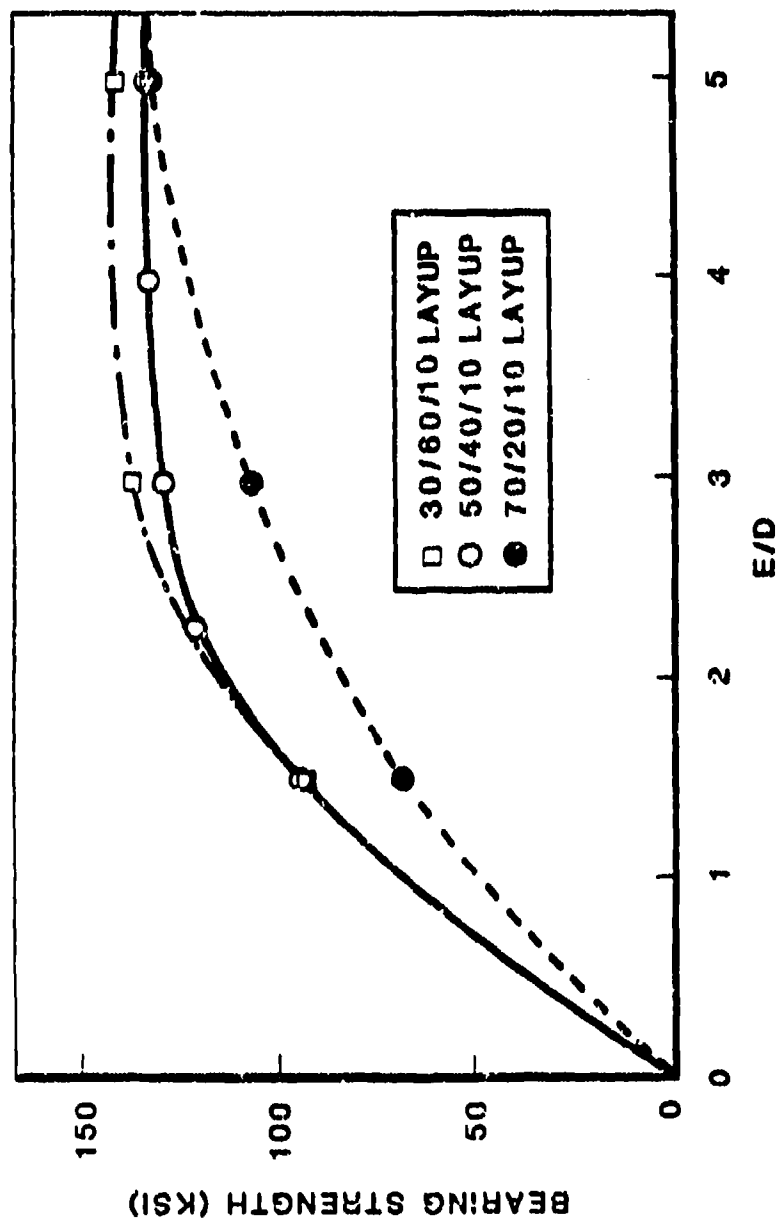
In summary, ensure that $D/t > 1$, $E/D > 3$, W/D (S_T/D) > 4 , $S_L/D > 4$, and the percentage of plies in any orientation is < 40 , to achieve a bearing failure mode and to realize the maximum joint strength.

(26)

2.8

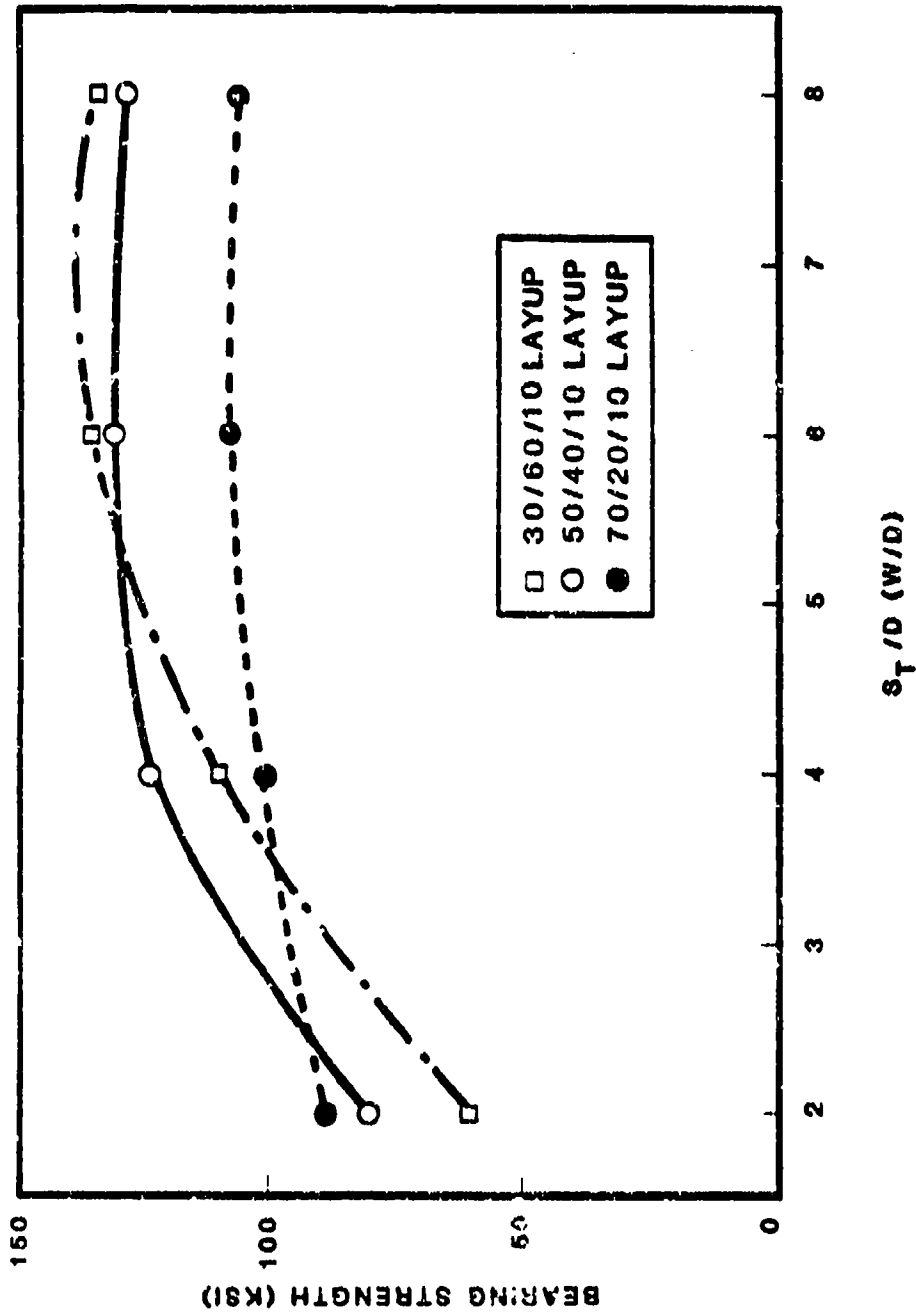
Joint Tailoring for Maximum Efficiency

The design of a joint should achieve the following objectives to be considered efficient: (1) It should be capable of transferring the design ultimate loads without failing any member; (2) It should possess the design life when subjected to the design spectrum fatigue loading; (3) It should be the least weight design that meets (1) and (2); and (4) The complexity of the design concept



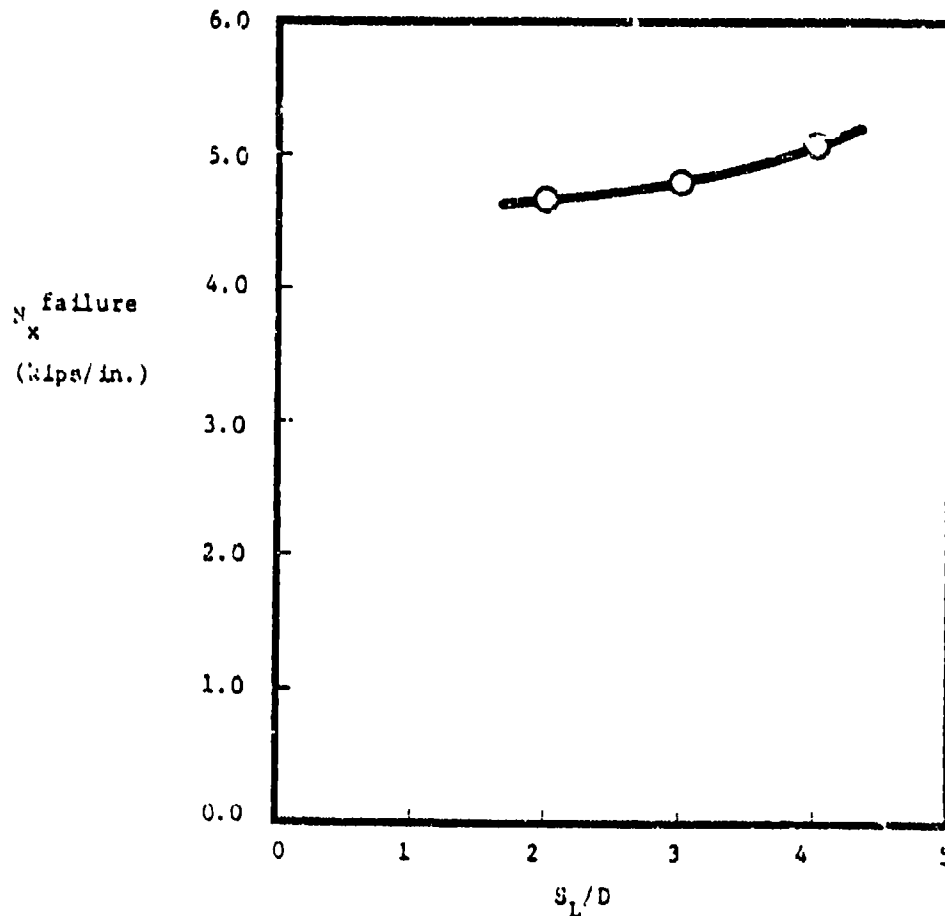
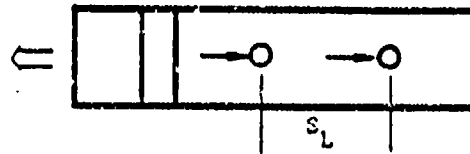
NOTE: Composite-to-metal joints in single shear; 20-ply AS1/3501-6 layups;
 0.31 in. aluminum plate; W/D = 6, Torque = 100 in-lbs; Protruding Head,
 steel fastener; D = 5/16 in.

Figure 20. Effect of E/D on the Bearing Strength of Bolted Laminates.



NOTE: Composite-to-metal joints in single shear; 20-ply, AS13601-6 layups;
 0.31 in. aluminum plate; $E/D = 3$; Protruding head steel fastener;
 $D = 5/16$ in.; Torque = 100 in-lbs

Figure 21. Effect of W/D on the Bearing Strength of Bolted Laminates.



Note: Composite-to-metal, two fasteners-in-a-row joint;
 20-ply, 50/40/10 layup; AS1/3501-6 graphite/epoxy;
 0.31 in. aluminum plate, single shear; RTD; static
 tension; $S_L/D=W/L=6$; protruding head steel fastener;
 $D=5/16$ in.; $T=100$ in-lbs.

Figure 22. Effect of S_L/D on the Strength of Bolted Laminates.

should be controlled to aid producibility and maintainability of the structural joint.

A joint can be tailored to improve its efficiency. For example, when the number of fastener rows (a row being perpendicular to the primary loading direction) is increased, the peak load fraction is generally carried by the innermost or outermost fastener row (see Figures 11 and 12). If the failure mode at the critical fastener location is bearing or net section, the thickness and width of the bolted plate at that location will influence the joint failure load. In an efficient design, the width and the thickness of the bolted plates will be tailored such that every fastener location is equally critical (see Figure 5). The peak bearing stress at the design ultimate load level will be lowered to a level that ensures a minimal bearing/by-pass interaction, if possible (see Figure 13).

Some experimental concepts have also been demonstrated to be efficient joint tailoring concepts, despite the difficulty they introduce in applying the concept at the production level. An example is shown in Figure 23, where the 0 degree plies in the bolted skin are replaced by + and -45 degree plies in the joint region (Reference 15). This causes a smaller fraction of the running load to be transferred at the joint location, and also increases the local bearing strength. An alternative, equivalent concept would be to replace the stiffer material by a tougher material at the joint location. For example, graphite/epoxy plies can be replaced by aramid fiber/epoxy plies at the joint location. It is reiterated, though, that these validated tailoring concepts are difficult to implement in a production environment.

The geometry of bolted laminates must be tailored, in the width and thickness directions, to render every fastener location equally critical.

(27)

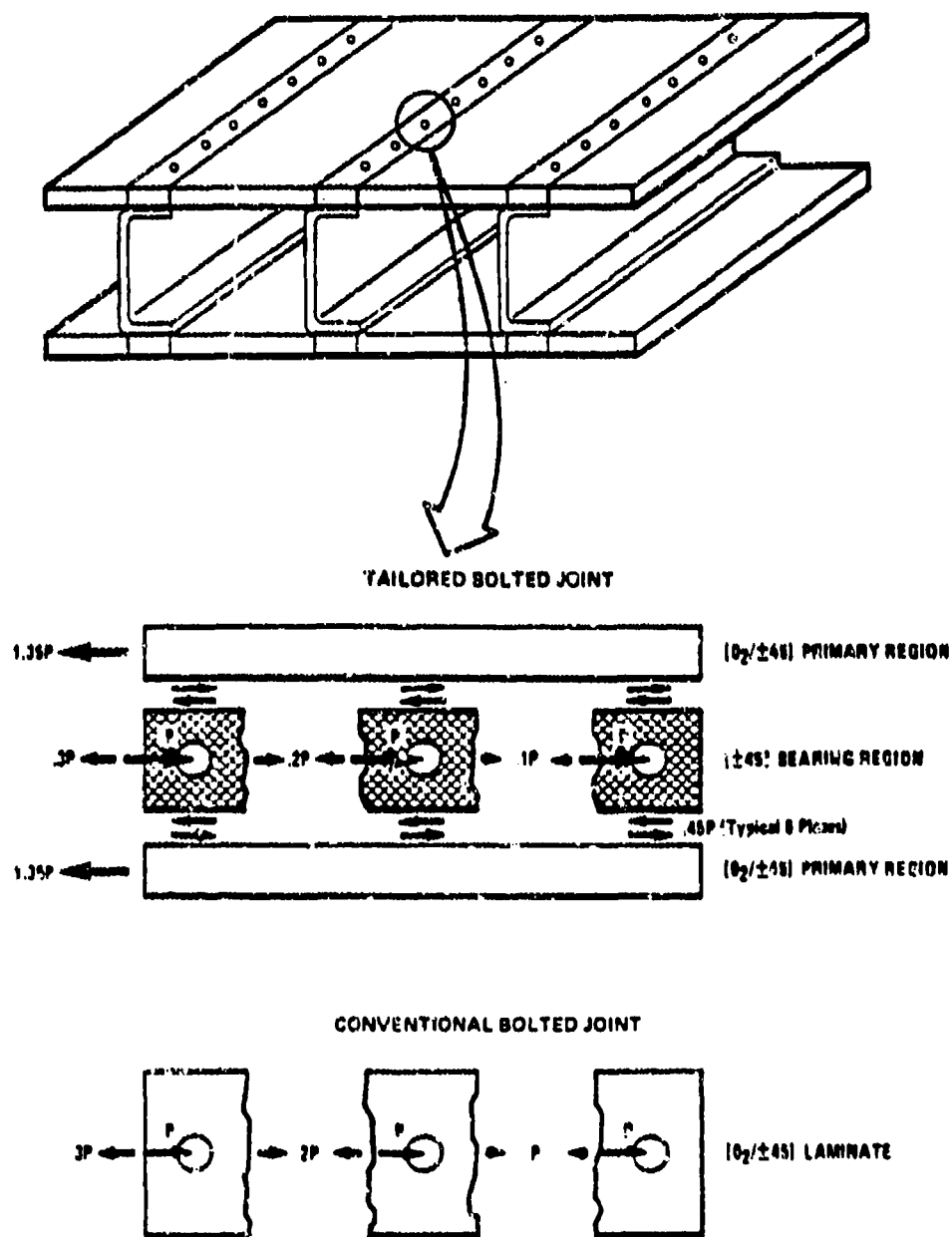


Figure 23. A Sample Tailored Joint.

2.9 Durability Considerations

The design of a bolted joint is currently based on an assumed design ultimate load level and a static strength analysis (see Section 3). The assumed design ultimate load level should account for durability considerations also. Generally, irrespective of the static failure mode, a bolted joint suffers fatigue failure via excessive hole elongation (bearing). This possible change in the failure mode from the static loading case to the fatigue loading case has been observed by many in the literature (see References 13 and 14).

If the joint statically fails in a bearing mode, it could suffer premature excessive hole elongation (fatigue failure) when subjected to the spectrum fatigue loading. Figures 24 and 25 present sample constant amplitude fatigue test results from Reference 14 for a fully reversed loading case ($R=-1$). Similar results should be used to approximately and conservatively estimate the fatigue life of a joint using a fatigue analysis (Miner's rule, for example). Based on the fatigue analysis, the bearing stress at the critical fastener location should be designed to be sufficiently lower than the static bearing strength, to ensure the design life of the joint. The final joint design, therefore, will be capable of statically transferring the design ultimate load, with the peak bearing stress value ensuring the design fatigue life.

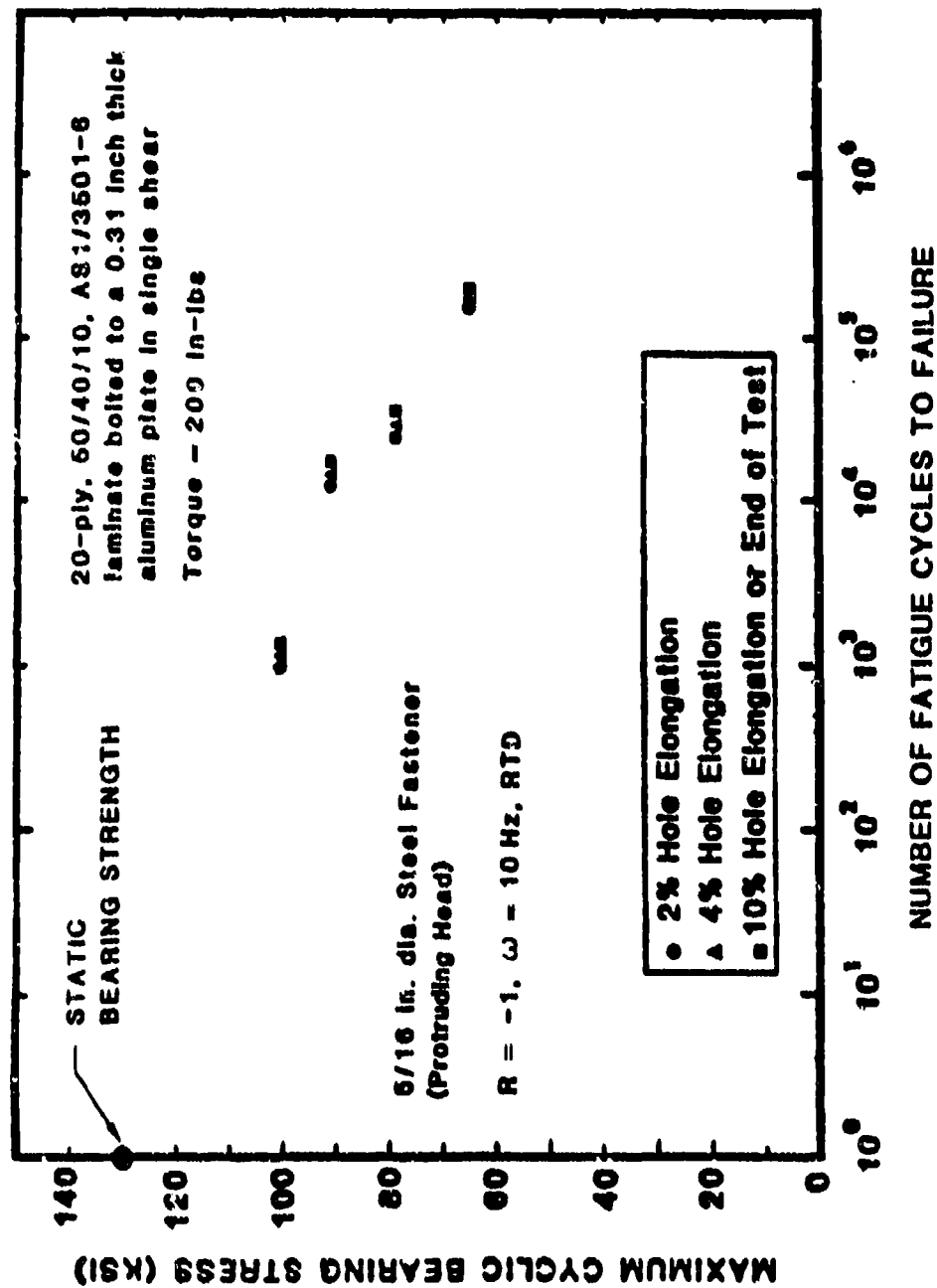


Figure 24. Effect of Maximum Cyclic Bearing Stress on the Number of R-1 Fatigue Cycles to Cause Specified Hole Elongations in a Bolted Laminate.

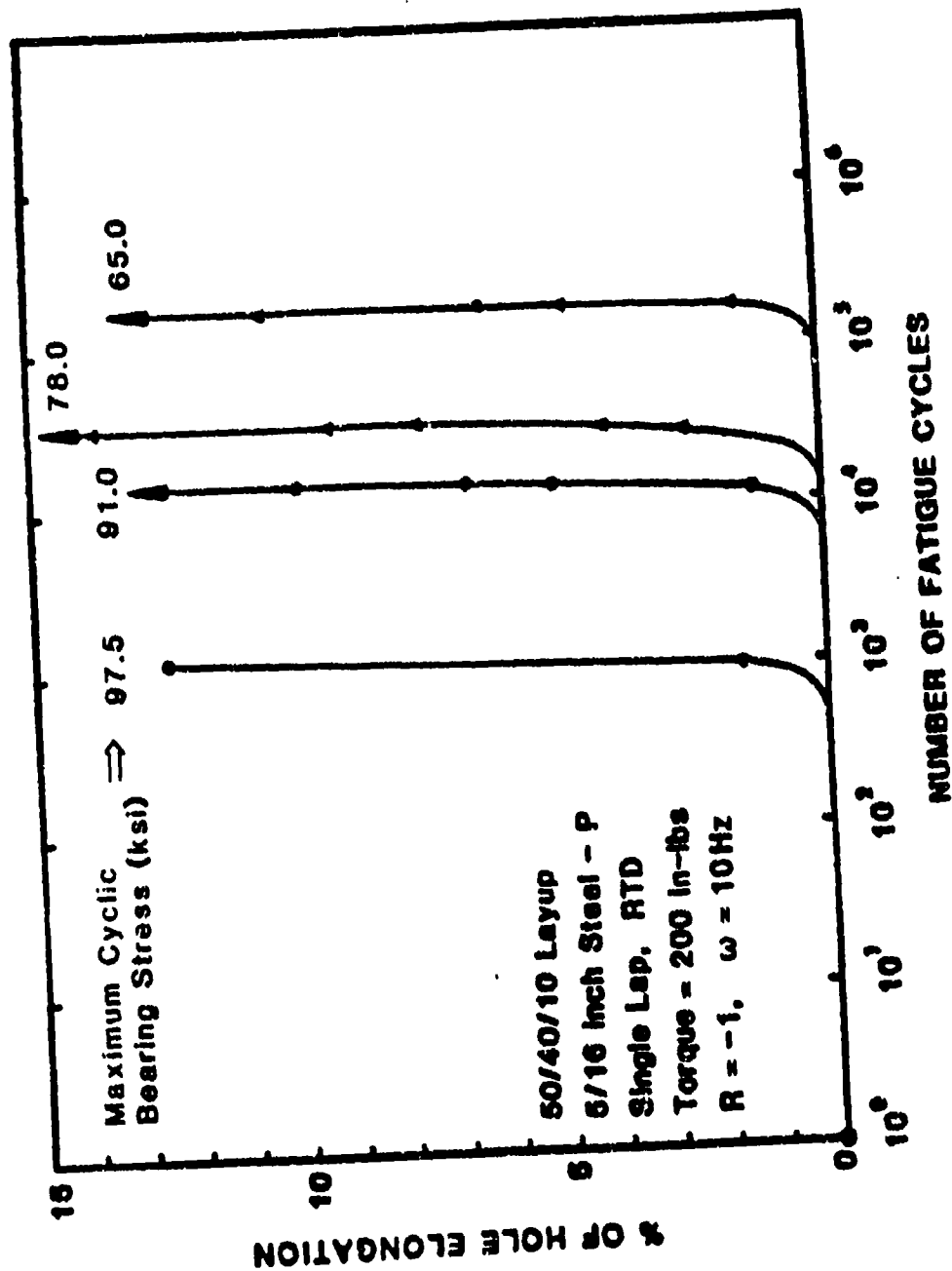


Figure 25. Effect of Maximum Cyclic Bearing Stress on the Hole Elongation Rate for a Bolted Laminate under $R=-1$ Loading.

SECTION 3

STRENGTH ANALYSIS OF BOLTED COMPOSITE STRUCTURES

As mentioned in Section 1.4, two computer codes were developed in this Northrop/AFWAL program to predict the strength of bolted joints containing a single fastener (SASCJ and SAMCJ) or multiple fasteners (SAMCJ). Most of the structural joints contain multiple fasteners, and SAMCJ is adequate for the design of these joints. SAMCJ is also capable of predicting the strength of single fastener joints, without accounting for the nonlinear joint load versus deflection behavior introduced by ply level failures. However, if the user wishes to interrogate an isolated fastener location, accounting for the nonlinear joint behavior due to progressive (two-stage) ply failures, the SASCJ code is useful. The reader is referred to References 6 and 7 for detailed descriptions of the SASCJ and SAMCJ analyses, respectively.

In the following sub-sections, brief descriptions of the analyses in the SASCJ and SAMCJ computer codes are presented, along with detailed instructions for the use of these analytical design tools.

3.1 Description of SASCJ Analysis

A two-dimensional anisotropic plate analysis that accounts for finite plate dimensions (FIGEOM), and a finite difference fastener analysis (FDFA), are incorporated into a progressive failure procedure to develop a strength analysis for single fastener joints in composite structures (SASCJ). An isolated fastener location in a bolted structures (see Figure 7) is primarily subjected to the loading shown in Figure 26. The general bolt bearing/by-pass situation can be analyzed as a superposition of an unloaded hole situation and a fully loaded hole situation (see Figure 26). The unloaded hole case is analyzed using the two-dimensional plate analysis (FIGEOM), and does not involve the

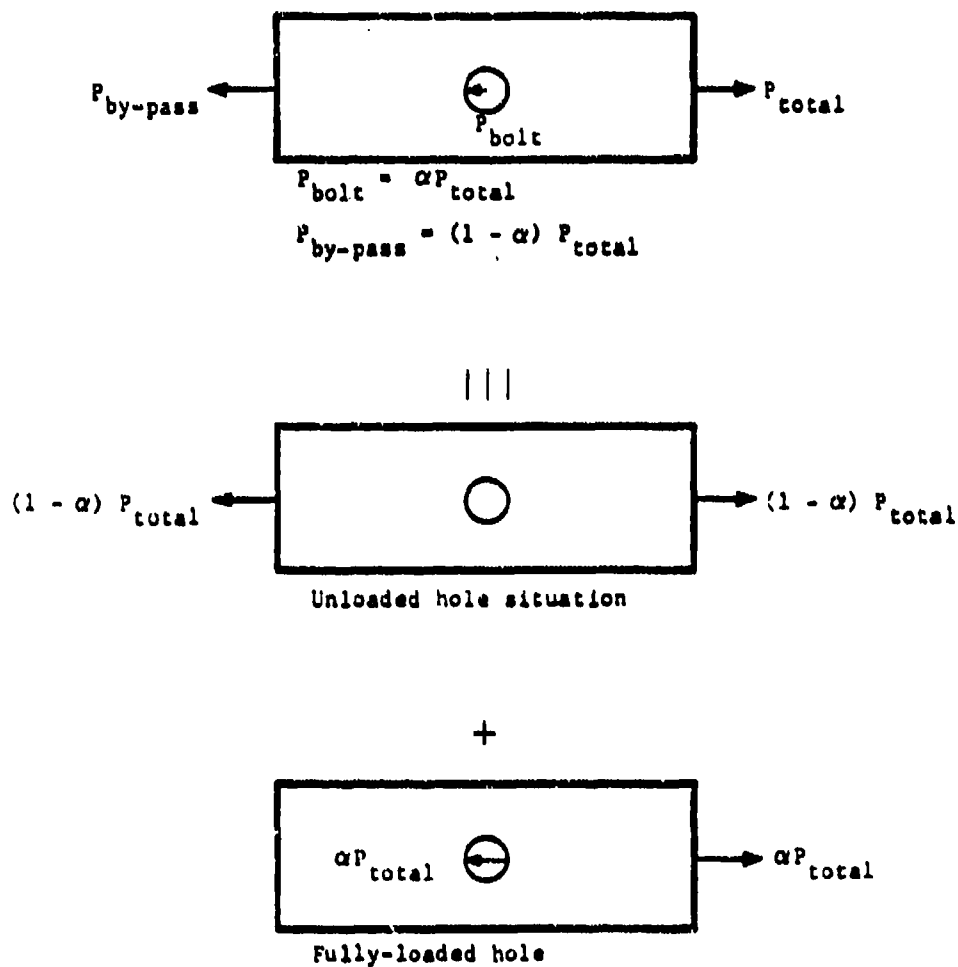


Figure 26. Schematic Representation of a General Single-Fastener Situation as a Superposition of Unloaded and Fully Loaded Hole Situations.

fastener analysis (FDFA). The fully-loaded hole situation is analyzed using a progressive failure procedure that predicts local ply failures and delaminations until the bolted plate cannot carry any additional applied load. The employed ply-failure criteria and the delamination criterion are discussed in Sections 3.1.3 and 3.1.4.

3.1.1 Strength Analysis Procedure for Fully-Loaded Holes

The strength of laminates with fully loaded holes is predicted using the procedure outlined in Figure 27. A two-dimensional stress analysis (FIGEOM), accounting for finite dimensions of the bolted plates, is initially performed on each bolted plate. Computed plate stresses are used to calculate the effective moduli of the various ply types in each bolted plate (see Reference 6). The inplane strains computed by the FIGEOM code are used to obtain the stress state in each ply. The ply stresses around the hole boundary are integrated to yield the bearing load in each ply (see Reference 6). The inplane stresses in each ply, per unit bearing load, are incorporated into selected failure criteria to compute the ply (bearing) loads corresponding to the various inplane failure modes.

The effective moduli and the ply bearing loads corresponding to the various failure modes, for all the plies in each bolted plate, are incorporated into the fastener analysis. The initial fastener analysis on the undamaged plates computes the distribution of the applied bearing load among the various plies. Comparing these ply loads with the stored failure values for inplane ply failures, the joint load corresponding to the earliest ply failure is obtained. The fastener analysis also computes approximate shear strain values at the interfacial locations between adjacent plies. Incorporating these into an interlaminar failure criterion, the joint load corresponding to the earliest interlaminar failure (delamination) is obtained. The smaller of the two joint loads, corresponding to the earliest inplane and interlaminar

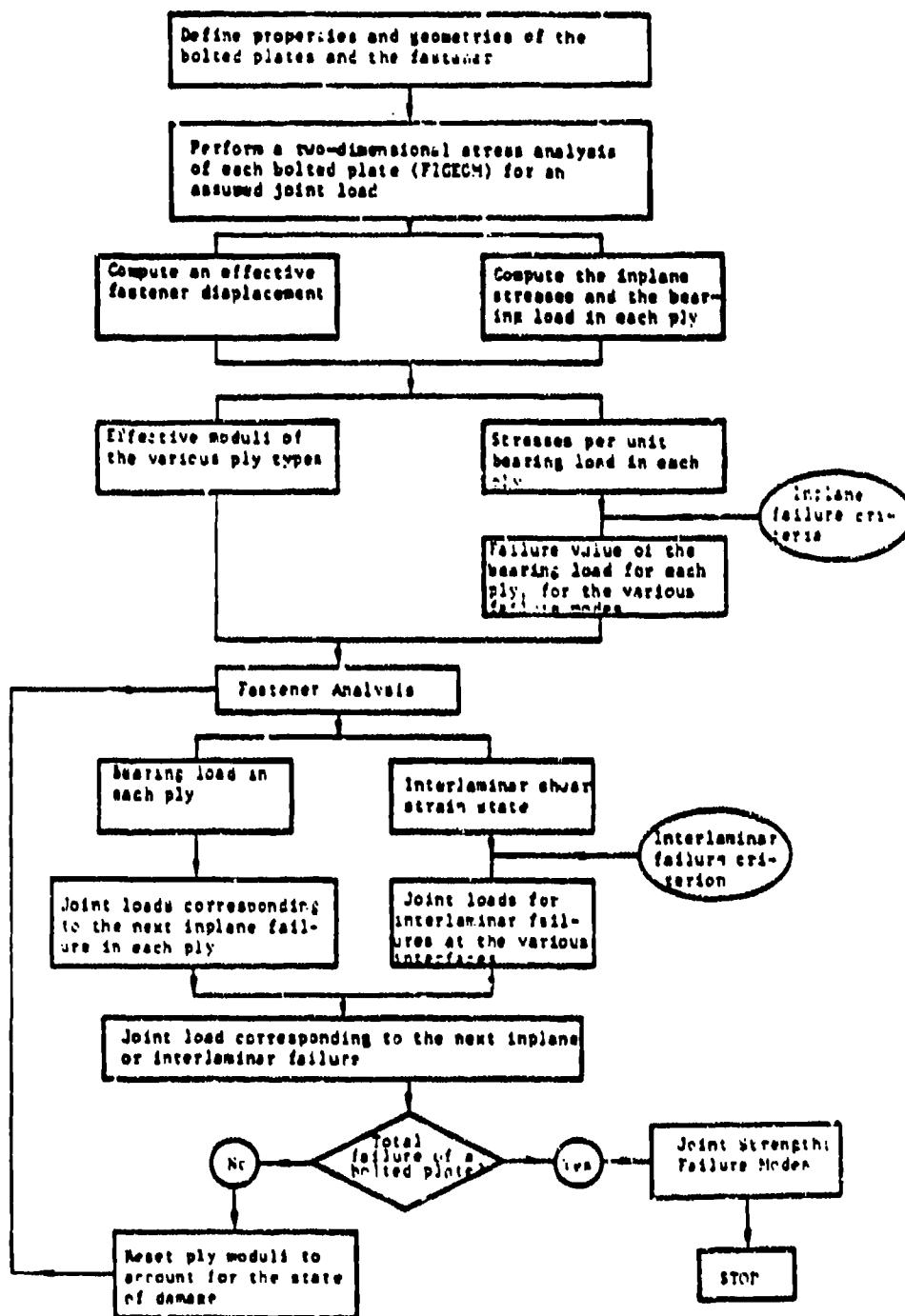


Figure 27. Flowchart for the Strength Analysis of Laminates with Fully Loaded Holes.

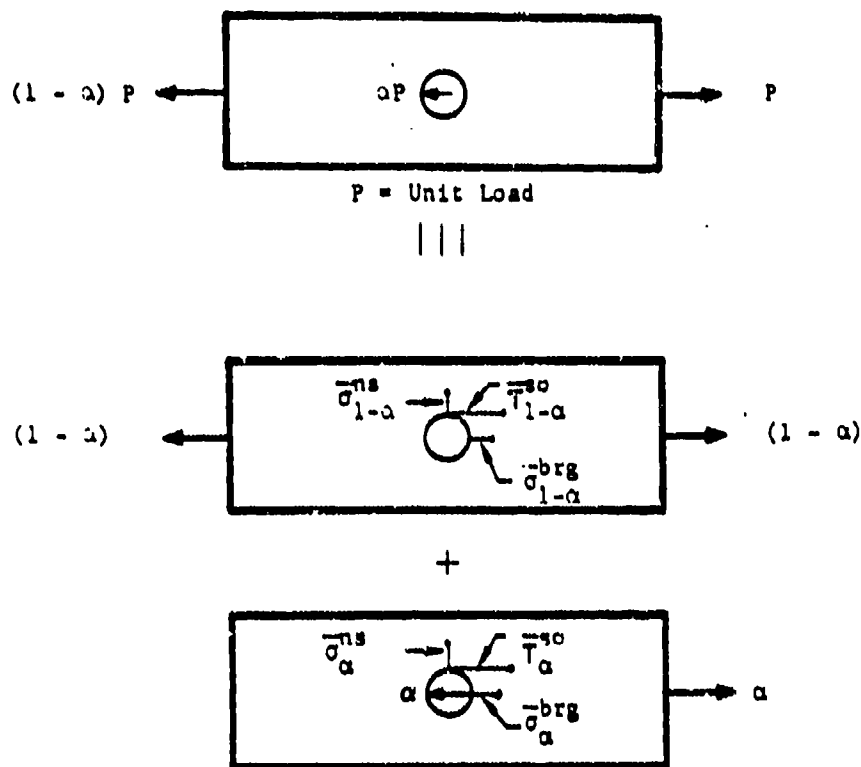
failures, determines the first failure in a bolted plate and the corresponding joint load value.

The effective moduli of the damaged plies are reset to appropriately represent the predicted failure modes. The revised moduli are incorporated into the fastener analysis, and the procedure is repeated to predict the next failure mode and the corresponding joint load. When any ply is predicted to fail totally, the analysis computes the redistribution of the corresponding joint load among the remaining effective plies, and determines if any other concomitant ply failure is precipitated. This process is repeated until one of the bolted plates becomes ineffective in transferring the applied load (joint failure).

The SASCF computer code is restricted to protruding head fasteners, and assumes that fastener failure is precluded. However, when a countersunk fastener is specified, SASCF assumes an appropriate boundary condition at the head location, and expects the user to input an equivalent (larger) uniform fastener diameter. It can analyze any combination of laminated and metallic plates, bolted together in a single-lap or double-lap configuration.

3.1.2 Strength Analysis Procedure for Partially-Loaded Holes

A general fastener location in a bolted plate transfers a fraction (α) of the total applied load via the fastener, the remainder ($1-\alpha$) being by-passed to the next fastener location (see Figures 7 and 26). In this case, the stress state at the fastener location is computed as a superposition of the stress states corresponding to the unloaded and fully-loaded hole situations. Figure 28, for example, presents a schematic representation of how the averaged stresses are obtained to predict net section, shear-out and bearing failures in the plies using average stress failure criteria. For a unit applied load, the averaged stresses in the laminate with an unloaded hole, when subjected to a load of $(1-\alpha)$, and the averaged stresses in the laminate with a fully loaded hole,



$$\bar{\sigma}^{ns} = \int_{D/2}^{D/2 + d_{ons}} \sigma_x(o, y) dy = \bar{\sigma}_{1-a}^{ns} + \bar{\sigma}_a^{ns}$$

$$\bar{\tau}^{so} = \int_0^{d_{oso}} \tau_{xy}(x, D/2) dx = \bar{\tau}_{1-a}^{so} + \bar{\tau}_a^{so}$$

$$\bar{\sigma}^{brg} = \int_{D/2}^{D/2 + d_{obrg}} \sigma_x(x, o) dx = \bar{\sigma}_{1-a}^{brg} + \bar{\sigma}_a^{brg}$$

Figure 28. Strength Analysis of Laminates with Partially-Loaded Holes using Average Stress Failure Criteria.

when subjected to a load of α , are computed separately and added. Incorporating the combined averaged stresses into the appropriate failure criteria, the applied load corresponding to a ply failure is computed.

In the case of fully loaded holes, progressive failure prediction involves the repetition of the fastener analysis with revised ply properties after every ply failure. The two-dimensional analysis (FIGEOM) is only carried out once. But, in the case of partially loaded holes, a ply failure will affect the unloaded and the fully loaded hole contributions to the local stresses. Hence, progressive failure prediction in the partially loaded case involves repeating FIGEOM and FDFA analyses after total ply failures.

3.1.3 Inplane Failure Criteria

The SASCJ code permits the user to select any of the following five failure criteria for the prediction of ply failures based on inplane stresses and strains: (1) point stress failure criterion, (2) average stress failure criterion, (3) maximum (fiber directional) strain criterion, (4) Hoffman criterion, and (5) Tsai-Hill criterion. The first two criteria predict three modes of failure in each ply-net section, shear-out and bearing. The maximum strain criterion predicts ply failure based on fiber failure. The Hoffman and Tsai-Hill criteria predict ply failure accounting for biaxial stress interaction that is ignored by the first three criteria.

The point stress failure criterion predicts net section, shear-out and bearing failures when the appropriate stress components at selected locations attain unnotched specimen failure values (see Figure 29). a_{ons} , a_{oso} and a_{obrg} are called characteristic distances. When $\sigma_x(0, D + a_{ons})$ exceeds the unnotched tensile or compressive strength of the ply, as appropriate, a net section ply failure is predicted. When $\sigma_x(D + a_{obrg}, 0)$ exceeds the unnotched compressive strength of the ply, a bearing mode of ply failure is

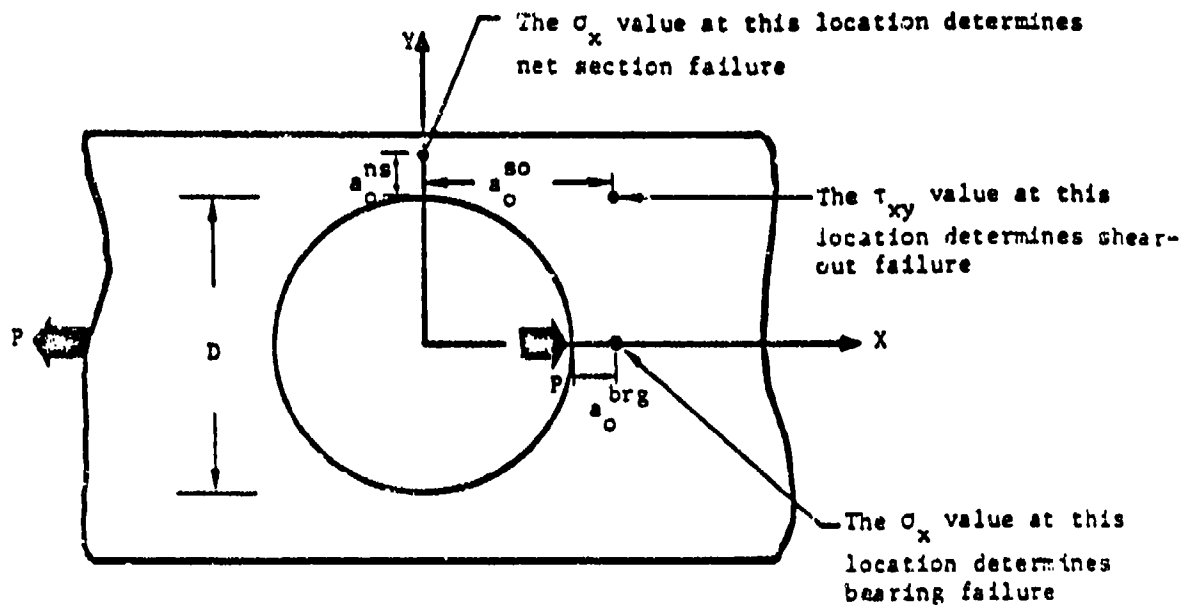


Figure 29. The Characteristic Distances used in the Point Stress Failure Criteria

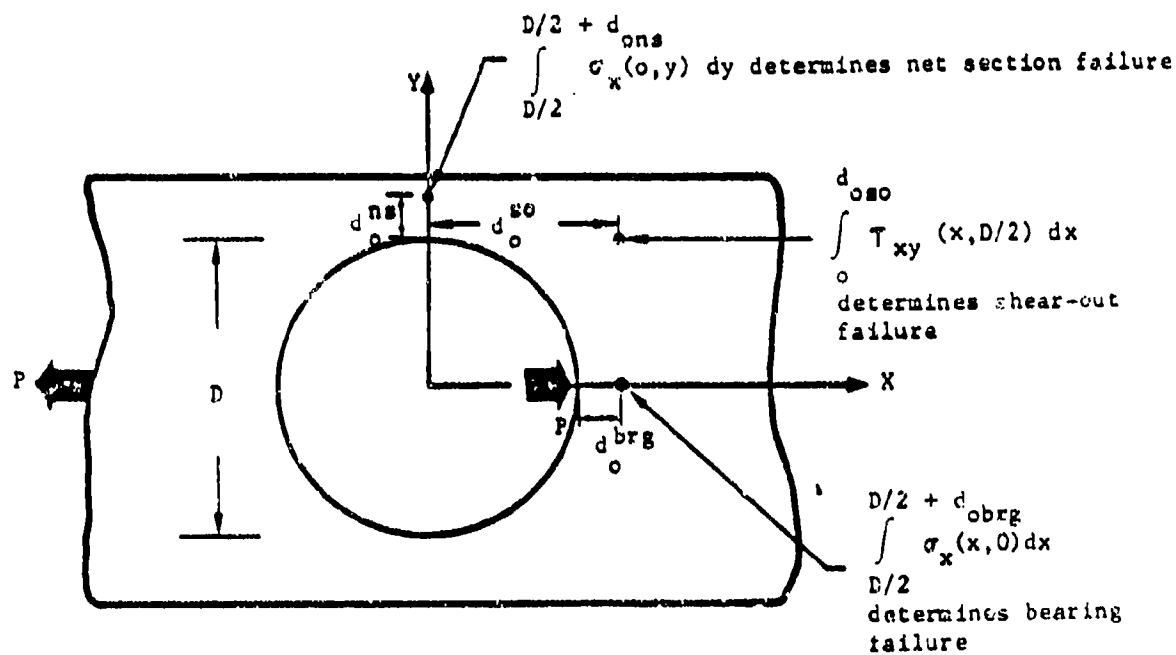


Figure 30. The Characteristic Distances Used in the Average Stress Failure Criteria.

predicted. When τ_{xy} (a_{oso} , $D/2$) exceeds the unnotched ply shear strength, a shear-out mode of ply failure is predicted. The average stress failure criterion predicts these failures based on averaged values of the mentioned stress components over selected characteristic distances (d_{ons} , d_{oso} , and d_{obrg}) that are larger in magnitude compared to those used in conjunction with the point stress criterion (see Figure 30).

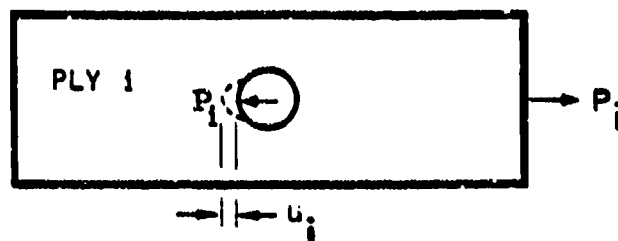
Of the three ply failure modes, only the net section mode causes the ply to become almost ineffective (total failure). The bearing mode of failure causes the ply to suffer a reduction in its effective modulus without losing its load-carrying capacity. The shear-out mode of failure causes a ply to become ineffective only when it is delaminated from the adjacent plies. When a ply suffers any of the above failures, its load versus deflection response is at the knee of the bilinear representation in Figure 31. The damaged ply can carry additional load until total ply failure is precipitated. The SASCF computer code automatically stores the damage state in every ply in the bolted plates, and reassigns values for ply moduli to appropriately represent predicted ply failures. When a ply suffers total failure, its modulus is set equal to zero, and the redistribution of the joint load among the remaining plies is computed. A typical overall load versus deflection behavior of the joint is shown in Figure 32, indicating the effects of local and total ply failures.

The maximum strain (fiber directional), Hoffman and Tsai-Hill criteria are applied along a path that is concentric to the fastener hole, at a characteristic distance (a_o) from the hole boundary (see Figure 33). The location along this path where the selected criterion is satisfied determines the failure location. The maximum strain criterion predicts fiber failure in a ply (total ply failure) when its fiber directional strain exceeds the failure values (ϵ_{11}^{tu} or ϵ_{11}^{cu}).

The Hoffman failure criterion, based on inplane ply

$$K_2 = aK_1$$

$$P_{\text{ULTIMATE}} = \beta P_{\text{INITIAL}}$$



$$q_i = P_i / h_i$$

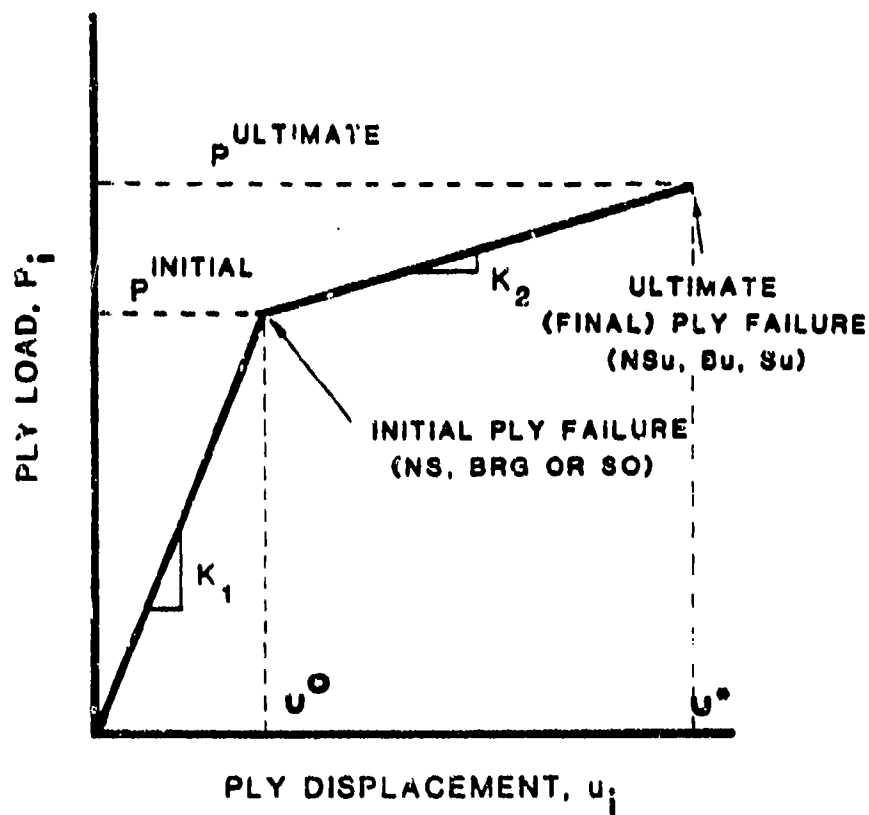


Figure 31. Bilinear Elastic Behavior of a Ply.

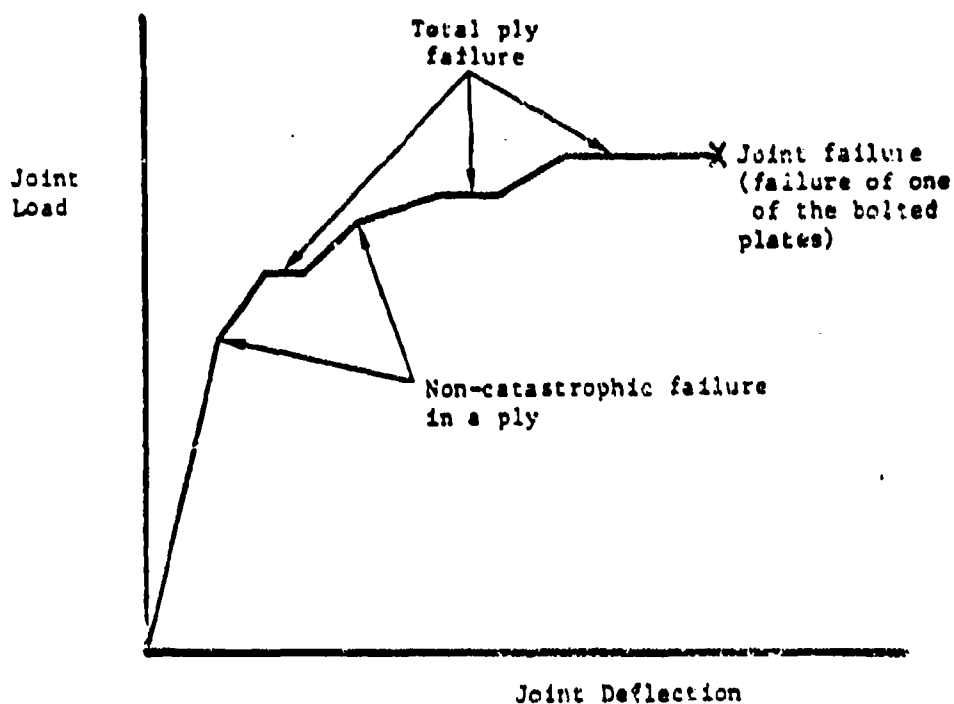


Figure 32. A Schematic Representation of the Overall Load Versus Deflection Response of the Joint.

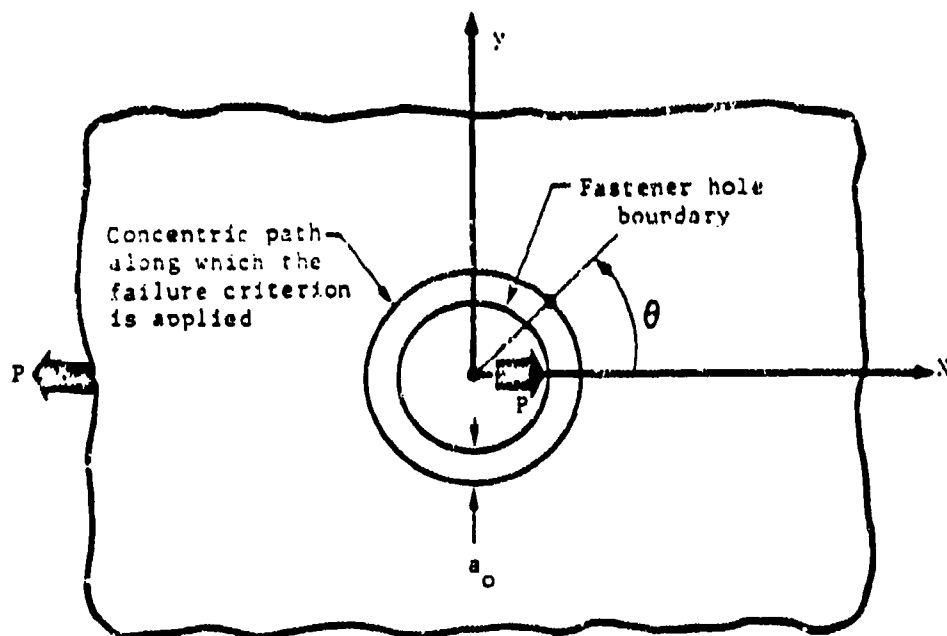


Figure 33. The Characteristic Distance (a_0) Defining the Region Where the Maximum Strain, Hoffman or Hill Criterion is Applied.

stresses, states that total ply failure will occur when the failure index (H) in the following equation reaches a value of unity:

$$\begin{aligned} & (\sigma_1^2 - \sigma_1\sigma_2)/X_cX_t + \sigma_1(X_c - X_t)/X_cX_t + \sigma_2^2/Y_cY_t + \sigma_2(Y_c - Y_t)/Y_cY_t + \\ & \sigma_6^2/S^2 = H \end{aligned}$$

In the above equation, σ_1 , σ_2 and σ_6 are the ply stresses in the fiber coordinate system, X_t and X_c are the uniaxial tensile and compressive material strengths along the fiber direction (1), Y_t and Y_c are the uniaxial tensile and compressive material strengths perpendicular to the fiber direction (2), and S is the material shear strength in the 1-2 plane.

In the SASCF code, the Hoffman criterion is applied along a path that is concentric to the fastener hole, defined by the characteristic distance a_0 (see Figure 33). At selected points along this path, the following expressions for the failure values of the ply load (P_f) are computed:

$$P_f = (-b \pm \sqrt{b^2 - 4ac})/2a$$

where

$$a = [(\sigma_1^2 - \sigma_1\sigma_2)/X_cX_t + \sigma_2^2/Y_cY_t + \sigma_6^2/S^2]/P_1^2$$

$$b = [(X_c - X_t)\sigma_1/X_cX_t + (Y_c - Y_t)\sigma_2/Y_cY_t]/P_1$$

$$c = -1, \text{ and}$$

$$P_1 = \text{ply load at which } \sigma_1, \sigma_2 \text{ and } \sigma_6 \text{ are computed}$$

The location where the smallest non-negative value for P_f is computed identifies the failure initiation point.

The Hoffman criterion predicts total ply failure and the failure location, but does not identify the mode of failure. The failure location, though, generally indicates the possible failure mode. Referring to Figure 33, if failure is predicted near $\theta=0^\circ$, a bearing mode of failure is suspected. If the failure location is near $\theta=90^\circ$, a net section mode of failure is suspected. And, intermediate values of θ indicate a shear-out mode of failure. The Tsai-Hill criterion can be obtained from the Hoffman criterion by setting $X_c = X_t$ and $Y_c = Y_t$. This criterion, therefore, does not account for different strengths under tension and compression. The ply failure load (P_f) in this case is computed to be $1/\sqrt{a}$.

3.1.4 Interlaminar Failure Criterion

Delamination between plies is predicted by incorporating computed shear strains at the interfacial locations into a maximum shear strain criterion. At the interface between plies i and j , for example, the shear strain is computed to be:

$$\gamma_{xz}^{i-j} = (u_i - u_j) / h_\alpha$$

where h_α is the ply thickness in the plate containing plies i and j . This expression for the shear strain is approximate. Plies i and j are assumed to delaminate when γ_{xz}^{i-j} exceeds a failure value. The failure value for γ_{xz} is determined by correlating predictions with observations for a sample test case.

3.2 SASCJ Input Description

SASCJ assumes a uniaxial tensile or compressive load to

be applied to a single fastener bolted joint, in a single or a double shear configuration (see Figures 34 and 35). The code requests information for a general bearing/by-pass situation. If the joint is a symmetric double shear configuration, only half the joint is analyzed (see Figure 35). For example, if plate 2 in Figure 35 is metallic, the input thickness should be half the actual value, and if plate 2 is a laminate, only the layup from the surface to its midplane should be input. The analysis accounts for the joint symmetry through appropriate symmetry conditions at the midplane location (see Figure 35).

A sample SASCH problem is now presented to describe the input requirements for the code. It addresses a steel-to-composite joint in a single shear configuration (see Figure 34). The input is requested by SASCH in an interactive mode. Figure 36 presents the code requests and the user replies for the sample joint. Though the information in Figure 36 is self-explanatory, a description of the input quantities is presented below.

The first input quantity specifies that the problem addresses a bearing/by-pass situation with a by-pass ratio of 0.99 --nearly an open hole situation. The second and third input quantities specify that a static tensile load is applied in a single shear configuration. Subsequently, the two bolted plates are specified to be either a composite laminate or a metal. If the bolted plate is a laminate, SASCH requests the user to specify the number of plies in that plate (20). Note again that, for a double shear configuration, only half the thickness of the second plate should be defined (see Figure 35). SASCH then requests the user to specify the thickness of the metallic plate (0.25). For the laminated plate, SASCH requests, in sequence, the average cured ply thickness (0.006), the number of distinct ply orientations (4), definition of the four orientations (0.0, +45.0, -45.0 and 90.0), and the laminate stacking sequence -- [(45/0/-45/0)₂/0/90]₀. SASCH automatically assumes a metallic plate to be divided into thirty identical layers. The number of layers in a laminate is controlled

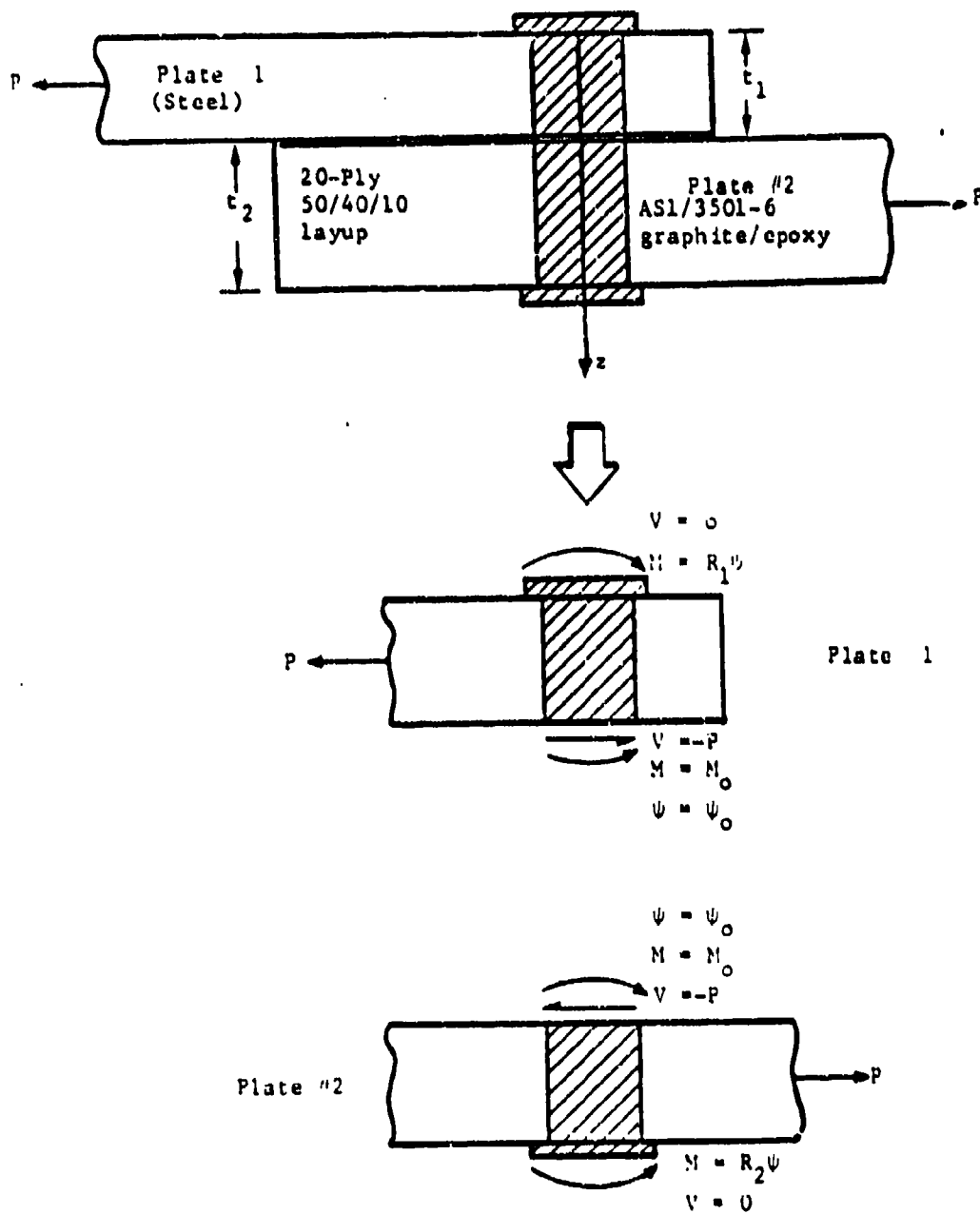


Figure 34. SASCJ Analysis of a Joint in Single Shear.

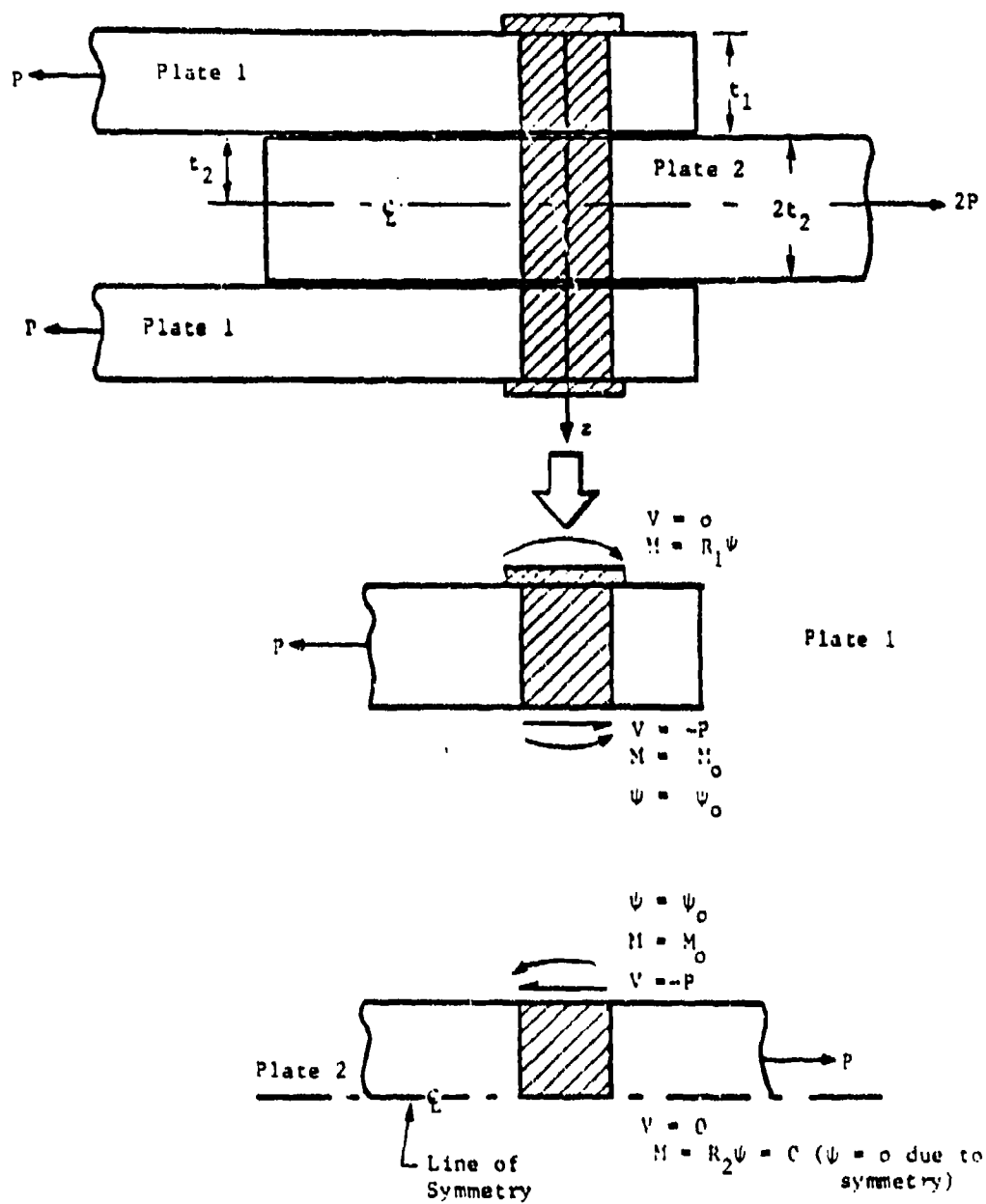


Figure 35. SASCJ Analysis of a Joint in Double Shear.

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171500061 TEMPERATURE ASSUMED AS MEMBERTEMP
PROGRAM SASCIJ
PROGRAM SASCIJ PREDICTS FAILURE LOADS OF
MECHANICALLY FASTENED COMPOSITE LAP JOINTS.
SINGLE OR DOUBLE LAP SHEAR JOINTS.
PROGRAM ASSURES THAT INPUT PARAMETERS ARE
IN INCHES UNITS - LENGTHS ARE INPUT
IN INCHES AND MODULI AND STRENGTHS ARE
EXPRESSED IN PSI.
ENTER BYPASS RATIO ALPHA:
ALPHA=0 FOR FULL BEARING
ALPHA=1 FOR OPEN HOLE
0 (ALPHA=1) FOR GENERAL BYPASS
?
0.99
ENTER:
1 FOR STATIC TENSION
2 FOR STATIC COMPRESSION
?
1
ENTER:
1 FOR SLS (SINGLE LAP SHEAR)
2 FOR BLS (DOUBLE LAP SHEAR)
?
1
IS THE TOP PLATE A COMPOSITE OR A METAL ?
ENTER C OR M IN THE FIRST FIELD
?
C
INPUT MATERIAL DESCRIPTION OF THIS PLATE
EX: AG4/3501-6
a4001
IS THE BOTTOM PLATE A COMPOSITE OR A METAL ?
ENTER C OR M IN THE FIRST FIELD
?
C
INPUT MATERIAL DESCRIPTION OF THIS PLATE
EX: AG4/3501-6
a41/3501-6 (105/0/-45/0/2/0/30)0
NOTE: FOR COMPUTATIONAL PURPOSES A
METALLIC PLATE IS REVELED AS P.30 PLY
LAMINATE OF 0 DEGREE PLYS WITH ISOTROPIC
MATERIAL PROPERTIES
NOTE: NUMERICAL DESIGNATIONS FOR THE
PLATES ARE:
TOP PLATE - NO 1
BOTTOM PLATE - NO 2
INPUT NUMBER OF PLYS IN PLATE NO 2
?
20
(n ) (p = 2)
INPUT THICKNESS OF PLATE NO 1
?
0.25
INPUT PLY THICKNESS IN PLATE NO 2
?
0.005
INPUT NUMBER OF DISTINCT PLY ORIENTATIONS
IN PLATE NO 2
?
4
FOR PLATE NUMBER 2 1
INPUT ORIENTATION OF PLY TYPE NO 1
?
0/0
INPUT ORIENTATION OF PLY TYPE NO 2
?
45/0
INPUT ORIENTATION OF PLY TYPE NO 3
?
-45/0
INPUT ORIENTATION OF PLY TYPE NO 4
?
90/0
INPUT TYPE OF PLY IN PLATE NO 2 FROM TOP
TO BOTTOM
PLY TYPE ORIENTATION
1 0/0 DEGREES
2 45/0 DEGREES
3 -45/0 DEGREES
4 90/0 DEGREES
INPUT TYPE OF PLY FOR PLY NO 1
?
1
INPUT TYPE OF PLY FOR PLY NO 2
?
2
INPUT TYPE OF PLY FOR PLY NO 3
?
1
INPUT TYPE OF PLY FOR PLY NO 4
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INPUT TYPE OF PLY FOR PLY NO 5
?
1
INPUT TYPE OF PLY FOR PLY NO 6
?
2
INPUT TYPE OF PLY FOR PLY NO 7
?
1
INPUT TYPE OF PLY FOR PLY NO 8
?
2
INPUT TYPE OF PLY FOR PLY NO 9
?
1
INPUT TYPE OF PLY FOR PLY NO 10
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2

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Figure 36. Sample SASCIJ Input.

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1 INPUT TYPE OF PLY FOR PLY NO 11
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3 INPUT TYPE OF PLY FOR PLY NO 12
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5 INPUT TYPE OF PLY FOR PLY NO 13
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7 INPUT TYPE OF PLY FOR PLY NO 14
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9 INPUT TYPE OF PLY FOR PLY NO 15
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11 INPUT TYPE OF PLY FOR PLY NO 16
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13 INPUT TYPE OF PLY FOR PLY NO 17
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15 INPUT TYPE OF PLY FOR PLY NO 18
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17 INPUT TYPE OF PLY FOR PLY NO 19
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19 INPUT TYPE OF PLY FOR PLY NO 20
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21 INPUT THE ENGINEERING PROPERTIES OF PLATE NO 1
22 INPUT YOUNG'S MODULUS AND POISSON'S RATIO
23 30.046 0.2
24 INPUT THE ENGINEERING PROPERTIES OF PLATE NO 2
25 INPUT YOUNG'S MODULUS, E1 AND E2
26 18.546 1.946
27 INPUT THE SHEAR MODULUS AND MAJOR POISSON'S RATIO
28 0.8546 0.3
29 INPUT MATERIAL DESCRIPTION FOR FASTENER
30 (a)
31 INPUT YOUNG'S MODULUS AND POISSON'S RATIO FOR
32 THE FASTENER
33 30.046 0.3
34 INPUT THE DIAMETER OF THE FASTENER
35 0.3185
36 FASTENER TYPE
37 ENTER 1 FOR PROTRUDING HEAD
38 2 FOR COUNTERSUNK HEAD
39
40 PLATE PROPERTIES ARE SPECIFIED BY
41 INPUTTING THE COORDINATES OF THE CORNER
42 LATHES. NOTE: THE ORIGIN IS AT THE FASTENER
43 CENTER; INPUT COORDINATES ACCORDINGLY
44
45 U3 U2
46 HOLE
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48 U4
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INPUT BETA1 FOR NET SECTION ULTIMATE
BETA2 FOR BEARING ULTIMATE
BETA3 FOR SHEAROUT ULTIMATE
1.02 1.5 1.12

TO AVOID LENGTHY RUN TIMES DUE TO
STRESS FIELD RECOMPUTATION SPECIFY THE
NUMBER OF ULTIMATE PLY FAILURES AFTER
WHICH JOINT FAILURE WILL BE PREDICTED
ENTER: NO OF ULTIMATE FAILURES
?

24
ENTER NO VALUES CORRESPONDING TO THE THREE
PLY FAILURE MODES IN PLATE NO 2

MODE - NET SECTION
MODE - BEARING
MODE - SHEAR OUT
INPUT MODE, MODE, MODE
?

0.1 0.025 0.00

FOR PLATE NUMBER 1 ENTER THE THREE STRENGTHS
REQUIRED TO PREDICT THE THREE FAILURE MODES
FST-UNNOTCHED STRENGTH IN TENSION
FSC-UNNOTCHED STRENGTH IN COMPRESSION
FSO-UNNOTCHED STRENGTH IN SHEAR-OUT

INPUT FST,FSC,FSO
?

250.000 300.000 200.000
FOR PLATE NO 2 ENTER FIBER ULTIMATE
STRAIN VALUES

EPSILON ULT IN COMPRESSION
EPSILON ULT IN TENSION
GAMMA ULT IN SHEAR
(UNITS: IN/IN)
?

0.014 0.011 0.012
SASCJ ASSUMES A BILINEAR PLY BEHAVIOR. THE
INITIAL MODULUS, E1, IS COMPUTED BY THE CODE.
THE REDUCED MODULUS, E2, FOR INITIAL FAILURE
IN NET SECTION, SHEAROUT OR BEARING IS COMPUTED
BY THE FORMULA $E2 = \alpha \cdot E1$.

FOR PLATE NUMBER 1 INPUT ALPHA VALUES FOR
NET SECTION, SHEAROUT AND BEARING FAILURE
?

0.01 0.1 0.1
INPUT SCALE FACTORS FOR P ULTIMATE
CALCULATION SUCH THAT $P1(ULT) = BETA1(INITIAL)$
INPUT BETA1 FOR NET SECTION ULTIMATE
BETA2 FOR BEARING ULTIMATE
BETA3 FOR SHEAROUT ULTIMATE
?

1.02 1.5 1.12
SASCJ ASSUMES A BILINEAR PLY BEHAVIOR. THE
INITIAL MODULUS, E1, IS COMPUTED BY THE CODE.
THE REDUCED MODULUS, E2, FOR INITIAL FAILURE
IN NET SECTION, SHEAROUT OR BEARING IS COMPUTED
BY THE FORMULA $E2 = \alpha \cdot E1$.

FOR PLATE NUMBER 2 INPUT ALPHA VALUES FOR
NET SECTION, SHEAROUT AND BEARING FAILURE
?

0.1 0.1 0.1
INPUT SCALE FACTORS FOR P ULTIMATE
CALCULATION SUCH THAT $P1(ULT) = BETA1(INITIAL)$

Figure 36. Sample SASCJ Input
(Concluded).

by the user. In Figure 36, each physical ply is modeled as one layer. For this sample problem, for example, the user could also specify each physical ply to be divided into two identical plies, by setting the number of plies in the laminate to be 40, the cured ply thickness to be 0.003 inch, and repeating each ply orientation in the stacking sequence twice.

Subsequent to the above information, SASCJ requests the material properties for plates 1 and 2 (Young's modulus and Poisson's ratio for a metal, and Young's modulus, shear modulus and the major Poisson's ratio for each lamina, in the fiber coordinate system). The fastener modulus, Poisson's ratio, diameter and head type (protruding head or countersunk) are requested next. Following that, the geometry of the bolted plates is defined by specifying the coordinates for the plate corners, assuming that the origin is located at the center of the fastener hole.

The last block of data addresses the selected failure criterion and the corresponding failure parameters. In the sample problem in Figure 36, the average stress failure criteria are selected for failure prediction (4). The characteristic distances for net section, bearing and shear-out modes of failure are then specified for the two plates. This is followed by the unnotched strengths for the two plates under tension, compression and inplane shear. Next, SASCJ requests the parameters that define the bilinear material behavior. These are the factors that define the modulus change after initial failure, and the ratio of the ultimate ply failure load to the initial ply failure load. Different factors may be specified for the three failure modes. Finally, the approximate ultimate shear strain value is requested for delamination prediction. A large value is generally specified for a metallic plate, to preclude the prediction of delaminations that are not applicable to these materials.

3.3 SASCJ Output Description

For the sample problem defined in Figure 36, SAS CJ provides the output shown in Figure 37. The input data for the bolted plates is initially reproduced for user verification. Subsequently, the sequence of failures in the bolted laminate and the corresponding joint load levels are printed. Note that the ultimate failure of a ply (shear-out of the 45 degree plies) does not necessarily imply joint failure. In the considered sample problem, shear-out of the 0 degree plies limits the load-carrying capacity of the joint. Every ply suffers a two-stage failure as described before (Figure 31).

When executed in some systems, SAS CJ could yield underflow messages after many plies have suffered total failure. This may occur when the double precision format is not followed in entering input data. Nevertheless, the user is advised to ignore these messages.

3.4 Description of SAMCJ Analysis

This section presents an overview of the strength analysis in the SAMCJ computer code, a description of the developed special finite elements, and the analytical procedure used in SAMCJ to predict fastener loads, the critical fastener or cut-out location, the corresponding joint strength and the failure mode.

A flow chart of SAMCJ operations is presented in Figure 38. As input, SAMCJ requires the user to specify how the bolted plates are divided into plain elements and elements with loaded or unloaded holes. The bolted plates are currently assumed by SAMCJ to be subjected to uniaxial tensile or compressive loading, in a single or double shear configuration. Additional input requirements for the SAMCJ code include the material properties of the bolted plates and fasteners, and the fastener size, location and torque. The material properties of the bolted laminates include the tensile and compressive failure strains in the fiber direction of the lamina, and the characteristic distances over which stresses are averaged to

NET SECTION ULTIMATE (TENS) -0.2500+06 PSI
NET SECTION ULTIMATE (COMP) -0.3000+06 PSI
BEARING ULTIMATE -0.3000+06 PSI
SHEAROUT ULTIMATE -0.2000+06 PSI

CHARACTERISTIC DISTANCES

ADMT - 0.2000+02 INCHES
RODR - 0.4000+00 INCHES
ROSO - 0.5000+02 INCHES

PLATE NUMBER 2

LAMINATE STRENGTH

NET SECTION ULTIMATE (TENS) -0.1230+06 PSI
NET SECTION ULTIMATE (COMP) -0.1500+06 PSI
BEARING ULTIMATE -0.1500+06 PSI
SHEAROUT ULTIMATE -0.1000+06 PSI

CHARACTERISTIC DISTANCES

ADMT - 0.1000+00 INCHES
RODR - 0.2500+01 INCHES
ROSO - 0.8000+01 INCHES

GEOMETRY OF PLATE NO 1

COORDINATES OF CORNER VERTICES

-3.000, 0.000 3.000, 0.000

-3.000, -0.000 3.000, -0.000

FASTENER HOLE DIAMETER - 0.3130+00 INCHES

E/D RATIO - 0.0000+01

U/D RATIO - 0.0000+01

GEOMETRY OF PLATE NO 8

COORDINATES OF CORNER VERTICES

PROGRAM SASCI

A SINGLE LAP SHEAR JOINT WILL BE ANALYZED

WITH A PARTIALLY LOADED HOLE

STRESS RATIO - 0.0000+00

LOADED IN STATIC TENSION

PLATE NO 1

STEEL

T - 0.2500+00 INCHES

MATERIAL PROPERTIES

E1 -0.3000+08 PSI

E2 -0.3000+08 PSI

G12 -0.1150+08 PSI

MU12 -0.3000+00

MU21 -0.3000+00

PLATE NO 2

AS1/2501-5 (145/0/-45/012/0/00)S

T - 0.1200+00 INCHES

MATERIAL PROPERTIES

E1 -0.1850+08 PSI

E2 -0.1850+08 PSI

G12 -0.0500+08 PSI

MU12 -0.3000+00

MU21 -0.3000+00

FASTENER:

DIAMETER - 0.3130+00 INCHES

MATERIAL PROPERTIES

E -0.3000+08 PSI

MU -0.3000+00

FAILURE ANALYSIS

AN AVERAGE STRESS CRITERION WILL BE USED

PLATE NUMBER 1

LAMINATE STRENGTH

Figure 37. SASCI Output for the Problem Defined in Figure 36.

13-00000

[illegible]

5503

534
535
536

1471167120150515

NO - NO ADDITIONAL DAMAGE AT CURRENT JOINT LOAD
DL - DELAMINATION
SO - SHEAR-OUT
BR - BEARING
MS - ME - SECTION
SUD - ULTIMATE FAILURE AFTER SO AND DL
SU - ULTIMATE FAILURE IN SO
BU - ULTIMATE FAILURE IN BR
MSU - ULTIMATE FAILURE IN MS
ULT - ULTIMATE FAILURE

000

Figure 37. SASCJ Output for the Problem Defined in Figure 36 (Concluded).

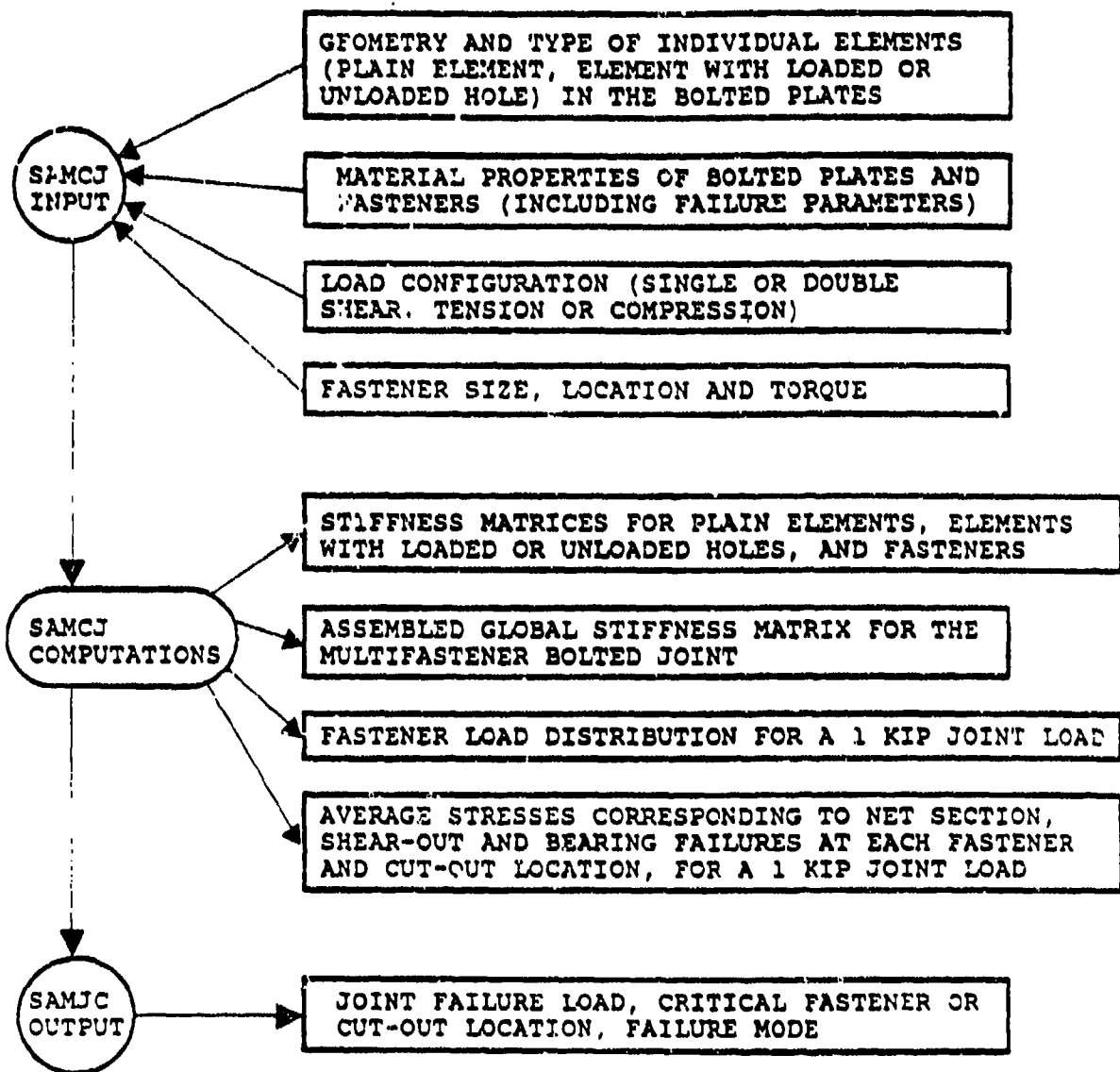


Figure 38. Flow Chart of SAMJC Operations.

predict net section, shear-out and bearing failures at the fastener or cut-out location.

With the above input, SAMCJ performs the following computations. It initially generates stiffness matrices for all the special finite elements, namely, plain elements, elements with loaded or unloaded holes, and effective fastener elements (see Reference 7). The individual stiffness matrices are subsequently assembled to obtain the global stiffness matrix for the bolted joint. A 1 kip uniaxial tensile or compressive joint load is imposed on the left end of the top plate, in accordance with the input instructions (see Figure 30). The nodes at the right end of the bottom plate are constrained from translating in the load direction, and one of these nodes is also constrained in the transverse direction, to preclude all rigid body translations. The solution to this finite element formulation of the bolted joint provides the axial and transverse components of the load at every fastener location, corresponding to a 1 kip joint load. Also computed are the average net section, shear-out and bearing stresses at every fastener and cut-out location, corresponding to a 1 kip joint load.

SAMCJ provides, as output, the failure value of the uniaxial joint load, the critical fastener or cut-out location, and the joint failure mode. These are obtained as follows. The tensile, compressive and shear strengths of the plain laminates are computed based on the input tensile and compressive failure strains in the fiber direction of the lamina. The ratios of the averaged stresses to the corresponding unnotched laminate strengths, at selected locations around each fastener and cut-out boundary, are compared to predict the failure mode, the critical fastener or cut-out location and the joint failure load. SAMCJ predicts net section, shear-out and bearing modes of failure at the laminate level. In the SASOJ code, similar failure predictions for single fastener joints in composites are made at the lamina level. Consequently, the failure parameters (characteristic distances for

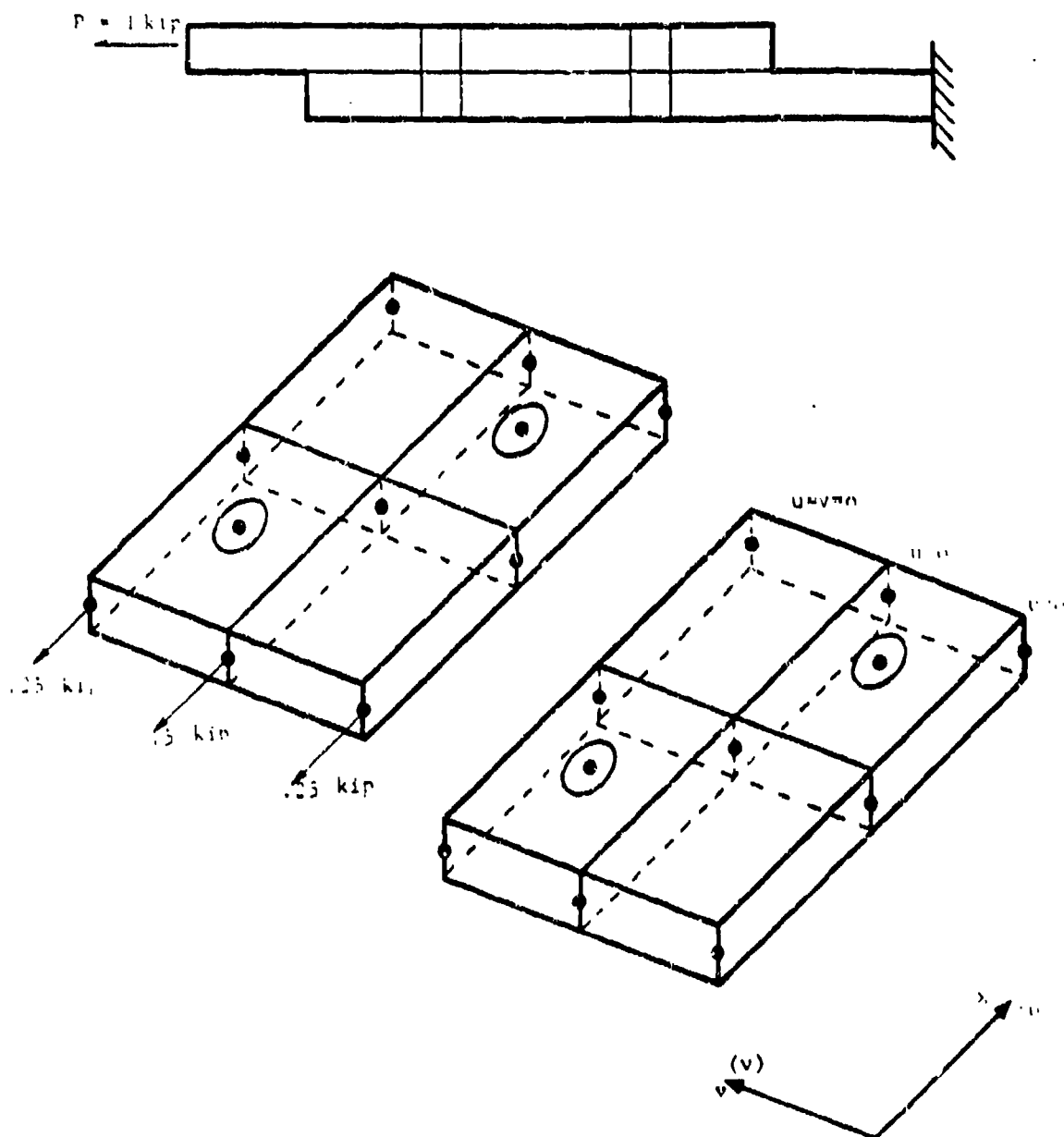


Figure 39. Application of Load and Displacement Boundary Conditions in the SAMCJ Code.

the three failure modes) used with SAMCJ are different from those used with SASCJ.

The incorporation of the transverse effective fastener stiffness values provides SAMCJ the capability to account for fastener flexibility, torque, and load eccentricity (single versus double shear load transfer). The FDFA code, developed in Reference 6, is used to compute the effective fastener transverse stiffnesses, along and perpendicular to the load direction (see Reference 7). The effect of the laminate stacking sequence is also accounted for in this analysis. SAMCJ executes FDFA twice to account for the layup variation (by 90 degrees) from the loading direction to the perpendicular direction.

SAMCJ accounts for stress concentration interaction effects introduced by neighboring cut-outs, free edges and proximate fastener locations. This is made possible by the use of the FIGEOM stress analysis, developed in Reference 6, to generate element stiffness matrices (see Reference 7). FIGEOM accounts for finite planform plate dimensions through a boundary collocation solution procedure (see Reference 6).

SAMCJ computes the magnitude and the orientation of the load at each fastener location. It is a two-dimensional load distribution analysis that does not rely on an experimental measurement of "joint stiffness." In a design situation, many fastener arrangements can be analytically and economically evaluated by SAMCJ to arrive at the best fastener pattern for the assumed loading conditions.

When the bolted plates are tapered, the SAMCJ user can input equivalent uniform thickness elements to approximate the tapering effect (see Figure 40). Adjacent elements in the tapered plate will have different thickness values. This feature is essential in the analysis of practical structural joints.

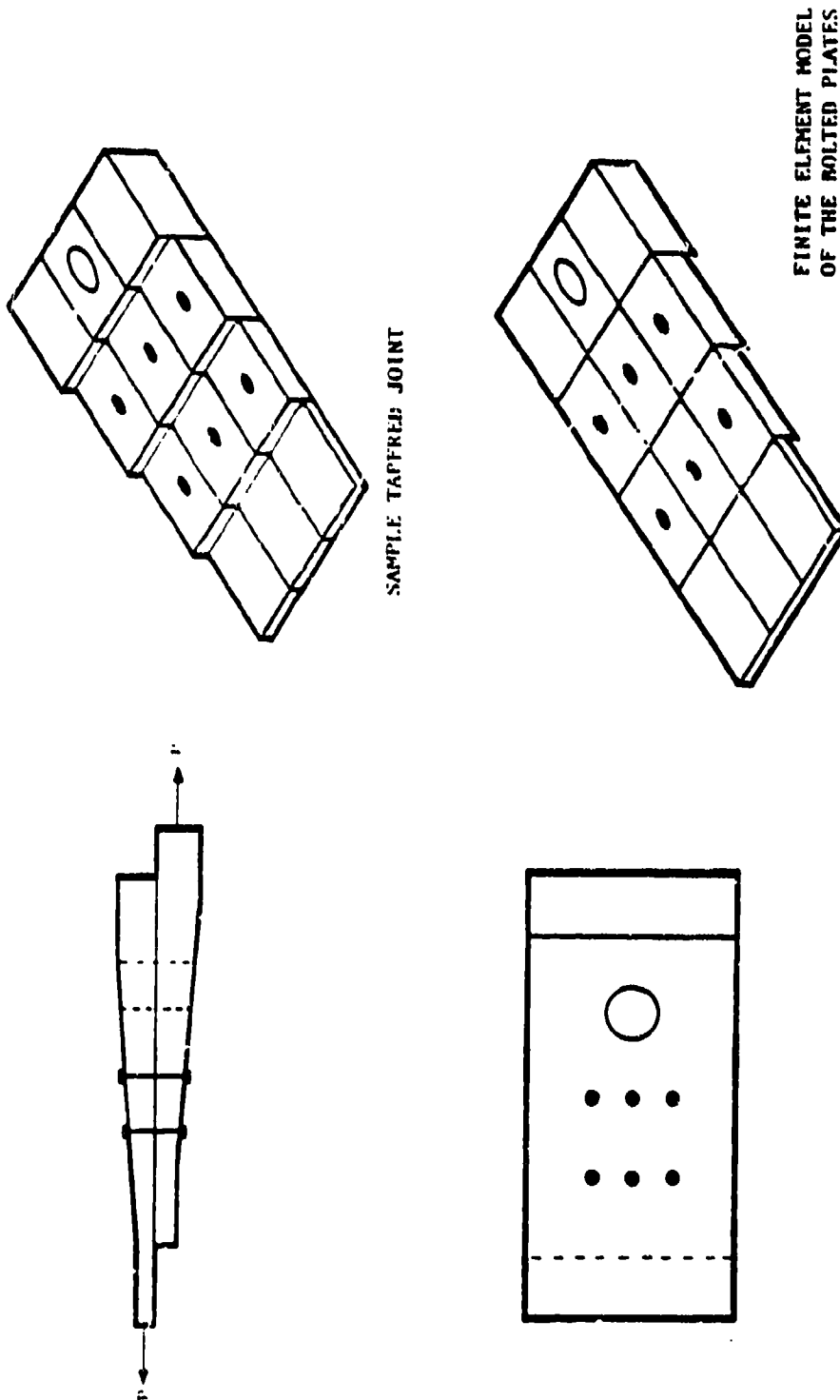
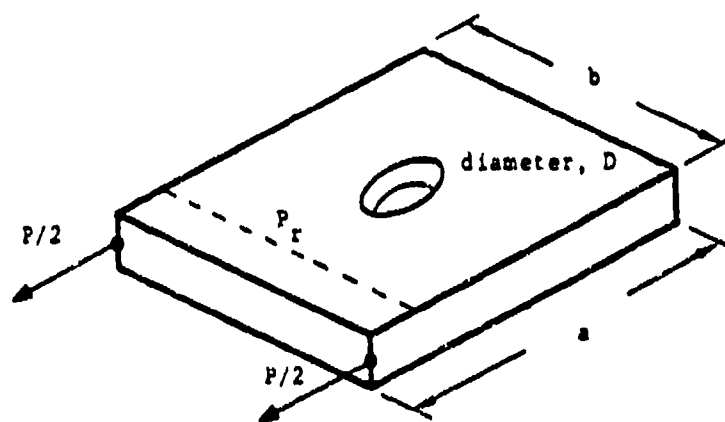


Figure 40. Finite Element Model of a Sample Tapered Bolted Joint.

SAMCJ has been developed for the strength prediction of bolted laminated structural parts. It currently assumes that the selected fasteners preclude fastener failure. Also, it applies the same failure procedure to both the bolted plates, accounting for net section, shear-out and bearing failures via the averaged stress failure criteria applied at the laminate level. Joint failure is assumed to be a one-step (catastrophic) process. The strength of a bolted plate corresponds to the initial failure at a fastener or a cut-out location, in the bearing, shear-out or net section failure.

The unnotched laminate strengths, under tension, compression and inplane shear, are computed by SAMCJ based on input fiber-directional failure strain values (tensile and compressive). Laminate strengths under N_x and N_{xy} loadings (inplane normal and shear stress resultants, respectively) are assumed to correspond to first fiber failure in a ply. This simplistic strength prediction procedure introduces inaccuracies that have been acknowledged and discussed in the literature. Nevertheless, SAMCJ adopts this procedure for lack of a validated alternative.

Despite its versatility, SAMCJ has limitations that the user should be aware of. Reference 7 discusses the limitations of the five-noded (10 degrees of freedom) loaded hole element and the four-noded (8 degrees of freedom) unloaded hole element. In addition, when dividing a bolted plate into many elements (loaded or unloaded hole elements, as well as plain elements), it is advisable to maintain element geometries that do not render the generated stiffness matrices inaccurate. Figure 41 presents results from a study conducted on a singly-fastened metallic plate. P_r is the recovered load that is obtained by integrating the stresses along a line transverse to the load direction as shown in Figure 41. P is the applied load or the sum of the nodal loads (especially in the interior elements in a general multifastened plate). The recovered load (P_r) approaches the applied load value (P) when the plate aspect ratio (a/b) increases beyond unity, and when a/D and b/D have a minimum value of approximately three. In predicting failure in



$t = 0.3125 \text{ inch}$

a/D	b/D	P_r/P
1.6	1.6	5.38
3.2	1.6	2.27
6.4	1.6	1.57
16.0	1.6	1.29
1.6	3.2	1.74
3.2	3.2	1.76
6.4	3.2	1.37
16.0	3.2	1.16
1.6	6.4	-0.0995
3.2	6.4	0.989
6.4	6.4	1.23
16.0	6.4	1.16
3.2	16.0	-0.46
6.4	16.0	0.029
16.0	16.0	1.23

Figure 41. Element Load Recovery for Various a/D and b/D Ratios.

the net section, bearing and shear-out modes, the computed average stress values are multiplied by P/P_r , to remove geometry (modeling) effects from the computed stresses.

3.5 SAMCJ Input Description

To familiarize the user with SAMCJ input requirements, a sample problem is presented here (see Figure 42). The sample problem considers a six fastener composite-to-metal joint, with a one inch diameter circular cut-out adjacent to the first row of fasteners. Figure 42 presents the assumed nine element model of each of the two bolted plates, analyzed by SAMCJ. Figure 43 presents SAMCJ requests and user input in response to these requests, for the sample problem in Figure 42.

Though self-explanatory, the interactively entered SAMCJ input in Figure 43, for the sample problem in Figure 42, is described here for completeness. The first entry (1) identifies the loading configuration to be a single shear configuration. The second entry (1) identifies the load to be in static tension. The next two entries say that the top plate is a metal (M), identified as "Aluminum." The two entries following these say that the bottom plate is a composite laminate (C), identified as follows: "(45/0/-45/0)2/0/90)2s." Subsequently, the Young's modulus (10.0D6) and Poisson's ratio (0.3) for aluminum, and the fiber-directional, transverse and shear moduli and Poisson's ratio (18.5D6, 0.85D6 and 0.3, respectively) for the composite lamina are input. The next five entries specify that four (4) different fiber orientations are present in the laminate (0, 45, -45 and 90 degrees with respect to the loading direction). The following three entries say that the elements in the bottom plate contain one (1) layup of forty (40) plies, of 0.006175 inch thickness each. The stacking sequence for this layup is input next, where 1, 2, 3 and 4 refer to 0, 45, -45 and 90 degree fiber orientations, respectively. Subsequently, the fastener is identified as "Steel," and its Young's modulus, Poisson's ratio, and head type (30.0D6, 0.3, 0.3125 and

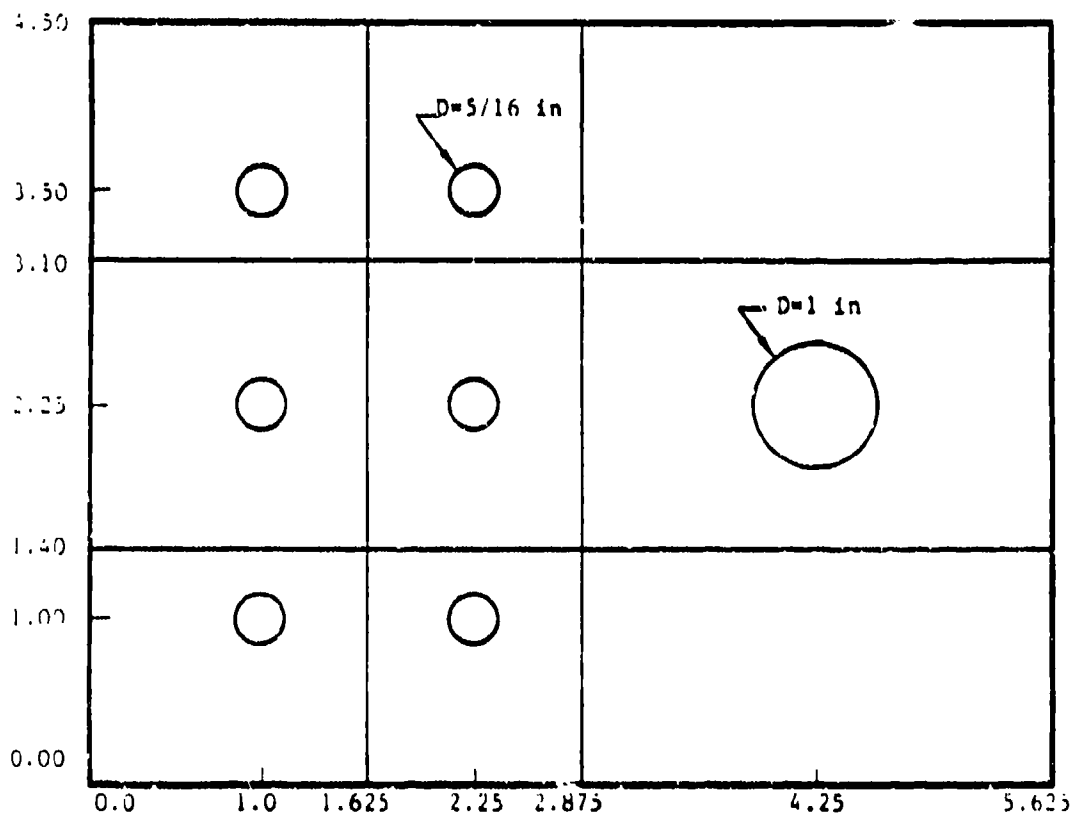
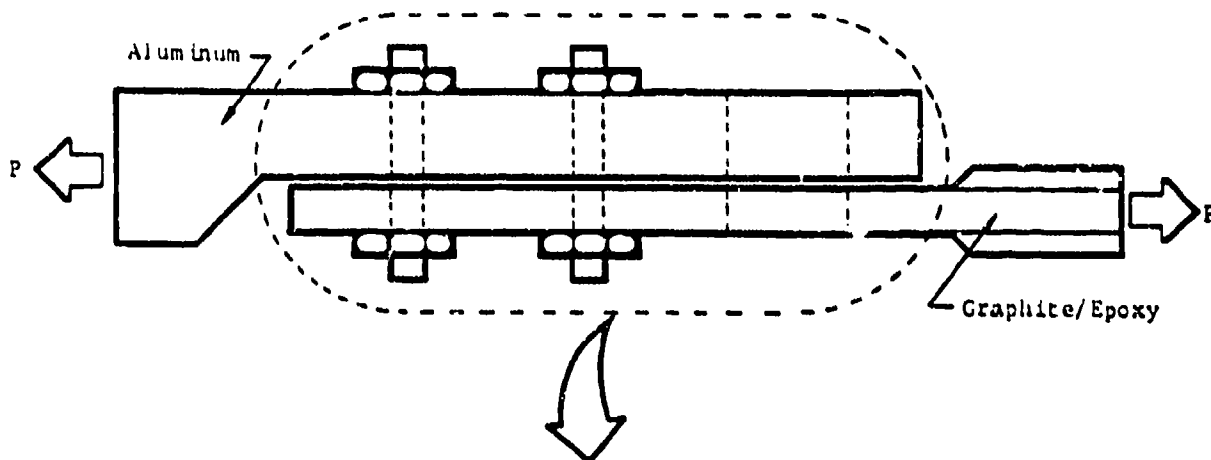


Figure 42. Nine Element Model of Each of the two Bolted Plates in the Sample Joint.

```

CALL NAMEJ
14.50061 TEMPERATURE ASSUMED AS "TEMPERATURE"

PROGRAM NAMEJ

PROGRAM NAMEJ PREDICTS THE FAILURE LOAD, FAILURE
LOCATION, AND FAILURE MODE IN MULTILAYER-FASTENED,
SINGLE OR DOUBLE LAP COMPOSITE SHEAR JOINTS.

THE ANALYSIS ASSUMES THAT INPUT PARAMETERS ARE
SPECIFIED IN ENGLISH UNITS - LENGTH IN INCHES,
MODULI AND STRENGTHS IN PSI.

ENTER:
1 FOR SLS (SINGLE LAP SHEAR)
2 FOR DLS (DOUBLE LAP SHEAR)

1
ENTER:
1 FOR STATIC TENSION
2 FOR STATIC COMPRESSION

1
1
ENTER:
1 IS THE TOP PLATE A COMPOSITE OR A METAL?
ENTER C OR M IN THE FIRST FIELD
0
INPUT MATERIAL DESCRIPTION OF THIS PLATE
0.001000
EX: ASA/3501-6
1 IS THE BOTTOM PLATE A COMPOSITE OR A METAL?
ENTER C OR M IN THE FIRST FIELD
0
INPUT MATERIAL DESCRIPTION OF THIS PLATE
0.001000
EX: ASA/3501-6
1:45/0/-45/0/12/0/0010

NOTE: FOR COMPUTATIONAL PURPOSES A
METALLIC PLATE IS MODELED AS A 30 PLY
LAMINATE OF 0 DEGREE PLYS WITH ISOTROPIC
MATERIAL PROPERTIES

INPUT THE ENGINEERING PROPERTIES OF THE TOP PLATE
INPUT YOUNG'S MODULUS AND POISSON'S RATIO
1
10.000 0.3
INPUT THE ENGINEERING PROPERTIES OF THE BOTTOM PLATE
INPUT YOUNG'S MODULUS, E1 AND E2
1
10.000 1.000
INPUT THE SHEAR MODULUS AND MAJOR POISSON'S RATIO
1
0.000 0.3
INPUT TOTAL NUMBER OF DISTINCT PLY
ORIENTATIONS IN THE BOTTOM PLATE
1
INPUT ORIENTATION OF PLY TYPE NO 1
0.0
  
```

Figure 43. NAMEJ Input for the Sample Problem in Figure 42.

```

0.3125
FASTENER TYPE
ENTER:
1 1 FINE PROUDING HEAD
2 2 FINE COUNTERSUNK HEAD

GRID LAYOUTS
ENTER NUMBER OF GRIDS IN TOP PLATE
7
22

ENTER 22 GRID POINTS
FORMAT: GRID ID, X AND Y COORDINATES
7
101 0.0 0.0
7
102 0.0 1.40
7
103 0.0 3.1
7
104 0.0 4.5
7
105 1.0 1.0
7
106 1.0 2.25
7
107 1.0 3.5
7
108 1.025 0.0
7
109 1.025 1.4
7
110 1.025 3.1
7
111 1.025 4.5
7
112 2.25 1.0
7
113 2.25 2.25
7
114 2.25 3.5
7
115 2.275 0.0
7
116 2.275 1.4
7
117 2.275 3.1
7
118 2.275 4.5
7
119 5.025 0.0
7
120 5.025 1.4

FASTENER DESCRIPTIONS
INPUT MATERIAL DESCRIPTION FOR FASTENER
Steel
INPUT YOUNG MODULUS AND POISSONS RATIO FOR
THE FASTENER
7
20.00E 0.3
INPUT THE DIAMETER OF THE FASTENER
7

```

Figure 43. SAMCJ Input for the Sample Problem in Figure 42 (Continued).

```

7 121 5.625 3.1
7 122 5.625 4.5
7 ENTER NUMBER OF CPIDS IN BOTTOM PLATE
82
7 ENTER 22 GRID POINTS
7 FORMAT: GRID ID, X AND Y COORDINATES
7 201 0.0 0.0
7 202 0.0 1.4
7 203 0.0 3.1
7 204 0.0 4.5
7 205 1.0 1.0
7 206 1.0 2.25
7 207 1.0 3.5
7 208 1.625 0.0
7 209 1.625 1.4
7 210 1.625 3.1
7 211 1.625 4.5
7 212 2.25 1.0
7 213 2.25 2.25
7 214 2.25 3.5
7 215 2.875 0.0
7 216 2.875 1.4
7 217 2.875 3.1
7 218 2.875 4.5
7 219 5.625 0.0
7 220 5.625 1.4
7 221 5.625 3.1
7 222 5.625 4.5
7 ELEMENT DESCRIPTION:

7 M2 M3
7 M1 M4

ELEMENT TYPES ARE DESIGNATED AS FOLLOWS:
7 4 MODE PLAIN ELEMENT TYPE NO. 1
7 5 MODE LOADED HOLE ELEMENT TYPE NO. 2
7 4 MODE OPEN HOLE ELEMENT TYPE NO. 3

7 (NOTE: ENTER M6-8 FOR FOUR MODE ELEMENTS)
7 ENTER NUMBER OF ELEMENTS IN TOP PLATE
7 9
7 FOR ELEMENT NO 1 ENTER: ELEMENT ID,M1,M2,M3,M4,M5,ELEMENT TYPE
7 101 102 100 100 105 2
7 ENTER ELEMENT THICKNESS
7 0.5
7 FOR ELEMENT NO 2 ENTER: ELEMENT ID,M1,M2,M3,M4,M5,ELEMENT TYPE
7 102 102 103 110 100 105 2
7 ENTER ELEMENT THICKNESS
7 0.5
7 FOR ELEMENT NO 3 ENTER: ELEMENT ID,M1,M2,M3,M4,M5,ELEMENT TYPE
7 103 103 104 111 110 107 2
7 ENTER ELEMENT THICKNESS
7 0.5
7 FOR ELEMENT NO 4 ENTER: ELEMENT ID,M1,M2,M3,M4,M5,ELEMENT TYPE
7 104 104 109 115 115 110 2
7 ENTER ELEMENT THICKNESS
7 0.5
7 FOR ELEMENT NO 5 ENTER: ELEMENT ID,M1,M2,M3,M4,M5,ELEMENT TYPE
7 105 105 110 117 116 113 8
7 ENTER ELEMENT THICKNESS
7 0.5
7 FOR ELEMENT NO 6 ENTER: ELEMENT ID,M1,M2,M3,M4,M5,ELEMENT TYPE
7 106 110 111 118 117 114 8
7 ENTER ELEMENT THICKNESS
7 0.5
    
```

PLATE ELEMENTS ARE NUMBERED
 CLOCKWISE AS SHOWN

Figure 43. SANCJ Input for the Sample Problem in Figure 42 (Continued).

```

FOR ELEMENT NO 7 ENTER: ELEMENT ID,M1,M2,M3,M4,M5,ELEMENT TYPE
107 115 116 120 119 0 1
ENTER ELEMENT THICKNESS
?
0 8
FOR ELEMENT NO 8 ENTER: ELEMENT ID,M1,M2,M3,M4,M5,ELEMENT TYPE
108 115 117 121 120 0 3
ENTER ELEMENT THICKNESS, X AND Y COORDINATES
OF OPEN HOLE AND HOLE RADIUS
?
0.5 4.25 2.25 0.5
FOR ELEMENT NO 9 ENTER: ELEMENT ID,M1,M2,M3,M4,M5,ELEMENT TYPE
109 117 118 122 121 0 1
ENTER ELEMENT THICKNESS
?
0.5
ENTER NUMBER OF ELEMENTS IN BOTTOM PLATE
?
9
FOR ELEMENT NO 10 ENTER: ELEMENT ID,M1,M2,M3,M4,M5,ELEMENT TYPE
201 201 208 209 208 205 2
ENTER ELEMENT LAYOUT NO
?
1
FOR ELEMENT NO 11 ENTER: ELEMENT ID,M1,M2,M3,M4,M5,ELEMENT TYPE
202 202 203 210 209 206 2
ENTER ELEMENT LAYOUT NO
?
1
FOR ELEMENT NO 12 ENTER: ELEMENT ID,M1,M2,M3,M4,M5,ELEMENT TYPE
203 203 204 211 210 207 2
ENTER ELEMENT LAYOUT NO
?
1
FOR ELEMENT NO 13 ENTER: ELEMENT ID,M1,M2,M3,M4,M5,ELEMENT TYPE
204 204 209 216 215 212 2
ENTER ELEMENT LAYOUT NO
?
1
FOR ELEMENT NO 14 ENTER: ELEMENT ID,M1,M2,M3,M4,M5,ELEMENT TYPE
205 209 210 217 219 213 2
ENTER ELEMENT LAYOUT NO
?
1
FOR ELEMENT NO 15 ENTER: ELEMENT ID,M1,M2,M3,M4,M5,ELEMENT TYPE
206 210 211 218 217 214 2
ENTER ELEMENT LAYOUT NO
?
1

FOR ELEMENT NO 16 ENTER: ELEMENT ID,M1,M2,M3,M4,M5,ELEMENT TYPE
207 215 216 220 219 0 1
ENTER ELEMENT LAYOUT NO
?
1
FOR ELEMENT NO 17 ENTER: ELEMENT ID,M1,M2,M3,M4,M5,ELEMENT TYPE
208 216 217 221 220 0 3
ENTER ELEMENT LAYOUT NUMBER, X AND Y
COORDINATES OF THE OPEN HOLE AND THE HOLE
RADIUS
?
1 4.25 2.25 0.5
FOR ELEMENT NO 18 ENTER: ELEMENT ID,M1,M2,M3,M4,M5,ELEMENT TYPE
209 217 218 222 221 0 1
ENTER ELEMENT LAYOUT NO
?
1
FASTENERS ARE MODELED BY EFFECTIVE
FASTER ELEMENTS WHICH PROVIDE THE
ELASTIC LINK BETWEEN THE TOP AND
BOTTOM PLATES
ENTER NUMBER OF FASTENERS IN JOINT
?
6

EFFECTIVE FASTENER ELEMENTS ARE
NUMBERED AS SHOWN:
M1 (TOP PLATE)
M2 (BOTTOM PLATE)

WHERE M1 AND M2 CORRESPOND TO THE CENTRAL
HOLES IN LOADED HOLE ELEMENTS

FORMAT: ELEMENT ID, M1, M2
ENTER ELEMENT NO 1
?
101 105 205
ENTER ELEMENT NO 2
?
102 105 205
ENTER ELEMENT NO 3
?
103 107 207
ENTER ELEMENT NO 4
?
104 112 218
ENTER ELEMENT NO 5
?
105 113 213
ENTER ELEMENT NO 6
?
106 114 214
  
```

Figure 43. SAMCJ Input for the Sample Problem in Figure 42 (Continued).

TO REDUCE RUN TIMES, ELEMENTS MAY BE
 GROUPED INTO SETS WHICH WILL BE ASSIGNED
 IDENTICAL STIFFNESS MATRICES.
 ENTER: 1 TO USE THIS OPTION
 2 OTHERWISE

1
 2
 3

FOR THE TOP PLATE INPUT NUMBER OF GROUPS
 FOR THE EFFECTIVE FASTENER, LOADED HOLE, UNLOADED
 HOLE AND PLAIN ELEMENT
 (INPUT 0 IF ELEMENT TYPE IS NOT USED)

1 6 1 1
 ENTER NUMBER OF EFFECTIVE FASTENER ELEMENTS
 ENTER NUMBER OF ELEMENTS IN GROUP NUMBER 1

5
 ENTER 6 ELEMENT IDS
 101 102 103 104 105 106

GROUPING OF LOADED HOLE ELEMENTS:
 ENTER NUMBER OF ELEMENTS IN GROUP NUMBER 1

1
 INPUT 1 ELEMENT ID6
 101
 ENTER NUMBER OF ELEMENTS IN GROUP NUMBER 2

1
 INPUT 1 ELEMENT ID5
 102
 ENTER NUMBER OF ELEMENTS IN GROUP NUMBER 3

1
 INPUT 1 ELEMENT ID5
 103
 ENTER NUMBER OF ELEMENTS IN GROUP NUMBER 4

1
 INPUT 1 ELEMENT ID5
 104
 ENTER NUMBER OF ELEMENTS IN GROUP NUMBER 5

1
 INPUT 1 ELEMENT ID6
 105
 ENTER NUMBER OF ELEMENTS IN GROUP NUMBER 6

1
 INPUT 1 ELEMENT ID6
 106
 ENTER NUMBER OF UNLOADED HOLE ELEMENTS
 ENTER NUMBER OF ELEMENTS IN GROUP NUMBER :

1
 ENTER 1 ELEMENT ID5
 107
 ENTER 8 ELEMENT IDS
 107 108
 FOR THE BOTTOM PLATE INPUT NUMBER OF GROUPS
 FOR THE LOADED HOLE, UNLOADED HOLE, AND PLAIN
 ELEMENTS
 (INPUT 0 IF AN ELEMENT TYPE IS NOT USED)

7
 6 1 1

GROUPING OF LOADED HOLE ELEMENTS:
 ENTER NUMBER OF ELEMENTS IN GROUP NUMBER 1

1
 INPUT 1 ELEMENT ID5
 201
 ENTER NUMBER OF ELEMENTS IN GROUP NUMBER 2

1
 INPUT 1 ELEMENT ID5
 202
 ENTER NUMBER OF ELEMENTS IN GROUP NUMBER 3

1
 INPUT 1 ELEMENT ID5
 203
 ENTER NUMBER OF ELEMENTS IN GROUP NUMBER 4

1
 INPUT 2 ELEMENT ID6
 204
 ENTER NUMBER OF ELEMENTS IN GROUP NUMBER 5

1
 INPUT 1 ELEMENT ID6
 205
 ENTER NUMBER OF ELEMENTS IN GROUP NUMBER 6

1
 INPUT 1 ELEMENT ID5
 206

Figure 43. SAHCL Input for the Sample Problem in Figure 42 (Continued).

```

GROUPING OF UNLOADED NOTCH ELEMENTS
ENTER NUMBER OF ELEMENTS IN GROUP NUMBER 1
1
ENTER 1 ELEMENT IDS
200
GROUPING OF PLAIN ELEMENTS:
ENTER NUMBER OF ELEMENTS IN GROUP NUMBER 1
1
ENTER 2 ELEMENT IDS
207 200
INPUT DATA FOR FAILURE ANALYSIS:
ENTER METALLIC STRENGTHS:
TENSILE STRENGTH
COMPRESSIVE STRENGTH
SHEAR STRENGTH
7
250.000 250.000 250.000
AN AVERAGE STRESS CRITERIA IS USED TO
PREDICT FAILURE. NO VALUES ARE REQUIRED AS
CHARACTERISTIC DISTANCES OVER WHICH STRESSES
ARE TO BE AVERAGED AND COMPARED TO UNNOTCHED
LAPINATE STRENGTHS TO PREDICT FAILURE
ENTER NO VALUES FOR STRESS AVERAGING
FOR EACH FAILURE MODE IN PLATE NO 1
1
MODE - NET SECTION
MODE - BEARING
MODE - SHEAROUT
7
0.5 0.5 0.5
ENTER FINEST ULTIMATE STRAIN VALUES
IN PLATE NO 2
EPSILON ULT IN COMPRESSION
EPSILON ULT IN TENSION
EPSILON ULT IN SHEAR
7
0.0175 0.012 0.012
ON AVERAGE STRESS CRITERIA IS USED TO
PREDICT FAILURE. NO VALUES ARE REQUIRED AS
CHARACTERISTIC DISTANCES OVER WHICH STRESSES
ARE TO BE AVERAGED AND COMPARED TO UNNOTCHED
LAPINATE STRENGTHS TO PREDICT FAILURE
ENTER NO VALUES FOR STRESS AVERAGING
FOR EACH FAILURE MODE IN PLATE NO 2
MODE - NET SECTION
MODE - BEARING
MODE - SHEAROUT
7
0.1 0.05 0.05
  
```

Figure 43. SAKCJ Input for the Sample Problem in Figure 42 (Concluded).

protruding head) are input.

Twenty-two (22) grid points each are specified in the top and bottom plates (101 to 122 and 201 to 222, respectively), along with their x and y coordinates (see Figure 42). Following this, nine (9) elements are specified in each plate, along with their nodal connectivity and element type information. Nodal connectivity is specified starting from the bottom left node, going clockwise around the element boundary, and ending at the fastener (internal) node. Element 101 in the top plate, for example, has 101, 102, 109 and 108 as its corner nodes, and 105 as its fastener node. The fifth node will be entered as 0 for plain and unloaded hole elements. The element type information follows the fifth node identification. It is 1, 2 and 3 for plain, loaded hole and unloaded hole elements, respectively. The element definitions are succeeded by the definition of six (6) effective fasteners (101 to 106). Fastener 101, for example, is identified as a fastener that connects node 105 in the top plate to node 205 in the bottom plate.

Following the above input, additional element data are specified for the two plates. These include the element thicknesses (for metallic plates) or layup identification number (for laminated plates), for plain and loaded hole elements, with additional information (x and y coordinates of the hole center and the hole radius) for unloaded hole elements. For the sample problem in Figure 42, all the elements in the top plate (metal) are specified to be 0.50 inch thick, and all the elements in the bottom plate (composite) are specified to contain the stacking sequence identified as one (1). Elements 108 and 208 specify the cut-out size and location. The one (1) following this states that groups of identical elements will be specified in the two plates. If two (2) is entered here, all elements will be assumed to be different from one another, resulting in larger computational costs. The entry "1 6 1 1" refers to the number of groups of effective fasteners, loaded hole, unloaded hole and plain elements, respectively, in the top plate. A zero (0) specifies the absence of an element type.

The number of elements in each group, and the corresponding element numbers, are input subsequently. Following this, the number of groups of loaded hole, unloaded hole and plain elements in the bottom plate (6, 1 and 1, respectively) is entered.

The last four lines of input introduce the failure parameters for the materials in the two plates. For metallic plates, the tensile, compressive and shear strengths (250.0D3 each), and the averaging distances for the net section, bearing and shear-out modes of failure (0.5 each) are input. Since the joints were designed to fail the laminated plates, and SAMCJ was developed primarily for the prediction of the strength of bolted laminates, the failure parameters for the metallic plates were input to be arbitrarily high. This information is followed by the failure parameters for the bottom (composite) plate. The first line specifies the fiber directional failure strains for the material under tension (0.012) and compression (0.0175). These values are used by SAMCJ to compute the unnotched laminate tensile, compressive and shear strengths, based on laminated plate theory and the assumption of laminate failure corresponding to the first fiber failure in any of its plies. The last line in Figure 43 specifies the distance over which the longitudinal (0.10 and 0.25) and shear (0.25) stress components are averaged, to predict net section, bearing and shear-out modes of failure, respectively.

3.6 SAMCJ Output Description

For the sample problem introduced in Section 3.5, the SAMCJ code yields the output presented in Figure 44. The initial part of the output reprints critical user-supplied information for verification purposes. Subsequently, SAMCJ prints the x and y components of the element nodal forces for all the elements in the bolted plates. This is followed by a list of the computed joint load levels that correspond to the three failure modes (net section, shear-out and bearing) at every loaded and unloaded hole element location. The smallest among these loads yields the joint failure

```

PROGRAM SANCJ

A SINGLE LAP SHEAR PANEL WILL BE ANALYZED
LOADED IN STATIC TENSION
  PLATE NO 1
ALUMINUM
MATERIAL PROPERTIES
  E1 -0.1000+00 PSI
  E2 -0.1000+00 PSI
  G12 -0.1079+00 PSI
  MU12 -0.3000+00
  MU21 -0.3000+00
  PLATE NO 2
AL1 3001-C
MATERIAL PROPERTIES
  E1 -0.1000+00 PSI
  E2 -0.1000+07 PSI
  G12 -0.1079+00 PSI
  MU12 -0.3000+00
  MU21 -0.3000+00
FASTENER DESCRIPTION
STEEL
DIAMETER - 0.3120+00 INCHES
MATERIAL PROPERTIES
  E -0.3000+00 PSI
  RU -0.3000+00
FAILURE ANALYSIS
AN ALUMINUM SHEAR CRITERION WILL BE USED
PLATE NUMBER 1
METALLIC STRENGTH
  TENSILE STRENGTH - 0.8000+00
  COMPRESSIVE STRENGTH - 0.8000+00
  SHEAR STRENGTH - 0.8000+00
CHARACTERISTIC DISTANCES
  AGENT - 0.5000+00 INCHES
  AGSM - 0.5000+00 INCHES
  AGSC - 0.5000+00 INCHES
  PLATE NUMBER 8
  FIBER STRAIN ULTIMATES
  EPSILON ULT COMP - 0.1750+01
  EPSILON ULT TEN - 0.1820+01
  GAMMA ULT SHEAR - 0.1200+01
CHARACTERISTIC DISTANCES
  AGENT - 0.1000+00 INCHES
  AGSM - 0.2500+00 INCHES
  AGSC - 0.2500+00 INCHES
PAUSE FOR STIFFNESS MATRIX CALCULATIONS
ELEMENT FORCES
  (P APPLIED - 10.000 LBS)
ELEMENT ID 101
  GRID FX FY
  101 -1.000+03 0.1750+12
  102 -1.000+03 -0.2840+02
  103 0.0000+00 0.1500+02
  104 0.7570+02 0.1250+02
  105 0.1000+01 0.1000+00
ELEMENT ID 102
  GRID FX FY
  102 -1.000+03 0.0240+00
  103 -1.000+03 -0.2840+02
  104 0.0000+00 -0.5000+01
  105 0.0000+00 0.0000+00
  106 0.1000+01 0.0000+00
ELEMENT ID 103
  GRID FX FY
  103 -1.000+03 0.0240+00
  104 -1.000+03 -0.2840+02
  105 0.0000+00 -0.5000+01
  106 0.0000+00 0.0000+00
  
```

Figure 44. SANCJ Output for the Problem Defined in Figure 43.

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 permit fully legible reproduction

Figure 44. SANCJ Output for the Problem Defined in Figure 43 (Continued).

ELEMENT ID	207		
CRIP	FX	FV	
215	1200+03	0.2560+02	
216	-1630+03	-1080+01	
217	0.1260+03	-1750+02	
219	0.1630+03	-1390+14	
ELEMENT ID	208		
CRIP	FX	FV	
216	-2120+03	0.5850+02	
217	-2110+03	-5910+02	
221	0.2120+03	-1690+02	
220	0.2110+03	0.1750+02	
ELEMENT ID	209		
CRIP	FX	FV	
217	-1590+03	0.8410+01	
218	-1290+03	-2530+02	
222	0.1620+03	-1780+13	
221	0.1260+03	0.1690+02	

JOINT LOAD LEVELS CORRESPONDING TO NET SECTION (NS), SHEAR-OUT (SO) AND BEARING (BR) FAILURES AT EVERY LOADED AND UNLOADED HOLE ELEMENT ARE PREDICTED AS FOLLOWS:

ELEMENT	NS	SO	BR
101	0.5750+06	0.0830+06	0.4590+06
102	0.5460+06	0.7000+06	0.5010+06
103	0.4170+06	0.6260+06	0.4920+06
104	0.2920+07	0.1510+07	0.7300+06
105	0.3110+07	0.1490+07	0.8370+06
106	0.2030+07	0.1710+07	0.7370+06
108	0.1480+08	0.8900+06	0.8270+06
201	0.3400+06	0.1230+06	0.1200+06
202	0.2950+06	0.9700+06	0.1950+06
203	0.5130+06	0.1200+06	0.1320+06
204	0.7040+05	0.6440+05	0.7700+05
205	0.5290+05	0.4110+05	0.5770+05
206	0.8080+05	0.5550+05	0.8130+05
208	0.3730+05	0.1740+06	0.3020+07

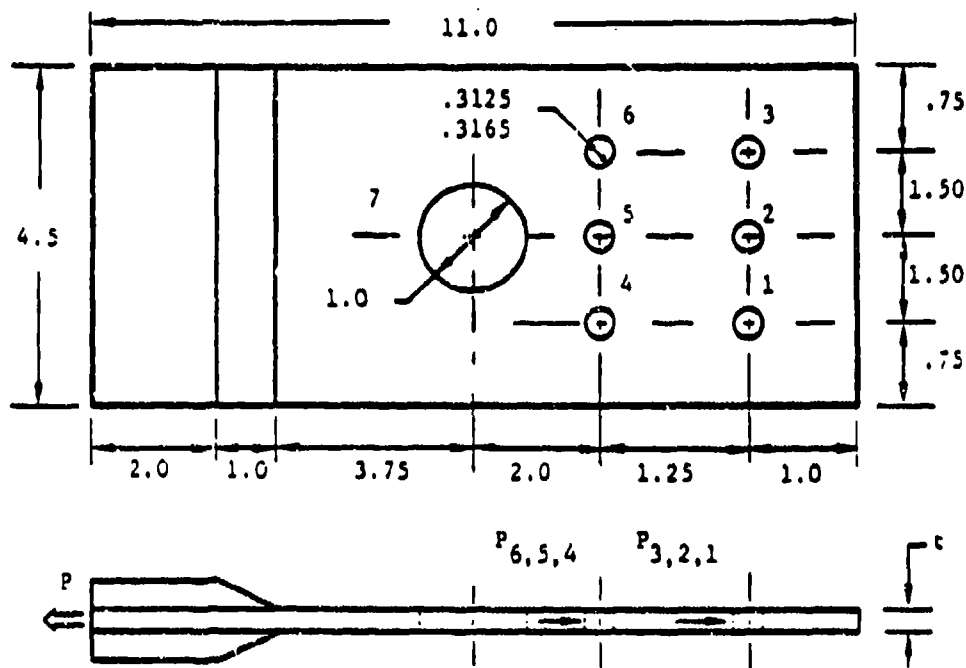
FAILURE IS PREDICTED TO OCCUR IN ELEMENT NUMBER 208 AT AN APPLIED JOINT LOAD VALUE OF 0.3720000+05 LBS

THE PREDICTED FAILURE MODE IS NET SECTION

Figure 44. SAMCJ Output for the Problem Defined in Figure 43 (Concluded).

load, the failure location and the failure mode. For the considered sample problem, a net section failure is predicted across the one inch diameter cut-out (element 208) in the graphite/epoxy plate, at a joint load level of 37.3 kips. Figure 45 compares SAMCJ predictions with test results from Reference 8.

Test Case 243, Static Tension, Single Lap
 40-Ply, 50/40/10 Laminate, $t=0.247$ in., $t_{AL}=0.50$ in.
 $D=5/16$ in., $H_D=1$ in., $S_L/D=S_T/D=4$, $W/D=14.4$, $E/D=3.2$



	SAMCJ PREDICTION	TEST RESULTS (Ref. 2)
P_1/P	0.165	0.162
P_2/P	0.191	0.150
P_3/P	0.165	0.168
P_4/P	0.167	0.177
P_5/P	0.175	0.161
P_6/P	0.168	0.165
$P_{failure}$ (kips)	37.3 (52.7)*	42.0
FAILURE LOCATION	7 (5)	7 and 4, 5, 6
FAILURE MODE(S)	NET SECTION (NET SECTION)	NET SECTION

* Next possible failure mode and location at a higher load level
 Figure 45. Comparison of SAMCJ Predictions for the Sample Problem with
 Test Results from Reference 8.

SECTION 4

DESIGN VERIFICATION OF A BOLTED STRUCTURAL ELEMENT

The design of a highly-loaded structural bolted joint is verified in this section using the analytical tool (SAMCJ computer code) proposed for the recommended design methodology (Section 1.3).

4.1 Description of the Bolted Structural Element

In Reference 5, a bolted joint concept was studied as an alternative to a highly loaded composite-to-titanium, step lap bonded joint. The vertical tail structure of the F/A-18A was used as the baseline for this study. A preliminary design of the bolted structural element, representative of the critical F/A-18A vertical tail root section, was performed based on approximate analyses and available test results. The test element was designed to transfer a design ultimate load of 70.2 kips (obtained from the F/A-18A empennage stress analysis report), and to survive two lifetimes of a representative design spectrum fatigue loading.

The design of the bolted structural element studied in Reference 5 differs from the existing F/A-18A vertical tail root joint significantly. It eliminates the graphite/epoxy skin-to-titanium bonded joint, and directly attaches the skins to the fuselage frame. In doing so, it also uses a light root rib, in contrast to the highly-loaded attachment root rib used in Reference 4. The AS4/3501-6 graphite/epoxy skins of the element have a 41-ply layup away from the attachment location. The skins increase in thickness to a 60-ply layup near the tab region that bolts the vertical tail skin to the fuselage frame. The graphite/epoxy tabs are machined, prior to assembly, to introduce a taper at the joint location. In Reference 5, the fuselage attachment fitting was made out of steel, and the skins were bolted to it using 3/8 inch diameter, countersunk high strength steel bolts. Figure 46 shows a

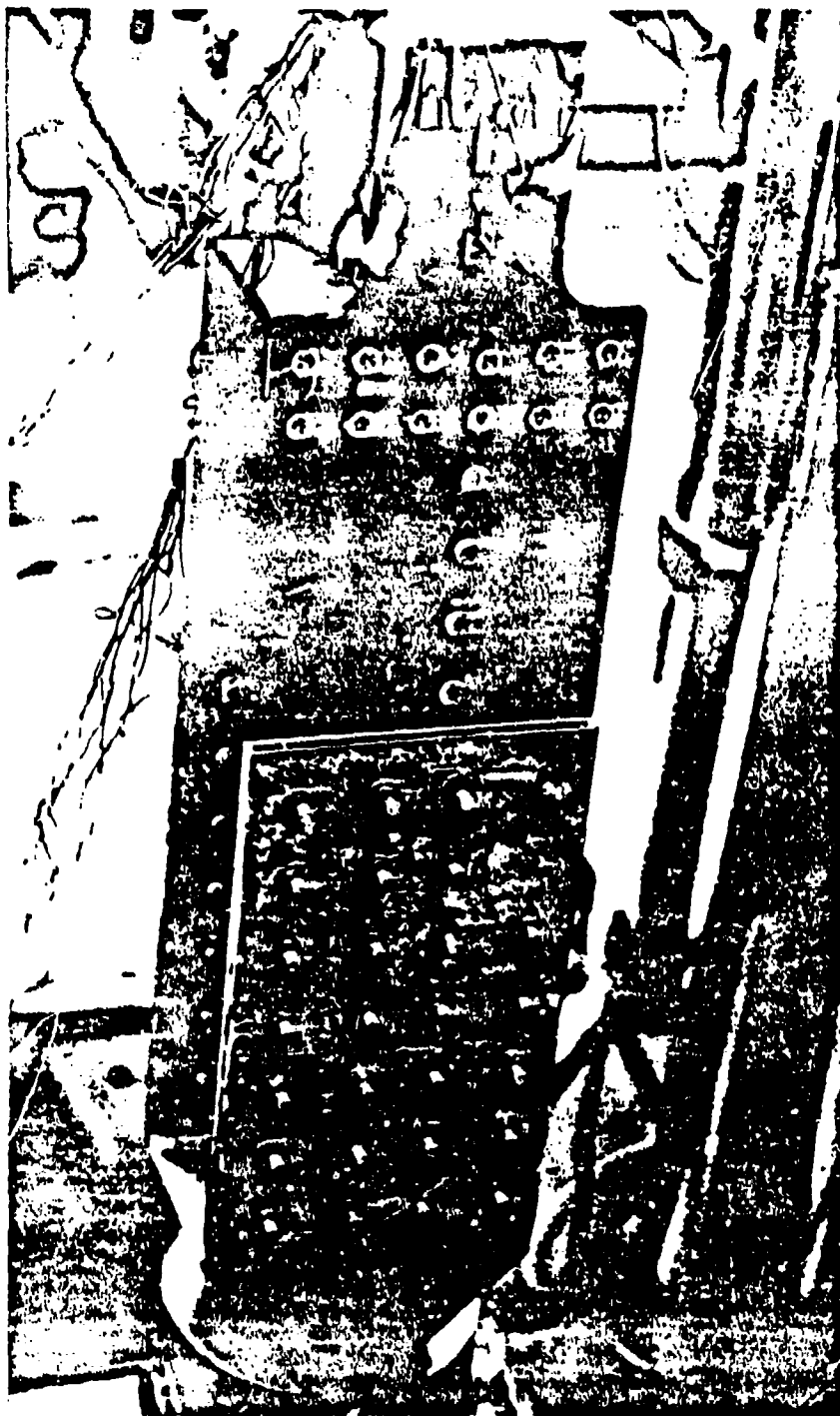


Figure 46. Photograph of an Assembled Test Element.

photograph of an assembled test element. The element spar and the root rib were fabricated using an aluminum alloy.

4.2 Test Results

Elements fabricated based on this preliminary design were subjected to static and fatigue loads in Reference 5. They survived two lifetimes of a spectrum fatigue load that was significantly more severe than the actual F/A-18A vertical tail design spectrum load, and their static strengths were approximately 30% larger than the design ultimate load. During the static test, failure occurred in the graphite/epoxy skin tab in a combined mode (see Figure 47). The observed failure modes were significantly influenced by the tilting or "digging in" of the countersunk fasteners - a phenomenon that cannot be accounted for by the fastener analysis in the SAMCJ computer code.

4.3 Design Verification of the Element Using SAMCJ

The critical vertical tail skin-to-fuselage joint region is analyzed below using the SAMCJ code that is recommended as an analytical design tool. Though the analysis was performed retrospectively, the assumed material and failure parameters are identical to those used in Reference 7.

Figure 48 presents the dimensions of the analyzed graphite/epoxy skin tabs and the fuselage attachment frame. The tapered skin has a $[0_{28}/\pm 45_{12}/90_7]_C$ layup at the top of the tab region. Across the top row of fasteners, it has an average of 58 plies, and across the bottom row of fasteners, it has an average of 52 plies. For analytical purposes, the tapered tab region is modeled as two uniform regions of different thicknesses. The top region is modeled to contain a $[0_{28}/\pm 45_{12}/90_6]_C$ layup, and the bottom region is assumed to be a $[0_{26}/\pm 45_{10}/90_6]_C$ laminate. The average thickness of a ply in the skin was measured to be 0.0049 inch. The fuselage attachment frame is, likewise, divided into a

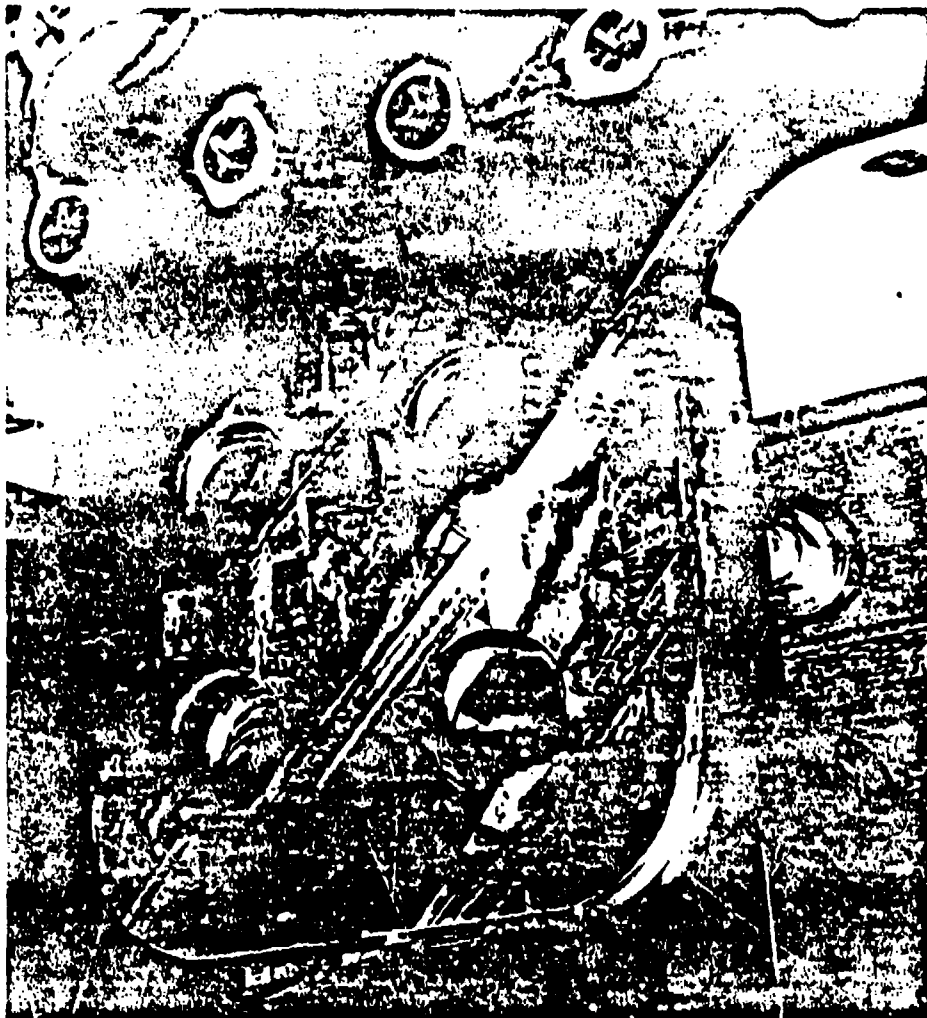


Figure 47. Photograph of the Tab Region of the Failed Element.

Fuselage
Attachment
Frame

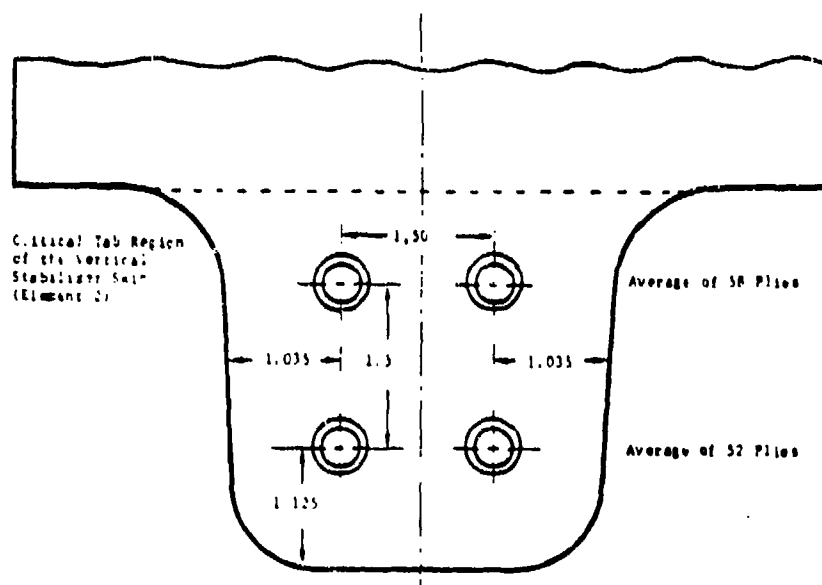
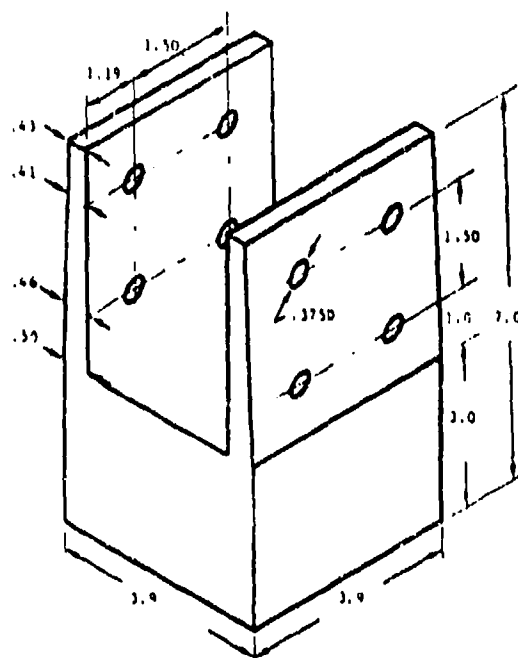


Figure 48. Dimensions of the Critical Skin Tab and the Fuselage Attachment Frame.

0.41 inch thick region and a 0.46 inch thick region (see Figure 48).

The modeled joint segment is half of the symmetric skin tab-to-fuselage attachment. The total joint failure load is, therefore, twice the predicted load. A single shear load transfer between the AS4/3501-6 graphite/epoxy skin tab and the steel attachment frame is analyzed. The graphite/epoxy tab and the steel plate are divided into four elements each. The average width of the slightly tapered tab is used in the analytical model (3.57 in.). The fiber-directional tensile and compressive failure strains for AS4/3501-6 graphite/epoxy are assumed to be 0.012 and 0.0175, respectively (References 7, 13). The characteristic distances for net section, bearing and shear-out failure modes are assumed to be 0.10, 0.25 and 0.25 inch, respectively (Reference 7). The basic AS4/3501-6 lamina properties are assumed to be 18.5 Msi, 1.9 Msi and 0.85 Msi for E_{11} , E_{22} and G_{12} , respectively, and 0.3 for the major Poisson's ratio.

The skins are attached to the fuselage frame by 3/8 inch diameter, countersunk fasteners (100 degree tension head). The fastener analysis in SAMCJ cannot accurately account for the effects of the countersunk head geometry. However, it approximates the actual effects by assuming free rotation at the fastener head location, and requires the user to input an equivalent protruding head fastener diameter. In the discussed element analysis, the average fastener diameter is assumed to be 0.458 inch, to account for the 100 degree tension head geometry.

Analytically predicted load distribution among the fasteners in each tab is presented in Figure 49. The symmetry in the fastener arrangement results in low values for the transverse components of fastener loads (perpendicular to the load direction). Also, the loads in the top row of fasteners are approximately 14% larger than those in the bottom row of fasteners. This leads to a prediction of failure initiation from the top row of fasteners (see Figure 49). The predicted failure site (critical location) is in

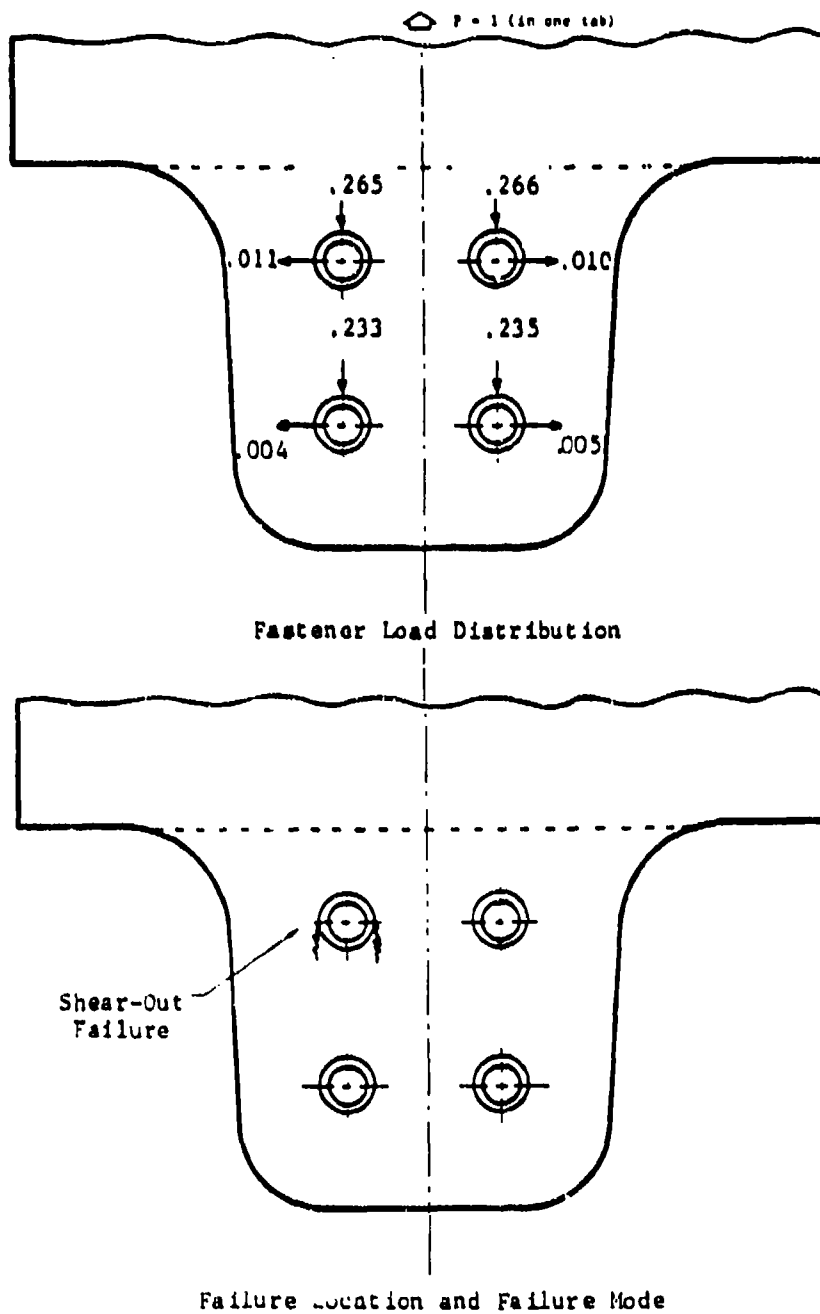


Figure 49. Load Distribution Among Fasteners, Failure Location and Failure Mode in the Graphite/Epoxy Tabs.

agreement with experimental observation.

Figure 50 presents the analytically predicted element load levels to precipitate net section, bearing and shear-out modes of failure at the various fastener locations. The lowest among these provides the element failure load, the failure location and the failure mode. SAMCJ predicts element failure to be caused by a shear-out mode of failure at the top left fastener location in Figure 50. The failure mode observed in Reference 5, however, was severe damage around the fastener hole, introduced by the tilting of the countersunk fasteners (see Figure 47). This included some amount of shear-out and local bearing, and severe delaminations around the fastener hole boundaries. Since SAMCJ cannot account for the severe local three-dimensional stress state introduced by the countersunk fasteners, the predicted failure mode (shear-out) does not correlate well with the observed combined failure mode (partial shear-out, local bearing, and severe delaminations).

Despite the approximate failure mode prediction, however, SAMCJ correctly predicts the failure location, and the failure load predicted by SAMCJ (96.0 kips) is only 7% larger than the measured value (91.8 kips). The approximation of the countersunk fasteners by equivalent protruding head fasteners (larger diameter, unconstrained at the head location), therefore, predicts the element failure load with adequate accuracy. The SAMCJ analysis and the test results in Reference 5 independently verify the 30% margin of safety in the static strength of the test element, due to the approximate analyses used in its preliminary design.

Tab Load = P; Element Load = 2P



The numbers below are the values of the applied element load (2P), in kips to precipitate the three failure modes at each fastener location.

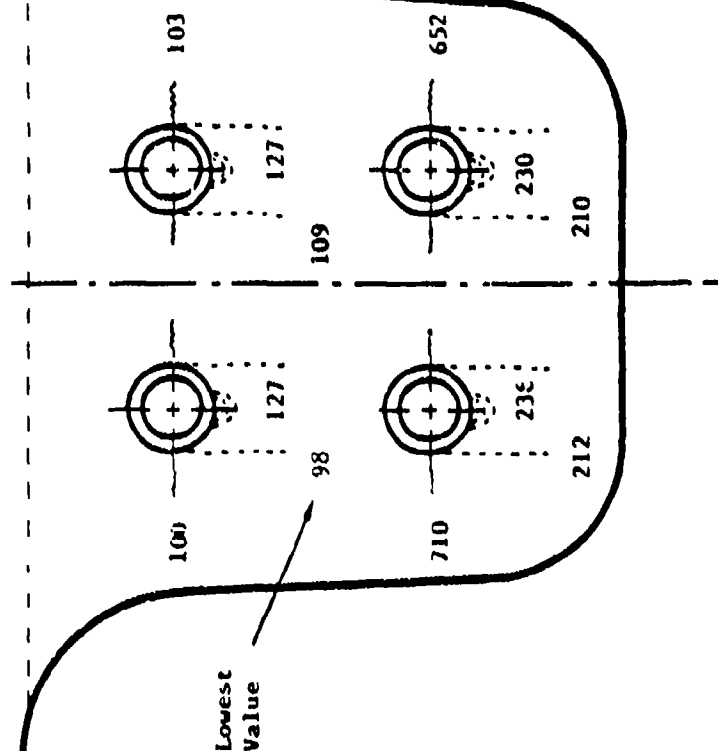


Figure 50. Analytically Predicted Element Load Levels to Precipitate Net Section, Bearing and Shear-Out Modes of Tab Failure at Each Fastener Location.

SECTION 5

CONCLUSIONS

A design guide was developed to enable the user in designing efficient bolted joints in composite structures. The guide highlights general design guidelines for the various parameters that are to be considered in selecting a bolted joint concept. A purely analytical design methodology is presented. It is devoid of complementary test requirements when a previously characterized material is used to fabricate the bolted structure. The design guide also illustrates the use of two computer codes (SASCJ and SAMCJ) that were developed in this Northrop/AFWAL program and are required for design purposes. A listing of these computer codes is appended to this report.

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APPENDIX A

SASCJ Program Listing

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```

WRITE(6,6,6)
876 FORMAT(///,10X,' PROGRAM SASCH',///,
M' PROGRAM SASCH PREDICTS FAILURE LOADS OF ',///,
M' MECHANICALLY FASTENED, COMPOSITE LAMINATE, ',///,
M' SINGLE OR DOUBLE LAP SHEAR JOINTS. ',///,
M' PROGRAM ASSUMES THAT INPUT PARAMETERS ARE ',///,
M' IN ENGLISH UNITS - LENGTHS ARE INPUT ',///,
M' IN INCHES AND MODULI AND STRENGTHS ARE ',///,
M' EXPRESSED IN PSI ',///)
WRITE(6,401)
401 FORMAT(' ENTER BYPASS RATIO ALPHA: ',///,
M' ALPHA=0 FOR FULL BEARING ',///,
M' ALPHA=1 FOR OPEN HOLE ',///,
M' 0<ALPHA<1 FOR GENERAL BYPASS ')
READ(5,M) BPR
WRITE(6,911)
911 FORMAT(' ENTER: ',///,
M' 1 FOR STATIC TENSION ',///,
M' 2 FOR STATIC COMPRESSION ',///)
READ(5,M) LTNCM
NLIM=1
IF(BPR.EQ.1.0) GO TO 380
NLIM=2
WRITE(6,400)
400 FORMAT(' ENTER: ',///,
M' 1 FOR SLS (SINGLE LAP SHEAR)',///,
M' 2 FOR DLS (DOUBLE LAP SHEAR)',///)
READ(5,M) NSDLS
106 FORMAT(A1)
380 CONTINUE
DO 300 K=1,NLIM
IF(K.EQ.1) WRITE(6,511)
611 FORMAT(' IS THE TOP PLATE A COMPOSITE OR A METAL ? ',///,
M' ENTER C OR M IN THE FIRST FIELD')
IF(K.EQ.2) WRITE(6,789)
789 FORMAT(' IS THE BOTTOM PLATE A COMPOSITE OR A METAL ? ',///,
M' ENTER C OR M IN THE FIRST FIELD')
READ(5,106) CM(K)
WRITE(6,203)
203 FORMAT(' INPUT MATERIAL DESCRIPTION OF THIS PLATE ',///,
M' EX: ASA/3501-6')
READ(5,204) (MTL(K,1),I=1,15)
204 FORMAT(15A4)
300 CONTINUE
IF(CM(1).NE.CMC.OR.CM(2).NE.CMC) WRITE(6,721)
721 FORMAT(//,' NOTE: FOR COMPUTATIONAL PURPOSES A ',///,
M' METALLIC PLATE IS MODELED AS A 30 PLY ',///,
M' LAMINATE OF 0 DEGREE PLIES WITH ISOTROPIC ',///,
M' MATERIAL PROPERTIES ',///)
IF(BPR.NE.1.0) WRITE(6,494)
IF(BPR.EQ.1.0) WRITE(6,495)
494 FORMAT(' NOTE: NUMERICAL DESIGNATIONS FOR THE ',///,
M' PLATES ARE: ',///,
M' TOP PLATE = NO 1 ',///,
M' BOTTOM PLATE = NO 2 ',///)
495 FORMAT(' NOTE: A SINGLE PLATE WITH AN OPEN ',///,
M' HOLE IS DESIGNATED AS PLATE NUMBER 1 ',///)
DO 301 K=1,NLIM
IF(CM(K).EQ.CMC) GO TO 15
NPLY(K)=30

```

```

00000560
00000570
00000580
00000590
00000600
00000610
00000620
00000630
00000640
00000650
00000660
00000670
00000680
00000690
00000700
00000710
00000720
00000730
00000740
00000750
00000760
00000770
00000780
00000790
00000800
00000810
00000820
00000830
00000840
00000850
00000860
00000870
00000880
00000890
00000900
00000910
00000920
00000930
00000940
00000950
00000960
00000970
00000980
00000990
00010000
00010100
00010200
00010300
00010400
00010500
00010600
00010700
00010800
00010900
00011000
00011100
00011200
00011300
00011400
00011500

```

GO TO 301	00001160
15 CONTINUE	00001170
IF(NSDLS.EQ.2.AND.K.EQ.1) WRITE(6,932)	00001180
932 FORMAT(/,' NOTE: FOR THE DOUBLE LAP SHEAR CASE HAVING ',/,	00001190
*' A COMPOSITE PLATE NUMBER 2, ENTER ONLY HALF',/,	00001200
*' OF THE LAYUP - IE HALF THE NUMBER OF ACTUAL',/,	00001210
*' PLIES ')	00001220
WRITE(6,205) K	00001230
205 FORMAT(' INPUT NUMBER OF PLIES IN PLATE NO',I5,/,	00001240
*' (N > OR = 2)')	00001250
READ(5,*) NPLY(K)	00001260
NLIM2=2*NPLY(K)+1	00001270
301 CONTINUE	00001280
DO 302 K=1,NLIM	00001290
IF(CM(K).EQ.CMC) GO TO 25	00001300
IF(NSDLS.EQ.2.AND.K.EQ.2) WRITE(6,933)	00001310
933 FORMAT(/,' FOR THE DOUBLE LAP SHEAR CASE HAVING',/,	00001320
*' A METALLIC PLATE NUMBER TWO, ENTER HALF THE ',/,	00001330
*' ACTUAL PLATE THICKNESS ')	00001340
WRITE(6,35) K	00001350
35 FORMAT(' INPUT THICKNESS OF PLATE NO',I5)	00001360
READ(5,*) A1	00001370
H(K)=A1/NPLY(K)	00001380
GO TO 302	00001390
25 CONTINUE	00001400
WRITE(6,260) K	00001410
260 FORMAT(' INPUT PLY THICKNESS IN PLATE NO',I5)	00001420
READ(5,*) H(K)	00001430
302 CONTINUE	00001440
DO 303 K=1,NLIM	00001450
IF(CM(K).EQ.CMC) GO TO 45	00001460
NMPLY(K)=1	00001470
GO TO 303	00001480
45 CONTINUE	00001490
WRITE(6,207) K	00001500
207 FORMAT(' INPUT NUMBER OF DISTINCT PLY ORIENTATIONS',/,	00001510
*' IN PLATE NO',I5,	00001520
READ(5,*) NMPLY(K)	00001530
303 CONTINUE	00001540
DO 209 K=1,NLIM	00001550
IF(CM(K).EQ.CMC) GO TO 55	00001560
ANG(1,K)=0.	00001570
GO TO 209	00001580
55 CONTINUE	00001590
WRITE(6,487) K	00001600
487 FORMAT(/,' FOR PLATE NUMBER',I5,' ',/,	00001610
N=NPLY(K)	00001620
DO 209 L=1,N	00001630
WRITE(6,206) L	00001640
206 FORMAT(' INPUT ORIENTATION OF PLY TYPE NO',I5)	00001650
READ(5,*) ANG(L,K)	00001660
209 CONTINUE	00001670
DO 305 K=1,NLIM	00001680
IF(CM(K).EQ.CMC) GO TO 65	00001690
NN=NPLY(K)	00001700
DO 75 IJ=1,NN	00001710
75 IPLY(IJ,K)=1	00001720
GO TO 305	00001730
65 CONTINUE	00001740
WRITE(6,210) K	00001750

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210 FORMAT(' INPUT TYPE OF PLY IN PLATE NO',I5,' FROM TOP',/,
  *' TO BOTTOM',/,
  *SX,'PLY TYPE',,OX,'ORIENTATION')
  N=NUMPLY(K)
  DO 212 L=1,N
    WRITE(6,713) L,ANG(L,K)
213 FORMAT(SX,I5,10X,F7.2,' DEGREES')
212 CONTINUE
    WRITE(6,711)
711 FORMAT(/)
    N=NPLY(K)
    DO 215 I=1,N
      WRITE(6,214) I
214 FORMAT(' INPUT TYPE OF PLY FOR PLY NO',I5)
      READ(5,*) IPLY(I,K)
215 CONTINUE
305 CONTINUE
      DO 306 K=1,NLIM
        WRITE(6,216) K
216 FORMAT(' INPUT THE ENGINEERING PROPERTIES OF PLATE NO',I5)
        IF(CMK).EQ.CMC) GO TO 35
        WRITE(6,95)
95 FORMAT(' INPUT YOUNGS MODULUS AND POISSONS RATIO')
        READ(5,*) E1(K),V12(K)
        E2(K)=E1(K)
        V21(K)=V12(K)*E2(K)/E1(K)
        G12(K)=E1(K)/(2.*(1.+V12(K)))
        GO TO 306
85 CONTINUE
        WRITE(6,217)
217 FORMAT(' INPUT YOUNGS MODULI, E1 AND E2')
        READ(5,*) E1(K),E2(K)
        WRITE(6,218)
218 FORMAT(' INPUT THE SHEAR MODULUS AND MAJOR POISSONS RATIO')
        READ(5,*) G12(K),V12(K)
        V21(K)=V12(K)*E2(K)/E1(K)
306 CONTINUE
        IF(UPR.NE.1.0) GO TO 930
        WRITE(6,844)
844 FORMAT(' INPUT HOLE DIAMETER')
        READ(5,*) FASD
        GO TO 360
930 CONTINUE
        WRITE(6,250)
250 FORMAT(' INPUT MATERIAL DESCRIPTION FOR FASTENER')
        READ(5,251) (MTL(J,I),I=1,15)
251 FORMAT(15A4)
        WRITE(6,252)
252 FORMAT(' INPUT YOUNG MODULUS AND POISSONS RATIO FOR',/,
  *' THE FASTENER')
        READ(5,*) FASE,FASV
        WRITE(6,253)
253 FORMAT(' INPUT THE DIAMETER OF THE FASTENER')
        READ(5,*) FASD
        WRITE(6,888)
888 FORMAT(/,' FASTENER TYPE',/,
  *' ENTER: 1 FOR PROTRUDING HEAD',/,
  *' 2 FOR COUNTERSUNK HEAD')
        READ(5,*) HFTYP
        R(1)=1.0D10
  
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R(2)=1.0E-1J
IF(NFTYP.EQ.1) GO TO 360
WRITE(6,889)
889 FORMAT(/,' ENTER PLATE WHICH CONTAINS THE COUNTERSUNK',/,
  *' HEAD (OPPOSITE PLATE ASSUMES THE NUT HEAD) ',/,
  *' ENTER: 1 FOR TOP PLATE ',/,
  *' 2 FOR BOTTOM PLATE ')
READ(5,M) N
R(N)=0.0D0
360 CONTINUE

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000
READ IN GEOMETRY AND BOUNDARY DATA

AX=FASD/2.0D0
BX=AX
WRITE(6,856)
856 FORMAT(' PLATE GEOMETRIES ARE SPECIFIED BY ',/,
  *' INPUTTING THE COORDINATES OF THE CORNER',/,
  *' VERTICIES. NOTE: THE ORIGIN IS AT THE FASTENER',/,
  *' CENTER ; INPUT COORDINATES ACCORDINGLY',/,
  *'
  *'          V3          V2
  *'          HOLE
  *'          CENTROID
  *'          V4          V1
  *' APPLIED LOAD CONVENTION:
  *' FOR PLATE NO 1 (TOP) NORMAL LOADS ARE APPLIED ',/,
  *' BETWEEN V3 AND V4
  *' FOR PLATE NO 2 (BOTTOM) NORMAL LOADS ARE APPLIED ',/,
  *' BETWEEN V1 AND V2
  *'
  DO 480 K=1,NLIM
  WRITE(6,734) K
734 FORMAT(' FOR PLATE NUMBER ',I5,' ',/,
  DO 110 I=1,4
  WRITE(6,290) I
290 FORMAT(' ENTER X,Y COORDINATES OF V',I4)
  READ(5,M) XC(K,I),YC(K,I)
110 CONTINUE
  IF(K.EQ.2) GO TO 841
  A1=XC(1,1)
  B1=YC(1,1)
  A2=XC(1,2)
  B2=YC(1,2)
  XC(1,1)=XC(1,4)
  YC(1,1)=YC(1,4)
  XC(1,2)=XC(1,3)
  YC(1,2)=YC(1,3)
  XC(1,4)=A1
  YC(1,4)=B1
  XC(1,3)=A2
  YC(1,3)=B2
841 CONTINUE
  WTH=YC(K,2)-YC(K,1)
480 CONTINUE
  IF(BPR.EQ.0.0.OR.BPR.EQ.1.0) GO TO 567
  WRITE(6,741)
741 FORMAT(' SELECT FAILURE CRITERION: ',/,
  *' ENTER 1 FOR POINT STRESS CRITERION ',/,
  *' ENTER 2 FOR AVERAGE STRESS CRITERION ')
  READ(5,M) NPT

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    IF(NPT.EQ.1) NOPT4=2
    IF(NPT.EQ.2) NOPT4=4
    GO TO 601
567 CONTINUE
    WRITE(6,220)
220 FORMAT(' SELECT FAILURE CRITERION ',//,
  *' ENTER 1 FOR HOFFMAN/ TSAI-HILL CRITERION ',//,
  *' ENTER 2 FOR POINT STRESS CRITERION ',//,
  *' ENTER 3 FOR MAXIMUM STRAIN CRITERION ',//,
  *' ENTER 4 FOR AVERAGE STRESS CRITERION')
    READ(5,*) NOPT4
601 CONTINUE
    IF(NOPT4.EQ.2.OR.NOPT4.EQ.4) GO TO 221
    DO 412 K=1,NLIM
    WRITE(6,222) K
222 FORMAT(' FOR PLATE NUMBER ',I5,' ENTER RADIUS OF
  * CHARACTERISTIC CIRCLE AT WHICH STRESSES ARE',//,
  * TO BE COMPUTED TO PREDICT FAILURE')
    READ(5,*) RCA(K)
    RCB(K)=RCA(K)
    HRCOUT(K)=50
    IF(NOPT4.EQ.3) GO TO 591
    WRITE(6,834)
834 FORMAT(' ENTER THE FAILURE INDEXES FOR THE ',//,
  * HOFFMAN/ TSAI-HILL CRITERIA
  * NOTE: FOR USING TSAI-HILL SET EQUAL THE COMPRESSION ',//,
  * AND TENSION ULTIMATES IN SIGMA X AND
  * ENTER: SIGMA X ULTIMATE-COMPRESSION
  * SIGMA X ULTIMATE-TENSION
  * SIGMA Y ULTIMATE-COMPRESSION
  * SIGMA Y ULTIMATE-TENSION
  * SIGMA XY ULTIMATE
    READ(5,*)(HFMCI,K),I=1,5)
    GO TO 412
591 CONTINUE
    WRITE(6,393) K
393 FORMAT(' ENTER MAXIMUM STRAIN ALLOWABLE FOR',//,
  * PLATE NUMBER ',I7,' (UNITS: IN/IN)')
    READ(5,*) SALON(K)
412 CONTINUE
    IF(NOPT4.EQ.3) GO TO 391
    IF(NOPT4.EQ.1) GO TO 391
    GO TO 262
221 CONTINUE
    IF(NOPT4.EQ.2) WRITE(6,555)
    IF(NOPT4.EQ.4) WRITE(6,556)
555 FORMAT(/,' POINT STRESS CRITERION ',//)
556 FORMAT(/,' AVERAGE STRESS CRITERION ',//,
  * AO IS THE CHARACTERISTIC DISTANCE OVER WHICH',//,
  * STRESSES ARE AVERAGED AND COMPARED WITH UNNOTCHED',//,
  * STRENGTHS TO PREDICT FAILURE')
    DO 226 K=1,NLIM
    IF(BPR.NE.0.0.AND.BPR.NE.1.0) GO TO 531
    WRITE(6,225) K
225 FORMAT(' INPUT AO FOR EACH OF THE THREE PLY FAILURE',//,
  * MODES OF PLATE NO',I5,/,
  * AONT = NET SECTION
  * AOBP = BEARING
  * AOSO = SHEAR OUT
    N=NUMPLY(K)

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WRITE(6,227)
227 FORMAT(' INPUT AONT,AOBR,AND AOSO')
READ(5,M) DONT(K),DOBR(K),DOSQ(K)
GO TO 226
531 CONTINUE
WRITE(6,532) K
532 FORMAT(' ENTER AO VALUES CORRESPONDING TO THE THREE',/,
  * PLY FAILURE MODES IN PLATE NO ',I2,/,
  * AONT = NET SECTION
  * AOBR = BEARING
  * AOSO = SHEAR OUT ')
WRITE(6,533)
533 FORMAT(' INPUT AONT, AOBR, AOSO ')
READ(5,M) AONT(1,K),AOBR(1,K),AOSO(1,K)
AONT(2,K)=AONT(1,K)
AOBR(2,K)=AOBR(1,K)
AOSO(2,K)=AOSO(1,K)
IF(K.EQ.1) WRITE(6,554)
554 FORMAT(/, ' TO AVOID LENGTHY RUN TIMES DUE TO ',/,
  * STRESS FIELD RECOMPUTATION SPECIFY THE ',/,
  * NUMBER OF ULTIMATE PLY FAILURES AFTER ',/,
  * WHICH JOINT FAILURE WILL BE PREDICTED ',/,
  * ENTER: NO OF ULTIMATE FAILURES ')
IF(K.EQ.1) READ(5,M) NULTF
226 CONTINUE
291 CONTINUE
NOPT1=1
IF(BPR.EQ.1.0) NOPT1=2
DO 229 K=1,NLIM
IF(BPR.NE.1.0.OR.(BPR.EQ.1.0.AND.NOPT1.EQ.2)) GO TO 670
GO TO 671
670 CONTINUE
IF(CM(K).EQ.CMC) GO TO 672
WRITE(6,228) K
228 FORMAT(' FOR PLATE NUMBER ',I5,' ENTER THE THREE STRENGTHS ',/,
  * REQUIRED TO PREDICT THE THREE FAILURE MODES ',/,
  * FNST=UNNOTCHED STRENGTH IN TENSION ',/,
  * FNSC=UNNOTCHED STRENGTH IN COMPRESSION',/,
  * FSO=UNNOTCHED STRENGTH IN SHEAR-OUT',/,
  * INPUT FNST,FNSC,FSO
  * )
READ(5,M) AF1,AF2,AF4
GO TO 673
672 WRITE(6,674) K
674 FORMAT(' FOR PLATE NO ',I5,' ENTER FIBER ULTIMATE',/,
  * STRAIN VALUES
  * EPSILON ULT IN COMPRESSION
  * EPSILON ULT IN TENSION
  * GAMMA ULT IN SHEAR
  * (UNITS: IN/IN)')
READ(5,M) ES1(K),ES2(K),ESS(K)
CALL STRTH(H,ES1,ES2,ESS,AF1,AF2,AF4,K)
675 CONTINUE
AF3=AF2
NP=NUMPLY(K)
DO 666 IL=1,NP
PSTC(1,IL,K)=AF1
PSTC(2,IL,K)=AF2
PSTC(3,IL,K)=AF3
PSTC(4,IL,K)=AF4
666 CONTINUE

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229	CONTINUE	00004160
671	CONTINUE	00004170
	IF(NOPT4.NE.4) GO TO 261	00004180
C		00004190
C	NUMBER OF DIVISIONS FOR STRESS AVERAGING	00004200
C	IS SET EQUAL TO 50	00004210
	NAVD=50	00004220
261	CONTINUE	00004230
	IF(BPR.EQ.1.0) GO TO 262	00004240
	DO 319 K=1,NLIM	00004250
	N=NUMPLY(K)	00004260
	WRITE(6,320) K	00004270
320	FORMAT(' SASCJ ASSUMES A BILINEAR PLY BEHAVIOR. THE ',,,	00004280
	*' INITIAL MODULUS, K1, IS COMPUTED BY THE CODE. ',,,	00004290
	*' THE REDUCED MODULUS, K2, FOR INITIAL FAILURE ',,,	00004300
	*' IN NET SECTION, SHEAROUT OR BEARING IS COMPUTED ',,,	00004310
	*' BY THE FORMULA K2=ALPHAK1. ',,,	00004320
	*' FOR PLATE NUMBER ',I5,' INPUT ALPHA VALUES FOR ',,,	00004330
	*' NET SECTION, SHEAROUT AND BEARING FAILURE ')	00004340
	READ(5,*) AF1,AF2,AF3	00004350
	DO 321 I=1,N	00004360
	DELNS(I,K)=AF1	00004370
	DELSR(I,K)=AF2	00004380
	DELSO(I,K)=AF3	00004390
321	CONTINUE	00004400
	WRITE(6,339)	00004410
339	FORMAT(' INPUT SCALE FACTORS FOR P ULTIMATE ',,,	00004420
	*' CALCULATION SUCH THAT P(ULT)=BETA*P(INITIAL) ',,,	00004430
	*' INPUT BETA1 FOR NET SECTION ULTIMATE ',,,	00004440
	*' BETA2 FOR BEARING ULTIMATE ',,,	00004450
	*' BETA3 FOR SHEAROUT ULTIMATE ')	00004460
	READ(5,*) PALT(3,K),PALT(2,K),PALT(1,K)	00004470
319	CONTINUE	00004480
391	CONTINUE	00004490
	IF(BPR.NE.0.0) GO TO 262	00004500
	DO 312 K=1,NLIM	00004510
	GAMD(K)=10.0	00004520
	IF(CMK(K).NE.CMC) GO TO 312	00004530
	WRITE(6,231) K	00004540
231	FORMAT(/,' INPUT THE APPROXIMATE INTERLAMINAR SHEAR STRAIN',,,	00004550
	*' ULTIMATE FOR DELAMINATION PREDICTION IN PLATE NO ',I5,,	00004560
	*' (UNITS: IN/IN) ')	00004570
	READ(5,*) GAMD(K)	00004580
312	CONTINUE	00004590
262	CONTINUE	00004600
C		00004610
C	CASE HEADING	00004620
C		00004630
	WRITE(6,143)	00004640
143	FORMAT(///,10X,'PROGRAM SASCJ',,,)	00004650
	IF(NSDLS.EQ.1.AND.BPR.NE.1.0) WRITE(6,633)	00004660
	IF(NSDLS.EQ.2.AND.BPR.NE.1.0) WRITE(6,634)	00004670
633	FORMAT(2X,'A SINGLE LAP SHEAR JOINT WILL BE ANALYZED',,,)	00004680
634	FORMAT(2X,'A DOUBLE LAP SHEAR JOINT WILL BE ANALYZED',,,)	00004690
	IF(BPR.EQ.0.0) WRITE(6,881)	00004700
	IF(BPR.EQ.1.0) WRITE(6,882)	00004710
	IF(BPR.NE.0.0.AND.BPR.NE.1.0) WRITE(6,883) BPR	00004720
881	FORMAT(2X,'WITH A LOADED HOLE',,,)	00004730
882	FORMAT(2X,'WITH AN OPEN HOLE',,,)	00004740
		00004750

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883 FORMAT(2X,'WITH A PARTIALLY LOADED HOLE',//,
      *2X,'BYPASS RATIO = ',D9.3,/)
      IF(LTNCM.EQ.1) WRITE(6,823)
      IF(LTNCM.EQ.2) WRITE(6,824)
823 FORMAT(2X,'LOADED IN STATIC TENSION',//)
824 FORMAT(2X,'LOADED IN STATIC COMPRESSION',//)
      DO 605 I=1,NLIM
      WRITE(6,600) I
600 FORMAT(10X,'PLATE NO ',I5,' ',//)
      WRITE(6,689) (MTL(I,J),J=1,15)
689 FORMAT(2X,15A4,/)
      HT=NPLY(I)*MH(I)
      WRITE(6,602) HT
602 FORMAT(2X,'T = ',D9.3,' INCHES',//)
      WRITE(6,603) E1(I),E2(I),G12(I),V12(I),V21(I)
603 FORMAT(2X,'MATERIAL PROPERTIES',//,
      *10X,'E1 = ',D9.3,' PSI',//,
      *10X,'E2 = ',D9.3,' PSI',//,
      *10X,'G12 = ',D9.3,' PSI',//,
      *10X,'NU12 = ',D9.3,/,
      *10X,'NU21 = ',D9.3,/)
605 CONTINUE
      IF(DPR.EQ.1.0) GO TO 708
      WRITE(6,606)
606 FORMAT(10X,'FASTENER',//)
      WRITE(6,607) (MTL(3,J),J=1,15)
607 FORMAT(2X,15A4,/)
      WRITE(6,608) FASD
608 FORMAT(2X,' DIAMETER = ',D9.3,' INCHES',//)
      WRITE(6,609) FASE,FASV
609 FORMAT(2X,' MATERIAL PROPERTIES',//,
      *10X,'E = ',D9.3,' PSI',//,
      *10X,'MU = ',D9.3,/)
708 CONTINUE
      WRITE(6,923)
923 FORMAT(//,10X,'FAILURE ANALYSIS',//)
      IF(NOPT4.EQ.2.OR.NOPT4.EQ.4) GO TO 621
      IF(NOPT4.EQ.3) GO TO 821
      WRITE(6,622)
622 FORMAT(2X,'THE HOFFMAN/ TSAI-HILL CRITERION WILL BE USED',//)
      DO 623 J=1,NLIM
      WRITE(6,624) J,RCA(J)
624 FORMAT(2X,'PLATE NUMBER ',I5,/,
      *2X,'CHARACTERISTIC RADIUS = ',D9.3,' INCHES')
      WRITE(6,790)
790 FORMAT(//,16X,' ULTIMATE STRESSES',
      *10X,'TENSION',18X,'COMPRESSION')
      WRITE(6,625) (HFMCI,I,J), I=1,5,HFMCS,J)
625 FORMAT(//,2X,'SIGMA X = ',D9.3,' PSI',5X,'SIGMA X = ',
      *D9.3,' PSI',//,
      *2X,'SIGMA Y = ',D9.3,' PSI',5X,'SIGMA Y = ',
      *D9.3,' PSI',//,
      *2X,'SIGMA S = ',D9.3,' PSI',5X,'SIGMA S = ',
      *D9.3,' PSI')
623 CONTINUE
      GO TO 627
621 CONTINUE
      IF(NOPT4.EQ.2) WRITE(6,628)
628 FORMAT(2X,'A POINT STRESS CRITERION WILL BE USED',//)
      IF(NOPT4.EQ.4) WRITE(6,558)

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      IF(IL.EQ.1) GO TO 23
      WRITE(6,871) K
871  FORMAT(/,5X,' GEOMETRY OF PLATE NO ',I5,' ',/)
      WRITE(6,872)
872  FORMAT(' COORDINATES OF CORNER VERTEXES ',/)
      IF(K.EQ.1) WRITE(6,873) XC(K,2),YC(K,2),XC(K,3),YC(K,3)
      IF(K.EQ.1) WRITE(6,874) XC(K,1),YC(K,1),XC(K,4),YC(K,4)
      IF(K.EQ.2) WRITE(6,873) XC(K,3),YC(K,3),XC(K,2),YC(K,2)
      IF(K.EQ.2) WRITE(6,874) XC(K,4),YC(K,4),XC(K,1),YC(K,1)
873  FORMAT(2X,F7.3,' ',F7.3,10X,F7.3,' ',F7.3,/)
874  FORMAT(2X,F7.3,' ',F7.3,10X,F7.3,' ',F7.3,/)
      AXD=AX*2.
      WRITE(6,875) AXD
875  FORMAT(' FASTENER HOLE DIAMETER = ',D9.3,' INCHES',/)
      ED=DABS(XC(K,3)-XC(K,1))/AXD
      WD=DABS(YC(K,3)-YC(K,1))/AXD
      WRITE(6,755) ED
755  FORMAT(' E/D RATIO = ',D9.3,/)
      WRITE(6,879) WD
879  FORMAT(' W/D RATIO = ',D9.3,/)
23  CONTINUE

C
C
C    PROCESS INPUT DATA ON PLATE GEOMETRIES
C
      WTH=YC(K,2)-YC(K,1)
      LMI=L0M
      CALL POLY(JK,K,XC,YC,W,AST,NCOL,LTNCH,BPR,IL)
      CALL CIRC(W,AST,JK,K,LTNCH,BPR,IL)
      IF(NOPT4.EQ.1.OR.NOPT4.EQ.3) CALL RCOUT(K)
      IF(NOPT4.EQ.2) CALL PSTRSS(K,LTNCH,BPR,IL)
      IF(NOPT4.EQ.4) CALL AVSTRS(K,LTNCH,NAVD,BPR,IL)

C
C
C    PERFORM FINITE GEOMETRY ANALYSIS FOR STRESS/DISPLACEMENT
    STATE, COMPUTE FOUNDATION MODULI AND FAILURE VALUES
C
      CALL FIOEM(W,K,NOPT4,ITT)
      IF(BPR.NE.0.0.AND.BPR.NE.1.0.AND.IL.EQ.1) GO TO 21
      IF(BPR.NE.1.0.AND.L0M.NE.1) CALL FBOLT(ANGK,H,K,NOPT1,LMI)
21  CALL FCRT(SALON,H,WTH,AST,K,NOPT1,NOPT4,BPR,NAVD,IL)
20  CONTINUE
22  CONTINUE
      IF(BPR.EQ.1.0) GO TO 410

C
C
C    PREPARE INPUT FOR SEQUENTIAL PLY FAILURE
    PREDICTION
C
      IF(L0M.GT.1) GO TO 61
      N=NPLY(1)
      DO 30 I=1,N
      M=IPLY(I,1)
30  PLYK(I)=ANGX(M,1)
      N=NPLY(2)
      DO 60 I=1,N
      N1=I+NPLY(1)
      N2=IPLY(I,2)
60  PLYK(N1)=ANGK(N2,2)
61  CONTINUE

C
C
C    CALCULATION OF FASTENER STIFFNESSES.

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C
FASQ=FASE/(2.*(1.+FASV))
FASLAM=5.*(1.0+FASV)/(7.+5.*FASV)
FASR=FASQ/2.
FASA=ACOS(-1.)*FASR**2
FASI=ACOS(-1.)*FASR**4/4.
FASSS=FASLAM*FASQ*FASA
FASBS=FASE*FASI

CCC
INITIALIZATION

IF(LCM.GT.1) GO TO 72
ITT=0
NTFL=0
JNT=1
P=0.
DELP=1000.
DO 5012 I=1,100
  NPNM(I,1)=I
  NPNM(I,2)=I*NPLY(1)
  UN(I)=0.
  QAMN(I)=0.
  MDAHPI(I)=0.
  MDAMI(I)=0.
  PN(I)=0.
  BARK(I)=0.
5012 BARU(I)=0.
72 CONTINUE

CCCC
INCREMENTAL LOADS TO PLY FAILURE, PLY FAILURE
MODES, AND FRACTIONAL STIFFNESS LOSSES ARE
CALCULATED FOR EACH PLY FROM TOP TO BOTTOM
UNTIL FINAL JOINT FAILURE

90 CONTINUE
ITT=ITT+1
CALL CENTD(R,H,FASSS,FASBS,P,DELP,ITT)
CALL SOLVE(U,H,P,DELP,NSDLS,ITT)
CALL FAIL(QAMDLS,H,P,DELP,BPR,AST,WTH,PFAIL,ANGLE,NODE,
  *IROUT,NOPT4,NULTE,JNT,ITT,NTFL)
CALL PRINT(U,P,DELP,PFAIL,ANGLE,BPR,NODE,IROUT,JNT,
  *NPN,NSDLS,ITT)
IF(JNT.EQ.0) GO TO 410
IF(NLIM2.EQ.1) GO TO 90
IF(NTFL.EQ.0.AND.NLIM2.GT.1) GO TO 90
71 CONTINUE
410 STOP
END

CCCC
SUBROUTINE STRTH(H,ES1,ES2,ESS,AF1,AF2,AF4,K)
IMPLICIT REAL*8(A-H,O-Z)
DIMENSION AINV(3,3),AVN(3),M(2),NV(3)
DIMENSION NPLY(2),NUMPLY(2),ANG(3,2),IPLY(100,2)
DIMENSION WK(25),PSMX(3),ES1(2),ES2(2),ESS(2)
DIMENSION E1(2),E2(2),G12(2),V12(2),V21(2)
COMMON/LYP/NPLY,NUMPLY,ANG,IPLY
COMMON/MCD/E1,E2,G12,V12,V21
COMMON/AM1/A

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C
C
C
C      COMPUTE LAMINATE FAILURE LOADS BASED ON MAXIMUM
      FIBER STRAINS FOR EACH FAILURE MODE
      CALL AMATRX(H,K)
      N=3
      IDOT=4
      IA=3
      CALL LINVZF(A,N,IA,AINV,IDOT,WK,IER)
      DO 100 KK=1,3
      DO 10 II=1,3
      NV(II)=0
10  AVN(II)=0.000
      IF(KK.EQ.1) NV(1)=1
      IF(KK.EQ.2) NV(1)=-1
      IF(KK.EQ.3) NV(3)=1
      DO 13 II=1,3
      DO 15 JJ=1,3
      AVN(IJ)=AVN(II)+AINV(IJ,JJ)*NV(JJ)
15  CONTINUE
      NP=NUMPLY(K)
      SMX=0.000
      RAD=DARCOS(-1.000)/180.000
      DO 25 II=1,NP
      TH=ANG(II,K)*RAD
      E11=DCOS(TH)*MX2+AVN(1)+AVN(2)*DSIN(TH)*MX2+
      *DCOS(TH)*DSIN(TH)*AVN(3)
      IF(KK.NE.1) GO TO 65
      EPRT=E11/ES2(K)
      GO TO 50
65  IF(KK.NE.2) GO TO 75
      EPRT=E11/ES1(K)
      GO TO 50
75  EPRT=E11/ES2(K)
50  CONTINUE
      IF(DABS(SMX).LT.DABS(EPRT)) SMX=EPRT
25  CONTINUE
      IF(DABS(SMX).GT.1.00-10) GO TO 555
      PSMX(KK)=ESS(K)*MO12(K)
      GO TO 100
555  CONTINUE
      PSMX(KK)=DABS(1.000/SMX)
100  CONTINUE
      AF1=PSMX(1)
      AF2=PSMX(2)
      AF4=PSMX(3)
      RETURN
      END

C
C
C
C      SUBROUTINE POLY(J,K,XC,YC,W,AST,NCOL,LTHCM,BPR,IL)
      IMPLICIT REAL*8(A-H,O-Z)
      DIMENSION XC(2,5),YC(2,5),A1(400),A2(400),XB(400)
      DIMENSION YB(400),T(400),A1A(4),A2A(4)
      COMMON/CMT1/XB,YB,A1,A2,T
      ARRAY COLLOCATION POINTS AROUND EXTERIOR BOUNDARY
      AND APPLY STRESS BOUNDARY CONDITIONS

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	DO 120 I=1,4	00007760
	A1A(I)=0.	00007770
	A2A(I)=0.	00007780
120	CONTINUE	00007790
	W=DABS(YC(K,2)-YC(K,1))	00007800
	IF(LTNCM.EQ.1) A1A(1)=1000.0	00007810
	IF(LTNCM.EQ.2) A1A(1)=-1000.0	00007820
	IF(BPR.NE.0.0) A1A(3)=A1A(1)	00007830
	IF(IL.EQ.2) A1A(3)=0.0	00007840
	AST=DABS(A1A(1))	00007850
	J=0	00007860
	XC(K,5)=XC(K,1)	00007870
	YC(K,5)=YC(K,1)	00007880
	PI=DARCOS(-1.000)	00007890
	DAT=PI/NCOL	00007900
	DO 10 I=1,4	00007910
	X=XC(K,I)-XC(K,I+1)	00007920
	Y=YC(K,I+1)-YC(K,I)	00007930
	IF(X.EQ.0.) X=1.D-6	00007940
	IF(Y.EQ.0.) Y=1.D-6	00007950
	TH=DATAH2(X,Y)	00007960
	TH=TH*180./DARCOS(-0.1D1)	00007970
	DX=(XC(K,I+1)-XC(K,I))/(NCOL+1)	00007980
	DY=(YC(K,I+1)-YC(K,I))/(NCOL+1)	00007990
	DO 20 II=1,NCOL	00008000
	J=J+1	00008010
	IF(I.EQ.1.OR.I.EQ.4) GO TO 23	00008020
	YB(J)=YC(K,I)	00008030
	XB(J)=XC(K,I)+DX*(II-5)	00008040
	IF(II.EQ.1) XB(J)=XC(K,I)+(DX/2.)	00008050
	GO TO 24	00008060
23	CONTINUE	00008070
	IF(XC(K,5).NE.0.0) GO TO 26	00008080
	IF(I.NE.3) GO TO 24	00008090
	ADT=DAT*II	00008100
	XB(J)=YC(K,3)*DCOS((PI/2.)*ADT)	00008110
	YB(J)=YC(K,3)*DSIN((PI/2.)*ADT)	00008120
	TH=((PI/2.)*ADT)*180./PI	00008130
	GO TO 24	00008140
26	CONTINUE	00008150
	YB(J)=YC(K,1)+DY*(II-5)	00008160
	IF(II.EQ.1) YB(J)=YC(K,1)+(DY/2.)	00008170
	XB(J)=XC(K,1)	00008180
24	T(J)=TH	00008190
	A1(J)=A1A(I)	00008200
	A2(J)=A2A(I)	00008210
20	CONTINUE	00008220
10	CONTINUE	00008230
	RETURN	00008240
	END	00008250
C		00008260
C		00008270
C		00008280
	SUBROUTINE CIRC(W,AST,JK,K,LTNCM,BPR,IL)	00008290
	IMPLICIT REAL*8(A-H,O-Z)	00008300
	DIMENSION X(400),Y(400),THTA(400),A1(400),A2(400)	00008310
	DIMENSION XB(400),YB(400)	00008320
	COMMON/FB1/BSTR,XSTR	00008330
	COMMON/CMT1/XB,YB,A1,A2,THTA	00008340
	COMMON/CMF2/X,Y,NPST,HAFT	00008350

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COMMON/EL/A,B,N                                00008360
ARRAY COLLOCATION POINTS AROUND INNER BOUNDARY  00008370
AND APPLY BEARING STRESS IN A COSINUSOIDAL    00008380
DISTRIBUTION                                  00008390
                                                00008400
CON=-1.0                                        00008410
XSTR=AST                                        00008420
BSTR=(2.*MMXSTR)/(DARCOS(CON)*B)              00008430
IF(BPR.NE.0.0.AND.BPR.NE.1.0.AND.IE.EQ.1) BSTR=0.0 00008440
IF(BPR.EQ.1.0) BSTR=0.                        00008450
NM4=N-4                                         00008460
NQ=NM4/4                                       00008470
DO 20 I=1,N                                   00008480
  JK=JK+1                                       00008490
  TH=((I-1)*2+1)*DARCOS(CON)/N                 00008500
  X(I)=AMDCOS(TH)                              00008510
  Y(I)=AMD SIN(TH)                              00008520
  CS=-X(I)*BMB/(Y(I)*MAA)                     00008530
  IF(Y(I).GT.0) THTA(JK)=DATAN(CS)-DARCOS(CON)/2. 00008540
  IF(Y(I).LT.0) THTA(JK)=DATAN(CS)+DARCOS(CON)/2. 00008550
  THTA(JK)=THTA(JK)*180./DARCOS(CON)           00008560
  IF(LTNCM.EQ.2) GO TO 25                      00008570
  IF(I.GT.(NQ+1).AND.I.LT.(N-NQ)) GO TO 204    00008580
  GO TO 30                                      00008590
25 IF(I.LE.(NQ+2).OR.I.GE.(N-NQ-1)) GO TO 204  00008600
30 CONTINUE                                    00008610
  A1(JK)=0.                                    00008620
  A2(JK)=0.                                    00008630
  XB(JK)=X(I)                                  00008640
  YB(JK)=Y(I)                                  00008650
  GO TO 204                                    00008660
204 IF(Y(I).GT.0.) TETA=ARCCOS(-1.)/DATAN(Y(I)/X(I)) 00008670
    IF(Y(I).LT.0.) TETA=ARCCOS(-1.)/DATAN(Y(I)/X(I)) 00008680
    A1(JK)=-1.0*BSTR*MDABS(DCOS(TETA))           00008690
    A2(JK)=0.                                    00008700
    XB(JK)=X(I)                                  00008710
    YB(JK)=Y(I)                                  00008720
    GO TO 204                                    00008730
20 CONTINUE                                    00008740
  RETURN                                        00008750
END                                              00008760
SUBROUTINE RCDUT(K)                             00008770
SPECIFY COORDINATES AROUND CHARACTERISTIC CIRCLE 00008780
AN WHICH STRESSES ARE NEEDED FOR THE HOFFMAN/  00008790
TSAI-HILL FAILURE CRITERIA                    00008800
IMPLICIT REAL*8(A-H,O-Z)                      00008810
DIMENSION X(400),Y(400),RCA(2),RCB(2),NRC(2) 00008820
COMMON/CMT2/X,Y,NPST,MAST                     00008830
COMMON/RC/RCA,RCB,NRC                         00008840
COMMON/ELP/AX,BX,NOUT                         00008850
RAD=DARCOS(-0.1D1)/180.                      00008860
N1=NRC(K)                                       00008870
DO 40 I=1,N1                                   00008880
  TINC=360./NRC(K)                             00008890
  THETA=(I-1)*TINC*PI/180.                     00008900
  C=DCOS(THETA)                                00008910
  S=DSIN(THETA)                                00008920
  X(I)=X(NPST)+C*(X(NPST)-X(NOUT))             00008930
  Y(I)=Y(NPST)+S*(Y(NPST)-Y(NOUT))             00008940
  RCA(I)=X(I)                                  00008950
  RCB(I)=Y(I)

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K=DSQRT(1./((CXX2/RCA(K)*X2)+(SXX2/RCB(K)*X2))
X(I+NGUT)=R*DCOS(THETA)
Y(I+NGUT)=R*DSIN(THETA)
40 CONTINUE
RETURN
END

SUBROUTINE PSTRSS(K,NCS,BPR,IL)

SPECIFY DISCRETE COORDINATES OF POINTS AT WHICH
STRESSES ARE REQUIRED FOR THE POINT STRESS
CRITERION

IMPLICIT REAL*8(A-H,O-Z)
DIMENSION X(400),Y(400),DONT(2),DOBR(2)
DIMENSION DOSO(2),NPLY(2),NUMPLY(2),ANG(5,2)
DIMENSION IPLY(100,2),AONT(2,2),AOBR(2,2),AOSO(2,2)
COMMON/ELP/AX,BX,NGUT
COMMON/CMT2/X,Y,NPST,NAST
COMMON/PSC1/DONT,DOBR,DOSO
COMMON/LYP/NPLY,NUMPLY,ANG,IPLY
COMMON/AOV/AONT,AOBR,AOSO
AHT=DONT(K)
ABR=DOBR(K)
ASO=DOSO(K)
IF(BPR.EQ.0.0.OR.BPR.L2.1.0) GO TO 25
AHT=AONT(IL,K)
ABR=AOBR(IL,K)
ASO=AOSO(IL,K)
5 CONTINUE
L=NGUT+1
SQ=1.0
IF(NCS.EQ.1) SQ=-1.0
X(L)=0.
Y(L)=AHT+BX
X(L+1)=SQ*(AX+ABR)
Y(L+1)=0
X(L+2)=SQ*(AX+ASO)
Y(L+2)=BX
NPST=3
DO 555 I=1,3
L=NGUT+1+(I-1)
RETURN
END

SUBROUTINE AVSTRS(K,NCS,NAVD,BPR,IL)

SPECIFY COORDINATES OF POINTS ALONG WHICH
STRESSES WILL BE AVERAGED FOR THE AVERAGE
STRESS CRITERION

IMPLICIT REAL*8(A-H,O-Z)
DIMENSION X(400),Y(400),DONT(2),DOBR(2)
DIMENSION DOSO(2),NPLY(2),NUMPLY(2),ANG(5,2)
DIMENSION IPLY(100,2),AONT(2,2),AOBR(2,2),AOSO(2,2)
COMMON/AOV/AONT,AOBR,AOSO

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C      AMATRX CALCULATES THE LAMINATE 'A' MATRIX
C      CALL AMATRX(H,KJ)
C      I=3
C      IDGT=4
C      A=3
C      LINVZF INVERTS THE 'A' MATRIX
C      CALL LINVZF(A,N,IA,AI,IDGT,HK,IER)
C      NDEG=4
C      AZ(1)=AI(1,1)/AI(2,2)
C      AZ(2)=-2.*AI(1,3)/AI(2,2)
C      AZ(3)=(2.*AI(1,2)+AI(3,3))/AI(2,2)
C      AZ(4)=-2.*AI(2,3)/AI(2,2)
C      AZ(5)=1.000
C      ZRPOLY FINDS THE ROOTS OF THE CHARACTERISTIC EQUATION
C      CALL ZRPOLY(AZ,NDEG,Z,IER)
C      Z(2) AND Z(4) ARE THE COMPLEX CONJUGATES OF Z(1)
C      AND Z(3) RESPECTIVELY
C      R1=Z(1)
C      R2=Z(3)
C      THE TWO ROOTS MUST BE CHECKED FOR A UNITARY COMPONENT
C      IN EITHER THE REAL OR IMAGINARY PART; SUCH AN
C      OCCURANCE SIGNIFIES A QUASI-ISOTROPIC LAYUP AND
C      THE VALUE MUST BE PERTURBED SLIGHTLY IN ORDER TO
C      AVOID A SINGULAR MATRIX
C      CH(1)=R1
C      CH(2)=(0.0,-1.0)*R1
C      CH(3)=R2
C      CH(4)=(0.0,-1.0)*R2
C      DO 30 IJK=1,4
C      IF(DABS(CH(IJK)).LT.1.0D-10) CH(IJK)=1.0D-10
C      AR=DABS(CH(IJK))
C      IF(AR.LE.1.0) GO TO 11
C      GO TO 32
31 IF((1.0-AR).LT.0.02) CH(IJK)=0.98
C      GO TO 30
32 IF((AR-1.0).LT.0.02) CH(IJK)=1.02
30 CONTINUE
C      R1=DCMPLX(CH(1),CH(2))
C      R2=DCMPLX(CH(3),CH(4))
C      CONSTANTS P1,P2,Q1,Q2 ARE NEEDED FOR STRESS CALCULATIONS
C      P1=AI(1,1)*R1**2+AI(1,2)*AI(1,3)*R1
C      P2=AI(1,1)*R2**2+AI(1,2)*AI(1,3)*R2
C      Q1=AI(2,2)/R1+AI(1,2)*R1-AI(2,3)
C      Q2=AI(2,2)/R2+AI(1,2)*R2-AI(2,3)
C      INPUTS AIN1(I),AIN2(I) ETC. REFER TO BOUNDARY CONDITIONS
C      NT4=4*NT
C      NT8=8*NT
  
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      NT8P4=8*NT+4
      NT8P2=8*NT+2
      NT8P1=8*NT+1
      NB2=2*NB
      NWK=NT8P1*(NT8P1+2)
      CALL CMAT(BC,CMCTCM,CMC,CM,RHS,GRHS,NT4,NT8,NT8P4,NT8P2,
      INT8P1,NB2,NWK,WA,WKK,AI,NOPT4,KJ,ITT)
      RETURN
      END

      SUBROUTINE AMATRX(H,K)
      ASSEMBLE THE A MATRIX

      IMPLICIT REAL*8(A-H,O-Z)
      DIMENSION A(3,3),ANG(5,2),H(2),NPLY(2),NUMPLY(2)
      DIMENSION E1(2),E2(2),Q12(2),V12(2),V21(2)
      DIMENSION IPLY(100,2)
      COMMON/MOD/E1,E2,Q12,V12,V21
      COMMON/LYP/NPLY,NUMPLY,ANG,IPLY
      COMMON/AMT/A
      THKNES=NPLY(K)*H(K)
      DENO=1.-E2(K)*V12(K)*H2/E1(K)
      Q11=E1(K)/DENO
      Q22=E2(K)/DENO
      Q12=V12(K)*Q22
      Q21=Q12
      Q33=Q12(K)
      DO 10 I=1,3
      DO 10 J=1,3
10  A(I,J)=0.
      NN=NPLY(K)
      DO 20 I=1,NN
      T=H(K)
      LP=IPLY(I,K)
      THTAI=ANG(LP,K)*DARCOS(-1.D0)/180.D0
      C=DCOS(THTAI)
      S=DSIN(THTAI)
      A(1,1)=(Q11*CMH4+2.*(Q12+2.*Q33)*CMCHMS+S+Q22*SMH4)*NT+A(1,1)
      A(2,2)=(Q11*SMH4+2.*(Q12+2.*Q33)*CMCHMS+S+Q22*CMH4)*NT+A(2,2)
      A(1,2)=(Q11+Q22-4.*Q33)*CMCHMS+Q12*(CMH4+SMH4)*NT+A(1,2)
      A(2,1)=A(1,2)
      A(3,3)=(Q11+Q22-2.*Q12-2.*Q33)*CMCHMS+Q33*(CMH4+SMH4)*NT+A(3,3)
      A(1,3)=(Q11-Q12-2.*Q33)*CMH3MS+(Q12-Q22+2.*Q33)*SMH3MC)*NT+A(1,3)
      A(2,3)=(Q11-Q12-2.*Q33)*SMH3MC+(Q12-Q22+2.*Q33)*CMH3MS)*NT+A(2,3)
      A(3,2)=A(2,3)
      A(3,1)=A(1,3)
20  CONTINUE
      DO 53 I=1,3
      DO 53 J=1,3
      A(I,J)=A(I,J)/THKNES
53  CONTINUE
      RETURN
      END

      SUBROUTINE CMAT(BC,CMCTCM,CMC,CM,RHS,GRHS,NT4,NT8,NT8P4,NT8P2,

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CCCCC

1HT8P1,NB2,NWK,WA,WKK,AMAT,NOPT4,KJ,ITT)

CMAT OUTPUTS STRESSES, STRAINS, AND DISPLACEMENTS
AT SPECIFIED COORDINATES

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IMPLICIT REAL*8(A-H,O-Z)
DIMENSION RCA(2),RCB(2),NRCOUT(2)
DIMENSION ASX(400),ASXY(400)
COMMON/XXY1/ASX,ASXY
COMMON/ROOTS/R1,R2
COMMON/TERMS/P1,Q1,P2,Q2
COMMON/CMT1/X,Y,AIN1,AIN2,THTA
COMMON/CMT2/XOUT,YOUT,NPST,NAST
COMMON/FB2/FUR,FTHT,FMR
COMMON/GMT/RTHT,REPX,REPY,REPLY
COMMON/RC/RCA,RCB,NRCOUT
COMMON/ELP/AX,BX,HOUT
COMMON/SER/NT,NB
DIMENSION THTA(400),X(400),Y(400),AMAT(3,3)
DIMENSION AIN1(400),AIN2(400),BC(NB2)
DIMENSION XOUT(400),YOUT(400),WKK(NT8P1)
DIMENSION FUR(400),FTHT(400),FMR(400)
DIMENSION RTHT(400),REPX(400),REPY(400),REPLY(400)
COMPLEX*16 CMCTCM(NT8P1,NT8P1),RHS(NT8P1),PH1D,PHI2D,XETA1,XETA2
COMPLEX*16 CM(NB2,NT8P4),CMC(NB2,NT8P1),Z1,Z2,Z11,Z22,R1,R2
COMPLEX*16 T11,T12,T21,T22,P11,P12,P21,P22
COMPLEX*16 P1,P2,Q1,Q2,DCMPLX,CO,CSUM,GRHS(NT8P2)
COMPLEX*16 PH1DP,PHI2DP,PHI1DN,PHI2DN
COMPLEX*16 PH1P,PHI2P,PHI1N,PHI2N,PHI1,PHI2
COMPLEX*16 SV11,SV12,SV21,SV22,RB11,RB21,RB1B,RB2B
COMPLEX*16 R1B,R2B,P1B,P2B,Q1B,Q2B,WA(NWK)
A=AX
B=BX
CO=(0.0,1.0)
RB11=(Q1-P1MR1)/(A*CMR1NB)
RB21=(Q2-P2MR2)/(A*CMR2NB)
REALR1=R1
REALR2=R2
REALP1=P1
REALP2=P2
REALQ1=Q1
REALQ2=Q2
RRB11=RB11
RRB21=RB21
AIMGR1=CMR1
AIMGR2=CMR2
AIMGP1=CMP1
AIMGP2=CMP2
AIMGQ1=CMQ1
AIMGQ2=CMQ2
ARB11=CO*RB11
ARB21=CO*RB21
R1B=DCMPLX(REALR1,AIMGR1)
R2B=DCMPLX(REALR2,AIMGR2)
P1B=DCMPLX(REALP1,AIMGP1)
P2B=DCMPLX(REALP2,AIMGP2)
Q1B=DCMPLX(REALQ1,AIMGQ1)
Q2B=DCMPLX(REALQ2,AIMGQ2)

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RB11B=DCMPLX(RRB11,ARB11) 00011960
RB21B=DCMPLX(RRB21,ARB21) 00011970
JJJ=0 00011980
DO 1000 I=1,NB 00011990
J=IM2 00012000
THTAI=THTA(I)*DARCOS(-1.0D0)/180.D0 00012010
C=DCOS(THTAI) 00012020
S=DSIN(THTAI) 00012030
P11=CXP1+SNQ1 00012040
P12=CXP2+SNQ2 00012050
P21=-SXP1+CXQ1 00012060
P22=-SXP2+CXQ2 00012070
T11=(CXCXR1*NR1+SN5-2.*CNSMR1) 00012080
T12=(CXCXR2*NR2+SN5-2.*CNSMR2) 00012090
T21=(-CNSMR1*NR1+CNS-(CXC-SN5)*NR1) 00012100
T22=(-CNSMR2*NR2+CNS-(CXC-SN5)*NR2) 00012110
Z1=X(I)+R1*Y(I) 00012120
Z2=X(I)+R2*Y(I) 00012130
Z11=CDSQRT(Z1*Z1-AXA-R1*NR1*NB) 00012140
Z22=CDSQRT(Z2*Z2-AXA-R2*NR2*NB) 00012150
REAL1=Z11 00012160
AIMO1=-CONZ11 00012170
IF(DABS(REAL1).LE.1.D-16)REAL1=0.0D0 00012180
IF(DABS(AIMO1).LE.1.D-16)AIMO1=0.0D0 00012190
Z11=DCMPLX(REAL1,AIMO1) 00012200
REAL2=Z22 00012210
AIMO2=-CONZ22 00012220
IF(DABS(REAL2).LE.1.D-16)REAL2=0.0D0 00012230
IF(DABS(AIMO2).LE.1.D-16)AIMO2=0.0D0 00012240
Z22=DCMPLX(REAL2,AIMO2) 00012250
XETA1=(Z1+Z11)/(A-COMR1*NB) 00012260
IF(CDABS(XETA1).LT.0.999) GO TO 300 00012270
GO TO 310 00012280
300 Z11=-Z11 00012290
XETA1=(Z1+Z11)/(A-COMR1*NB) 00012300
310 XETA2=(Z2+Z22)/(A-COMR2*NB) 00012310
IF(CDABS(XETA2).LT.0.999) GO TO 320 00012320
GO TO 330 00012330
320 Z22=-Z22 00012340
XETA2=(Z2+Z22)/(A-COMR2*NB) 00012350
330 CONTINUE 00012360
JJJ=JJJ+1 00012370
C 00012380
C 00012390
C 00012400
NORMAL & TANGENTIAL STRESS BOUNDARY CONDITIONS ARE IMPOSED 00012410
DO 5 N=1,NT 00012420
NP=N 00012430
CM(J-1,N)=NP*XETA1*NNPNT11/Z11 00012440
CM(J-1,2*NT+N)=NP*XETA2*NNPNT12/Z22 00012450
CM(J,N)=NP*XETA1*NNPNT21/Z11 00012460
CM(J,2*NT+N)=NP*XETA2*NNPNT22/Z22 00012470
NN=-N 00012480
CM(J-1,NT+N)=NN*XETA1*NNNT11/Z11 00012490
CM(J-1,3*NT+N)=NN*XETA2*NNNT12/Z22 00012500
CM(J,NT+N)=NN*XETA1*NNNT21/Z11 00012510
CM(J,3*NT+N)=NN*XETA2*NNNT22/Z22 00012520
5 CONTINUE 00012530
CM(J-1,NT8+1)=T11/Z11 00012540
CM(J-1,NT8+2)=T12/Z22 00012550
CM(J,NT8+1)=T21/Z11

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1000 CM(J,NT8+2)=T22/Z22
      CONTINUE
      DO 195 I=1,NB2
      DO 196 J=1,NT4
      REAL1=CM(I,J)
      AIMQ1=-CONJG(CM(I,J))
      IF(DABS(REAL1).LE.1.D-16)REAL1=0.000
      IF(DABS(AIMQ1).LE.1.D-16)AIMQ1=0.000
      CM(I,J)=DCMPLX(REAL1,AIMQ1)
      AIMQ2=-AIMQ1
      CM(I,NT4+J)=DCMPLX(REAL1,AIMQ2)
196 CONTINUE
195 CONTINUE
      DO 295 I=1,NB2
      DO 296 J=1,2
      REAL1=CM(I,NT8+J)
      AIMQ1=-CONJG(CM(I,NT8+J))
      IF(DABS(REAL1).LE.1.D-16)REAL1=0.000
      IF(DABS(AIMQ1).LE.1.D-16)AIMQ1=0.000
      CM(I,NT8+J)=DCMPLX(REAL1,AIMQ1)
      AIMQ2=-AIMQ1
      CM(I,NT8+2+J)=DCMPLX(REAL1,AIMQ2)
296 CONTINUE
295 CONTINUE
      SV11=(P2*Q1B-Q2*P1B)/(Q1*P2-Q2*P1)
      SV12=(P2*Q2B-Q2*P2B)/(Q1*P2-Q2*P1)
      SV21=(Q1*P1B-Q1B*P1)/(Q1*P2-Q2*P1)
      SV22=(Q1*P2B-Q2B*P1)/(Q1*P2-Q2*P1)
      DO 139 I=1,NB2
      IMPOSE RIGID BODY ROTATION CONDITION
      CM(I,2*NT+1)=-CM(I,1)*RB21/RB11+CM(I,2*NT+1)
      CM(I,4*NT+1)=-CM(I,1)*RB11B/RB11+CM(I,4*NT+1)
      CM(I,6*NT+1)=-CM(I,1)*RB21B/RB11+CM(I,6*NT+1)
      CM(I,1)=(0.0,0.0)
      IMPOSE SINGLE-VALUEDNESS CONDITION
      CM(I,NT8+3)=CM(I,NT8+1)*SV11+CM(I,NT8+3)
      CM(I,NT8+4)=CM(I,NT8+1)*SV12+CM(I,NT8+4)
      CM(I,NT8+3)=CM(I,NT8+2)*SV21+CM(I,NT8+3)
      CM(I,NT8+4)=CM(I,NT8+2)*SV22+CM(I,NT8+4)
      CM(I,NT8+1)=(0.0,0.0)
      CM(I,NT8+2)=(0.0,0.0)
139 CONTINUE
      DO 141 I=1,NB2
      DO 142 J=2,NT8
142 CM(I,J-1)=CM(I,J)
      CM(I,NT8)=CM(I,NT8+3)
      CM(I,NT8+1)=CM(I,NT8+4)
141 CONTINUE
      DO 95 I=1,NB2
      DO 96 J=1,NT8P1
      REAL1=CM(I,J)
      AIMQ1=-CONJG(CM(I,J))
      IF(DABS(REAL1).LE.1.D-16)REAL1=0.000
      IF(DABS(AIMQ1).LE.1.D-16)AIMQ1=0.000
      CM(I,J)=DCMPLX(REAL1,AIMQ1)
      AIMQ2=-AIMQ1

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    CMC(I,J)=DCMPLX(REAL1,AIMG2)
96 CONTINUE
95 CONTINUE
    DO 120 I=1,NB
      J=IM2
      BC(J-1)=AIN1(I)
      BC(J)=AIN2(I)
120 DO 100 I=1,NT8P1
      DO 100 J=1,NT8P1
      CSUM=(0.0,0.0)
      DO 110 K=1,NB2
110 CSUM=CMC(K,I)*CMC(K,J)+CSUM
      CMCTCM(I,J)=CSUM
100 CONTINUE
      DO 130 I=1,NT8P1
      CSUM=(0.0,0.0)
      DO 140 K=1,NB2
140 CSUM=CMC(K,I)*BC(K)+CSUM
130 RHS(1)=CSUM
      IJOB=0
      M=1
      CALL LEQ2C(CMCTCM,NT8P1,NT8P1,RHS,M,NT8P1,IJOB,WA,WKK,IER)
      ORHS(1)=-(RHS(2*NT)+RB21+RHS(4*NT)*RB11B+RHS(6*NT)*RB21B)/RB11
      ORHS(8*NT+1)=RHS(8*NT)*SV11+RHS(8*NT+1)*MSV12
      ORHS(8*NT+2)=RHS(8*NT)*SV21+RHS(8*NT+1)*MSV22
      DO 151 I=2,NT8
151 ORHS(I)=RHS(I-1)

C
C
C      STRESS AND STRAIN CALCULATION
      NRC3=NOUT+1
      IF(NOPT4.EQ.1.OR.NOPT4.EQ.3) NRCF=NOUT+NRCOUT(KJ)
      IF(NOPT4.EQ.2) NRCF=NOUT+NPST
      IF(NOPT4.EQ.4) NRCF=NOUT+NAST
      DO 190 K=1,NRCF
      Z1=XOUT(K)+R1*YOUT(K)
      Z2=XOUT(K)+R2*YOUT(K)
      Z11=CDSQRT(Z1*Z1-A*A-R1*R1*NMB)
      Z22=CDSQRT(Z2*Z2-A*A-R2*R2*NMB)
      XETA1=(Z1+Z11)/(A-C0*R1*B)
      IF(CDABS(XETA1).LT.0.999) GO TO 400
      GO TO 410
400 Z11=-Z11
      XETA1=(Z1+Z11)/(A-C0*R1*B)
410 XETA2=(Z2+Z22)/(A-C0*R2*B)
      IF(CDABS(XETA2).LT.0.999) GO TO 420
      GO TO 430
420 Z22=-Z22
      XETA2=(Z2+Z22)/(A-C0*R2*B)
430 CONTINUE
      PH11DP=(0.0,0.0)
      PH12DP=(0.0,0.0)
      PH11DN=(0.0,0.0)
      PH12DN=(0.0,0.0)
      PH11P=(0.0,0.0)
      PH12P=(0.0,0.0)
      PH11N=(0.0,0.0)
      PH12N=(0.0,0.0)
      DO 170 N=1,NT
      NP=N
      00013150
      00013170
      00013180
      00013190
      00013200
      00013210
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CALCULATE DELEFF

COMPUTE PLY STRESSES FROM LAMINATE STRAINS

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NN=NPLY(K)
DO 100 J=1,NN
LP=IPLY(J,K)
THETA=ANG(LP,K)*RAD
LI1=1
LI2=NOUT
NCAS=1
CALL QMATX(K,LI1,LI2,NCAS,NOPT1,RAD,THETA)

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C      INDIVIDUAL PLY LOADS AND COMPUTE FOUNDATION          00014960
C      MODULI                                                00014970
C                                                                00014980
C                                                                00014990
C                                                                00015000
C      NNN=LI2-1                                           00015010
C      PLOADX=0.                                           00015020
C      WK=0.                                               00015030
C      DO 70 I=LI1,NNN                                     00015040
C      TH1=ATETAA(I+1)-ATETAA(I)                          00015050
C      TH2=(ATETAA(I)+ATETAA(I+1))/2.                    00015060
C      THETA=TH2*HRAD                                       00015070
C      C=DCOS(THETA)                                       00015080
C      S=DSIN(THETA)                                       00015090
C      R=DSQRT(1./(C**2/AX**2+S**2/BX**2))                00015100
C      FORCR=((ASIGR(I)+ASIGR(I+1))/2.)*HRNTH1*HRAD*H(K)  00015110
C      FORCRT=((ASIGRT(I)+ASIGRT(I+1))/2.)*HRNTH1*HRAD*H(K) 00015120
C      PLOADX=PLOADX+FORCR*C-FORCRT*S                     00015130
C 70  CONTINUE                                           00015140
C      FK(I)=DABS(PLOADX/(H(K)*DELEFF))                 00015150
C      PLX(I)=(K-1)*NPLY(1)+PLOADX                      00015160
C 100 CONTINUE                                           00015170
C      NT=NN*NPLY(K)                                       00015180
C      NN=NPLY(K)                                           00015190
C      DO 310 I=1,NT                                       00015200
C      DO 310 II=1,NN                                       00015210
C      IF(IPLY(II,K).EQ.I) ANGK(I,K)=FK(II)              00015220
C      IF(IPLY(II,K).EQ.I) PLXPT(I)=PLX(II+(K-1)*NPLY(1)) 00015230
C 310 CONTINUE                                           00015240
C      NP=NN*NPLY(K)                                       00015250
C                                                                00015260
C                                                                00015270
C                                                                00015280
C                                                                00015290
C      COMPUTE TOTAL BEARING LOAD                         00015300
C                                                                00015310
C      IF(K.EQ.1) GO TO 611                               00015320
C      PLXTOT=0.                                           00015330
C      TH=H(1)*NPLY(1)+H(2)*NPLY(2)                     00015340
C      BLOAD=(BSTR*DA*CO5(-1.0D0)*8X*TH)/2.              00015350
C      NN=NPLY(1)+NPLY(2)                                  00015360
C      DO 212 I=1,NN                                       00015370
C      PLXTOT=PLXTOT+PLX(I)                               00015380
C 212 CONTINUE                                           00015390
C 611 CONTINUE                                           00015400
C      RETURN                                              00015410
C      END                                                 00015420
C                                                                00015430
C      SUBROUTINE FCRT(SALOH,H,WTH,AST,K,NOPT1,NOPT4,BPR,NAVD,IL) 00015440
C                                                                00015450
C                                                                00015460
C                                                                00015470
C      FAILURE LOAD CALCULATION                          00015480
C                                                                00015490
C                                                                00015500
C                                                                00015510
C      IMPLICIT REAL*8(A-H,O-Z)                          00015520
C      DIMENSION ASIG1(400),ASIG2(400),ASIG6(400),ASIGR(400) 00015530
C      DIMENSION UR(400),FSMR(400),ATETAA(400),ASIGRT(400),NPLY(2) 00015540
C      DIMENSION ANG(5,2),IPLY(100,2),HFMC(3,2),PLXPT(100),NPLY(2) 00015550
C      DIMENSION PNS(5,2),PBR(5,2),P50(5,2),PALT(3,2)

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DIMENSION H(2), SALOW(2), SX(400), SXY(400), RCA(2), RCB(2), NRC(2), 00015560
DIMENSION AEP51(400), PFL(5,2), PSTC(5,5,2) 00015570
DIMENSION BPSTS(2,10,2,3) 00015580
COMMON/BP1,BPSTS 00015590
COMMON/RC/RCA,RCB,NRC 00015600
COMMON/FB1/BSTR,XSTR 00015610
COMMON/STRSS2/AEP51 00015620
COMMON/FB2/UR,ATETAA,FSMR 00015630
COMMON/FCT/FLXPT 00015640
COMMON/ELP/AX,BX,NOUT 00015650
COMMON/HFF/HFMC 00015660
COMMON/LYP/NPLY,NUMPLY,ANG,IPLY 00015670
COMMON/FAL1/PHS,PBR,PJO,PALT 00015680
COMMON/FAL3/PFL 00015690
COMMON/STRESS/ASIOR,ASIOR1,ASIOR2,ASIOR6 00015700
COMMON/PSC2/PSTC 00015710
COMMON/PSC3/SX,SXY 00015720
RAD=DARCO3(-0.1D1)/180. 00015730
IF(1.EQ.1.AND.(BPR.EQ.0.0.OR.BPR.EQ.1.0)) WRITE(6,19) K 00015740
19 FORMAT(//, ' ANALYSIS OF PLATE NO', I3, ' ', //) 00015750
IF(NOPT4.EQ.2) GO TO 20 00015760
IF(NOPT4.EQ.3) GO TO 30 00015770
IF(NOPT4.EQ.4) GO TO 40 00015780
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HOFFMAN/TSAI-HILL CRITERIA

WRITE(6,10)
10 FORMAT(//, ' HOFFMAN/TSAI-HILL CRITERION', //)
LI1=NOUT+1
LI2=LI1+NRC(K)
NCAS=2
NN=NUMPLY(K)
DO 402 I=1,NN
THETA=ANG(I,K)*RAD
CALL QMATX(K,LI1,LI2,NCAS,NOPT1,RAD,THETA)
PFAIL=1.0D10
N1=NRC(K)
IF(BPR.EQ.0.0) DSB=DABS(PLXPT(I))
IF(BPR.EQ.1.0) DSB=DABS(XSTR)
DO 404 J=1,N1
S1=ASIOR1(J)/DSB
S2=ASIOR2(J)/DSB
S6=ASIOR6(J)/DSB
CALL HOFF(S1,S2,S6,A,B,K)
NNN=NOUT+J

FOR EACH PLY TYPE FIND THE LOCATION AND MAGNITUDE
OF THE HIGHEST HOFFMAN/TSAI-HILL FAILURE INDEX VALUE

P1=(-B+DSQRT(B**2+4*A))/(2*MA)
P2=(-B-DSQRT(B**2+4*A))/(2*MA)
IF(P1.LT.0.D0) PF=P2
IF(P2.LT.0.D0) PF=P1
IF(P1.LT.P2.AND.P1.GT.0.D0) PF=P1
IF(P2.LT.P1.AND.P2.GT.0.D0) PF=P2
IF(DABS(PF).GT.PFAIL) GO TO 480

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PFAIL=DABJ(PF)
AMAX=A
BMAX=B
LOC=J
480 CONTINUE
404 CONTINUE
A=AMAX
B=BMAX
THTA=ATETAA(NOUT+LOC)

THE CORRESPONDING FAILURE LOAD IS OBTAINED FROM THE INDEX VALUE

XULT=HFMC(1,K)
IF(ASIO1(LOC).LT.0.) XULT=-HFIC(2,K)
YULT=HFMC(3,K)
IF(ASIC2(LOC).LT.0.) YULT=-HFMC(4,K)
SULT=HFIC(5,K)
IF(ASIO6(LOC).LT.0.) SULT=-HFMC(5,K)
SR1=ASIO1(LOC)/XULT
SR2=ASIO2(LOC)/YULT
SR6=ASIO6(LOC)/SULT
IF(BPR.EQ.0.0) WRITE(6,405) I,ANO(I,K),THTA,PFAIL,SR1,SR2,SR6
405 FORMAT(//,' FOR PLY TYPE NO ',I5,' ( ',D9.3,' DEGREES ),//,
  * THE HIGHEST FAILURE INDEX WAS FOUND AT ',D9.3,' DEGREES',//,
  * THE CORRESPONDING FAILURE LOAD = ',D9.3,' LBS',//,
  * THE STRESS RATIOS AT THIS LOCATION ARE',//,
  * SIG1/XULT = ',D9.3,/,
  * SIG2/YULT = ',D9.3,/,
  * SIG6/SULT = ',D9.3,/,
  PFL(I,K)=PFAIL
402 CONTINUE
IF(BPR.EQ.0.0) GO TO 80
SFAIL=1.0D10
DO 110 I=1,NH
  IF(SFAIL.GT.PFL(I,K)) NPY=I
110 IF(SFAIL.GT.PFL(I,K)) SFAIL=PFL(I,K)
  PLFL=SFAIL*WTHM(K)*NPLY(K)
  WRITE(6,771) PLFL
771 FORMAT(//,' FOR THE OPEN HOLE LAMINATE, FAILURE',//,
  * IS PREDICTED AT A JOINT LOAD OF ',D9.3,' LBS',//)
  GO TO 80
20 CONTINUE

POINT STRESS CRITERION

IF(IL.EQ.1.AND.(BPR.EQ.0.0.OR.BPR.EQ.1.0)) WRITE(6,50)
50 FORMAT(//,' POINT STRESS CRITERION ',//)
NH=NUMPLY(K)
NCAS=2
LI1=NOUT+1
LI2=LI1+2
IF(BPR.EQ.1.0.AND.NOPT1.EQ.2) NH=I
DO 100 I=1,NH
  THETA=ANO(I,K)*RAD
  CALL QMATX(K,LI1,LI2,NCAS,NOPT1,RAD,THETA)
  IF(BPR.EQ.0.0.OR.BPR.EQ.1.0) GO TO 705

```

```

IF(IL.EQ.1) FAC=BPR
IF(IL.EQ.2) FAC=(1.-BPR)/PLXPT(I)
BPSTS(K,I,IL,1)=SX(1)*FAC/(AST*WTH*H(K)*NPLY(K))
BPSTS(K,I,IL,2)=SX(2)*FAC/(AST*WTH*H(K)*NPLY(K))
BPSTS(K,I,IL,3)=SX(3)*FAC/(AST*WTH*H(K)*NPLY(K))
GO TO 100
705 CONTINUE
IF(BPR.EQ.0.0) DSN=DABS(PLXPT(I))
IF(BPR.EQ.1.0) DSN=XSTR
PNT=DSNMPSTC(1,I,K)/DABS(SX(1))
IF(SX(1).LT.0.) PNT=DSNMPSTC(2,I,K)/DABS(SX(1))
PBN=DSNMPSTC(3,I,K)/DABS(SX(2))
PSH=DSNMPSTC(4,I,K)/DABS(SX(3))
IF(BPR.EQ.0.0) WRITE(6,70) I,ANG(I,K),PNT,PBN,PSH
70 FORMAT(/,' FOR PLY TYPE NUMBER ',I5,' WITH ',
* A PLY ORIENTATION OF ',D9.3,' DEGREES ',
* NET SECTION FAILURE LOAD = ',D9.3,' LBS ',
* BEARING FAILURE LOAD = ',D9.3,' LBS ',
* SHEAROUT FAILURE LOAD = ',D9.3,' LBS ',
PNS(I,K)=PNT
PBR(I,K)=PBN
PSO(I,K)=PSH
100 CONTINUE
IF(BPR.EQ.0.0) GO TO 80
N=NPLY(K)
IF(BPR.EQ.1.0.AND.NOPT1.EQ.2) N=1
PFAIL1=1.0D10
PFAIL2=1.0D10
PFAIL3=1.0D10
DO 781 I=1,N
IF(PFAIL1.GT.PNS(I,K)) NPY1=I
IF(PFAIL1.GT.PNS(I,K)) PFAIL1=PNS(I,K)
IF(PFAIL2.GT.PBR(I,K)) NPY2=I
IF(PFAIL2.GT.PBR(I,K)) PFAIL2=PBR(I,K)
IF(PFAIL3.GT.PSO(I,K)) NPY3=I
781 IF(PFAIL3.GT.PSO(I,K)) PFAIL3=PSO(I,K)
IF(PFAIL1.GE.PFAIL2.OR.PFAIL1.GE.PFAIL3) GO TO 813
PFAIL1=PFAIL1*WTH*H(K)*NPLY(K)
WRITE(6,982) PFAIL1
WRITE(6,814)
GO TO 811
813 IF(PFAIL2.GE.PFAIL1.OR.PFAIL2.GE.PFAIL3) GO TO 812
PFAIL2=PFAIL2*WTH*H(K)*NPLY(K)
WRITE(6,982) PFAIL2
WRITE(6,815)
GO TO 811
812 IF(PFAIL3.GE.PFAIL1.OR.PFAIL3.GE.PFAIL2) GO TO 811
PFAIL3=PFAIL3*WTH*H(K)*NPLY(K)
WRITE(6,982) PFAIL3
WRITE(6,816)
811 CONTINUE
982 FORMAT(/,' FOR THE LAMINATE WITH AN OPEN HOLE, FAILURE ',
* IS PREDICTED AT A JOINT LOAD OF ',D9.3,' LBS ',
814 FORMAT(' PREDICTED FAILURE MODE IS NET SECTION',/)
815 FORMAT(' PREDICTED FAILURE MODE IS BEARING FAILURE',/)
816 FORMAT(' PREDICTED FAILURE MODE IS SHEAR-OUT FAILURE',/)
GO TO 80
C
C
C
AVERAGE STRESS CRITERION

```

40 CONTINUE

IF(IL.EQ.1.AND.(BPR.EQ.0.0.OR.BPR.EQ.1.0)) WRITE(6,55)
55 FORMAT(/,' AVERAGE STRESS CRITERION ',//)
LI1=NOUT+1
NN=NUMPLY(K)
NCAS=2
IF(BPR.EQ.1.0.AND.NOPT1.EQ.2) NN=1
DO 105 I=1,NN
LI2=NOUT+3*NAVD
THETA=ANG(I,K)*RAD
CALL QMATX(K,LI1,LI2,NCAS,NOPT1,RAD,THETA)

CALCULATE AVERAGE STRESS

SUM=0.
N1=1
N2=NAVD
DO 200 IJ=N1,N2
200 SUM=SUM+5X(IJ)
AS1=SUM/FLOAT(NAVD)
N1=NAVD+1
N2=2*NAVD
SUM=0.
DO 215 IJ=N1,N2
215 SUM=SUM+5XY(IJ)
AS2=SUM/FLOAT(NAVD)
SUM=0.
N1=2*NAVD+1
N2=3*NAVD
DO 220 IJ=N1,N2
220 SUM=SUM+5X(IJ)
AS3=SUM/FLOAT(NAVD)
IF(BPR.EQ.0.0.OR.BPR.EQ.1.0) GO TO 720
IF(IL.EQ.1) FAC=BPR
IF(IL.EQ.2) FAC=(1.-BPR)/PLXPT(1)
BPSTS(K,I,IL,1)=AS1*FAC/(ASTMWTM(K)*NPLY(K))
BPSTS(K,I,IL,2)=AS3*FAC/(ASTMWTM(K)*NPLY(K))
BPSTS(K,I,IL,3)=AS2*FAC/(ASTMWTM(K)*NPLY(K))
GO TO 105
720 CONTINUE
IF(BPR.EQ.0.0) DSN=DABS(PLXPT(1))
IF(BPR.EQ.1.0) DSN=XSTR
PNT=DSN*PSTC(1,I,K)/DABS(AS1)
IF(AS1.LT.0.) PNT=DSN*PSTC(2,I,K)/DABS(AS1)
PBN=DSN*PSTC(3,I,K)/DABS(AS3)
PSH=DSN*PSTC(4,I,K)/DABS(AS2)
IF(BPR.EQ.0.0) WRITE(6,75) I,ANG(I,K),PNT,PBN,PSH
75 FORMAT(/,' FOR PLY TYPE NUMBER ',I5,' WITH ',//
' A PLY ORIENTATION OF ',D3,' DEGREES ',//
' NET SECTION FAILURE LOAD = ',D9.3,' LBS ',//
' BEARING FAILURE LOAD = ',D9.3,' LBS ',//
' SHEAROUT FAILURE LOAD = ',D9.3,' LBS ',//)
PNS(I,K)=PNT
PBR(I,K)=PBN
PSO(I,K)=PSH
105 CONTINUE
IF(BPR.EQ.0.0) GO TO 80
N=NUMPLY(K)

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IF(BPR.EQ.1.0.AND.NOPT1.EQ.2) N=1
PFAIL1=1.0D10
PFAIL2=1.0D10
PFAIL3=1.0D10
DO 718 I=1,N
IF(PFAIL1.GT.PNS(I,K)) NPY1=I
IF(PFAIL1.GT.PNS(I,K)) PFAIL1=PNS(I,K)
IF(PFAIL2.GT.PBR(I,K)) NPY2=I
IF(PFAIL2.GT.PBR(I,K)) PFAIL2=PBR(I,K)
IF(PFAIL3.GT.PSO(I,K)) NPY3=I
718 IF(PFAIL3.GT.PSO(I,K)) PFAIL3=PSO(I,K)
IF(PFAIL1.GE.PFAIL2.OR.PFAIL1.GE.PFAIL3) GO TO 883
PFAIL1=PFAIL1*WTH*H(K)*NPLY(K)
WRITE(6,478) PFAIL1
WRITE(6,834)
GO TO 881
883 IF(PFAIL2.GE.PFAIL1.OR.PFAIL2.GE.PFAIL3) GO TO 882
PFAIL2=PFAIL2*WTH*H(K)*NPLY(K)
WRITE(6,473) PFAIL2
WRITE(6,835)
GO TO 881
882 IF(PFAIL3.GE.PFAIL1.OR.PFAIL3.GE.PFAIL2) GO TO 881
PFAIL3=PFAIL3*WTH*H(K)*NPLY(K)
WRITE(6,478) PFAIL3
WRITE(6,886)
881 CONTINUE
478 FORMAT(//,' FOR THE LAMINATE WITH THE OPEN HOLE, FAILURE',//,
* ' IS PREDICTED AT A JOINT LOAD OF ',D9.3,' LBS',//)
884 FORMAT(' PREDICTED FAILURE MODE IS NET SECTION',//)
885 FORMAT(' PREDICTED FAILURE MODE IS BEARING FAILURE',//)
886 FORMAT(' PREDICTED FAILURE MODE IS SHEAR-OUT FAILURE',//)
GO TO 80

C
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C
90 CONTINUE
PAPP=XSTR*H(K)*NPLY(K)*WTH
WRITE(6,772)
772 FORMAT(//,' MAXIMUM STRAIN CRITERION ',//)
L11=NOUT+1
L12=L11+NRC(K)
NCAS=2
NN=NUMPLY(K)
DO 210 I=1,NN
THETA=ANG(I,K)*RAD
CALL GMATX(K,L11,L12,NCAS,NOPT1,RAD,THETA)
STMAX=-1.0D10
N1=NRC(K)
DO 510 J=1,N1
IF(STMAX.LT.DABS(AEPS1(J))) LOC=J
510 IF(STMAX.LT.DABS(AEPS1(J))) STMAX=DABS(AEPS1(J))
THTA=ATETAA(NOUT+LOC)
IF(BPR.EQ.1.0) GO TO 511
PFL(I,K)=DABS(PLXPT(I)*SALOW(K)/STMAX)
WRITE(6,774) I,ANG(I,K),THTA,PFL(I,K)
774 FORMAT(' FOR PLY TYPE NUMBER ',I5,' WITH ',//,
* ' A PLY ORIENTATION OF ',D9.3,' DEGREES ',//,
* ' FAILURE IS PREDICTED AT ',D9.3,' DEGREES ',//,
* ' AT A PLY LOAD OF ',D9.3,' LBS',//)
GO TO 210

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511 PFL(I,K)=ABS(PAPP*SALOW(K)/STMAX)
210 CONTINUE
    IF(BPR.EQ.0.0) GO TO 80
    A=1.0D10
    NN=NUMPLY(K)
    DO 514 I=1,NN
      IF(A.GT.PFL(I,K)) NPY=I
514  IF(A.GT.PFL(I,K)) A=PFL(I,K)
      WRITE(6,778) A
778  FORMAT(//,' FOR THE OPEN HOLE LAMINATE, FAILURE IS',/,
* ' PREDICTED AT A JOINT LOAD OF ',D9.3,' LBS',//)
80  RETURN
    END

SUBROUTINE QMATX(K,LI1,LI2,NCAS,NOPT1,RAD,THETA)

QMATX PERFORMS BASIC STRESS AND STRAIN
TRANSFORMATIONS

IMPLICIT REAL*8(A-H,O-Z)
DIMENSION ASIGR(400),ASIGRT(400),ASIG1(400),ASIG2(400),ASIG6(400)
DIMENSION ATETAA(400),AEPSX(400),AEPSY(400),AEPSXY(400)
DIMENSION E11(2),E22(2),ESS(2),PMU12(2),PMU21(2),SX(400),SXY(400)
DIMENSION AEPS1(400)
DIMENSION ASX(400),ASXY(400)
COMMON/XXY1/ASX,ASXY
COMMON/MOD/E11,E22,ESS,PMU12,PMU21
COMMON/STRSS2/AEPS1
COMMON/STRESS/ASIGR,ASIGRT,ASIG1,ASIG2,ASIG6
COMMON/QMT/ATETAA,AEPSX,AEPSY,AEPSXY
COMMON/PSC3/SX,SXY
J=0
Q11=E11(K)/(1.0-PMU12(K)*PMU21(K))
Q12=(PMU21(K)*E11(K))/(1.0-PMU12(K)*PMU21(K))
Q22=E22(K)/(1.0-PMU12(K)*PMU21(K))
Q66=ESS(K)
C=DCOS(THETA)
S=DSIN(THETA)
BQ11=(Q11*(C**4))+(2.*(Q12+(2.*Q66))*(C**2)*(S**2))+(Q22*(S**4))
BQ12=((Q11+Q22-(4.*Q66))*S*(C**2)*(C**2))+(Q12*(S**4+C**4))
BQ16=((Q11-Q12-(2.*Q66))*S*(C**3))+(Q12-Q22+(2.*Q66))*S*(C**3)*C
*)
BQ22=(Q11*(S**4))+(2.*(Q12+(2.*Q66))*(S**2)*(C**2))+(Q22*(C**4))
BQ26=((Q11-Q12-(2.*Q66))*C*(S**3))+(Q12-Q22+(2.*Q66))*S*(C**3))
BQ66=((Q11+Q22-(2.*(Q12+Q66)))*(S**2)*(C**2))+(Q66*(C**4)+(S**4))
*)
DO 40 I=LI1,LI2
  J=J+1
  IF(NCAS.EQ.1) THETA=ATETAA(I)*RAD
  C=DCOS(THETA)
  S=DSIN(THETA)
  SIGX=BQ11*AEPSX(I)+BQ12*AEPSY(I)+BQ16*AEPSXY(I)
  SIGY=BQ12*AEPSX(I)+BQ22*AEPSY(I)+BQ26*AEPSXY(I)
  SIGXY=BQ16*AEPSX(I)+BQ26*AEPSY(I)+BQ66*AEPSXY(I)
  SX(J)=SIGX
  SXY(J)=SIGXY
  IF(NOPT1.EQ.2) SX(J)=ASX(I)
  IF(NOPT1.EQ.2) SXY(J)=ASXY(I)

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ASIGR(I)=SIGX*CM2+SIGY*SM2+2.*SIGY*SMC      00019160
ASIGRT(I)=-SIGX*SMC+SIGY*CM2+SIGY*(CM2-SM2)  00019170
MSIG1(J)=SIGX*CM2+SIGY*SM2+2.*SMC*SIGY      00019180
ASIG2(J)=SIGX*SM2+SIGY*CM2-2.*SMC*SIGY      00019190
ASIG6(J)=-C*S*SIGX+SIGY*CM2+(CM2-SM2)*SIGY  00019200
AEPS1(J)=AEPSX(I)*CM2+AEPSY(I)*SM2+AEPSXY(I)*SMC 00019210
40 CONTINUE                                00019220
RETURN                                    00019230
END                                        00019240
COC 00019250
COC 00019260
COC 00019270
SUBROUTINE HOFF(S1,S2,S6,A,B,K)              00019280
IMPLICIT REAL*8(A-H,O-Z)                   00019290
DIMENSION HPMC(5,2)                         00019300
COMMON/HFF/HPMC                             00019310
COC 00019320
COC 00019330
COMPUTE THE HOFFMAN/TSAL-HILL FAILURE INDEX 00019340
A=0.0D0                                     00019350
B=0.0D0                                     00019360
XC=HPMC(1,K)                                00019370
XT=HPMC(2,K)                                00019380
YC=HPMC(3,K)                                00019390
YT=HPMC(4,K)                                00019400
STC=HPMC(5,K)                               00019410
A=(S1*S2-S1*XT)/(XC*XT)+(S2*S2)/(YC*YT)+(S6/STC)*S2 00019420
IF(XC.EQ.XT.AND.YC.EQ.YT) GO TO 10          00019430
B=((XC-XT)/(XC*XT))*S1+((YC-YT)/(YC*YT))*S2 00019440
GO TO 20                                     00019450
10 CONTINUE                                00019460
B=0.0D0                                     00019470
20 CONTINUE                                00019480
RETURN                                       00019490
END                                          00019500
COC 00019510
COC 00019520
COC 00019530
COC 00019540
COC 00019550
COC 00019560
SUBROUTINE CENTD(RF,H,FASSS,FASBS,P,DELP,ITT) 00019570
COC 00019580
COC 00019590
IMPLICIT REAL*8(A-H,O-Z)                   00019600
DIMENSION PLYK(100),BARK(100),BARU(100),F(100) 00019610
DIMENSION H(2),RF(2)                       00019620
DIMENSION AII(100,100),A(2),B(2)           00019630
DIMENSION NPLY(2)                           00019640
COMMON/PBB/PLYK,BARK,BARU                  00019650
COMMON/AFM/AII,F                           00019660
COMMON/LYP/NPLY                             00019670
COC 00019680
SET UP THE CENTRAL DIFFERENCE EQUATIONS     00019690
DO 3 I=1,100                                00019700
DO 3 J=1,100                                00019710
3 AII(I,J)=0.                               00019720
COC 00019730
COC 00019740
NECESSARY CONSTANTS ARE FORMED              00019750

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C      DO 7 I=1,2
      A(I)=H(I)*H2/FASSS
7      B(I)=H(I)*H4/FASBS
      H2=H(1)/H(2)
      A1=H(1)*H2/FASSS
      A2=H(2)*H2/FASSS
      NP=NPLY(1)+NPLY(2)

CCCC   SHEAR AT TOP OF JOINT EQUALS ZERO
      AII(1,1)=1.
      AII(1,2)=-2.+A1*PLYK(2)
      AII(1,4)=2.+A1*PLYK(2)
      AII(1,5)=-1.
      F(1)=0.0

CC      MOMENT CONDITION AT TOP
      IF(RF(1).GE.1.D10) GO TO 50
      Z=1.
      R=RF(1)
      GO TO 60
50      Z=0.
      R=1.
60      AII(2,1)=R
      AII(2,2)=(Z*H2.*H(1)*FASSS)+R*(-2.-A1*PLYK(2)+(H(1)*H2
      *FASSS)/FASBS)
      AII(2,3)=-Z*(4.*H(1)*FASSS+(2*H(1)*H2*PLYK(1)*H(1)))
      AII(2,4)=Z*H2.*H(1)*FASSS+R*(2.+A1*PLYK(2)-(H(1)*H2
      *FASSS)/FASBS)
      AII(2,5)=-R
      F(2)=Z*H2.*H(1)*H3*BARK(1)*BARU(1)

CC      GOVERNING EQUATIONS FOR THE TOP PLATE
      H2=NPLY(1)
      DO 55 J=1,N2
      I=J+2
      AII(I,J)=1.
      IF(J.EQ.1) GO TO 56
      AII(I,J+1)=-4.-A(1)*PLYK(J-1)
      GO TO 57
56      AII(I,J+1)=-4.-A(1)*PLYK(2)
57      AII(I,J+2)=6.+(2.*A(1)+B(1))*PLYK(J)
      IF(J.EQ.N2) GO TO 61
      AII(I,J+3)=-4.-A(1)*PLYK(J+1)
      GO TO 62
61      AII(I,J+3)=-4.-A(1)*PLYK(NPLY(1)-1)
62      AII(I,J+4)=1.
      IF(J.EQ.1) GO TO 58
      IF(J.EQ.N2) GO TO 63
      F(1)=A(1)*BARK(J-1)*BARU(J-1)
      M=-(2.*A(1)+B(1))*BARK(J)*BARU(J)
      *A(1)*BARK(J+1)*BARU(J+1)
      GO TO 59
58      F(1)=2.*A(1)*BARK(2)*BARU(2)
      M=-(2.*A(1)+B(1))*BARK(1)*BARU(1)
      GO TO 59
  
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	63 F(I)=2.*H(1)*MBARK(NPLY(1)-1)*MBARU(NPLY(1)-1)	00020360
	X=(2.*A(1)+B(1))*BARK(J)*BARU(J)	00020370
	59 CONTINUE	00020380
	55 CONTINUE	00020390
C	INTERFACE SHEAR ON TOP PLATE = P+DELP	00020400
C	I=NPLY(1)+3	00020410
C	J=NPLY(1)	00020420
	AII(I,J)=1.	00020430
	AII(I,J+1)=-((2.+A1*PLYK(NPLY(1)-1))	00020440
	AII(I,J+3)=2.+A1*PLYK(NPLY(1)-1)	00020450
	AII(I,J+4)=-1.	00020460
	F(I)=(-2.*H(1)*X3*(P+DELP))/FASBS	00020470
		00020480
C	SLOPE CONTINUITY	00020490
C	I=NPLY(1)+4	00020500
C	J=NPLY(1)	00020510
	AII(I,J)=1.	00020520
	AII(I,J+1)=-((2.+A1*PLYK(NPLY(1)-1)-H(1)*X2*FASBS/FASBS)	00020530
	AII(I,J+3)=2.+A1*PLYK(NPLY(1)-1)-H(1)*X2*FASBS/FASBS	00020540
	AII(I,J+4)=-1.	00020550
	AII(I,J+5)=-H12*X3	00020560
	AII(I,J+6)=H12*X3*(2.+A2*PLYK(NPLY(1)+2)-H(2)*X2*FASBS/FASBS)	00020570
	AII(I,J+8)=-H12*X3*(2.+A2*PLYK(NPLY(1)+2)-H(2)*X2*FASBS/FASBS)	00020580
	AII(I,J+9)=H12*X3	00020590
	F(I)=0.	00020600
C	MOMENT CONTINUITY	00020610
C	I=NPLY(1)+5	00020620
C	J=NPLY(1)+1	00020630
	AII(I,J)=1.	00020640
	AII(I,J+1)=-((2.+A1*PLYK(NPLY(1)))	00020650
	AII(I,J+2)=1.	00020660
	AII(I,J+5)=-H12*X2	00020670
	AII(I,J+6)=H12*X2*(2.+A2*PLYK(NPLY(1)+1))	00020680
	AII(I,J+7)=-H12*X2	00020690
	F(I)=A1*(BARK(NPLY(1))*BARU(NPLY(1))-BARK(NPLY(1)+1)*	00020700
	*BARU(NPLY(1)+1))	00020710
C	INTERFACE SHEAR ON BOTTOM PLATE	00020720
C	I=NPLY(1)+6	00020730
C	J=NPLY(1)+5	00020740
	AII(I,J)=-1.	00020750
	AII(I,J+1)=(2.+A2*PLYK(NPLY(1)+2))	00020760
	AII(I,J+3)=-((2.+A2*PLYK(NPLY(1)+2))	00020770
	AII(I,J+4)=1.	00020780
	F(I)=2.*H(2)*X3*(P+DELP)/FASBS	00020790
C	GOVERNING EQUATIONS FOR THE BOTTOM PLATE	00020800
C	N1=NPLY(1)+7	00020810
	N2=NPLY(1)+NPLY(2)+6	00020820
	DO 70 I=N1,N2	00020830
	J=I-2	00020840
	AII(I,J)=1.	00020850
	IF(I.EQ.N1) GO TO 71	00020860
		00020870
		00020880
		00020890
		00020900
		00020910
		00020920
		00020930
		00020940
		00020950

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      AII(I,J+1,4,-4,-A(2)*PLYK(J-5)
      GO TO 72
71 AII(I,J+1)=-4,-A(2)*PLYK(NPLY(1)+2)
72 AII(I,J+2)=6,+(2,MA(2)+B(2))*PLYK(J-4)
      IF(I.EQ.N2) GO TO 75
      AII(I,J+3)=-4,-A(2)*PLYK(J-3)
      GO TO 74
75 AII(I,J+3)=-4,-A(2)*PLYK(J-5)
76 AII(I,J+4)=1
      IF(I.EQ.N1) GO TO 73
      IF(I.EQ.N2) GO TO 77
      F(I)=A(2)*BARK(J-5)*BARU(J-5)
      M=(2,MA(2)+B(2))*BARK(J-4)*BARU(J-4)
      N+A(2)*BARK(J-3)*BARU(J-3)
      GO TO 74
73 F(I)=2*MA(2)*BARK(NPLY(1)+2)*BARU(NPLY(1)+2)
      M=(2,MA(2)+B(2))*BARK(J-4)*BARU(J-4)
      GO TO 74
77 F(I)=2*MA(2)*BARK(J-5)*BARU(J-5)
      M=(2,MA(2)+B(2))*BARK(J-4)*BARU(J-4)
74 CONTINUE
70 CONTINUE

      SHEAR ON BOTTOM PLATE EQUALS ZERO

      NP=NPLY(1)+NPLY(2)
      I=NP+7
      J=NP+4
      AII(I,J)=1
      AII(I,J+1)=(2,+A2*PLYK(NP-1))
      AII(I,J+3)=(2,+A2*PLYK(NP-1))
      AII(I,J+4)=1
      F(I)=0

      MOMENT BOUNDARY CONDITION ON BOTTOM PLATE

      I=NP+8
      IF(RF(2).OE.1.D10) GO TO 85
      Z=1
      R=RF(2)
      GO TO 95
85 Z=0
      R=1
95 AII(I,J)=-R
      AII(I,J+1)=Z*(2,MH(2)*FASSS)+R*(2,+A2*PLYK(NP-1))
      M=M(2)*M2*FASSS/FASBS
      AII(I,J+2)=-Z*(4,MH(2)*FASSS+2,MH(2)*M3*PLYK(NP))
      AII(I,J+3)=Z*(2,MH(2)*FASSS)+R*(-2,-A2*PLYK(NP-1))
      M+M(2)*M2*FASSS/FASBS
      AII(I,J+4)=R
      F(I)=Z*(2,MH(2)*M3*BARK(NP)*BARU(NP))
      RETURN
      END

      SUBROUTINE SOLVE(U,H,P,DELP,NSDLS,ITT)

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      IMPLICIT REAL*8(A-H,O-Z)                                00021560
      DIMENSION A(100,100),B(100),NPLY(2),U(100),F(100)      00021570
      DIMENSION SX(100),PLYK(100),H(2)                        00021580
      DIMENSION BARK(100),BARU(100)                           00021590
      COMMON/LYP/NPLY                                           00021600
      COMMON/AFM/A,F                                             00021610
      COMMON/PBB/PLYK,BARK,BARU                                00021620
      SOLUTION OF THE SYSTEM: 9AI(U)=(B)                        00021630
      NP=NPLY(1)+NPLY(2)+8                                       00021640
      DO 444 I=1,NP                                             00021650
444    B(I)=F(I)                                                00021660
                                                                00021670
                                                                00021680
                                                                00021690
                                                                00021700
                                                                00021710
                                                                00021720
                                                                00021730
                                                                00021740
                                                                00021750
      APPLYING QUASSIAN ELIMINATION TO THE                     00021760
      MATRIX OF COEFFICIENTS                                     00021770
      DO 2001 I=1,NP                                           00021780
      IR=I                                                        00021790
2042  IF(A(IR,I).NE.0.) GO TO 2041                               00021800
      IR=IR+1                                                    00021810
      IF(IR.GT.NP) GO TO 2001                                     00021820
      GO TO 2042                                                 00021830
2041  NN=IR+1                                                    00021840
      DO 2002 L=NN,NP                                           00021850
      IF(DABS(A(L,I)).GT.1.D-30) GO TO 2009                     00021860
      A(L,I)=0.                                                  00021870
      GO TO 2002                                                 00021880
2009  CF=-A(IR,I)/A(L,I)                                         00021890
      DO 2003 J=I,NP                                           00021900
      A(L,J)=A(L,J)+CF+A(IR,J)                                   00021910
      IF(DABS(A(L,J)).LT.1.D-30) A(L,J)=0.0                    00021920
2003  CONTINUE                                                  00021930
      B(L)=B(L)+CF+B(I)                                         00021940
2002  CONTINUE                                                  00021950
2001  CONTINUE                                                  00021960
                                                                00021970
      BACK SUBSTITUTION                                         00021980
      DO 2011 I=1,NP                                           00021990
      L=NP+1-I                                                  00022000
      SUM=0.                                                     00022010
      IF(A(L,L).EQ.0.) GO TO 2112                               00022020
      N=L+1                                                      00022030
      IF(N.GT.NP) GO TO 2013                                     00022040
      DO 2013 J=N,NP                                           00022050
      SUM=SUM-A(L,J)*SX(J)                                       00022060
2013  CONTINUE                                                  00022070
      SX(L)=(B(L)+SUM)/A(L,L)                                    00022080
      GO TO 2011                                                 00022090
2112  CONTINUE                                                  00022100
      SX(L)=0.                                                  00022110
2011  CONTINUE                                                  00022120
                                                                00022130
      EQUILIBRIUM CHECK                                         00022140
                                                                00022150
      NPTS=NPLY(1)+NPLY(2)+8
  
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PT=P+DELP
N1=NPLY(1)+2
N2=NPLY(1)+7
NN=NPLY(1)+NPLY(2)+6
SUM4=0.
SUM5=0.
DO 1444 I=3,N1
J=I-2
U(J)=SX(I)
SUM4=SUM4+SX(I)*PLYK(J)*MH(1)
1444 CONTINUE
DO 1555 I=N2,NN
J=I-6
U(J)=SX(I)
SUM5=SUM5+SX(I)*PLYK(J)*MH(2)
1555 CONTINUE
IF(NSDLS.EQ.1) GO TO 810
PT=PT*2.
SUM4=SUM4*2.
SUM5=SUM5*2.
810 CONTINUE
NP=NPLY(1)+NPLY(2)
N=NPLY(1)+NPLY(2)
II=1
DO 311 I=1,N
IF(I.GT.NPLY(1)) II=2
PL=U(I)*PLYK(II)*MH(II)
IF(I.LE.NPLY(1)) GO TO 311
311 CONTINUE
RETURN
END

C
C
SUBROUTINE FAIL(GAMD, U, H, P, DELP, BPR, AST, WTH, PFAIL, ANGLE, NODE,
C
C
C
MIROUT, NOPT4, NULYF, JNT, ITT, NTFL)

IMPLICIT REAL*8(A-H,O-Z)
DIMENSION NPLY(2), MDAMP(100), H(2), PLYK(100), U(100)
DIMENSION BARK(100), BARU(100)
DIMENSION PN(100), MDAMI(100), GAMD(2), GAMN(100)
DIMENSION DELNS(5,2), DELBR(5,2), DELSO(5,2)
DIMENSION UN(100), PFL(5,2), PSTC(5,5,2)
DIMENSION IPLY(100,2), ANG(5,2), NUMPLY(2)
DIMENSION PNS(5,2), PBR(5,2), PSO(5,2), PALT(3,2)
DIMENSION BPSTS(2,10,2,3)
DIMENSION NPNM(100,2)
COMMON/COUNT/NPNM
COMMON/BP1/BPSTS
COMMON/PSC2/PSTC
COMMON/FAL1/PNS, PBR, PSO, PALT
COMMON/FAL3/DELNS, DELBR, DELSO
COMMON/FAL4/UN, GAMN, MDAMP, MDAMI, PN
COMMON/FAL5/PFL
COMMON/PBB/PLYK, BARK, BARU
COMMON/PRT/NDAM, INPLY, ITYP
COMMON/LYP/NPLY, HUMPLY, ANG, IPLY
NP=NPLY(1)+NPLY(2)

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	MODEF=MODE	00023360
	NCC=1	00023370
	GO TO 140	00023380
200	NMN=1	00023390
	IF(I.GT.NPLY(1)) NMN=I-NPLY(1)	00023400
	NX=IPLY(NMN,K)	00023410
	IF(PBR(NX,K).LT.PSO(NX,K).OR.PNS(NX,K).LT.PSO(NX,K)) GO TO 700	00023420
	MODE=1	00023430
	PF=PSO(NX,K)	00023440
	IF(MDAMP(I).EQ.1) MODE=5	00023450
	IF(MDAMP(I).EQ.1) PF=PALT(1,K)*PF	00023460
	GO TO 25	00023470
700	IF(PNS(NX,K).LT.PBR(NX,K)) GO TO 710	00023480
	MODE=2	00023490
	PF=PBR(NX,K)	00023500
	IF(MDAMP(I).EQ.2) MODE=6	00023510
	IF(MDAMP(I).EQ.2) PF=PALT(2,K)*PF	00023520
	GO TO 25	00023530
710	MODE=3	00023540
	PF=PNS(NX,K)	00023550
	IF(MDAMP(I).EQ.3) MODE=7	00023560
	IF(MDAMP(I).EQ.3) PF=PALT(3,K)*PF	00023570
25	CONTINUE	00023580
	NCC=0	00023590
	IF(DELPH.EQ.0) GO TO 210	00023600
	IF(DABS(PL).LT.DABS(PF)) GO TO 210	00023610
	PFP=0.	00023620
	IIMPLY=1	00023630
	DEF=MODE	00023640
	NCC=1	00023650
	GO TO 1212	00023660
210	CONTINUE	00023670
C		00023680
C	DETERMINE INCREMENTAL JOINT LOAD TO CAUSE	00023690
C	PLY FAILURE	00023700
	IF(ITT.LE.1) GO TO 21	00023710
	IF(DABS(DABS(U(I)/UN(I))-1.).LT.1.0D-10) GO TO 20	00023720
21	CONTINUE	00023730
	DELPH=(PF-DABS(PN(I)))*1000./((DABS(PL)-DABS(PN(I)))	00023740
C		00023750
C	A NEGATIVE VALUE OF DELPH INDICATES UNLOADING	00023760
C	IN A PLY. THIS NODE IS THEN SKIPPED	00023770
	IF(DELPH.LT.0.) GO TO 20	00023780
C		00023790
C	RECORD LOWEST JOINT FAILURE LOAD INCREMENT,	00023800
C	PLY IN WHICH IT OCCURS, AND MODE	00023810
	PFP2=PFP	00023820
	IF(DELPH.EQ.0) PFP2=1.	00023830
	IF(DELPH.GT.PFP2) GO TO 20	00023840
	PFP=DELPH	00023850
	IIMPLY=1	00023860
	MODEF=MODE	00023870
20	CONTINUE	00023880
C		00023890
C	LOOP OVER ALL INTERFACES TO FIND LOAD	00023900
C	AND LOCATION OF NEXT DELAMINATION	00023910
		00023920
		00023930
		00023940
		00023950

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NN=NPLY(1),NPLY(2)-2                                00023960
DO 30 J=1,NN                                           00023970
C-----IF INTERFACE HAS ALREADY FAILED, GO TO NEXT    00023980
C-----IF(MDAMI(J).EQ.1) GO TO 50                     00023990
C-----DETERMINE WHICH PLATE INTERFACE IS IN          00024000
C-----K=1                                              00024010
C-----IF(J.GE.NPLY(1)) K=2                            00024020
C-----CALCULATE INTERFACE SHEAR STRAIN FOR CURRENT   00024030
C-----JOINT LOAD                                     00024040
C-----OAMJ=(U(J+K-1)-U(J+K))/H(K)                   00024050
C-----DETERMINE INCREMENTAL JOINT LOAD TO CAUSE     00024060
C-----INTERFACE FAILURE                              00024070
C-----IF(ITT.EQ.1) GO TO 47                           00024080
C-----IF(DABS(DABS(OAMJ/OAMN(J))-1.),LT.1.0D-10) GO TO 50 00024090
47 CONTINUE                                           00024100
DELPP=(OAMD(K)-DABS(OAMN(J)))/(DABS(OAMJ)-DABS(OAMN(J)))*1000. 00024110
IF(DELPP.LT.0.) GO TO 50                             00024120
C-----RECORD LOWEST JOINT FAILURE LOAD INCREMENT,    00024130
C-----PLY OF INTERFACE IN WHICH IT OCCURS, AND MODE  00024140
C-----PPF2=PPF                                         00024150
C-----IF(DELPP.EQ.0) PPF2=1.                          00024160
C-----IF(DELPP.GT.PPF2) GO TO 50                     00024170
PPF=DELPP                                             00024180
INPLY=J                                              00024190
MODEF=9                                              00024200
50 CONTINUE                                           00024210
C-----DETERMINE VALUES AT END OF INCREMENT          00024220
C-----JOINT LOAD AT FAILURE                           00024230
C-----IF(MODEF.EQ.0) GO TO 325                       00024240
P=P+PPF*DELPP/1000.                                00024250
325 CONTINUE                                           00024260
C-----NODAL DISPLACEMENTS AND PLY LOADS            00024270
C-----NN=NPLY(1)+NPLY(2)                             00024280
C-----DO 55 I=1,NN                                   00024290
C-----UN(I)=UN(I)+(U(I)-UN(I))*PPF/1000.            00024300
C-----UPDATE UN                                       00024310
C-----IF(NCC.EQ.1) UN(I)=U(I)                        00024320
C-----K=1                                              00024330
C-----IF(I.GT.NPLY(1)) K=2                            00024340
C-----PN(I)=-H(K)*NPLY(K)*UN(I)+BARK(I)*BARU(I)     00024350
55 CONTINUE                                           00024360
C-----

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C	INTERFACE SHEAR STRAINS	00024560
C		00024570
	NH=NPLY(1)+NPLY(2)-2	00024580
	DO 60 J=1,NH	00024590
	K=1	00024600
	IF(J.GE.NPLY(1)) K=2	00024610
	GAMN(J)=(UN(J+K-1)-UN(J+K))/H(K)	00024620
60	CONTINUE	00024630
1212	CONTINUE	00024640
C		00024650
C	PLY STIFFNESSES, DAMAGE STATES, AND NEXT LOAD	00024660
C	INCREMENT	00024670
		00024680
	K=1	00024690
	IF(INPLY.GT.NPLY(1)) K=2	00024700
	NMN=INPLY	00024710
	IF(INPLY.GT.NPLY(1)) NMN=INPLY-NPLY(1)	00024720
	NX=INPLY(NMN,K)	00024730
	IF(MODEF.NE.0) GO TO 70	00024740
	DELP=1000.	00024750
	NDAM=1	00024760
	GO TO 65	00024770
70	IF(MODEF.NE.1) GO TO 80	00024780
	IF(INPLY.EQ.1.AND.MDAMI(INPLY).EQ.1) GO TO 75	00024790
	IF(INPLY.EQ.NPLY(1).AND.MDAMI(INPLY-1).EQ.1) GO TO 75	00024800
	IF(INPLY.EQ.(NPLY(1)+1).AND.MDAMI(INPLY-1).EQ.1) GO TO 75	00024810
	IF(INPLY.EQ.(NPLY(1)+NPLY(2)).AND.MDAMI(NPLY(1)+NPLY(2)-1-2).EQ.1) GO TO 75	00024820
	KK=0	00024830
	IF(INPLY.GT.NPLY(1)) KK=1	00024840
	IF(MDAMI(INPLY-KK-1).EQ.1.AND.MDAMI(INPLY-KK).EQ.1) GO TO 75	00024850
	MDAMP(INPLY)=1	00024860
	TEMPK=PLYK(INPLY)	00024870
	PLYK(INPLY)=DELSO(NX,K)*TEMPK	00024880
	BARK(INPLY)=(1.-DELSO(NX,K))*TEMPK	00024890
	BARU(INPLY)=UN(INPLY)	00024900
	DELP=0.	00024910
	NDAM=2	00024920
	GO TO 65	00024930
75	PLYK(INPLY)=0.0	00024940
	BARK(INPLY)=0.	00024950
	BARU(INPLY)=UN(INPLY)	00024960
	MDAMP(INPLY)=10	00024970
	DELP=0.	00024980
	NDAM=2	00024990
	GO TO 65	00025000
80	IF(MODEF.NE.2) GO TO 85	00025010
	TEMPK=PLYK(INPLY)	00025020
	PLYK(INPLY)=DELS(NX,K)*TEMPK	00025030
	BARK(INPLY)=(1.-DELS(NX,K))*TEMPK	00025040
	BARU(INPLY)=UN(INPLY)	00025050
	MDAMP(INPLY)=2	00025060
	DELP=0.	00025070
	NDAM=4	00025080
	GO TO 65	00025090
85	IF(MODEF.NE.3) GO TO 90	00025100
	TEMPK=PLYK(INPLY)	00025110
	PLYK(INPLY)=DELSR(NX,K)*TEMPK	00025120
	BARK(INPLY)=(1.-DELSR(NX,K))*TEMPK	00025130
	BARU(INPLY)=UN(INPLY)	00025140
		00025150

MDAMP(INPLY)=3	00025160
DELP=0.	00025170
NDAM=5	00025180
GO TO 65	00025190
90 IF(MODEF.NE.4) GO TO 140	00025200
PLYK(INPLY)=0.0	00025210
BARK(INPLY)=0.	00025220
BARU(INPLY)=UN(INPLY)	00025230
MDAMP(INPLY)=10	00025240
DELP=0.	00025250
NDAM=6	00025260
GO TO 65	00025270
140 IF(MODEF.NE.5) GO TO 110	00025280
PLYK(INPLY)=0.	00025290
BARK(INPLY)=0.	00025300
BARU(INPLY)=UN(INPLY)	00025310
MDAMP(INPLY)=10	00025320
DELP=0.	00025330
NDAM=6	00025340
GO TO 65	00025350
110 IF(MODEF.NE.6) GO TO 115	00025360
PLYK(INPLY)=0.0	00025370
BARK(INPLY)=0.0	00025380
BARU(INPLY)=UN(INPLY)	00025390
MDAMP(INPLY)=10	00025400
NDAM=7	00025410
DELP=0.	00025420
GO TO 65	00025430
115 IF(MODEF.NE.7) GO TO 120	00025440
PLYK(INPLY)=0.	00025450
BARK(INPLY)=0.	00025460
BARU(INPLY)=UN(INPLY)	00025470
MDAMP(INPLY)=10	00025480
DELP=0.	00025490
NDAM=8	00025500
GO TO 65	00025510
120 IF(MODEF.NE.8) GO TO 125	00025520
PLYK(INPLY)=0.	00025530
BARK(INPLY)=0.	00025540
BARU(INPLY)=UN(INPLY)	00025550
MDAMP(INPLY)=10	00025560
NDAM=9	00025570
DELP=0.	00025580
GO TO 65	00025590
125 IF(MODEF.NE.9) GO TO 65	00025600
DELP=1000.	00025610
MDAMI(INPLY)=1	00025620
NDAM=10	00025630
GO TO 65	00025640
600 CONTINUE	00025650
PARTIAL BEARING FAILURE ANALYSIS	00025660
IROUT=2	00025670
NPL=NPLY(1)+NPLY(2)	00025680
AJFLNS=1.0D10	00025690
AJFLBR=1.0D10	00025700
AJFLSD=1.0D10	00025710
DO 550 I=1,NPL	00025720
K=1	00025730
	00025740
	00025750

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IF(I.GT.NPLY(1)) K=2                                00025760
PL=-H(K)*PLY(K)*MU(I)+BARK(I)*BARU(I)                00025770
IF(PL.LT.0.0.AND.K.EQ.1) GO TO 550                  00025780
IF(PL.OT.0.0.AND.K.EQ.2) GO TO 550                  00025790
PL=DABS(PL)*ASTM*WTH*H(K)*NPLY(K)/1000.            00025800
IJ=I                                                    00025810
IF(I.OT.NPLY(1)) IJ=IJ-NPLY(1)                      00025820
IP=IPLY(IJ,K)                                          00025830
NT=I                                                    00025840
IF(BPSTS(K,IP,2,1).LT.0.0) NT=2                     00025850
F1=DABS(PSTC(NT,IP,K)/(BPSTS(K,IP,1,1)+PLMBPSTS(K,IP,2,1))) 00025860
F2=DABS(PSTC(3,IP,K)/(BPSTS(K,IP,1,2)+PLMBPSTS(K,IP,2,2))) 00025870
F3=DABS(PSTC(4,IP,K)/(BPSTS(K,IP,1,3)+PLMBPSTS(K,IP,2,3))) 00025880
IF(MDAMP(I).EQ.5) F1=PALT(3,K)*F1                  00025890
IF(MDAMP(I).EQ.4) F2=PALT(2,K)*F2                  00025900
IF(MDAMP(I).EQ.2) F3=PALT(1,K)*F3                  00025910
IF(AJFLNS.OT.F1) NF1=I                                00025920
IF(AJFLNS.OT.F1) AJFLNS=F1                          00025930
IF(AJFLBR.OT.F2) NF2=I                                00025940
IF(AJFLBR.OT.F2) AJFLBR=F2                          00025950
IF(AJFLSO.OT.F3) NF3=I                                00025960
IF(AJFLSO.OT.F3) AJFLSO=F3                          00025970
550 CONTINUE                                          00025980
IF(AJFLNS.OT.AJFLBR.OR.AJFLNS.OT.AJFLSO) GO TO 560 00025990
INPLY=NF1                                              00026000
551 NDAM=5                                             00026010
IF(MDAMP(INPLY).EQ.2) GO TO 571                      00026020
IF(MDAMP(INPLY).EQ.4) GO TO 561                      00026030
IF(MDAMP(INPLY).EQ.5) NDAM=8                        00026040
MDAMP(INPLY)=NDAM                                    00026050
PFAIL=AJFLNS                                          00026060
GO TO 64                                              00026070
560 IF(AJFLBR.OT.AJFLNS.OR.AJFLBR.OT.AJFLSO) GO TO 571 00026080
INPLY=NF2                                              00026090
561 NDAM=4                                             00026100
IF(MDAMP(INPLY).EQ.5) GO TO 551                      00026110
IF(MDAMP(INPLY).EQ.2) GO TO 571                      00026120
IF(MDAMP(INPLY).EQ.4) NDAM=7                        00026130
MDAMP(INPLY)=NDAM                                    00026140
PFAIL=AJFLBR                                          00026150
GO TO 64                                              00026160
570 IF(AJFLSO.OT.AJFLNS.OR.AJFLSO.OT.AJFLBR) GO TO 64 00026170
INPLY=NF3                                              00026180
571 NDAM=2                                             00026190
IF(MDAMP(INPLY).EQ.5) GO TO 551                      00026200
IF(MDAMP(INPLY).EQ.4) GO TO 561                      00026210
IF(MDAMP(INPLY).EQ.2) NDAM=6                        00026220
MDAMP(INPLY)=NDAM                                    00026230
PFAIL=AJFLSO                                          00026240
64 CONTINUE                                          00026250
K=I                                                    00026260
IF(INPLY.OT.NPLY(1)) K=2                              00026270
IPL=INPLY                                              00026280
IF(IPL.OT.NPLY(1)) IPL=IPL-NPLY(1)                  00026290
IPLP=IPLY(IPL,K)                                      00026300
ANGLE=ANG(IPLP,K)                                    00026310
NODE=NPHM(IPL,K)                                      00026320
IF(MDAMP(INPLY).EQ.6) GO TO 107                      00026330
IF(MDAMP(INPLY).EQ.5) AR=DELNS(IPLP,K)              00026340
IF(MDAMP(INPLY).EQ.4) AR=DELBK(IPLP,K)              00026350

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IF(MDAMP(INPLY).EQ.2) AR=DELSO(IPLP,K) 00026360
TEMPK=PLYK(INPLY) 00026370
PLYK(INPLY)=AR*TEMPK 00026380
BARK(INPLY)=(1.-AR)*TEMPK 00026390
BARU(INPLY)=U(INPLY) 00026400
ITYP=IPLY(IPL,K) 00026410
NTFL=0 00026420
GO TO 103 00026430
107 CONTINUE 00026440
IF(K.EQ.1) NPLY(1)=NPLY(1)-1 00026450
IF(K.EQ.2) NPLY(2)=NPLY(2)-1 00026460
NP=INPLY 00026470
IF(K.EQ.2) NP=INPLY-NPLY(1) 00026480
N=NPLY(K)-NP+2 00026490
ITYP=IPLY(NP,K) 00026500
DO 101 I=1,N 00026510
IPLY(NP+I-1,K)=IPLY(NP+I,K) 00026520
NPNM(NP+I-1,K)=NPNM(NP+I,K) 00026530
101 CONTINUE 00026540
N=NPL-INPLY 00026550
DO 102 I=1,N 00026560
MDAMP(INPLY+I-1)=MDAMP(INPLY+I) 00026570
PLYK(INPLY+I-1)=PLYK(INPLY+I) 00026580
BARK(INPLY+I-1)=BARK(INPLY+I) 00026590
102 BARU(INPLY+I-1)=BARU(INPLY+I) 00026600
NTFL=1 00026610
NULTF=NULTF-1 00026620
IF(NULTF.EQ.0) JNT=0 00026630
IF(NPLY(1).EQ.2.OR.NPLY(2).EQ.2) JNT=0 00026640
103 CONTINUE 00026650
RETURN 00026660
65 CONTINUE 00026670
C C INCREMENT LOAD IF JOINT HAS NOT FAILED 00026680
C C 00026690
T1=0. 00026700
T2=0. 00026710
N1=NPLY(1) 00026720
N2=NPLY(2) 00026730
DO 135 I=1,N1 00026740
135 T1=T1+PLYK(I) 00026750
DO 126 I=1,N2 00026760
N3=NPLY(1)+I 00026770
126 T2=T2+PLYK(N3) 00026780
IF(T1.EQ.0.0.OR.T2.EQ.0.0) GO TO 130 00026790
RETURN 00026800
130 JNT=0 00026810
RETURN 00026820
END 00026830
C C 00026840
C C 00026850
C C 00026860
SUBROUTINE PRINT(U,P,DELP,PFAIL,ANGLE,BPR,NODE,IROUT,JNT, 00026870
MNP,NSDLS,ITY) 00026880
00026890
00026900
00026910
IMPLICIT REAL*8(A-H,O-Z) 00026920
DIMENSION U(100),PLYK(100) 00026930
DIMENSION NPLY(2),NUNPLY(2),ANG(5,2),IPLY(100,2) 00026940
DIMENSION BARK(100),BARU(100) 00026950

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DIMENSION NPNM(100.2)
COMMON/COUNT/NPNM
COMMON/LYP/NPLY,NUMPLY,ANG,IPLY
COMMON/PBB/PLYK,DARK,BARU
COMMON/PRT/NDAM,INPLY,ITYP

PRINT VALUES AT END OF INCREMENT

IF(ITT.EQ.1) WRITE(6,10)
10 FORMAT(//,10X,'FAILURE MODE ABBREVIATIONS'
*10X,'ND' = NO ADDITIONAL DAMAGE AT CURRENT JOINT LOAD'
*10X,'DL' = DELAMINATION
*10X,'SO' = SHEAR-OUT
*10X,'BR' = BEARING
*10X,'NS' = NET SECTION
*10X,'SUD' = ULTIMATE FAILURE AFTER SO AND DL
*10X,'SU' = ULTIMATE FAILURE IN SO
*10X,'BU' = ULTIMATE FAILURE IN BR
*10X,'NSU' = ULTIMATE FAILURE IN NS
*10X,'ULT' = ULTIMATE FAILURE
*4X,'INCREMENT NO',3X,'JOINT LOAD',5X,'NODE',8X,'PLY TYPE',
*8X,'MODE',/)
PL=P
IF(ROUT.EQ.2) PL=PFail
IF(NSDLS.EQ.2) PL=2.*PL
IF(ITT.EQ.1) PFAILP=0.0D0
IF(PFAILP.LT.PL) PFAILP=PL
IF(JNT.EQ.0.AND.PFAILP.EQ.0.0D0) PFAILP=PL
K=1
IF(INPLY.GT.NPLY(1)) K=2
N=IPLY(INPLY,K)
IF(K.EQ.2) N=IPLY((INPLY-NPLY(1)),K)
IF(ROUT.EQ.2) N=ITYP
IF(BPR.EQ.0.0.OR.BPR.EQ.1.0) ANGLE=ANG(N,K)
IF(BPR.EQ.0.0.OR.BPR.EQ.1.0) NODE=INPLY
IF(NDAM.EQ.1) WRITE(6,20) ITT,PL
IF(NDAM.EQ.2) WRITE(6,30) ITT,PL,NODE,ANGLE
IF(NDAM.EQ.3) WRITE(6,40) ITT,PL,NODE,ANGLE
IF(NDAM.EQ.4) WRITE(6,50) ITT,PL,NODE,ANGLE
IF(NDAM.EQ.5) WRITE(6,60) ITT,PL,NODE,ANGLE
IF(NDAM.EQ.6) WRITE(6,70) ITT,PL,NODE,ANGLE
IF(NDAM.EQ.7) WRITE(6,80) ITT,PL,NODE,ANGLE
IF(NDAM.EQ.8) WRITE(6,90) ITT,PL,NODE,ANGLE
IF(NDAM.EQ.9) WRITE(6,100) ITT,PL,NODE,ANGLE
IF(NDAM.EQ.10) WRITE(6,110) ITT,PL,NODE,ANGLE
20 FORMAT(5X,15,10X,D9.3,3X,15,5X,F7.3,' DEGREE ',5X,' SO')
30 FORMAT(5X,15,10X,D9.3,3X,15,5X,F7.3,' DEGREE ',5X,' SUD')
40 FORMAT(5X,15,10X,D9.3,3X,15,5X,F7.3,' DEGREE ',5X,' BR')
50 FORMAT(5X,15,10X,D9.3,3X,15,5X,F7.3,' DEGREE ',5X,' NS')
60 FORMAT(5X,15,10X,D9.3,3X,15,5X,F7.3,' DEGREE ',5X,' SU')
70 FORMAT(5X,15,10X,D9.3,3X,15,5X,F7.3,' DEGREE ',5X,' BU')
80 FORMAT(5X,15,10X,D9.3,3X,15,5X,F7.3,' DEGREE ',5X,' NSU')
90 FORMAT(5X,15,10X,D9.3,3X,15,5X,F7.3,' DEGREE ',5X,' ULT')
110 FORMAT(5X,15,10X,D9.3,3X,15,5X,F7.3,' DEGREE ',5X,' DL')
IF(JNT.EQ.0) GO TO 220
GO TO 250
220 CONTINUE
WRITE(6,240) PFAILP
240 FORMAT(//,' THE PREDICTED JOINT FAILURE ',/,

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*1 LOAD IS 'D14.7' LBS',//)
250 CONTINUE
RETURN
END
C
C
C
SUBROUTINE LINV2F (A,N,IA,AINV,IDGT,WKAREA,IER)
C
DOUBLE PRECISION A(IA,N),AINV(IA,N),WKAREA(1),ZERO,ONE
DATA ONE/1.0D0/,ZERO/0.0D0/
C
C FIRST EXECUTABLE STATEMENT
C INITIALIZE IER
IER=0
C
C SET AINV TO THE N X N
C IDENTITY MATRIX
DO 10 I = 1,N
DO 5 J = 1,N
AINV(I,J) = ZERO
5 CONTINUE
AINV(I,I) = ONE
10 CONTINUE
C
C COMPUTE THE INVERSE OF A
CALL LEQ2F (A,N,N,IA,AINV,IDGT,WKAREA,IER)
IF (IER.EQ.0) GO TO 9005
9000 CONTINUE
CALL UERTST (IER,6HLINV2F)
9005 RETURN
END
C
C
C
SUBROUTINE LEQ2F (A,M,N,IA,B,IDGT,WKAREA,IER)
C
DIMENSION A(IA,1),B(IA,1),WKAREA(1)
DOUBLE PRECISION A,B,WKAREA,D1,D2,WA
C
C FIRST EXECUTABLE STATEMENT
C INITIALIZE IER
IER=0
JER=0
J = N*N+1
K = J+N
MM = K+N
KK = 0
MM1 = MM-1
JJ=1
DO 5 L=1,N
DO 5 I=1,N
WKAREA(JJ)=A(I,L)
JJ=JJ+1
5 CONTINUE
C
C DECOMPOSE A
CALL LUDATN (WKAREA,N,N,A,IA,IDGT,D1,D2,WKAREA(J),WKAREA(K),
* WA,IER)
IF (IER.GT.128) GO TO 25
IF (IDGT.EQ.0 .OR. IER.NE.0) KK = 1
DO 15 I = 1,M
C
C PERFORMS THE ELIMINATION PART OF
C AX = B
CALL LUELNM (A,IA,N,B(1,I),WKAREA(J),WKAREA(MM))
C REFINEMENT OF SOLUTION TO AX = B

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	IF (KK .NE. 0)	00028160
*	CALL LUREFN (WKAREA,N,N,A,IA,B(1,1),IDOT,WKAREA(J),WKAREA(MM),	00028170
*	WKAREA(K),WKAREA(K),JER).	00028180
	DO 10 II=1,N	00028190
	B(II,I) = WKAREA(MM+II)	00028200
10	CONTINUE	00028210
	IF (JER.NE.0) GO TO 20	00028220
15	CONTINUE	00028230
	GO TO 25	00028240
20	IER = 131	00028250
25	JJ=1	00028260
	DO 30 J = 1,N	00028270
	DO 30 I = 1,N	00028280
	A(I,J)=WKAREA(JJ)	00028290
	JJ=JJ+1	00028300
30	CONTINUE	00028310
	IF (IER .EQ. 0) GO TO 9005	00028320
9000	CONTINUE	00028330
	CALL UERTST (IER,6HLEQT2F)	00028340
9005	RETURN	00028350
	END	00028360
		00028370
		00028380
		00028390
	SUBROUTINE LUDATF (A,LU,N,IA,IDOT,D1,D2,IPVT,EQUIL,WA,IER)	00028400
		00028410
	DIMENSION A(IA,1),LU(IA,1),IPVT(1),EQUIL(1)	00028420
	DOUBLE PRECISION A,LU,D1,D2,EQUIL,WA,ZERO,ONE,FOUR,SIXTH,SIXTH,	00028430
*	RN,WREL,BIGA,BIG,P,SUM,AI,WI,T,TEST,Q	00028440
	DATA ZERO,ONE,FOUR,SIXTH,SIXTH/O.DO,1.DO,4.DO,	00028450
*	16.DO,.0625DO/	00028460
	FIRST EXECUTABLE STATEMENT	00028470
	INITIALIZATION	00028480
	IER = 0	00028490
	RN = N	00028500
	WREL = ZERO	00028510
	D1 = ONE	00028520
	D2 = ZERO	00028530
	BIGA = ZERO	00028540
	DO 10 I=1,N	00028550
	BIG = ZERO	00028560
	DO 5 J=1,N	00028570
	P = A(I,J)	00028580
	LU(I,J) = P	00028590
	P = DABS(P)	00028600
	IF (P .GT. BIG) BIG = P	00028610
5	CONTINUE	00028620
	IF (BIG .GT. BIGA) BIGA = BIG	00028630
	IF (BIG .EQ. ZERO) GO TO 110	00028640
	EQUIL(I) = ONE/BIG	00028650
10	CONTINUE	00028660
	DO 105 J=1,N	00028670
	JM1 = J-1	00028680
	IF (JM1 .LT. 1) GO TO 40	00028690
	COMPUTE U(I,J), I=1,...,J-1	00028700
	DO 35 I=1,JM1	00028710
	SUM = LU(I,J)	00028720
	IM1 = I-1	00028730
	IF (IDOT .EQ. 0) GO TO 25	00028740
	WITH ACCURACY TEST	00028750

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    AI = DABS(SUM)
    WI = ZERO
    IF (IM1 .LT. 1) GO TO 20
    DO 15 K=1,IM1
      T = LU(I,K)*LU(K,J)
      SUM = SUM-T
      WI = WI+DABS(T)
15  CONTINUE
    LU(I,J) = SUM
20  WI = WI+DABS(SUM)
    IF (AI .EQ. ZERO) AI = BIGA
    TEST = WI/AI
    IF (TEST .GT. WREL) WREL = TEST
    GO TO 35
C 25  IF (IM1 .LT. 1) GO TO 35
      DO 30 K=1,IM1
        SUM = SUM-LU(I,K)*LU(K,J)
30  CONTINUE
    LU(I,J) = SUM
35  CONTINUE
40  P = ZERO
C      COMPUTE U(J,J) AND L(I,J), I=J+1,...,
    DO 70 I=J,N
      SUM = LU(I,J)
      IF (IDOT .EQ. 0) GO TO 55
C      WITH ACCURACY TEST
      AI = DABS(SUM)
      WI = ZERO
      IF (JM1 .LT. 1) GO TO 50
      DO 45 K=1,JM1
        T = LU(I,K)*LU(K,J)
        SUM = SUM-T
        WI = WI+DABS(T)
45  CONTINUE
      LU(I,J) = SUM
50  WI = WI+DABS(SUM)
      IF (AI .EQ. ZERO) AI = BIGA
      TEST = WI/AI
      IF (TEST .GT. WREL) WREL = TEST
      GO TO 65
C      WITHOUT ACCURACY TEST
55  IF (JM1 .LT. 1) GO TO 65
      DO 60 K=1,JM1
        SUM = SUM-LU(I,K)*LU(K,J)
60  CONTINUE
      LU(I,J) = SUM
65  Q = EQUIL(I)*DABS(SUM)
      IF (P .GE. Q) GO TO 70
      P = Q
      IMAX = I
70  CONTINUE
C      TEST FOR ALGORITHMIC SINGULARITY
      IF (RN+P .EQ. RN) GO TO 110
      IF (J .EQ. IMAX) GO TO 80
C      INTERCHANGE ROWS J AND IMAX
      D1 = -D1
      DO 75 K=1,N
        P = LU(IMAX,K)
        LU(IMAX,K) = LU(J,K)

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	LU(J,K) = P	00029360
75	CONTINUE	00029370
	EQUIL(IMAX) = EQUIL(J)	00029380
80	IPVT(J) = IMAX	00029390
	D1 = DIMLU(J,J)	00029400
85	IF (DABS(D1) .LE. ONE) GO TO 90	00029410
	D1 = DIMSIXTH	00029420
	D2 = D2+FOUR	00029430
	GO TO 85	00029440
90	IF (DABS(D1) .GE. SIXTH) GO TO 95	00029450
	D1 = DIMSIXTH	00029460
	D2 = D2-FOUR	00029470
	GO TO 90	00029480
95	CONTINUE	00029490
	JPI = J+1	00029500
	IF (JPI .GT. N) GO TO 105	00029510
C		00029520
	P = LU(J,J)	00029530
	DO 100 I=JPI,N	00029540
	LU(I,J) = LU(I,J)/P	00029550
100	CONTINUE	00029560
105	CONTINUE	00029570
C		00029580
	IF (IDOT .EQ. 0) GO TO 9005	00029590
	P = 3*H+3	00029600
	WA = P*WREL	00029610
	IF (WA+10.D0*H*(-IDOT) .NE. WA) GO TO 9005	00029620
	IER = 34	00029630
	GO TO 9000	00029640
C		00029650
	ALGORITHMIC SINGULARITY	00029660
110	IER = 129	00029670
	D1 = ZERO	00029680
	D2 = ZERO	00029690
9000	CONTINUE	00029700
C		00029710
	PRINT ERROR	00029720
9005	CALL UERTST(IER,6HLDATF)	00029730
	RETURN	00029740
	END	00029750
C		00029760
	SUBROUTINE LUELMN (A,IA,N,B,APVT,X)	00029770
C		00029780
	DIMENSION A(IA,1),B(1),APVT(1),X(1)	00029790
	DOUBLE PRECISION A,B,X,SUM,APVT	00029800
C		00029810
	FIRST EXECUTABLE STATEMENT	00029820
C		00029830
	SOLVE LY = B FOR Y	00029840
	DO 5 I=1,N	00029850
5	X(I) = B(I)	00029860
	IW = 0	00029870
	DO 20 I=1,N	00029880
	IP = APVT(I)	00029890
	SUM = X(IP)	00029900
	X(IP) = X(I)	00029910
	IF (IW .EQ. 0) GO TO 15	00029920
	IM1 = I-1	00029930
	DO 10 J=IW,IM1	00029940
	SUM = SUM-A(I,J)*X(J)	00029950
10	CONTINUE	
	GO TO 20	
15	IF (SUM .NE. 0.D0) IW = I	

<pre> C 20 X(I) = SUM DO 30 I=1,N I = N+1-IB IP1 = I+1 SUM = X(I) IF (IP1 .GT. N) GO TO 30 DO 25 J=IP1,N SUM = SUM-A(I,J)*X(J) 25 CONTINUE 30 X(I) = SUM/A(I,I) RETURN END C C SUBROUTINE LUREFN (A,IA,N,UL,IUL,B,IDOT,APVT,X,RES,DX,IER) C C DIMENSION A(IA,1),UL(IUL,1),B(1),X(1),RES(1),DX(1) C DIMENSION APVT(1) C DIMENSION ACCXT(2) C DOUBLE PRECISION A,ACCXT,B,UL,X,RES,DX,ZERO,XNORM,DXNORM,APVT C DATA ITMAX/75/,ZERO/0.00/ C C IER=0 C XNORM = ZERO C DO 10 I=1,N C XNORM = DMAX1(XNORM,DABS(X(I))) 10 CONTINUE C IF (XNORM .NE. ZERO) GO TO 20 C IDOT = 50 C GO TO 9005 20 DO 45 ITER=1,ITMAX C DO 30 I=1,N C ACCXT(1) = 0.000 C ACCXT(2) = 0.000 C CALL VXADD(B(I),ACCXT) C DO 25 J=1,N C CALL VXMUL(-A(I,J),X(J),ACCXT) 25 CONTINUE C CALL VXSTO(ACCXT,RES(I)) 30 CONTINUE C CALL LUELMN (UL,IUL,N,RES,APVT,DX) C DXNORM = ZERO C XNORM = ZERO C DO 35 I=1,N C X(I) = X(I) + DX(I) C DXNORM = DMAX1(DXNORM,DABS(DX(I))) C XNORM = DMAX1(XNORM,DABS(X(I))) 35 CONTINUE C IF (ITER .NE. 1) GO TO 40 C IDOT = 50 C IF (DXNORM .NE. ZERO) IDOT = -DLOG10(DXNORM/XNORM) 40 IF (XNORM+DXNORM .EQ. XNORM) GO TO 9005 45 CONTINUE C C IER = 129 9000 CONTINUE C CALL UERTST(IER,6HLUREFN) 9005 RETURN END </pre>	<pre> 00029763 00029970 00029980 00029990 00030000 00030010 00030020 00030030 00030040 00030050 00030060 00030070 00030080 00030090 00030100 00030110 00030120 00030130 00030140 00030150 00030160 00030170 00030180 00030190 00030200 00030210 00030220 00030230 00030240 00030250 00030260 00030270 00030280 00030290 00030300 00030310 00030320 00030330 00030340 00030350 00030360 00030370 00030380 00030390 00030400 00030410 00030420 00030430 00030440 00030450 00030460 00030470 00030480 00030490 00030500 00030510 00030520 00030530 00030540 00030550 </pre>
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C      00031080
C      00031090
C      00031100
C      00031110
C      00031120
C      00031130
C      00031140
C      00031150

SUBROUTINE UERTST (IER,NAME)
C      SPECIFICATIONS FOR ARGUMENTS
C      INTEGER IER
C      INTEGER NAME(1)
C      SPECIFICATIONS FOR LOCAL VARIABLES
C      INTEGER I,IEQ,IEQDF,IOUNIT,LEVEL,LEVOLD,NAMEQ(4),
M      NAMSET(6),NAMUPK(6),NIN,NMT9
C      DATA NAMSET/1HU,1HE,1HR,1HS,1HE,1HT/
C      DATA NAMEQ/6*1H/
C      DATA LEVEL/4/,IEQDF/0/,IEQ/1H=/
C      UNPACK NAME INTO NAMUPK
C      FIRST EXECUTABLE STATEMENT
C      CALL USPDK (NAME,6,NAMUPK,NMT9)
C      GET OUTPUT UNIT NUMBER
C      CALL UGETIO(1,NIN,IOUNIT)
C      CHECK IER
C      IF (IER.GT.999) GO TO 25
C      IF (IER.LT.-32) GO TO 55
C      IF (IER.LE.128) GO TO 5
C      IF (LEVEL.LT.1) GO TO 30
C      PRINT TERMINAL MESSAGE
C      IF (IEQDF.EQ.1) WRITE(IOUNIT,35) IER,NAMEQ,IEQ,NAMUPK
C      IF (IEQDF.EQ.0) WRITE(IOUNIT,35) IER,NAMUPK
C      GO TO 30
5      IF (IER.LE.64) GO TO 10
C      IF (LEVEL.LT.2) GO TO 30
C      PRINT WARNING WITH FIX MESSAGE
C      IF (IEQDF.EQ.1) WRITE(IOUNIT,40) IER,NAMEQ,IEQ,NAMUPK
C      IF (IEQDF.EQ.0) WRITE(IOUNIT,40) IER,NAMUPK
C      GO TO 30
10      IF (IER.LE.32) GO TO 15
C      PRINT WARNING MESSAGE
C      IF (LEVEL.LT.3) GO TO 30
C      IF (IEQDF.EQ.1) WRITE(IOUNIT,45) IER,NAMEQ,IEQ,NAMUPK
C      IF (IEQDF.EQ.0) WRITE(IOUNIT,45) IER,NAMUPK
C      GO TO 30
15      CONTINUE
C      CHECK FOR UERSET CALL
C      DO 20 I=1,6
C      IF (NAMUPK(I).NE.NAMSET(I)) GO TO 25
20      CONTINUE
C      LEVOLD = LEVEL
C      LEVEL = IER
C      IER = LEVOLD
C      IF (LEVEL.LT.0) LEVEL = 4
C      IF (LEVEL.GT.4) LEVEL = 4
C      GO TO 30
25      CONTINUE
C      IF (LEVEL.LT.4) GO TO 30
C      PRINT NON-DEFINED MESSAGE
C      IF (IEQDF.EQ.1) WRITE(IOUNIT,50) IER,NAMEQ,IEQ,NAMUPK
C      IF (IEQDF.EQ.0) WRITE(IOUNIT,50) IER,NAMUPK
30      IEQDF = 0
C      RETURN
35      FORMAT(19H *** TERMINAL ERROR,10X,7H( IER = ,I3,
1      20H) FROM IMSL ROUTINE ,6A1,A1,6A1)
40      FORMAT(27H *** WARNING WITH FIX ERROR,2X,7H( IER = ,I3,

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	1	20H) FROM IMSL ROUTINE ,6A1,A1,6A1)	00031160
	45	FORMAT(18H *** WARNING ERROR,11X,7H(IER = ,IS,	00031170
	1	20H) FROM IMSL ROUTINE ,6A1,A1,6A1)	00031180
	50	FORMAT(20H *** UNDEFINED ERROR,9X,7H(IER = ,IS,	00031190
	1	20H) FROM IMSL ROUTINE ,6A1,A1,6A1)	00031200
			00031210
		SAVE P FOR P = R CASE	00031220
		P IS THE PAGE NAMUPK	00031230
		R IS THE ROUTINE NAMUPK	00031240
	55	IEQDF = 1	00031250
		DO 60 I=1,6	00031260
	60	NAMEQ(I) = NAMUPK(I)	00031270
	65	RETURN	00031280
		END	00031290
			00031300
			00031310
			00031320
		SUBROUTINE UGETID(IOPT,NIN,NOUT)	00031330
		SPECIFICATIONS FOR ARGUMENTS	00031340
	INTEGER	IOPT,NIN,NOUT	00031350
		SPECIFICATIONS FOR LOCAL VARIABLES	00031360
	INTEGER	NIND,NOUTD	00031370
	DATA	NIND/5/,NOUTD/6/	00031380
		FIRST EXECUTABLE STATEMENT	00031390
		IF (IOPT.EQ.3) GO TO 10	00031400
		IF (IOPT.EQ.2) GO TO 5	00031410
		IF (IOPT.NE.1) GO TO 9005	00031420
		NIN = NIND	00031430
		NOUT = NOUTD	00031440
		GO TO 9005	00031450
	5	NIND = NIN	00031460
		GO TO 9005	00031470
	10	NOUTD = NOUT	00031480
	9005	RETURN	00031490
		END	00031500
			00031510
			00031520
		SUBROUTINE VXADD(A,ACC)	00031530
			00031540
		SPECIFICATIONS FOR ARGUMENTS	00031550
	DOUBLE PRECISION	A,ACC(2)	00031560
		SPECIFICATIONS FOR LOCAL VARIABLES	00031570
	DOUBLE PRECISION	X,Y,Z,ZZ	00031580
		FIRST EXECUTABLE STATEMENT	00031590
		X = ACC(1)	00031600
		Y = A	00031610
		IF (DABS(ACC(1)).GE.DABS(A)) GO TO 1	00031620
		X = A	00031630
		Y = ACC(1)	00031640
		COMPUTE Z+ZZ = ACC(1)+A EXACTLY	00031650
	1	Z = X+Y	00031660
		ZZ = (X-Z)+Y	00031670
		COMPUTE ZZ+ACC(2) USING DOUBLE	00031680
		PRECISION ARITHMETIC	00031690
		ZZ = ZZ+ACC(2)	00031700
		COMPUTE ACC(1)+ACC(2) = Z+ZZ EXACTLY	00031710
		ACC(1) = Z+ZZ	00031720
		ACC(2) = (Z-ACC(1))+ZZ	00031730
		RETURN	00031740
			00031750

```

C      END
C
C      SUBROUTINE VXMUL (A,B,ACC)
C      DOUBLE PRECISION A,B,ACC(2)
C      DOUBLE PRECISION X,HA,TA,HB,TB
C      INTEGER IX(2),I
C      LOGICAL M1
C      EQUIVALENCE (X,LX(1),IX(1)),(I,LI(1))
C      DATA I/O/
C
C      SPECIFICATIONS FOR ARGUMENTS
C      SPECIFICATIONS FOR LOCAL VARIABLES
C
C      SPLIT A = HA+TA
C      B = HB+TB
C      FIRST EXECUTABLE STATEMENT
C
C      X = A
C      LI(4) = LX(5)
C      IX(2) = 0
C      I = (I/16)*16
C      LX(5) = LI(4)
C      HA=X
C      TA=A-HA
C      X = B
C      LI(4) = LX(5)
C      IX(2) = 0
C      I = (I/16)*16
C      LX(5) = LI(4)
C      HB = X
C      TB = B-HB
C
C      COMPUTE HAHB,HANTB,TANHb, AND TANTB
C      AND CALL VXADD TO ACCUMULATE THE
C      SUM
C
C      X = TANTB
C      CALL VXADD(X,ACC)
C      X = HANTB
C      CALL VXADD(X,ACC)
C      X = TANHb
C      CALL VXADD(X,ACC)
C      X = HAHB
C      CALL VXADD(X,ACC)
C      RETURN
C      END
C
C      SUBROUTINE VXSTO (ACC,D)
C      DOUBLE PRECISION ACC(2),D
C      D = ACC(1)+ACC(2)
C      RETURN
C      END
C
C      SUBROUTINE ZRPOLY (A,NDEO,Z,IER)
C      INTEGER NDEO,IER
C      DOUBLE PRECISION A(1),Z(1)

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C	INTEGER	N, NN, J, JJ, I, NM1, ICNT, N2, L, N2, NPI	00032360
	REAL	ETA, RMRE, RINFP, REPSR, RADIX, RLO, XX, YY, SINR,	00032370
	1 REAL	COSR, RMAX, RMIN, X, SC, XM, FT, DX, DF, BND, XXX, ARE	00032380
	DOUBLE PRECISION	PT(101)	00032390
	1 DOUBLE PRECISION	TEMP(101), P(101), QP(101), RK(101), GK(101),	00032400
		SVK(101)	00032410
		SR, SI, U, V, RA, RB, C, D, A1, A2, A3,	00032420
		A6, A7, E, F, G, H, SZR, SZI, RLZR, RLZI,	00032430
		T, AA, BU, CC, FACTOR, REPSR1, ZERO, ONE, FN	00032440
	2 LOGICAL	ZEROK	00032450
	COMMON /ZRPQLJ/	P, QP, RK, QK, SVK, SR, SI, U, V, RA, RB, C, D, A1, A2, A3, A6,	00032460
	1	A7, E, F, G, H, SZR, SZI, RLZR, RLZI, ETA, ARE, RMRE, N, NN	00032470
		THE FOLLOWING STATEMENTS SET MACHINE	00032480
		CONSTANTS USED IN VARIOUS PARTS OF	00032490
		THE PROGRAM. THE MEANING OF THE	00032500
		FOUR CONSTANTS ARE - REPSR1 THE	00032510
		MAXIMUM RELATIVE REPRESENTATION	00032520
		ERROR WHICH CAN BE DESCRIBED AS	00032530
		THE SMALLEST POSITIVE FLOATING	00032540
		POINT NUMBER SUCH THAT 1.+REPSR1 IS	00032550
		GREATER THAN 1	00032560
		RINFP THE LARGEST FLOATING-POINT	00032570
		NUMBER	00032580
		REPSR THE SMALLEST POSITIVE	00032590
		FLOATING-POINT NUMBER IF THE	00032600
		EXPONENT RANGE DIFFERS IN SINGLE	00032610
		AND DOUBLE PRECISION THEN REPSR	00032620
		AND RINFP SHOULD INDICATE THE	00032630
		SMALLER RANGE	00032640
		RADIX THE BASE OF THE FLOATING-POINT	00032650
		NUMBER SYSTEM USED	00032660
	DATA	RINFP/27FFFFFFF/	00032670
	DATA	REPSR/200100000/	00032680
	DATA	RADIX/16.0/	00032690
	DATA	REPSR1/234100000000000000/	00032700
	DATA	ZERO/0.000/ ONE/1.000/	00032710
		ZRPOLY USES SINGLE PRECISION	00032720
		CALCULATIONS FOR SCALING, BOUNDS	00032730
		AND ERROR CALCULATIONS.	00032740
		FIRST EXECUTABLE STATEMENT	00032750
	IER = 0		00032760
	IF (NDEG .GT. 100 .OR. NDEG .LT. 1) GO TO 165		00032770
	ETA = REPSR1		00032780
	ARE = ETA		00032790
	RMRE = ETA		00032800
	RLO = REPSR/ETA		00032810
		INITIALIZATION OF CONSTANTS FOR	00032820
		SHIFT ROTATION	00032830
	XX = .7071068		00032840
	YY = -XX		00032850
	SINR = .9975641		00032860
	COSR = -.06975647		00032870
	N = NDEG		00032880
	NN = N+1		00032890
		ALGORITHM FAILS IF THE LEADING	00032900
		COEFFICIENT IS ZERO.	00032910
	IF (A(1).NE.ZERO) GO TO 5		00032920
	IER = 130		00032930
	GO TO 9000		00032940
			00032950

C		REMOVE THE ZEROS AT THE ORIGIN AND ANY	00032960
C	5 IF (A(NN).NE.ZERO) GO TO 10		00032970
	J = NDEG-N+1		00032980
	JJ = J+NDEG		00032990
	Z(J) = ZERO		00033000
	Z(JJ) = ZERO		00033010
	NN = NN-1		00033020
	N = N-1		00033030
	IF (NN.EQ.1) GO TO 9005		00033040
	GO TO 5		00033050
C		MAKE A COPY OF THE COEFFICIENTS	00033060
	10 DO 15 I=1,NN		00033070
	P(I) = A(I)		00033080
	15 CONTINUE		00033090
C		START THE ALGORITHM FOR ONE ZERO	00033100
	20 IF (N.GT.2) GO TO 30		00033110
	IF (N.LT.1) GO TO 9005		00033120
C		CALCULATE THE FINAL ZERO OR PAIR OF ZEROS	00033130
	IF (N.EQ.2) GO TO 25		00033140
	Z(NDEG) = -P(2)/P(1)		00033150
	Z(NDEG+NDEG) = ZERO		00033160
	GO TO 145		00033170
	25 CALL ZRPQ(I (P(1),P(2),P(3),Z(NDEG-1),Z(NDEG+NDEG-1),Z(NDEG),		00033180
	1 Z(NDEG+NDEG))		00033190
	GO TO 145		00033200
C		FIND LARGEST AND SMALLEST MODULI OF COEFFICIENTS.	00033210
	30 RMAX = 0.		00033220
	RMIN = RINFP		00033230
	DO 35 I=1,NN		00033240
	X = ABS(SNGL(P(I)))		00033250
	IF (X.GT.RMAX) RMAX = X		00033260
	IF (X.NE.0. AND X.LT.RMIN) RMIN = X		00033270
	35 CONTINUE		00033280
C		SCALE IF THERE ARE LARGE OR VERY SMALL COEFFICIENTS COMPUTES A SCALE FACTOR TO MULTIPLY THE COEFFICIENTS OF THE POLYNOMIAL. THE SCALING IS DONE TO AVOID OVERFLOW AND TO AVOID UNDETECTED UNDERFLOW INTERFERING WITH THE CONVERGENCE CRITERION. THE FACTOR IS A POWER OF THE BASE	00033290
	SC = RLO/RMIN		00033300
	IF (SC.GT.1.0) GO TO 40		00033310
	IF (RMAX.LT.10.) GO TO 55		00033320
	IF (SC.EQ.0.) SC = REPSM*RADIX*RADIX		00033330
	GO TO 45		00033340
	40 IF (RINFP/SC.LT.RMAX) GO TO 55		00033350
	45 L = ALOG(SC)/ALOG(RADIX)+.5		00033360
	IF (L.EQ.0) GO TO 55		00033370
	FACTOR = DBLE(RADIX)**L		00033380
	DO 50 I=1,NN		00033390
	50 P(I) = FACTOR*P(I)		00033400
C		COMPUTE LOWER BOUND ON MODULI OF ZEROS.	00033410
	55 DO 60 I=1,NN		00033420
	60 PT(I) = ABS(SNGL(P(I)))		00033430
			00033440
			00033450
			00033460
			00033470
			00033480
			00033490
			00033500
			00033510
			00033520
			00033530
			00033540
			00033550

C	PT(NN) = -PT(NN)	COMPUTE UPPER ESTIMATE OF BOUND	00033560
	X = EXP((ALOG(-PT(NN))-ALOG(PT(1)))/N)		00033570
	IF (PT(N).EQ.0.) GO TO 65		00033580
C		IF NEWTON STEP AT THE ORIGIN IS	00033590
C		BETTER, USE IT.	00033600
	XN = -PT(NN)/PT(N)		00033610
	IF (XM.LT.X) X = XM		00033620
C		CHOP THE INTERVAL (0,X) UNTIL FF.LE.	00033630
65	XM = XM.1		00033640
	FF = PT(1)		00033650
	DO 70 I=2,NN		00033660
70	FF = FF*XM+PT(I)		00033670
	IF (FF.LE.0.) GO TO 75		00033680
	X = XM		00033690
	GO TO 65		00033700
75	DX = X		00033710
C		DO NEWTON ITERATION UNTIL X	00033720
C		CONVERGES TO TWO DECIMAL PLACES	00033730
80	IF (ABS(DX/X).LE..005) GO TO 90		00033740
	FF = PT(1)		00033750
	DF = FF		00033760
	DO 85 I=2,N		00033770
	FF = FF*X+PT(I)		00033780
	DF = DF*X+FF		00033790
85	CONTINUE		00033800
	FF = FF*X+PT(NN)		00033810
	DX = FF/DF		00033820
	X = X-DX		00033830
	GO TO 80		00033840
90	BND = X		00033850
C		COMPUTE THE DERIVATIVE AS THE INITIAL	00033860
C		K POLYNOMIAL AND DO 5 STEPS WITH	00033870
C		NO SHIFT	00033880
	NM1 = N-1		00033890
	FN = ONE/N		00033900
	DO 95 I=2,N		00033910
95	RK(I) = (NN-I)*P(I)*FN		00033920
	RK(1) = P(1)		00033930
	AA = P(NN)		00033940
	BB = P(N)		00033950
	ZEROK = RK(N).EQ.ZERO		00033960
	DO 115 JJ=1,5		00033970
	CC = RK(N)		00033980
	IF (ZEROK) GO TO 105		00033990
C		USE SCALED FORM OF RECURRENCE IF	00034000
C		VALUE OF K AT 0 IS NONZERO	00034010
	T = -AA/CC		00034020
	DO 100 I=1,NM1		00034030
	J = NN-I		00034040
	RK(J) = T*RK(J-1)+P(J)		00034050
100	CONTINUE		00034060
	RK(1) = P(1)		00034070
	ZEROK = DABS(RK(N)).LE.DABS(BB)*ETAN10.		00034080
	GO TO 115		00034090
C		USE UNSCALED FORM OF RECURRENCE	00034100
105	DO 110 I=1,NM1		00034110
	J = NN-I		00034120
	RK(J) = RK(J-1)		00034130
110	CONTINUE		00034140
			00034150

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      RK(1) = ZERO
      ZEROK = RK(N).EQ.ZERO
115 CONTINUE
      DO 120 I=1,N
120 TEMP(I) = RK(I)
      DO 140 ICNT=1,20
      XXX = COSRXXX-SINRYYY
      YY = SINRXXX+COSRYYY
      XX = XXX
      SR = BNDXXX
      SI = BNDYYY
      U = -SR-SR
      V = BNDMBND
      CALL ZRPQLB (20*ICNT,NZ)
      IF (NZ.EQ.0) GO TO 130
      J = NDEG-N+1
      JJ = J+NDEG
      Z(J) = SZR
      Z(JJ) = SZI
      NN = NN-NZ
      N = NN-1
      DO 125 I=1,NN
125 P(I) = QP(I)
      IF (NZ.EQ.1) GO TO 20
      Z(J+1) = RLZR
      Z(JJ+1) = RLZI
      GO TO 20
      DO 135 I=1,N
135 RX(I) = TEMP(I)
140 CONTINUE
      IER = 131
      DO 145 I=1,NDEG
      NPI = NDEG+I
      P(I) = Z(NPI)
150 CONTINUE
      NZ = NDEG+NDEG
      J = NDEG
  
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DO 155 I=1,NDEO                                00034760
  Z(N2-1) = Z(J)                                00034770
  Z(N2) = P(J)                                  00034780
  N2 = N2-2                                     00034790
  J = J-1                                       00034800
155 CONTINUE                                    00034810
IF (IER.EQ. 0) GO TO 9005                        00034820
C SET UNFOUND ROOTS TO MACHINE INFINITY 00034830
  N2 = 2*(NDEO-NN)+3                            00034840
DO 160 I=1,N                                    00034850
  Z(N2) = RINFP                                  00034860
  Z(N2+1) = RINFP                                00034870
  N2 = N2+2                                       00034880
160 CONTINUE                                    00034890
GO TO 9000                                       00034900
165 IER = 129                                    00034910
9000 CONTINUE                                    00034920
CALL UERTST (IER,6H ZRPOLY)                     00034930
9005 RETURN                                      00034940
END                                              00034950
C                                              00034960
C                                              00034970
C                                              00034980
SUBROUTINE ZRPQLB (L2,NZ)                        00034990
C SPECIFICATIONS FOR ARGUMENTS                 00035000
  INTEGER L2,NZ                                00035010
C SPECIFICATIONS FOR LOCAL VARIABLES           00035020
  INTEGER N,NN,J,ITYPE,I,IFLAG                 00035030
  REAL ARE,BETAS,BETAV,ETA,OSS,OTS,OTV,OVV,RMRE,SS, 00035040
  TS,TSS,TV,TVV,VV                             00035050
  1 DOUBLE PRECISION P(101),QP(101),RK(101),QK(101),SVK(101) 00035060
  DOUBLE PRECISION SR,SI,U,V,RA,RB,C,D,A1,A2,A3, 00035070
  1 A6,A7,E,F,G,H,SZR,SZI,RLZR,RLZI,           00035080
  2 SVU,SVV,UI,VI,S,ZERO                        00035090
  LOGICAL VPASS,SPASS,VTRY,STRY                00035100
  COMMON /ZRPQLJ/ P,QP,RK,QK,SVK,SR,SI,U,V,RA,RB,C,D,A1,A2,A3,A6, 00035110
  1 A7,E,F,G,H,SZR,SZI,RLZR,RLZI,ETA,ARE,RMRE,N,NN 00035120
  DATA ZERO/0.000/                            00035130
C FIRST EXECUTABLE STATEMENT                  00035140
  NZ = 0                                         00035150
C COMPUTES UP TO L2 FIXED SHIFT                00035160
C K-POLYNOMIALS, TESTING FOR                   00035170
C CONVERGENCE IN THE LINEAR OR                 00035180
C QUADRATIC CASE. INITIATES ONE OF             00035190
C THE VARIABLE SHIFT ITERATIONS AND           00035200
C RETURNS WITH THE NUMBER OF ZEROS             00035210
C FOUND.                                         00035220
  L2 = LIMIT OF FIXED SHIFT STEPS              00035230
  NZ = NUMBER OF ZEROS FOUND                   00035240
C EVALUATE POLYNOMIAL BY SYNTHETIC              00035250
C DIVISION                                     00035260
  CALL ZRPQLH (NN,U,V,P,QP,RA,RB)              00035270
  CALL ZRPQL (ITYPE)                           00035280
DO 40 J=1,L2                                    00035290
  CALCULATE NEXT K POLYNOMIAL AND              00035300
  ESTIMATE V                                    00035310
  00035320
  00035330
  00035340
  00035350

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	CALL ZRPQLF (ITYPE)	00035360
	CALL ZRPQLE (ITYPE)	00035370
	CALL ZRPQLG (ITYPE,UI,VI)	00035380
	VV = VI	00035390
C	ESTIMATE S	00035400
	SS = 0.	00035410
	IF (RK(N).NE.ZERO) SS = -P(NN)/RK(N)	00035420
	TV = 1.	00035430
	TS = 1.	00035440
	IF (J.EQ.1.OR.ITYPE.EQ.3) GO TO 35	00035450
C	COMPUTE RELATIVE MEASURES OF	00035460
C	CONVERGENCE OF S AND V SEQUENCES	00035470
	IF (VV.NE.0.) TV = ABS((VV-OVV)/VV)	00035480
	IF (SS.NE.0.) TS = ABS((SS-OSS)/SS)	00035490
C	IF DECREASING, MULTIPLY TWO MOST	00035500
C	RECENT CONVERGENCE MEASURES	00035510
	TVV = 1.	00035520
	IF (TV.LT.OTV) TVV = TV*OTV	00035530
	TSS = 1.	00035540
	IF (TS.LT.OTS) TSS = TS*OTS	00035550
C	COMPARE WITH CONVERGENCE CRITERIA	00035560
	VPASS = TVV.LT.BETAV	00035570
	SPASS = TSS.LT.BETAS	00035580
	IF (.NOT.(SPASS.OR.VPASS)) GO TO 35	00035590
C	AT LEAST ONE SEQUENCE HAS PASSED THE	00035600
C	CONVERGENCE TEST. STORE VARIABLES	00035610
C	BEFORE ITERATING	00035620
	SVU = U	00035630
	SVV = V	00035640
	DO 5 I=1,N	00035650
5	SVK(I) = RK(I)	00035660
	S = SS	00035670
C	CHOOSE ITERATION ACCORDING TO THE	00035680
C	FASTEST CONVERGING SEQUENCE	00035690
	VTRY = .FALSE.	00035700
	STRY = .FALSE.	00035710
	IF (SPASS AND ((.NOT.VPASS).OR.TSS.LT.TVV)) GO TO 20	00035720
10	CALL ZRPQLC (UI,VI,NZ)	00035730
	IF (NZ.GT.0) RETURN	00035740
C	QUADRATIC ITERATION HAS FAILED. FLAG	00035750
C	THAT IT HAS BEEN TRIED AND	00035760
C	DECREASE THE CONVERGENCE	00035770
C	CRITERION.	00035780
	VTRY = TRUE.	00035790
	BETAU = BETAU*.25	00035800
C	TRY LINEAR ITERATION IF IT HAS NOT	00035810
C	BEEN TRIED AND THE S SEQUENCE IS	00035820
C	CONVERGING	00035830
	IF (STRY.OR.(.NOT.SPASS)) GO TO 25	00035840
	DO 15 I=1,N	00035850
15	RK(I) = SVK(I)	00035860
20	CALL ZRPQLD (S,NZ,IFLAG)	00035870
	IF (NZ.GT.0) RETURN	00035880
C	LINEAR ITERATION HAS FAILED. FLAG	00035890
C	THAT IT HAS BEEN TRIED AND	00035900
C	DECREASE THE CONVERGENCE CRITERION	00035910
	STRY = .TRUE.	00035920
	BETAS = BETAS*.25	00035930
	IF (IFLAG.EQ.0) GO TO 25	00035940
C	IF LINEAR ITERATION SIGNALS AN	00035950

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C      IF ( DABS(DABS(SZR)-DABS(RLZR)).GT.PT01*DABS(RLZR)) RETURN 00036560
C      EVALUATE POLYNOMIAL BY QUADRATIC 00036570
C      SYNTHETIC DIVISION 00036580
      CALL ZRPQLH (NN,U,V,P,QP,RA,RB) 00036590
      RMP = DABS(RA-SZR*RB)+DABS(SZI*RB) 00036600
C      COMPUTE A RIGOROUS BOUND ON THE 00036610
C      ROUNDING ERROR IN EVALUTING P 00036620
      ZM = SQRT(ABS(SNGL(V))) 00036630
      EL = 2.*ABS(SNGL(QP(1))) 00036640
      T = -SZR*RB 00036650
      DO 10 I=2,N 00036660
10    EE = EE*ZM+ABS(SNGL(QP(I))) 00036670
      EE = EE*ZM+ABS(SNGL(RA)+T) 00036680
      EE = (5.*RMPRE+4.*ARE)*EE-(5.*RMPRE+2.*ARE)*(ABS(SNGL(RA)+T)+ 00036690
1    ABS(SNGL(RB))*ZM)+2.*ARE*ABS(T) 00036700
C      ITERATION HAS CONVERGED SUFFICIENTLY 00036710
C      IF THE POLYNOMIAL VALUE IS LESS 00036720
C      THAN 20 TIMES THIS BOUND 00036730
      IF (RMP.GT.20.*EE) GO TO 15 00036740
      NZ = 2 00036750
      RETURN 00036760
15    J = J+1 00036770
C      STOP ITERATION AFTER 20 STEPS 00036780
      IF (J.GT.20) RETURN 00036790
      IF (J.LT.2) GO TO 25 00036800
      IF (RELSTP.GT..01.OR.RMP.LT.OMP.OR.TRIED) GO TO 25 00036810
C      A CLUSTER APPEARS TO BE STALLING THE 00036820
C      CONVERGENCE. FIVE FIXED SHIFT 00036830
C      STEPS ARE TAKEN WITH A U,V CLOSE 00036840
C      TO THE CLUSTER 00036850
      IF (RELSTP.LT.ETA) RELSTP = ETA 00036860
      RELSTP = SQRT(RELSTP) 00036870
      U = U-UM*RELSTP 00036880
      V = V+VM*RELSTP 00036890
      CALL ZRPQLH (NN,U,V,P,QP,RA,RB) 00036900
      DO 20 I=1,5 00036910
        CALL ZRPQLE (ITYPE) 00036920
        CALL ZRPQLF (ITYPE) 00036930
20    CONTINUE 00036940
      TRIED = .TRUE. 00036950
      J = 0 00036960
25    OMP = RMP 00036970
C      CALCULATE NEXT K POLYNOMIAL AND NEW 00036980
C      U AND V 00036990
      CALL ZRPQLE (ITYPE) 00037000
      CALL ZRPQLF (ITYPE) 00037010
      CALL ZRPQLE (ITYPE) 00037020
      CALL ZRPQLG (ITYPE,UI,VI) 00037030
C      IF VI IS ZERO THE ITERATION IS NOT 00037040
C      CONVERGING 00037050
      IF (VI.EQ.ZERO) RETURN 00037060
      RELSTP = DABS((VI-V)/VI) 00037070
      U = UI 00037080
      V = VI 00037090
      GO TO 5 00037100
      END 00037110
C      00037120
C      00037130
C      00037140
C      00037150

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C      SUBROUTINE ZRPQLD (SSS,NZ,IFLAG)                                00037160
C      INTEGER          NZ,IFLAG      SPECIFICATIONS FOR ARGUMENTS      00037170
C      DOUBLE PRECISION SSS           00037180
C                                     SPECIFICATIONS FOR LOCAL VARIABLES 00037190
C      INTEGER          N,NN,J,I      00037200
C      REAL             ARE,EE,ETA,OMP,PMR,RMS,RMRE                     00037210
C      DOUBLE PRECISION P(101),QP(101),RK(101),QK(101),SVK(101)         00037220
C      DOUBLE PRECISION SR,SI,U,V,RA,RB,C,D,A1,A2,A3,                   00037230
C      DOUBLE PRECISION A6,A7,E,F,G,H,SZR,SZI,RLZR,RLZI,                00037240
C      PV,RKV,T,S,ZERO,PT001                                             00037250
C      COMMON /ZRPQLJ/ P,QP,RK,QK,SVK,SR,SI,U,V,RA,RB,C,D,A1,A2,A3,A6, 00037260
C      A7,E,F,G,H,SZR,SZI,RLZR,RLZI,ETA,ARE,RMRE,N,NN                  00037270
C      DATA            ZERO/0.000/,PT001/0.00100/                      00037280
C                                     VARIABLE-SHIFT H POLYNOMIAL        00037290
C                                     ITERATION FOR A REAL ZERO SSS -     00037300
C                                     STARTING ITERATE                   00037310
C                                     NZ - NUMBER OF ZERO FOUND          00037320
C                                     IFLAG - FLAG TO INDICATE A PAIR OF   00037330
C                                     ZEROS NEAR REAL AXIS                00037340
C                                     FIRST EXECUTABLE STATEMENT          00037350
C                                     00037360
C      NZ = 0                                                            00037370
C      S = SSS                                                            00037380
C      IFLAG = 0                                                         00037390
C      J = 0                                                             00037400
C                                     MAIN LOOP                            00037410
C      5 PV = P(1)                                                       00037420
C                                     EVALUATE P AT S                     00037430
C      QP(1) = PV                                                         00037440
C      DO 10 I=2,NN                                                       00037450
C      PV = PV*S+P(I)                                                     00037460
C      QP(I) = PV                                                         00037470
C      10 CONTINUE                                                         00037480
C      RMP = DABS(PV)                                                      00037490
C                                     COMPUTE A RIGOROUS BOUND ON THE      00037500
C                                     ERROR IN EVALUATING P               00037510
C      RMS = DABS(S)                                                       00037520
C      EE = (RMRE/(ARE+RMRE))*ABS(SNGL(QP(1)))                            00037530
C      DO 15 I=2,NN                                                       00037540
C      15 EE = EE*RMS+ABS(SNGL(QP(I)))                                     00037550
C                                     ITERATION HAS CONVERGED SUFFICIENTLY 00037560
C                                     IF THE POLYNOMIAL VALUE IS LESS     00037570
C                                     THAN 20 TIMES THIS BOUND            00037580
C      IF (RMP.GT.20.*((ARE+RMRE)*EE-RMRE*RMP)) GO TO 20                00037590
C      NZ = 1                                                             00037600
C      SZR = S                                                            00037610
C      SZI = ZERO                                                         00037620
C      RETURN                                                             00037630
C      20 J = J+1                                                         00037640
C                                     STOP ITERATION AFTER 10 STEPS        00037650
C      IF (J.GT.10) RETURN                                                 00037660
C      IF (J.LT.2) GO TO 25                                                00037670
C      IF (DABS(T).GT.PT001*DABS(S-T).OR.RMP.LE.OMP) GO TO 25           00037680
C      A CLUSTER OF ZEROS NEAR THE REAL                                00037690
C      AXIS HAS BEEN ENCOUNTERED RETURN                                00037700
C      WITH IFLAG SET TO INITIATE A                                    00037710
C      QUADRATIC ITERATION                                              00037720
C      IFLAG = 1                                                         00037730
C      SSS = S                                                            00037740
C      RETURN                                                             00037750

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C		RETURN IF THE POLYNOMIAL VALUE HAS	00037760
C		INCREASED SIGNIFICANTLY	00037770
C	25 OMP = RMP	COMPUTE T, THE NEXT POLYNOMIAL, AND	00037780
C		THE NEW ITERATE	00037790
	RKV = RK(1)		00037800
	QK(1) = RKV		00037810
	DO 30 I=2,N		00037820
	RKV = RKVMS+RK(I)		00037830
	QK(I) = RKV		00037840
	50 CONTINUE		00037850
	IF (DABS(RKV).LE.DABS(RK(N))*10.*ETA) GO TO 40		00037860
C		USE THE SCALED FORM OF THE	00037870
C		RECURRENCE IF THE VALUE OF K AT S	00037880
C		IS NONZERO	00037890
	T = -PV/RKV		00037900
	RK(1) = QP(1)		00037910
	DO 35 I=2,N		00037920
	35 RK(I) = T*QK(I-1)+QP(I)		00037930
	GO TO 50		00037940
C		USE UNSCALED FORM	00037950
	40 RK(1) = ZERO		00037960
	DO 45 I=2,N		00037970
	45 RK(I) = QK(I-1)		00037980
	50 RKV = RK(1)		00037990
	DO 55 I=2,N		00038000
	55 RKV = RKVMS+RK(I)		00038010
	T = ZERO		00038020
	IF (DABS(RKV).GT.DABS(RK(N))*10.*ETA) T = -PV/RKV		00038030
	S = S+T		00038040
	GO TO 5		00038050
	END		00038060
C			00038070
C			00038080
C			00038090
C			00038100
C	IMSL ROUTINE NAME - ZRPQLE		00038110
C			00038120
C			00038130
C	COMPUTER - IBM/DOUBLE		00038140
C			00038150
C	LATEST REVISION - JANUARY 1, 1978		00038160
C			00038170
C	SUBROUTINE ZRPQLE (ITYPE)		00038180
C		SPECIFICATIONS FOR ARGUMENTS	00038190
C	INTEGER ITYPE		00038200
C		SPECIFICATIONS FOR LOCAL VARIABLES	00038210
	INTEGER N,NN		00038220
	REAL ARE,ETA,RMRE		00038230
	DOUBLE PRECISION P(101),QP(101),RK(101),QK(101),SVK(101)		00038240
	DOUBLE PRECISION SR,S1,U,V,RA,RB,C,D,A1,A2,A3,		00038250
	1 A6,A7,E,F,G,H,SZR,SZI,RLZR,RLZI		00038260
	1 COMMON /ZRPQLJ/		00038270
	1 P,QP,RK,QK,SVK,SR,S1,U,V,RA,RB,C,D,A1,A2,A3,A6,		00038280
	A7,E,F,G,H,SZR,SZI,RLZR,RLZI,ETA,ARE,RMRE,N,NN		00038290
C		THIS ROUTINE CALCULATES SCALAR	00038300
C		QUANTITIES USED TO COMPUTE THE	00038310
C		NEXT K POLYNOMIAL AND NEW	00038320
C		ESTIMATES OF THE QUADRATIC	00038330
C		COEFFICIENTS	00038340
C		ITYPE - INTEGER VARIABLE SET HERE	00038350

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    RK(1) = ZERO                                00038960
    RK(2) = -A7*QP(1)                          00038970
    DO 5 I=3,N                                00038980
  5 RK(I) = A3*QK(I-2)-A7*QP(I-1)              00038990
    RETURN                                     00039000
C                                     USE SCALED FORM OF THE RECURRENCE
10 A7 = A7/A1                                00039010
    A3 = A3/A1                                00039020
    RK(1) = QP(1)                             00039030
    RK(2) = QP(2)-A7*QP(1)                   00039040
    DO 15 I=3,N                               00039050
  15 RK(I) = A3*QK(I-2)-A7*QP(I-1)+QP(I)      00039060
    RETURN                                     00039070
C                                     USE UNSCALED FORM OF THE RECURRENCE
C                                     IF TYPE IS 3
20 RK(1) = ZERO                                00039080
    RK(2) = ZERO                              00039090
    DO 25 I=3,N                               00039100
  25 RK(I) = QK(I-2)                          00039110
    RETURN                                     00039120
    END                                         00039130
                                         00039140
                                         00039150
                                         00039160
                                         00039170
                                         00039180
                                         00039190
IMSL ROUTINE NAME - ZRPQLO                    00039200
                                         00039210
-----
COMPUTER - IBM/DOUBLE                        00039220
                                         00039230
LATEST REVISION - JANUARY 1, 1978           00039240
                                         00039250
SUBROUTINE ZRPQLO (ITYPE,UU,VV)             00039260
C                                     SPECIFICATIONS FOR ARGUMENTS
    INTEGER ITYPE                            00039270
    DOUBLE PRECISION UU,VV                  00039280
C                                     SPECIFICATIONS FOR LOCAL VARIABLES
    INTEGER N,NN                            00039290
    REAL ARE,ETA,RMRE                       00039300
    DOUBLE PRECISION P(101),QP(101),RK(101),QK(101),SVK(101) 00039310
    DOUBLE PRECISION SP,SI,U,V,RA,RB,C,D,A1,A2,A3, 00039320
    1 A6,A7,E,F,G,H,SZR,SZI,RLZR,RLZI,      00039330
    2 A4,A5,B1,B2,C1,C2,C3,C4,TEMP,ZERO     00039340
    COMMON /ZRPQLJ/ P,QP,RK,QK,SVK,SR,SI,U,V,RA,RB,C,D,A1,A2,A3,A6, 00039350
    1 A7,E,F,G,H,SZR,SZI,RLZR,RLZI,ETA,ARE,RMRE,N,NN 00039360
    DATA ZERO/0.0D0/                      00039370
                                         00039380
                                         00039390
                                         00039400
                                         00039410
                                         00039420
                                         00039430
                                         00039440
                                         00039450
                                         00039460
                                         00039470
                                         00039480
                                         00039490
                                         00039500
                                         00039510
                                         00039520
                                         00039530
                                         00039540
                                         00039550
    IF (ITYPE.EQ.3) GO TO 15
    IF (ITYPE.EQ.2) GO TO 5
    A4 = RA+U*RB+H*F
    A5 = C+(U+V*F)*D
    GO TO 10
  5 A4 = (RA+G)*F+H
    A5 = (F+U)*C+V*D
C                                     EVALUATE NEW QUADRATIC COEFFICIENTS.

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C	10	B1 = -RK(N)/P(NN)	00039560
		B2 = -(RK(N-1)+B1*P(N))/P(NN)	00039570
		C1 = VMB2*A1	00039580
		C2 = B1*A7	00039590
		C3 = B1*B1*A3	00039600
		C4 = C1-C2-C3	00039610
		TEMP = A5+B1*A4-C4	00039620
		IF (TEMP.EQ.ZERO) GO TO 15	00039630
		UU = U-(U*(C3+C2)+V*(B1*A1+B2*A7))/TEMP	00039640
		VV = V*(1+C4/TEMP)	00039650
		RETURN	00039660
C		IF TYPE=3 THE QUADRATIC IS ZEROED	00039670
	15	UU = ZERO	00039680
		VV = ZERO	00039690
		RETURN	00039700
		END	00039710
C			00039720
C			00039730
C			00039740
C		SUBROUTINE ZRPOLH (NN,U,V,P,Q,RA,RB)	00039750
		SPECIFICATIONS FOR ARGUMENTS	00039760
C		INTEGER NN	00039770
		DOUBLE PRECISION P(NN),Q(NN),U,V,RA,RB	00039780
C		SPECIFICATIONS FOR LOCAL VARIABLES	00039790
		INTEGER I	00039800
		DOUBLE PRECISION C	00039810
C			00039820
C		DIVIDES P BY THE QUADRATIC 1,U,V	00039830
C		PLACING THE QUOTIENT IN Q AND THE	00039840
C		REMAINDER IN A,B	00039850
C		FIRST EXECUTABLE STATEMENT	00039860
		RB = P(1)	00039870
		Q(1) = RB	00039880
		RA = P(2)-U*RB	00039890
		Q(2) = RA	00039900
		DO 5 I=3,NN	00039910
		C = P(I)-U*RA-V*RB	00039920
		Q(I) = C	00039930
		RB = RA	00039940
		RA = C	00039950
	5	CONTINUE	00039960
		RETURN	00039970
		END	00039980
C			00039990
C			00040000
C			00040010
C		IMSL ROUTINE NAME - ZRPQLI	00040020
C			00040030
C			00040040
C			00040050
C		COMPUTER - IBM/DOUBLE	00040060
C			00040070
C		LATEST REVISION - JANUARY 1, 1978	00040080
C			00040090
C		SUBROUTINE ZRPQLI (RA,B1,C,SR,SI,RLR,RLI)	00040100
		SPECIFICATIONS FOR ARGUMENTS	00040110
		DOUBLE PRECISION RA,B1,C,SR,SI,RLR,RLI	00040120
C		SPECIFICATIONS FOR LOCAL VARIABLES	00040130
		DOUBLE PRECISION RE,D,E,ZERO,ONE,TWO	00040140
		DATA ZERO,ONE,TWO/0.000,1.000,2.000/	00040150

U

C	IER = 0	FIRST EXECUTABLE STATEMENT	00040760
	N1 = N+1		00040770
	N2 = N+2		00040780
	IF (IJOB .EQ. 2) GO TO 15		00040790
C	DO 10 I = 1,N	SAVE MATRIX A	00040800
	DO 5 J = 1,N		00040810
	WA(I,J) = A(I,J)		00040820
	CONTINUE		00040830
5	CONTINUE		00040840
10	CONTINUE		00040850
C	CALL LEQ1C(WA,N,N,B,M,1B,1,WK,IER)	FACTOR MATRIX A	00040860
	IF (IER .NE. 0) GO TO 9000		00040870
	IF (IJOB .EQ. 1) GO TO 9005		00040880
C	DO 65 J = 1,M	SAVE THE RIGHT HAND SIDES	00040890
	DO 20 I = 1,N		00040900
	WA(I,N1) = B(I,J)		00040910
20	CONTINUE		00040920
C	CALL LEQ1C(WA,N,N,WA(1,N1),1,N,2,WK,IER)	OBTAIN A SOLUTION	00040930
C	XNORM = ZERO	COMPUTE THE NORM OF THE SOLUTION	00040940
	DO 25 I = 1,N		00040950
	TEMPA = WA(I,N1)		00040960
	XNORM = DMAX1(XNORM,DABS(AR),DABS(AI))		00040970
25	CONTINUE		00040980
	IF (XNORM .EQ. ZERO) GO TO 65		00040990
C	DO 50 ITER = 1,ITMAX	COMPUTE RESIDUALS	00041000
	DO 40 I = 1,N		00041010
	TEMPB = B(I,J)		00041020
	ACC(1) = 0.000		00041030
	ACC(2) = 0.000		00041040
	CALL VXADD(BR,ACC)		00041050
	DO 30 JJ = 1,N		00041060
	TEMPA = A(I,JJ)		00041070
	TEMPB = WA(JJ,N1)		00041080
	CALL VXMUL(-AR,PR,ACC)		00041090
	CALL VXMUL(AI,BI,ACC)		00041100
30	CONTINUE		00041110
	CALL VXSTO(ACC,CR)		00041120
	TEMPB = B(I,J)		00041130
	ACC(1) = 0.000		00041140
	ACC(2) = 0.000		00041150
	CALL VXADD(BI,ACC)		00041160
	DO 35 JJ = 1,N		00041170
	TEMPA = A(I,JJ)		00041180
	TEMPB = WA(JJ,N1)		00041190
	CALL VXMUL(-AR,BI,ACC)		00041200
	CALL VXMUL(-BR,AI,ACC)		00041210
35	CONTINUE		00041220
	CALL VXSTO(ACC,C1)		00041230
	WA(I,N2) = TEMPC		00041240
40	CONTINUE		00041250
	CALL LEQ1C(WA,N,N,WA(1,N2),1,N,2,WK,IER)		00041260
	DXNORM = ZERO		00041270
	DO 45 I = 1,N	UPDATE THE SOLUTION	00041280
			00041290
			00041300
			00041310
			00041320
			00041330
			00041340
			00041350

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      WA(I,N1) = WA(I,N1)+WA(I,N2)
      TEMPA = WA(I,N2)
      DXNORM = DMAX1(DXNORM,DABS(AR),DABS(AI))
45  CONTINUE
      IF (XNORM+DXNORM .EQ. XNORM) GO TO 55
50  CONTINUE
      IER = 130
C   STORE THE SOLUTION
55  DO 60 JK = 1,N
      B(JK,J) = WA(JK,N1)
60  CONTINUE
      IF (IER .NE. 0) GO TO 9000
65  CONTINUE
      GO TO 9005
9000 CONTINUE
      CALL UERTST(IER,AHLEQ2C)
9005 RETURN
      END
C
C
C   SUBROUTINE LEQ1C (A,N,IA,B,M,IB,IJOB,WA,IER)
C   SPECIFICATIONS FOR ARGUMENTS
      INTEGER      N,IA,M,IB,IJOB,IER
      COMPLEX*16   A(IA,N),B(1B,M)
      DOUBLE PRECISION WA(N)
C   SPECIFICATIONS FOR LOCAL VARIABLES
      DOUBLE PRECISION P,Q,ZERO,ONE,T(2),RN,BIG
      COMPLEX*16      SUM,TEMP
      INTEGER          I,J,JM1,IM1,K,IMAX,JP1,IN,N1
      EQUIVALENCE     (SUM,T(1))
      DATA            ZERO/0.0D0/,ONE/1.0D0/
C   INITIALIZATION
C   FIRST EXECUTABLE STATEMENT
      IER = 0
      IF (IJOB .EQ. 2) GO TO 75
      RN = N
C   FIND EQUILIBRATION FACTORS
      DO 10 I=1,N
        BIG = ZERO
        DO 5 J=1,N
          TEMP = A(I,J)
          P = CDABS(TEMP)
          IF (P .GT. BIG) BIG = P
5    CONTINUE
          IF (BIG .EQ. ZERO) GO TO 105
          WA(I) = ONE/BIG
10  CONTINUE
C   L-U DECOMPOSITION
      DO 70 J = 1,N
        JM1 = J-1
        IF (JM1 .LT. 1) GO TO 25
C   COMPUTE U(I,J), I=1,...,J-1
      DO 20 I=1,JM1
        SUM = A(I,J)
        IM1 = I-1
        IF (IM1 .LT. 1) GO TO 20
        DO 15 K=1,IM1
          SUM = SUM-A(I,K)*A(K,J)
15  CONTINUE

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00041360
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 00041950

	A(I,J) = SUM	00041960
20	CONTINUE	00041970
25	P = ZERO	00041980
C	COMPUTE U(J,J) AND L(I,J), I=J+1,....	00041990
	DO 45 I=J,N	00042000
	SUM = A(I,J)	00042010
	IF (JMI .LT. 1) GO TO 40	00042020
	DO 35 K=1,JMI	00042030
	SUM = SUM-A(I,K)*A(K,J)	00042040
35	CONTINUE	00042050
	A(I,J) = SUM	00042060
40	Q = WA(I)*CDABS(SUM)	00042070
	IF (P .GE. Q) GO TO 45	00042080
	P = Q	00042090
	IMAX = I	00042100
45	CONTINUE	00042110
C	TEST FOR ALGORITHMIC SINGULARITY	00042120
	Q = RN+P	00042130
	IF (Q .EQ. RN) GO TO 105	00042140
	IF (J .EQ. IMAX) GO TO 60	00042150
C	INTERCHANGE ROWS J AND IMAX	00042160
	DO 50 K=1,N	00042170
	TEMP = A(IMAX,K)	00042180
	A(IMAX,K) = A(J,K)	00042190
	A(J,K) = TEMP	00042200
50	CONTINUE	00042210
	WA(IMAX) = WA(J)	00042220
60	WA(J) = IMAX	00042230
	JPI = J+1	00042240
	IF (JPI .GT. N) GO TO 70	00042250
C	DIVIDE BY PIVOT ELEMENT U(J,J)	00042260
	TEMP = A(J,J)	00042270
	DO 65 I = JPI,N	00042280
	A(I,J) = A(I,J)/TEMP	00042290
65	CONTINUE	00042300
70	CONTINUE	00042310
75	IF (IJOB .EQ. 1) GO TO 9005	00042320
	DO 103 K = 1,M	00042330
C	SOLVE UX = Y FOR X	00042340
	IW = 0	00042350
	DO 90 I = 1,N	00042360
	IMAX = WA(I)	00042370
	SUM = B(IMAX,K)	00042380
	B(IMAX,K) = B(I,K)	00042390
	IF (IW .EQ. 0) GO TO 85	00042400
	IM1 = I-1	00042410
	DO 80 J = IW,IM1	00042420
	SUM = SUM-A(I,J)*B(J,K)	00042430
80	CONTINUE	00042440
	GO TO 88	00042450
85	IF (T(1) .NE. ZERO .OR. T(2) .NE. ZERO) IW = I	00042460
88	B(I,K) = SUM	00042470
90	CONTINUE	00042480
C	SOLVE LY = B FOR Y	00042490
	N1 = N+1	00042500
	DO 100 IW = 1,N	00042510
	I = N1-IW	00042520
	JPI = I+1	00042530
	SUM = B(I,K)	00042540
	IF (JPI .GT. N) GO TO 98	00042550

```
          DO 95 J = JP1,N
            SUM = SUM-A(I,J)*B(J,K)
95        CONTINUE
98        B(I,K) = SUM/A(I,I)
100       CONTINUE
103      CONTINUE
        GO TO 9005
C
105 IER = 129
9000 CONTINUE
C
        CALL UERTST(IER,6HLEQTIC)
9005 RETURN
        END
```

ALGORITHMIC SINGULARITY

PRINT ERROR

00042560
00042570
00042580
00042590
00042600
00042610
00042620
00042630
00042640
00042650
00042660
00042670
00042680
00042690

APPENDIX B
SAMCJ Program Listing

התאחדות העובדים

CCCCC

```

COMMON/RT/R
COMMON/MFS/F5CD
COMMON/NTP/NELTYP
COMMON/NPT/NOPT2,NOPT6,NOPT7,NOPT8
COMMON/MD/E1,E2,G12,V12,V21
COMMON/LYP/NPLY,NUMPLY,ANG,IPLY
COMMON/ELP/AX,8X,NOUT,NSTS
COMMON/FCC/ELWDTH,ELTHK,ELLOAD
COMMON/NCST/NCASE,NTYPE
COMMON/DISP/ANR2
COMMON/PBB/PLYK,BARK,BARU
COMMON/ELS/ELSTFF,ELSTSS
COMMON/CMT2/XOUT,YOUT
COMMON/SER/NT,NB
DATA Y,Y,Y
DATA CMC,'C'

      READ IN REQUIRED INPUT DATA

      WRITE(6,876)
876  FORMAT(///,10X,' PROGRAM SAMCJ',///,
  * ' PROGRAM SAMCJ PREDICTS THE FAILURE LOAD, FAILURE ',/,
  * ' LOCATION, AND FAILURE MODE IN MULTIPLY-FASTENED, ',/,
  * ' SINGLE OR DOUBLE LAP COMPOSITE SHEAR JOINTS. ',/,
  * ' THE ANALYSIS ASSUMES THAT INPUT PARAMETERS ARE ',/,
  * ' SPECIFIED IN ENGLISH UNITS - LENGTH IN INCHES, ',/,
  * ' MODULI AND STRENGTHS IN PSI. ',/)
      WRITE(6,900)
900  FORMAT(' ENTER: ',/,
  * ' 1 FOR SLS (SINGLE LAP SHEAR)',/,
  * ' 2 FOR DLS (DOUBLE LAP SHEAR)',/)
      READ(5,*) NSDLS
      WRITE(6,911)
911  FORMAT(' ENTER: ',/,
  * ' 1 FOR STATIC TENSION ',/,
  * ' 2 FOR STATIC COMPRESSION',/)
      READ(5,*) LTNCM
106  FORMAT(A1)
380  CONTINUE
      DO 300 K=1,2
      IF(K.EQ.1) WRITE(6,912)
      IF(K.EQ.2) WRITE(6,913)
912  FORMAT(' IS THE TOP PLATE A COMPOSITE OR A METAL?')
913  FORMAT(' IS THE BOTTOM PLATE A COMPOSITE OR A METAL?')
      WRITE(6,914)
914  FORMAT(' ENTER C OR M IN THE FIRST FIELD')
      READ(5,106) CM(K)
      WRITE(6,203)
203  FORMAT(' INPUT MATERIAL DESCRIPTION OF THIS PLATE ',/,
  * ' EX: ASA/3501-6')
      READ(5,204) (MTL(K,I),I=1,15)
204  FORMAT(15A4)
300  CONTINUE
      IF(CM(1).NE.CMC.OR.CM(2).NE.CMC) WRITE(6,754)
754  FORMAT(//,' NOTE: FOR COMPUTATIONAL PURPOSES A ',/,
  * ' METALLIC PLATE IS MODELED AS A 30 PLY ',/,
  * ' LAMINATE OF 0 DEGREE PLYS WITH ISOTROPIC ',/,
  * ' MATERJ' ' PROPERTIES',/)

```

```

DO 306 K=1,2
IF(K.EQ.1) WRITE(6,216)
IF(K.EQ.2) WRITE(6,555)
216 FORMAT(' INPUT THE ENGINEERING PROPERTIES OF THE TOP PLATE')
555 FORMAT(' INPUT THE ENGINEERING PROPERTIES OF THE BOTTOM PLATE')
IF(CM(K).EQ.CMC) GO TO 85
WRITE(6,95)
95 FORMAT(' INPUT YOUNGS MODULUS AND POISSONS RATIO')
READ(5,M) E1(K),V12(K)
E2(K)=E1(K)
G12(K)=E1(K)/(2.0D0*V12(K))
V21(K)=V12(K)*E2(K)/E1(K)
GO TO 306
85 CONTINUE
WRITE(6,217)
217 FORMAT(' INPUT YOUNGS MODULI, E1 AND E2')
READ(5,M) E1(K),E2(K)
WRITE(6,213)
218 FORMAT(' INPUT THE SHEAR MODULUS AND MAJOR POISSONS RATIO')
READ(5,M) G12(K),V12(K)
V21(K)=V12(K)*E2(K)/E1(K)
306 CONTINUE
307 CONTINUE
290 CONTINUE
DO 303 K=1,2
IF(CM(K).EQ.CMC) GO TO 45
NUMPLY(K)=1
GO TO 303
45 CONTINUE
IF(K.EQ.1) WRITE(6,207)
IF(K.EQ.2) WRITE(6,702)
207 FORMAT(' INPUT TOTAL NUMBER OF DISTINCT PLY ',/,
* ' ORIENTATIONS IN THE TOP PLATE')
702 FORMAT(' INPUT TOTAL NUMBER OF DISTINCT PLY ',/,
* ' ORIENTATIONS IN THE BOTTOM PLATE')
READ(5,M) NUMPLY(K)
303 CONTINUE
DO 209 K=1,2
IF(CM(K).EQ.CMC) GO TO 55
ANG(1,K)=0.
GO TO 209
55 CONTINUE
N=NUMPLY(K)
DO 209 L=1,N
WRITE(6,206) L
206 FORMAT(' INPUT ORIENTATION OF PLY TYPE NO',I5)
READ(5,M) ANG(L,K)
209 CONTINUE
WRITE(6,1823)
1823 FORMAT(/, ' THICKNESS VARIATIONS MAY BE APPROXIMATED',/,
* ' BY ASSIGNING DIFFERENT LAYUPS TO ELEMENTS',/,
* ' IN A COMPOSITE PLATE OR BY SPECIFYING DIFFERENT',/,
* ' THICKNESSES TO ELEMENTS IN A METALLIC PLATE',/)
IF(NSDLS.EQ.2) WRITE(6,789)
789 FORMAT(/, ' NOTE: FOR THE DOUBLE LAP SHEAR CASE, FOR',/,
* ' THE BOTTOM PLATE, ENTER ONLY HALF FOR THE ',/,
* ' LAYUP FOR A COMPOSIT OR HALF THE THICKNESS ',/,
* ' FOR A METALLIC',/)
DO 811 I=1,2
IF(CM(I).EQ.CMC) GO TO 891
00001160
00001170
00001180
00001190
00001200
00001210
00001220
00001230
00001240
00001250
00001260
00001270
00001280
00001290
00001300
00001310
00001320
00001330
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00001380
00001390
00001400
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00001470
00001480
00001490
00001500
00001510
00001520
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00001590
00001600
00001610
00001620
00001630
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00001650
00001660
00001670
00001680
00001690
00001700
00001710
00001720
00001730
00001740
00001750

```

NELPLS(I,1)=30	00001760
DO 892 III=1,30	00001770
892 NELPT(I,1,III)=1	00001780
GO TO 811	00001790
891 CONTINUE	00001800
IF(I.EQ.1) WRITE(6,812)	00001810
IF(I.EQ.2) WRITE(6,813)	00001820
812 FORMAT(/, ' ENTER NUMBER OF DIFFERENT LAYUPS IN THE ',/,	00001830
* ' TOP PLATE')	00001840
813 FORMAT(/, ' ENTER NUMBER OF DIFFERENT LAYUPS IN THE ',/,	00001850
* ' BOTTOM PLATE')	00001860
READ(5,M) NL	00001870
DO 814 J=1,NL	00001880
WRITE(6,815) J	00001890
815 FORMAT(' ENTER NUMBER OF PLIES IN LAYUP NO ',I5)	00001900
READ(5,M) NELPLS(I,J)	00001910
WRITE(6,816)	00001920
816 FORMAT(' ENTER PLY THICKNESS FOR THIS LAYUP')	00001930
READ(5,M) PLYTHK(I,J)	00001940
NN=NELPLS(I,J)	00001950
WRITE(6,818)	00001960
818 FORMAT(' ENTER SEQUENCE OF PLY TYPES FROM TOP TO BOTTOM')	00001970
DO 817 K=1,NN	00001980
READ(5,M) NELPT(I,J,K)	00001990
817 CONTINUE	00002000
814 CONTINUE	00002010
811 CONTINUE	00002020
WRITE(6,855)	00002030
855 FORMAT(/, ' FASTENER DESCRIPTION:',/,)	00002040
WRITE(6,250)	00002050
250 FORMAT(' INPUT MATERIAL DESCRIPTION FOR FASTENER')	00002060
READ(5,251) (MTL(3,I),I=1,15)	00002070
251 FORMAT(15A4)	00002080
WRITE(6,252)	00002090
252 FORMAT(' INPUT YOUNG'S MODULUS AND POISSON'S RATIO FOR',/,	00002100
* ' THE FASTENER')	00002110
READ(5,M) FASE,FASV	00002120
WRITE(6,253)	00002130
253 FORMAT(' INPUT THE DIAMETER OF THE FASTENER')	00002140
READ(5,M) FASD	00002150
WRITE(6,888)	00002160
888 FORMAT(/, ' FASTENER TYPE',/,	00002170
* ' ENTER: 1 FOR PROTRUDING HEAD',/,	00002180
* ' 2 FOR COUNTERSUNK HEAD')	00002190
READ(5,M) NFTYP	00002200
R(1)=1.0D10	00002210
R(2)=1.0D10	00002220
IF(NFTYP.EQ.1) GO TO 360	00002230
WRITE(6,889)	00002240
889 FORMAT(/, ' ENTER PLATE WHICH CONTAINS THE COUNTERSUNK',/,	00002250
* ' HEAD (OPPOSITE PLATE ASSUMES THE NUT HEAD)',/,	00002260
* ' ENTER: 1 FOR TOP PLATE',/,	00002270
* ' 2 FOR BOTTOM PLATE')	00002280
READ(5,M) N	00002290
R(N)=0.0D0	00002300
360 CONTINUE	00002310
WRITE(6,477)	00002320
477 FORMAT(/, ' GRID LAYOUT:',/,)	00002330
C INPUT GRIDS, ELEMENT CONNECTIVITY AND PROPERTIES	00002340
	00002350

00002360
00002370
00002380
00002390
00002400
00002410
00002420
00002430
00002440
00002450
00002460
00002470
0000248
00002490
00002500
00002510
00002520
00002530
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00002580
00002590
00002600
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00002680
00002690
00002700
00002710
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00002750
00002760
00002775
00002780
00002790
00002800
00002810
00002820
00002830
00002840
00002850
00002860
00002870
00002880
00002890
00002900
00002910
00002920
00002930
00002940
00002950

IF(CM(1).EQ.CMC) GO TO 627	00002960
IF(NELTYP(I).NE.1) GO TO 721	00002970
WRITE(6,1721)	00002980
1721 FORMAT(' ENTER ELEMENT THICKNESS')	00002990
READ(5,X) ATH	00003000
721 IF(NELTYP(I).NE.2) GO TO 722	00003010
WRITE(6,723)	00003020
723 FORMAT(' ENTER ELEMENT THICKNESS')	00003030
READ(5,X) ATH	00003040
FSCD(I,1)=GCOORD(NELCHA(I,6),1)	00003050
FSCD(I,2)=GCOORD(NELCHA(I,6),2)	00003060
FSCD(I,3)=FASD/2.000	00003070
722 IF(NELTYP(I).NE.3) GO TO 724	00003080
WRITE(6,725)	00003090
725 FORMAT(' ENTER ELEMENT THICKNESS, X AND Y COORDINATES',/,	00003100
* OF OPEN HOLE AND HOLE RADIUS')	00003110
READ(5,X) ATH,(FSCD(I,J),J=1,3)	00003120
724 ELTHK(I)=ATH/30.000	00003130
PLYTHK(I,1)=ATH/30.000	00003140
LYPN(I)=1	00003150
GO TO 474	00003160
627 CONTINUE	00003170
IF(NELTYP(I).NE.1) GO TO 726	00003180
WRITE(6,727)	00003190
727 FORMAT(' ENTER ELEMENT LAYUP NO')	00003200
READ(5,X) LYPN(I)	00003210
726 IF(NELTYP(I).NE.2) GO TO 728	00003220
WRITE(6,729)	00003230
729 FORMAT(' ENTER ELEMENT LAYUP NO')	00003240
READ(5,X) LYPN(I)	00003250
FSCD(I,1)=GCOORD(NELCHA(I,6),1)	00003260
FSCD(I,2)=GCOORD(NELCHA(I,6),2)	00003270
FSCD(I,3)=FASD/2.000	00003280
728 IF(NELTYP(I).NE.3) GO TO 730	00003290
WRITE(6,731)	00003300
731 FORMAT(' ENTER ELEMENT LAYUP NUMBER, X AND Y ',/,	00003310
* COORDINATES OF THE OPEN HOLE AND THE HOLE',/,	00003320
* RADIUS')	00003330
READ(5,X) LYPN(I),(FSCD(I,J),J=1,3)	00003340
730 ELTHK(I)=PLYTHK(1,LYPN(I))	00003350
474 CONTINUE	00003360
WRITE(6,688)	00003370
688 FORMAT(/,' ENTER NUMBER OF ELEMENTS IN BOTTOM PLATE '	00003380
READ(5,X) NEL2	00003390
NELTOT=NEL1+NEL2	00003400
NP1=NEL1+1	00003410
DO 611 I=NP1,NELTOT	00003420
WRITE(6,800) I	00003430
800 FORMAT(' FOR ELEMENT NO',I5,	00003440
* ENTER: ELEMENT ID,N1,N2,N3,N4,N5,ELEMENT TYPE')	00003450
READ(5,X) (NELCON(I,J),J=1,6),NELTYP(I)	00003460
DO 593 IL=2,6	00003470
IC=0	00003480
NELCHA(I,1)=NELCON(I,1)	00003490
NIN=NP1+1	00003500
DO 594 KL=NIN,NGTOT	00003510
IF(NELCON(I,IL).EQ.NGRID(KL)) IC=1	00003520
IF(NELCON(I,IL).EQ.NGRID(KL)) NELCHA(I,IL)=KL	00003530
IF(IC.EQ.1) GO TO 593	00003540
594 CONTINUE	00003550

593	CONTINUE	00003560
	IF(CM(2).EQ.CMC) GO TO 927	00003570
	IF(NELTYP(I).NE.1) GO TO 921	00003580
	WRITE(6,1921)	00003590
1921	FORMAT(' ENTER ELEMENT THICKNESS')	00003600
	READ(5,M) ATH	00003610
921	IF(NELTYP(I).NE.2) GO TO 922	00003620
	WRITE(6,1923)	00003630
1923	FORMAT(' ENTER ELEMENT THICKNESS')	00003640
	READ(5,M) ATH	00003650
	FSCD(I,1)=GCOORD(NELCNA(I,6),1)	00003660
	FSCD(I,2)=GCOORD(NELCNA(I,6),2)	00003670
	FSCD(I,3)=FASD/2.0D0	00003680
	WRITE(6,3443) I,NELCNA(I,6),FSCD(I,1),FSCD(I,2)	00003690
3443	FORMAT(' C I NELCNA FSCD12',I5,2X,5(D9.3,2X))	00003700
922	IF(NELTYP(I).NE.3) GO TO 924	00003710
	WRITE(6,925)	00003720
925	FORMAT(' ENTER ELEMENT THICKNESS, X AND Y COORDINATES',/,	00003730
	* OF OPEN HOLE AND HOLE RADIUS')	00003740
	READ(5,M) ATH,(FSCD(I,J),J=1,3)	00003750
924	ELTHK(I)=ATH/30.0D0	00003760
	PLYTHK(2,1)=ATH/30.0D0	00003770
	LYPN(I)=1	00003780
	GO TO 611	00003790
927	CONTINUE	00003800
	IF(NELTYP(I).NE.1) GO TO 926	00003810
	WRITE(6,1927)	00003820
1927	FORMAT(' ENTER ELEMENT LAYUP NO')	00003830
	READ(5,M) LYPN(I)	00003840
926	IF(NELTYP(I).NE.2) GO TO 928	00003850
	WRITE(6,929)	00003860
929	FORMAT(' ENTER ELEMENT LAYUP NO')	00003870
	READ(5,M) LYPN(I)	00003880
	FSCD(I,1)=GCOORD(NELCNA(I,6),1)	00003890
	FSCD(I,2)=GCOORD(NELCNA(I,6),2)	00003900
	FSCD(I,3)=FASD/2.0D0	00003910
928	IF(NELTYP(I).NE.3) GO TO 930	00003920
	WRITE(6,931)	00003930
931	FORMAT(' ENTER ELEMENT LAYUP NUMBER, X AND Y ',/,	00003940
	* COORDINATES OF THE OPEN HOLE AND THE HOLE',/,	00003950
	* RADIUS')	00003960
	READ(5,M) LYPN(I),(FSCD(I,J),J=1,3)	00003970
930	ELTHK(I)=PLYTHK(2,LYPN(I))	00003980
611	CONTINUE	00003990
	WRITE(6,1741)	00004000
1741	FORMAT(/,' FASTENERS ARE MODELED BY EFFECTIVE ',/,	00004010
	* FASTENER ELEMENTS WHICH PROVIDE THE ',/,	00004020
	* ELASTIC LINK BETWEEN THE TOP AND ',/,	00004030
	* BOTTOM PLATES',/,)	00004040
	WRITE(6,1711)	00004050
1711	FORMAT(' ENTER NUMBER OF FASTENERS IN JOINT ')	00004060
	READ(5,M) NUMF	00004070
	WRITE(6,,16)	00004080
716	FORMAT(/,'	00004090
	* EFFECTIVE FASTENER ELEMENTS ARE	00004100
	* NUMBERED AS SHOWN:	00004110
	* N1 (TOP PLATE)	00004120
	* N2 (BOTTOM PLATE)	00004130
		00004140
		00004150

```

X' WHERE N1 AND N2 CORRESPOND TO THE CENTRAL
X' NODES IN LOADED HOLE ELEMENTS
X' FORMAT: ELEMENT ID, N1, N2
DO 717 I=1,NUMF
  WRITE(6,711) I
711 FORMAT(' ENTER ELEMENT NO',I5)
  READ(5,X) (NELFAS(I,J),J=1,3)
717 CONTINUE

  DETERMINE GRID STORAGE LOCATIONS FOR
  ELEMENT NUDES

  DO 612 I=1,NEL1
    N=6
    IF(NELTYP(I).NE.2) N=5
    NELCNA(I,1)=NELCON(I,1)
    DO 613 J=2,N
      IC=0
      DO 614 K=1,NOP1
        IF(NELCON(I,J).EQ.NGRID(K)) IC=1
        IF(NELCON(I,J).EQ.NGRID(K)) NELCNA(I,J)=K
        IF(IC.EQ.1) GO TO 613
      614 CONTINUE
    613 CONTINUE
    612 CONTINUE
    NP1=NEL1+1
    DO 395 I=NP1,NELTOT
      N=6
      IF(NELTYP(I).NE.2) N=5
      NELCNA(I,1)=NELCON(I,1)
      DO 616 J=2,N
        IC=0
        NIN=NOPI+1
        DO 617 K=NIN,NOPTOT
          IF(NELCON(I,J).EQ.NGRID(K)) IC=1
          IF(NELCON(I,J).EQ.NGRID(K)) NELCNA(I,J)=K
          IF(IC.EQ.1) GO TO 616
        617 CONTINUE
      616 CONTINUE
    395 CONTINUE
    DO 741 J=1,NUMF
      N=2
      NELFSA(I,1)=NELFAS(I,1)
      DO 242 J=1,N
        IC=0
        DO 243 K=1,NOPTOT
          IF(NELFAS(I,J+1).EQ.NGRID(K)) IC=1
          IF(NELFAS(I,J+1).EQ.NGRID(K)) NELFSA(I,J+1)=K
          IF(IC.EQ.1) GO TO 242
        243 CONTINUE
      242 CONTINUE
    741 CONTINUE

    COMPUTE ELEMENT WIDTHS

    DO 239 I=1,NELTOT
      ELWIDTH(I)=DABS(GCOORD(NELCNA(I,3),2)-GCOORD(NELCNA(I,2),2))
    239 CONTINUE
  
```

```

00004160
00004170
00004180
00004190
00004200
00004210
00004220
00004230
00004240
00004250
00004260
00004270
00004280
00004290
00004300
00004310
00004320
00004330
00004340
00004350
00004360
00004370
00004380
00004390
00004400
00004410
00004420
00004430
00004440
00004450
00004460
00004470
00004480
00004490
00004500
00004510
00004520
00004530
00004540
00004550
00004560
00004570
00004580
00004590
00004600
00004610
00004620
00004630
00004640
00004650
00004660
00004670
00004680
00004690
00004700
00004710
00004720
00004730
00004740
00004750
  
```

```

C      GROUP ELEMENTS TO AVOID THE DUPLICATE
C      CALCULATION OF IDENTICAL STIFFNESS
C      MATRICIES
      WRITE(6,3000)
3000  FORMAT(/,' TO REDUCE RUN TIMES, ELEMENTS MAY BE ',/,
* ' GROUPED INTO SETS WHICH WILL BE ASSIGNED',/,
* ' IDENTICAL STIFFNESS MATRICIES
* ' ENTER:      1 TO USE THIS OPTION
* '            2 OTHERWISE ' )
      READ(5,*) NOPT
      IF(NOPT.EQ.1) GO TO 3001
      N1=0
      N2=0
      N3=0
      DO 3002 I=1,NEL1
      IF(NELTYP(I).EQ.1) N1=N1+1
      IF(NELTYP(I).EQ.2) N2=N2+1
3002  IF(NELTYP(I).EQ.3) N3=N3+1
      NEF(1)=NUMF
      NLH(1)=N2
      NOH(1)=N3
      NPL(1)=N1
      N=NUMF
      DO 3003 I=1,N
3003  NOFF(1,1,1)=NELFAS(I,1)
      IC=0
      DO 3004 I=1,NEL1
      IF(NELTYP(I).EQ.2) IC=IC+1
      IF(NELTYP(I).EQ.2) NGLH(1,IC,1)=NELCON(I,1)
      IF(NELTYP(I).EQ.2) NUMLH(1,IC)=1
3004  CONTINUE
      IC=0
      DO 3005 I=1,NEL1
      IF(NELTYP(I).EQ.3) IC=IC+1
      IF(NELTYP(I).EQ.3) NOOH(1,IC,1)=NELCON(I,1)
      IF(NELTYP(I).EQ.3) NUMOH(1,IC)=1
3005  CONTINUE
      IC=0
      DO 3006 I=1,NEL1
      IF(NELTYP(I).EQ.1) IC=IC+1
      IF(NELTYP(I).EQ.1) NGPL(1,IC,1)=NELCON(I,1)
      IF(NELTYP(I).EQ.1) NUMPL(1,IC)=1
3006  CONTINUE
      N=NEL1+1
      N1=0
      N2=0
      N3=0
      DO 3007 I=N,NELTOT
      IF(NELTYP(I).EQ.1) N1=N1+1
      IF(NELTYP(I).EQ.2) N2=N2+1
3007  IF(NELTYP(I).EQ.3) N3=N3+1
      NEF(2)=NUMF
      NLH(2)=N2
      NOH(2)=N3
      NPL(2)=N1
      N=NUMF
      DO 3008 I=1,N
3008  NGEF(2,1,1)=NELFAS(I,1)
      IC=0

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N=NEL1+1
DO 3010 I=N,NELTOT
IF(NELTYP(1).EQ.2) IC=IC+1
IF(NELTYP(1).EQ.2) NGLH(2,IC,1)=NELCON(1,1)
IF(NELTYP(1).EQ.2) NUMLH(2,IC)=1
3010 CONTINUE
IC=0
N=NEL1+1
DO 3011 I=N,NELTOT
IF(NELTYP(1).EQ.1) IC=IC+1
IF(NELTYP(1).EQ.1) NGPL(2,IC,1)=NELCON(1,1)
IF(NELTYP(1).EQ.1) NUMPL(2,IC)=1
3011 CONTINUE
IC=0
N=NEL1+1
DO 3012 I=N,NELTOT
IF(NELTYP(1).EQ.3) IC=IC+1
IF(NELTYP(1).EQ.3) NMOH(2,IC,1)=NELCON(1,1)
IF(NELTYP(1).EQ.3) NUMOH(2,IC)=1
3012 CONTINUE
GO TO 3013
3001 CONTINUE
WRITE(6,3015)
3015 FORMAT(/,' FOR THE TOP PLATE INPUT NUMBER OF GROUPS',/,
* ' FOR THE EFFECTIVE FASTENER, LOADED HOLE, UNLOADED',/,
* ' HOLE AND PLAIN ELEMENT',/,
* ' (INPUT 0 IF ELEMENT TYPE IS NOT USED)')
READ(5,M) NEF(1),NLH(1),NOH(1),NPL(1)
WRITE(6,3016)
3016 FORMAT(' GROUPING OF EFFECTIVE FASTENER ELEMENTS:')
N=NEF(1)
DO 3017 I=1,N
WRITE(6,3018) I
3018 FORMAT(' ENTER NUMBER OF ELEMENTS IN GROUP NUMBER',I8)
READ(5,M) NUMEF(1,I)
N1=NUMEF(1,I)
WRITE(6,3019) N1
3019 FORMAT(' ENTER ',I8,' ELEMENT IDS')
READ(5,M) (NOEF(1,I,J),J=1,N1)
3017 CONTINUE
WRITE(6,3020)
3020 FORMAT(/,' GROUPING OF LOADED HOLE ELEMENTS:')
N=NLH(1)
DO 3021 I=1,N
WRITE(6,3021) I
3021 FORMAT(' ENTER NUMBER OF ELEMENTS IN GROUP NUMBER',I8)
READ(5,M) NUMLH(1,I)
N1=NUMLH(1,I)
WRITE(6,3022) N1
3022 FORMAT(' INPUT',I8,' ELEMENT IDS')
READ(5,M) (NGLH(1,I,J),J=1,N1)
3020 CONTINUE
IF(NOH(1).EQ.0) GO TO 4071
WRITE(6,3023)
3023 FORMAT(' GROUPING OF UNLOADED HOLE ELEMENTS')
N=NOH(1)
DO 3024 I=1,N
WRITE(6,3025) I
3025 FORMAT(' ENTER NUMBER OF ELEMENTS IN GROUP NUMBER',I8)
READ(5,M) NUMOH(1,I)

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N1=NUMOH(1,I)
WRITE(6,3026) N1
3026 FORMAT(' ENTER',I8,' ELEMENT IDS')
READ(5,*) (NGOH(1,I,J),J=1,N1)
3024 CONTINUE
4071 IF(NPL(1).EQ.0) GO TO 4072
WRITE(6,3027)
3027 FORMAT(' GROUPING OF PLAIN ELEMENTS:')
N=NPL(1)
DO 3031 I=1,N
WRITE(6,3032) I
3032 FORMAT(' ENTER NUMBER OF ELEMENTS IN GROUP NUMBER',I8)
READ(5,*) NUMPL(1,I)
N1=NUMPL(1,I)
WRITE(6,3033) N1
3033 FORMAT(' ENTER',I8,' ELEMENT IDS')
READ(5,*) (NOPL(1,I,J),J=1,N1)
3031 CONTINUE
4072 CONTINUE
WRITE(6,4015)
4015 FORMAT(' FOR THE BOTTOM PLATE INPUT NUMBER OF GROUPS',/,
* FOR THE LOADED HOLE, UNLOADED HOLE, AND PLAIN ',/,
* ELEMENTS ',/,
* (INPUT 0 IF AN ELEMENT TYPE IS NOT USED)')
READ(5,*) NLH(2),NOH(2),NPL(2)
NEF(2)=NEF(1)
N=NEF(1)
DO 4017 I=1,N
NUMEF(2,I)=NUMEF(1,I)
N1=NUMEF(1,I)
DO 4019 J=1,N1
4019 NOEF(2,I,J)=NOEF(1,I,J)
4017 CONTINUE
WRITE(6,4088)
4088 FORMAT(' GROUPING OF LOADED HOLE ELEMENTS:')
N=NLH(2)
DO 4020 I=1,N
WRITE(6,4021) I
4021 FORMAT(' ENTER NUMBER OF ELEMENTS IN GROUP NUMBER ',I8)
READ(5,*) NUMLH(2,I)
N1=NUMLH(2,I)
WRITE(6,4022) N1
4022 FORMAT(' INPUT',I8,' ELEMENT IDS')
READ(5,*) (NGLH(2,I,J),J=1,N1)
4020 CONTINUE
IF(NOH(2).EQ.0) GO TO 4073
WRITE(6,4023)
4023 FORMAT(' GROUPING OF UNLOADED HOLE ELEMENTS')
N=NOH(2)
DO 4024 I=1,N
WRITE(6,4025) I
4025 FORMAT(' ENTER NUMBER OF ELEMENTS IN GROUP NUMBER',I8)
READ(5,*) NUMOH(2,I)
N1=NUMOH(2,I)
WRITE(6,4026) N1
4026 FORMAT(' ENTER',I8,' ELEMENT IDS')
READ(5,*) (NGOH(2,I,J),J=1,N1)
4024 CONTINUE
4073 IF(NPL(2).EQ.0) GO TO 4074
WRITE(6,4027)

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4027 FORMAT(' GROUPING OF PLAIN ELEMENTS:')
      N=NPL(2)
      DO 4031 I=1,N
      WRITE(6,4032) I
4032 FORMAT(' ENTER NUMBER OF ELEMENTS IN GROUP NUMBER',I$)
      READ(5,*) NUMPL(2,I)
      N1=NUMPL(2,I)
      WRITE(6,4033) N1
4033 FORMAT(' ENTER',I$, ' ELEMENT IDS')
      READ(5,*) (NOPL(2,I,J),J=1,N1)
4031 CONTINUE
4074 CONTINUE
3013 CONTINUE
      WRITE(6,3737)
3737 FORMAT(//, ' INPUT DATA FOR FAILURE ANALYSIS',//)
      DO 226 K=1,2
      IF(CM(K).NE.CMC) GO TO 2226
      WRITE(6,532) K
532 FORMAT(' ENTER FIBER ULTIMATE STRAIN VALUES ',//,
* ' IN PLATE NO ',I$,//,
* ' EPSILON ULT IN COMPRESSION ',//,
* ' EPSILON ULT IN TENSION ',//,
* ' GAMMA ULT IN SHEAR ',//)
      READ(5,*) (STULT(I,K),I=1,3)
      GO TO 2227
2226 CONTINUE
      WRITE(6,2229)
2229 FORMAT(' ENTER METALLIC STRENGTHS: ',//,
* ' TENSILE STRENGTH ',//,
* ' COMPRESSIVE STRENGTH ',//,
* ' SHEAR STRENGTH')
      READ(5,*) STM(1),STM(2),STM(3)
2227 CONTINUE
      WRITE(6,4054)
4054 FORMAT(//, ' AN AVERAGE STRESS CRITERIA IS USED TO ',//,
* ' PREDICT FAILURE. AO VALUES ARE REQUIRED AS ',//,
* ' CHARACTERISTIC DISTANCES OVER WHICH STRESSES ',//,
* ' ARE TO BE AVERAGED AND COMPARED TO UNNOTCHED ',//,
* ' LAMINATE STRENGTHS TO PREDICT FAILURE',//)
      WRITE(6,5432) K
5432 FORMAT(' ENTER AO VALUES FOR STRESS AVERAGING',//,
* ' FOR EACH FAILURE MODE IN PLATE NO',I$,//,
* ' AONT = NET SECTION ',//,
* ' AOBK = BEARING ',//,
* ' AOSO = SHEAROUT ')
      READ(5,*) AONT(K),AOBK(K),AOSO(K)
226 CONTINUE
C
C CASE HEADING
C
      WRITE(6,143)
143 FORMAT(///,10X,'PROGRAM SAMCJ',//)
      IF(NSDLS.EQ.1) WRITE(6,633)
      IF(NCDLS.EQ.2) WRITE(6,634)
633 FORMAT(2X,'A SINGLE LAP SHEAR PANEL WILL BE ANALYZED',//)
634 FORMAT(2X,'A DOUBLE LAP SHEAR PANEL WILL BE ANALYZED',//)
      IF(LTHCM.EQ.1) WRITE(6,823)
      IF(LTHCM.EQ.2) WRITE(6,824)
823 FORMAT(2X,'LOADED IN STATIC TENSION',//)
824 FORMAT(2X,'LOADED IN STATIC COMPRESSION',//)

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DO 241 I=1,2
WRITE(6,600) I
600 FORMAT(10X,'PLATE NO ',I5,' ',/)
WRITE(6,601) (MTL(I,J),J=1,15)
601 FORMAT(2X,15A4,/)
HT=NELPLS(I,1)*PLYTHK(I,1)
WRITE(6,691) E1(I),E2(I),G12(I),V12(I),V21(I)
691 FORMAT(2X,'MATERIAL PROPERTIES',/,/,
  *10X,'E1' =',D9.3,' PSI',/,
  *10X,'E2' =',D9.3,' PSI',/,
  *10X,'G12' =',D9.3,' PSI',/,
  *10X,'NU12' =',D9.3,/,
  *10X,'NU21' =',D9.3,/)
241 CONTINUE
WRITE(6,606)
606 FORMAT(10X,'FASTENER DESCRIPTION',/,/)
WRITE(6,607) (MTL(3,J),J=1,15)
607 FORMAT(2X,15A4,/)
WRITE(6,647) FASD
647 FORMAT(2X,' DIAMETER = ',D9.3,' INCHES',/)
WRITE(6,609) FASE,FASV
609 FORMAT(2X,' MATERIAL PROPERTIES',/,/,
  *10X,'E' =',D9.3,' PSI',/,
  *10X,'MU' =',D9.3,/)
708 CONTINUE
WRITE(6,923)
923 FORMAT(10X,'FAILURE ANALYSIS',/,/)
WRITE(6,558)
558 FORMAT(2X,' AN AVERAGE STRESS CRITERION WILL BE USED',/,)
DO 631 I=1,2
WRITE(6,632) I
632 FORMAT(2X,' PLATE NUMBER',I5,/)
NP=NUMPLY(I)
IF(CM(1,1,1,1)) GO TO 3112
WRITE(6,713)
713 FORMAT(2X,' FIBER STRAIN ULTIMATES',/,)
776 WRITE(6,677) (STULT(LL,I),LL=1,3)
677 FORMAT(2X,' EPSILON ULT COMP = ',D9.3,/,
  *2X,' EPSILON ULT TEN = ',D9.3,/,
  *2X,' GAMMA ULT SHEAR = ',D9.3,/)
GO TO 3113
3112 CONTINUE
WRITE(6,3114)
3114 FORMAT(' METALLIC STRENGTHS ',/,)
WRITE(6,3115) STM(1),STM(2),STM(3)
3115 FORMAT(2X,' TENSILE STRENGTH = ',D9.3,/,
  *2X,' COMPRESSIVE STRENGTH = ',D9.3,/,
  *2X,' SHEAR STRENGTH = ',D9.3,/)
3113 CONTINUE
WRITE(6,1563)
1563 FORMAT(10X,' CHARACTERISTIC DISTANCES',/,)
WRITE(6,564) AONT(I),AOBR(I),AOSO(I)
564 FORMAT(' AONT = ',D9.3,' INCHES',/,
  *' AOBR = ',D9.3,' INCHES',/,
  *' AOSO = ',D9.3,' INCHES',/)
631 CONTINUE
C THE JOINT LOAD DISTRIBUTION IS CALCULATED USING THE
C FINITE ELEMENT METHOD WITH SPECIAL PROBLEM-ADAPTED
C ELEMENTS WHICH EFFECTIVELY REPRESENT THE STIFFNESS

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C      PROPERTIES OF FASTENERS, LOADED HOLES, AND OPEN
C      HOLE REGIONS IN THE JOINT
C
C      INTERNAL APPLIED LOAD SET TO 1 KIP
C
C      APP=1000.0
C      IF(LTNCM.EQ.2) APP=-APP
C      NELTOT=NEL1+NEL2
C      NGTOT=NGP1+NGP2
C
C      INITIALIZE ARRAYS
C
C      DO 1 I=1,50
C      DO 3 J=1,4
C      DO 3 K=1,4
C      3 ELSTFF(I,J,K)=0.
C      1 CONTINUE
C      DO 4 I=1,200
C      PBC(I)=0.
C      HHS(I)=0.
C      AHR(I)=0.
C      AHR2(I)=0.
C      DO 5 J=1,200
C      GLSTFF(I,J)=0.
C      ASQM(I,J)=0.
C      5 CONTINUE
C      4 CONTINUE
C
C      CALCULATION OF EFFECTIVE FASTENER ELEMENT
C      STIFFNESS MATRICIES
C
C      WRITE(6,8418)
C      8418 FORMAT(/,' PAUSE FOR STIFFNESS MATRIX CALCULATIONS',/)
C      NLOOP=NEF(1)
C      DO 444 I=1,NLOOP
C      NEL=NGEF(1,1)
C      DO 5001 II=1,NUMF
C      5001 IF(NEL.EQ.NELFAS(II,1)) IEL=II
C
C      SEARCH FOR LOADED HOLE ELEMENTS CONNECTED TO
C      FASTENER ELEMENT
C
C      NTOP=0
C      NBOT=0
C      DO 643 J=1,NEL1
C      643 IF(NELFAS(IEL,2).EQ.NELCON(J,6)) NTOP=J
C      NP1=NEL1+1
C      DO 446 J=NP1,NELTOT
C      446 IF(NELFAS(IEL,3).EQ.NELCON(J,6)) NBOT=J
C      NPLY(1)=NELPLS(1,LYPN(NTOP))
C      H(1)=ELTHK(NTOP)
C      DO 910 JJJ=1,50
C      910 IPLY(JJJ,1)=NELPT(1,LYPN(NTOP),JJJ)
C      NPLY(2)=NELPLS(2,LYPN(NBOT))
C      H(2)=ELTHK(NBOT)
C      DO 113 JJJ=1,50
C      113 IPLY(JJJ,2)=NELPT(2,LYPN(NBOT),JJJ)
C
C      INITIALIZE PARAMETERS FOR COLLOCATION

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	NT=7	00008360
	NOUT=57	00008370
	NCLL=10	00008380
	NB=NOUT+4*NCLL	00008390
	AX=FSCD(NTOP,3)	00008400
	BX=FSCD(NTOP,3)	00008410
	DO 570 L=1,2	00008420
	PHI=0.000	00008430
	IF(L.EQ.2) PHI=90.00	00008440
	DO 530 K=1,2	00008450
	NTB=NTOP	00008460
	IF(K.EQ.2) NTB=NBOT	00008470
		00008480
	ELEMENT VERTEXES ARE INTERNALLY	00008490
	NUMBERED AS:	00008500
	3 2	00008510
	4 1	00008520
		00003530
		00008540
	SFX=(GCOORD(NELCNA(NTB,5),1)+GCOORD(NELCNA(NTB,2),1))/2.000	00008550
	SFY=(GCOORD(NELCNA(NTB,3),2)+GCOORD(NELCNA(NTB,2),2))/2.000	00008560
	DO 128 JJ=1,4	00008570
	XC(JJ)=GCOORD(NELCNA(NTB,6-JJ),1)-FSCD(NTB,1)	00008580
	YC(JJ)=GCOORD(NELCNA(NTB,6-JJ),2)-FSCD(NTB,2)	00008590
128	CONTINUE	00008600
	XC(5)=XC(1)	00008610
	YC(5)=YC(1)	00008620
	N=ELWDTH(NTB)	00008630
	AST=1000.0	00008640
	CALL POLY(N,AST,JK,K,NCLL,LTNCM)	00008650
	CALL CIRC(N,AST,JK,K,LTNCM)	00008660
	NOPT4=1	00008670
	NCASE=1	00008680
	NTYPE=NELTYP(1EL)	00008690
	CALL FIGEOM(H,PHI,K,NOPT4,NCLL)	00008700
	CALL FBOLT(ANGK,H,PHI,K)	00008710
580	CONTINUE	00008720
	N=NPLY(1)	00008730
	DO 30 II=1,N	00008740
	N=IPLY(II,1)	00008750
30	PLYK(II)=ANGK(M,1)	00008760
	N=NPLY(2)	00008770
	DO 61 II=1,N	00008780
	N1=II+NPLY(1)	00008790
	N2=IPLY(II,2)	00008800
61	PLYK(N1)=ANGK(N2,2)	00008810
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	FASG=FASE/(2.*(1+FA5V))
	FASLAM=5.*(1.0+FA5V)/(7.+6.*FA5V)
	FASR=FA5D/2.
	FASA=ACOS(-1.)*FASR**2
	FASI=ACOS(-1.)*FASR**4/4.
	FASSS=FASLAM*FASG*FASA
	FASBS=FA5E*FASI
	P=1000.
	CALL CENTD(H,FASSS,FASBS,P)
	CALL SOLVE(H,P,U1,U2)
	IF(L.EQ.2) GO TO 666

```

RDSTFF(IEL,1)=DABS(P/(U1+U2))
GO TO 570
666 RDSTFF(IEL,2)=DABS(P/(U1+U2))
570 CONTINUE
IF(NUMEF(1,I).EQ.1) GO TO 444
N=NUMEF(1,1)
DO 5023 K=2,N
DO 5024 L=1,NUM?
5024 IF(NGEF(1,I,K).EQ.NELFAS(L,1)) IEL2=L
RDSTFF(IEL2,1)=RDSTFF(IEL,1)
RDSTFF(IEL2,2)=RDSTFF(IEL,2)
5023 CONTINUE
444 CONTINUE
8584 CONTINUE
CCCCC
CALCULATION OF LOADED HOLE AND UNLOADED HOLE
ELEMENT STIFFNESS MATRICIES
CCCCC
INITIALIZE GAUSSIAN QUADRATURE POINTS AND WEIGHTS
NGP=5
GSSX(1)=-0.9739065285
GSSX(2)=-0.8650633666
GSSX(3)=-0.6794095682
GSSX(4)=-0.4333953941
GSSX(5)=-0.1488743389
GSSW(1)=0.0666713443
GSSW(2)=0.1494513491
GSSW(3)=0.2190863625
GSSW(4)=0.2492467193
GSSW(5)=0.295542247
DO 588 I=1,NGP
GSSX(I+NGP)=-GSSX(NGP-I+1)
GSSW(I+NGP)=GSSW(NGP-I+1)
588 CONTINUE
NAVD=10
DO 420 KJ=1,2
ISLH=1
ISOH=1
ISPL=1
NLOOP=NLH(KJ)+NOH(KJ)+NPL(KJ)
NCLH=0
NCOH=0
NCPL=0
DO 400 L=1,NLOOP
IF(NCLH.EQ.NLH(KJ)) ISLH=0
IF(NCLH.EQ.NLH(KJ)) GO TO 6010
NCLH=NCLH+1
IEL2=NLH(KJ,NCLH,1)
GO TO 6011
6010 IF(NCOH.EQ.NOH(KJ)) ISOH=0
IF(NCOH.EQ.NOH(KJ)) GO TO 6020
NCOH=NCOH+1
IEL2=NOH(KJ,NCOH,1)
GO TO 6011
6020 IF(NCPL.EQ.NPL(KJ)) ISPL=0
IF(NCPL.EQ.NPL(KJ)) GO TO 400
NCPL=NCPL+1
IEL2=NOH(KJ,NCPL,1)

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6011 CONTINUE
DO 6030 KK=1,NELTOT
6030 IF(IEL2.EQ.NELCON(KK,1)) IEL=KK
      H(KJ)=ELTHK(IEL)
      NPLY(KJ)=NELPLS(KJ,LYPN(IEL))
      DO 919 JJJ=1,50
        IPLY(JJJ,KJ)=NELPT(KJ,LYPN(IEL),JJJ)
919 CONTINUE
      NRNK=5
      IF(NELTYP(IEL).EQ.2) NRNK=7

      INTERNAL NUMBERING OF ELEMENT VERTICIES:

              3      2
              4      1

      SFX=(GCOORD(NELCNA(IEL,5),1)+GCOORD(NELCNA(IEL,2),1))/2.000
      SFY=(GCOORD(NELCNA(IEL,3),2)+GCOORD(NELCNA(IEL,2),2))/2.000
      DO 440 K=1,4
        XC(K)=GCOORD(NELCNA(IEL,6-K),1)-FSCD(IEL,1)
        IF(NELTYP(IEL).EQ.1) XC(K)=GCOORD(NELCNA(IEL,6-K),1)-SFX
        YC(K)=GCOORD(NELCNA(IEL,6-K),2)-FSCD(IEL,2)
        IF(NELTYP(IEL).EQ.1) YC(K)=GCOORD(NELCNA(IEL,6-K),2)-SFY
440 CONTINUE
        XC(5)=XC(1)
        YC(5)=YC(1)
        AX=FSCD(IEL,3)
        IF(NELTYP(IEL).EQ.1) AX=0.1
        BX=AX
        PI=DARCOS(-1.000)
        RAD=PI/180.00
        NOAUSS=2*NGP
        NGPT=4*NOAUSS*2
        IC=0
        NCPT=2*NGP

      DETERMINE COORDINATES AT WHICH STRESSES AND
      DISPLACEMENTS ARE TO BE COMPUTED.
      ELEMENT NATURAL FLEXIBILITY MATRICIES
      ARE COMPUTED BY INTEGRATING STRESSES
      FOR EACH LOAD CASE IN THE NATURAL
      MODE METHOD. THE ELEMENTS ARE DIVIDED
      INTO FOUR REGIONS AND THE GAUSSIAN POINTS
      ARE SCALED TO EACH REGION SIZE

      REGION 1

      DO 15 II=1,NCPT
      DO 15 JJ=1,NCPT
      IC=IC+1
      XOUT(IC)=((-AX-XC(1))/2.)*GSSX(II)+(-AX+XC(3))/2.
      YOUT(IC)=(YC(3)-YC(1))/2.)*GSSX(JJ)+(YC(3)+YC(4))/2.
      NGHT(IC)=GSSW(II)*GSSW(JJ)*(YC(3)-YC(4))*(-AX-XC(3))/4.00
15 CONTINUE

      REGION 2

```

	DO 16 II=1,NCPT	00010160
	DO 16 JJ=1,NCPT	00010170
	IC=IC+1	00010180
	XOUT(IC)=AX*GSSX(II)	00010190
	YI=DSQRT(AX**2-XOUT(IC)**2)	00010200
	YOUT(IC)=((YC(2)-YI)/2.)*GSSX(JJ)+(YC(2)+YI)/2.	00010210
	WOHT(IC)=GSSW(II)*GSSW(JJ)*(YC(2)-YI)*AX/2.0D0	00010220
16	CONTINUE	00010230
C		00010240
C	REGION 3	00010250
C		00010260
	DO 17 II=1,NCPT	00010270
	DO 17 JJ=1,NCPT	00010280
	IC=IC+1	00010290
	XOUT(IC)=AX*GSSX(II)	00010300
	YI=-DSQRT(AX**2-XOUT(IC)**2)	00010310
	YOUT(IC)=((YI-YC(1))/2.)*GSSX(JJ)+(YI+YC(1))/2.	00010320
	WOHT(IC)=GSSW(II)*GSSW(JJ)*(YI-YC(1))*AX/2.0D0	00010330
17	CONTINUE	00010340
C		00010350
C	REGION 4	00010360
C		00010370
	DO 18 II=1,NCPT	00010380
	DO 18 JJ=1,NCPT	00010390
	IC=IC+1	00010400
	XOUT(IC)=((XC(1)-AX)/2.)*GSSX(II)+(XC(1)+AX)/2.	00010410
	YOUT(IC)=((YC(2)-YC(1))/2.)*GSSX(JJ)+(YC(2)+YC(1))/2.	00010420
	WOHT(IC)=GSSW(II)*GSSW(JJ)*(YC(2)-YC(1))*(XC(1)-AX)/4.0D0	00010430
18	CONTINUE	00010440
	NINT=IC	00010450
	N=4*(NCPT**2)	00010460
C		00010470
C	ADD COORDINATES ALONG WHICH STRESSES WILL	00010480
C	BE AVERAGED	00010490
C		00010500
	ANT=AONT(KJ)	00010510
	ABR=AOBR(KJ)	00010520
	ASO=AOSI(KJ)	00010530
	SG=1.0	00010540
	IF(LTHCM.EQ.2) SG=-1.0	00010550
	IF(KJ.EQ.2) SG=-SG	00010560
C		00010570
C	NET SECTION	00010580
C		00010590
	ANDO=ANT/FLOAT(NAVD)	00010600
	DO 21 II=1,NAVD	00010610
	IC=IC+1	00010620
	XOUT(IC)=0.0D0	00010630
	YOUT(IC)=BX+ANDO/2.+(II-1)*ANDO	00010640
21	CONTINUE	00010650
C		00010660
C	SHEAROUT	00010670
C		00010680
	ANSO=ASO/FLOAT(NAVD)	00010690
	DO 31 II=1,NAVD	00010700
	IC=IC+1	00010710
	XOUT(IC)=SG*(BX+ANSO/2.+(II-1)*ANSO)	00010720
	YOUT(IC)=BX	00010730
31	CONTINUE	00010740
C		00010750

C	BEARING	00010760
C	ANBR=ABR/FLOAT(NAVD)	00010770
	DO 41 II=1,NAVD	00010780
	IC=IC+1	00010790
	XOUT(IC)=30*(AX+ANBR/2.+(II-1)*ANBR)	00010800
	YOUT(IC)=0.	00010810
41	CONTINUE	00010820
C	ADD COORDINATES ALONG WHICH ELEMENT LOAD	00010830
C	RECOVERY WILL BE COMPUTED	00010840
C	DO 3332 III=1,10	00010850
	IC=IC+1	00010860
	IF(KJ.EQ.1) XOUT(IC)=XC(3)+0.1*AX	00010870
	IF(KJ.EQ.2) XOUT(IC)=XC(1)-0.1*AX	00010880
	YOUT(IC)=((YC(2)-YC(1))/2.000)*OSSX(III)+(YC(2)+YC(1))/2.000	00010890
C	STRESSES ARE SINGULAR AT THETA = 180 DEG OR Y = 0	00010900
C	IF(DABS(YOUT(IC)).LT.0.01) YOUT(IC)=YOUT(IC-1)	00010910
3332	CONTINUE	00010920
4891	CONTINUE	00010930
	NSTS=4*NAVD	00010940
	NOUT=4*(NOAUSS**2)	00010950
C	CALCULATION OF LOADED HOLE, UNLOADED HOLE, AND	00010960
C	PLAIN ELEMENT STIFFNESS MATRICIES	00010970
C	THN=DARCOS(-1.000)/FLOAT(NOP)	00010980
	NN=IEL	00010990
	DO 410 J=1,NNRK	00011000
	NOPT4=5	00011010
	NT=7	00011020
	NCLL=10	00011030
	NB=52+4*NCLL	00011040
	HT=H(KJ)*NPLY(KJ)	00011050
	HCASE=J	00011060
	NTYPE=NELTYP(IEL)	00011070
	CALL MOED(HT,W,AST,J,NN,KJ,NEL,NCLL)	00011080
	CALL MCIR(W,AST,NN,J,NCLL)	00011090
	PHI=0.000	00011100
	CALL AMATRX(H,PHI,KJ)	00011110
	CALL FIGEOM(H,PHI,KJ,NOPT4,NCLL)	00011120
	CALL INFLN(WOHT,H,NNRK,J,KJ,NN,NOPT)	00011130
410	CONTINUE	00011140
4000	CONTINUE	00011150
C	COMPUTE ELEMENT FAILURE VALUES BASED	00011160
C	ON MAXIMUM FIBER STRAIN ALLOWABLES	00011170
C	HT=H(KJ)*NPLY(KJ)	00011180
	IF(NELTYP(IEL).EQ.2) CALL SMAX(HT,KJ,IEL)	00011190
	IF(NELTYP(IEL).EQ.3) CALL SMAX(HT,KJ,IEL)	00011200
	IF(ISLH.EQ.0) GO TO 6040	00011210
	NL=NUMLH(KJ,NCLH)	00011220
	IF(NL.EQ.1) GO TO 400	00011230
	DO 6041 K=2,NL	00011240
	DO 6042 LL=1,NELTOT	00011250
6042	IF(NOLH(KJ,NCLH,K).EQ.NELCON(LL,1)) IEL2=LL	00011260
		00011270
		00011280
		00011290
		00011300
		00011310
		00011320
		00011330
		00011340
		00011350

DO 6043 ILM=1,10	00011360
DO 6043 ILK=1,10	00011370
6043 ELSTFF(IEL2,ILM,ILK)=ELSTFF(IEL,ILM,ILK)	00011380
DO 6044 KK=1,4	00011390
6044 PSMX(IEL2,KK)=PSMX(IEL,KK)	00011400
NNN=4*NAVD	00011410
DO 6045 ILM=1,NNN	00011420
DO 6045 ILK=1,10	00011430
6045 ELSTSS(IEL2,ILM,ILK)=ELSTSS(IEL,ILM,ILK)	00011440
6041 CONTINUE	00011450
GO TO 400	00011460
6040 IF(ISOH.EQ.0) GO TO 6046	00011470
NL=NUMOH(KJ,NGOH)	00011480
IF(NL.EQ.1) GO TO 400	00011490
DO 6047 K=2,NL	00011500
DO 6048 LL=1,NELTOT	00011510
6048 IF(NGOH(KJ,NGOH,K).EQ.NELCON(LL,1)) IEL2=LL	00011520
DO 6049 ILM=1,10	00011530
DO 6049 ILK=1,10	00011540
6049 ELSTFF(IEL2,ILM,ILK)=ELSTFF(IEL,ILM,ILK)	00011550
DO 6050 KK=1,4	00011560
6050 PSMX(IEL2,KK)=PSMX(IEL,KK)	00011570
NNN=4*NAVD	00011580
DO 6051 ILM=1,NNN	00011590
DO 6051 ILK=1,10	00011600
6051 ELSTSS(IEL2,ILM,ILK)=ELSTSS(IEL,ILM,ILK)	00011610
6047 CONTINUE	00011620
GO TO 400	00011630
6046 IF(ISPL.EQ.0) GO TO 400	00011640
NL=NUMPL(KJ,NCPL)	00011650
IF(NL.EQ.1) GO TO 400	00011660
DO 6053 K=2,NL	00011670
DO 6054 LL=1,NELTOT	00011680
6054 IF(NCPL(KJ,NCPL,K).EQ.NELCON(LL,1)) IEL2=LL	00011690
DO 6055 ILM=1,10	00011700
DO 6055 ILK=1,10	00011710
6055 ELSTFF(IEL2,ILM,ILK)=ELSTFF(IEL,ILM,ILK)	00011720
DO 6056 KK=1,4	00011730
6056 PSMX(IEL2,KK)=PSMX(IEL,KK)	00011740
NNN=4*NAVD	00011750
DO 6057 ILM=1,NNN	00011760
DO 6057 ILK=1,10	00011770
6057 ELSTSS(IEL2,ILM,ILK)=ELSTSS(IEL,ILM,ILK)	00011780
6053 CONTINUE	00011790
400 CONTINUE	00011800
420 CONTINUE	00011810
C DETERMINE ELEMENT ARRANGEMENT IN TOP	00011820
C AND BOTTOM PLATES	00011830
C	00011840
DO 681 KJ=1,2	00011850
IF(KJ.EQ.2) GO TO 501	00011860
L1=1	00011870
L2=NGP1	00011880
L3=1	00011890
L4=NEL1	00011900
GO TO 502	00011910
501 L1=NGP1+1	00011920
L2=NGTOT	00011930
L3=NEL1+1	00011940
	00011950

L4=NELTOT	00011960
502 CONTINUE	00011970
AXMIN=1.D10	00011980
AYMIN=1.D10	00011990
DO 503 I=L1,L2	00012000
IF(AXMIN.GT.GCOORD(I,1)) AXMIN=GCOORD(I,1)	00012010
IF(AYMIN.GT.GCOORD(I,2)) AYMIN=GCOORD(I,2)	00012020
IF(AXMIN.EQ.GCOORD(I,1).AND.AYMIN.EQ.GCOORD(I,2)) NC=I	00012030
503 CONTINUE	00012040
DO 574 I=L3,L4	00012050
574 IF(NELCON(I,2).EQ.NORID(NC)) IEL=I	00012060
NELGRD(KJ,1,1)=IEL	00012070
DO 504 I=1,25	00012080
DO 505 J=1,25	00012090
IPL=0	00012100
DO 506 K=L3,L4	00012110
506 IF(NELCON(K,2).EQ.NELCON(NELORD(KJ,J,I),3)) IEL=K	00012120
IF(IEL.EQ.0) GO TO 507	00012130
NELORD(KJ,J+1,I)=IEL	00012140
505 CONTINUE	00012150
507 CONTINUE	00012160
IF(KJ.EQ.1) NROW1=J	00012170
IF(KJ.EQ.2) NROW2=J	00012180
IEL=0	00012190
DO 508 L=L3,L4	00012200
508 IF(NELCON(NELORD(KJ,L,I),5).EQ.NELCON(	00012210
L,2)) IEL=L	00012220
IF(IEL.EQ.0) GO TO 509	00012230
NELORD(KJ,L+1,I)=IEL	00012240
504 CONTINUE	00012250
509 CONTINUE	00012260
IF(KJ.EQ.1) NCOL1=I	00012270
IF(KJ.EQ.2) NCOL2=I	00012280
681 CONTINUE	00012290
CCC	00012300
COMPUTE NODAL DEGREES OF FREEDOM	00012310
IC=0	00012320
DO 540 KJ=1,2	00012330
IF(KJ.EQ.1) NR=NROW1	00012340
IF(KJ.EQ.1) NC=NCOL1	00012350
IF(KJ.EQ.2) NR=NROW2	00012360
IF(KJ.EQ.2) NC=NCOL2	00012370
NELDIS(NELORD(KJ,1,1),1,1)=IC+1	00012380
NELDIS(NELORD(KJ,1,1),1,2)=IC+2	00012390
NELDIS(NELORD(KJ,1,1),2,1)=IC+3	00012400
NELDIS(NELORD(KJ,1,1),2,2)=IC+4	00012410
IC=IC+4	00012420
IF(NR.EQ.1) GO TO 549	00012430
DO 541 I=2,NR	00012440
NELDIS(NELORD(KJ,I,1),1,1)=NELDIS(NELORD(KJ,I-1	00012450
,1),2,1)	00012460
NELDIS(NELORD(KJ,I,1),1,2)=NELDIS(NELORD(KJ,I-1	00012470
,1),2,2)	00012480
NELDIS(NELORD(KJ,I,1),2,1)=IC+1	00012490
NELDIS(NELORD(KJ,I,1),2,2)=IC+2	00012500
IC=IC+2	00012510
541 CONTINUE	00012520
549 CONTINUE	00012530
DO 542 I=1,NC	00012540
	00012550

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DO 543 J=1,NR
  IF(I.EQ.1) GO TO 544
  NELDIS(NELORD(KJ,J,I),1,1)=NELDIS(NELORD(
    *KJ,J,I-1),4,1)
  NELDIS(NELORD(KJ,J,I),1,2)=NELDIS(NELORD(
    *KJ,J,I-1),4,2)
  NELDIS(NELORD(KJ,J,I),2,1)=NELDIS(NELORD(
    *KJ,J,I-1),3,1)
  NELDIS(NELORD(KJ,J,I),2,2)=NELDIS(NELORD(
    *KJ,J,I-1),3,2)
544 CONTINUE
  IF(J.EQ.1) GO TO 561
  NELDIS(NELORD(KJ,J,I),4,1)=NELDIS(NELORD(KJ,J-1,I),3,1)
  NELDIS(NELORD(KJ,J,I),4,2)=NELDIS(NELORD(KJ,J-1,I),3,2)
  GO TO 562
561 CONTINUE
  NELDIS(NELORD(KJ,J,I),4,1)=IC+1
  NELDIS(NELORD(KJ,J,I),4,2)=IC+2
  IC=IC+2
562 CONTINUE
  IF(NELTY(NELORD(KJ,J,I)).NE.2) GO TO 543
  NELDIS(NELORD(KJ,J,I),5,1)=IC+1
  NELDIS(NELORD(KJ,J,I),5,2)=IC+2
  IC=IC+2
543 CONTINUE
  NELDIS(NELORD(KJ,J,I),3,1)=IC+1
  NELDIS(NELORD(KJ,J,I),3,2)=IC+2
  IC=IC+2
543 CONTINUE
542 CONTINUE
540 CONTINUE
C
C      DETERMINE BOUNDARY NODES AND VALUES
C
  NRD=2*(NGP1+NGP2)
  DO 165 I=1,100
165 PBC(I)=0.
C
C      DISTRIBUTE APPLIED LOAD
C
  ATOT=GCOORD(NELCNA(NELORD(1,NROW1,1),3),2)
  *GCCORD(NELCNA(NELORD(1,1,1),2),2)
  APL=APP/ATOT
  SG=1.0
  IF(LTHCM.EQ.1) SG=-1.0
  DO 178 I=1,NROW1
  A1=GCOORD(NELCNA(NELORD(1,I,1),3),2)
  *GCCORD(NELCNA(NELORD(1,I,1),2),2)
  M1=NELDIS(NELORD(1,I,1),1,1)
  M2=NELDIS(NELORD(1,I,1),2,1)
  PBC(M1)=PBC(M1)+SG*(0.5*DABS(APL*M1))
  PBC(M2)=PBC(M2)+SG*(0.5*DABS(APL*M2))
178 CONTINUE
1119 CONTINUE
C
C      ASSEMBLE GLOBAL STIFFNESS MATRIX
C
  DO 220 NI=1,NELTOT
  IR=5
  IF(NELTY(NI).NE.2) IR=4

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00012570
00012580
00012590
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00013090
00013100
00013110
00013120
00013130
00013140
00013150

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C		00013160
C	TOP AND BOTTOM PLATE LOADED HOLE AND	00013170
C	UNLOADED HOLE ELEMENTS	00013180
	IC1=0	00013190
	DO 425 N2=1,IR	00013200
	DO 425 N3=1,2	00013210
	M1=NELDIS(N1,N2,N3)	00013220
	IC1=IC1+1	00013230
	IC2=0	00013240
	DO 425 N4=1,IR	00013250
	DO 425 N5=1,2	00013260
	M2=NELDIS(N1,N4,N5)	00013270
	IC2=IC2+1	00013280
	OLSTFF(M1,M2)=OLSTFF(M1,M2)+ELSTFF(N1,IC1,IC2)	00013290
425	CONTINUE	00013300
220	CONTINUE	00013310
C		00013320
C	ADD EFFECTIVE FASTENER ELEMENTS	00013330
	DO 260 I=1,NUMF	00013340
	DO 1541 J=1,NEL1	00013350
1541	IF(NELFAS(I,2).EQ.NELCON(J,6)) N=J	00013360
	N1=NELDIS(N,5,1)	00013370
	N2=NELDIS(N,5,2)	00013380
	NL=NEL1+1	00013390
	DO 1562 J=NL,NELTOT	00013400
1562	IF(NELFAS(I,3).EQ.NELCON(J,6)) N=J	00013410
	N3=NELDIS(N,5,1)	00013420
	N4=NELDIS(N,5,2)	00013430
	OLSTFF(N1,N1)=OLSTFF(N1,N1)+RDSTFF(I,1)	00013440
	OLSTFF(N1,N3)=OLSTFF(N1,N3)+RDSTFF(I,1)	00013450
	OLSTFF(N2,N2)=OLSTFF(N2,N2)+RDSTFF(I,2)	00013460
	OLSTFF(N2,N4)=OLSTFF(N2,N4)+RDSTFF(I,2)	00013470
	OLSTFF(N3,N3)=OLSTFF(N3,N3)+RDSTFF(I,1)	00013480
	OLSTFF(N3,N1)=OLSTFF(N3,N1)+RDSTFF(I,1)	00013490
	OLSTFF(N4,N4)=OLSTFF(N4,N4)+RDSTFF(I,2)	00013500
	OLSTFF(N4,N2)=OLSTFF(N4,N2)+RDSTFF(I,2)	00013510
260	CONTINUE	00013520
	NP=2*(NCP1+NCP2)	00013530
C		00013540
C	GLOBAL BOUNDARY CONDITIONS	00013550
	DO 415 I=1,NP	00013560
	RHS(I)=PBC(I)	00013570
415	CONTINUE	00013580
C		00013590
	IC=1	00013600
	NZERO(IC)=NELDIS(NELORD(2,1,NCOL2),4,1)	00013610
	DO 437 I=1,NROW2	00013620
	IC=IC+1	00013630
	NZERO(IC)=NELDIS(NELORD(2,1,NCOL2),3,1)	00013640
437	CONTINUE	00013650
	IC=IC+1	00013660
	NZERO(IC)=NELDIS(NELORD(2,1,NCOL2),4,2)	00013670
	NUMZ=IC	00013680
C		00013690
C	RESTORE REDUCED STIFFNESS MATRIX	00013700
	ICF=0	00013710
		00013720
		00013730
		00013740
		00013750

[illegible]

0000

	GO TO 2011	00014360
3112	CONTINUE	00014370
	ANR(L)=0.	00014380
2011	CONTINUE	00014390
C		00014400
C	CALCULATE NODAL LOADS	00014410
C		00014420
	IC=0	00014430
	DO 44 I=1,NRD	00014440
	DO 54 J=1,NUMZ	00014450
	IF(I.NE.NZERO(J)) GO TO 54	00014460
	ANR2(I)=0.0D0	00014470
	GO TO 44	00014480
54	CONTINUE	00014490
	IC=IC+1	00014500
	ANR2(I)=ANR(IC)	00014510
44	CONTINUE	00014520
	WRITE(6,3712)	00014530
3712	FORMAT(/,10X,'ELEMENT FORCES',//)	00014540
	DO 500 K=1,HELTOF	00014550
	NID=HELCON(K,1)	00014560
	WRITE(6,3947) NID	00014570
8947	FORMAT(/,' ELEMENT ID',IS,/,	00014580
	*6X,'GRID',9X,'FX',9X,'FY',/)	00014590
	IR=5	00014600
	KL=K	00014610
	IF(KL.GT.NE1) AL=K-NEL	00014620
	IF(NE1YP(K).NE.2) IR=4	00014630
	DO 510 I=1,IR	00014640
	SUMU=0.	00014650
	SUMV=0.	00014660
	N=2*I-1	00014670
	DO 520 J=1,IR	00014680
	N1=NELDIS(K,J,1)	00014690
	N2=NELDIS(K,J,2)	00014700
	SUMU=SUMU+ELSTFF(K,N,(2*J-1))*ANR2(N1)+	00014710
	*ELSTFF(K,N,(2*J))*ANR2(N2)	00014720
520	CONTINUE	00014730
	N=2*I	00014740
	DO 530 J=1,IR	00014750
	N1=NELDIS(K,J,1)	00014760
	N2=NELDIS(K,J,2)	00014770
	SUMV=SUMV+ELSTFF(K,N,(2*J-1))*ANR2(N1)+	00014780
	*ELSTFF(K,N,(2*J))*ANR2(N2)	00014790
530	CONTINUE	00014800
C		00014810
C	STORE ELEMENT LOADS FOR CHECK ON ELEMENT	00014820
C	LOAD RECOVERY	00014830
C		00014840
	IF(K.LE.NEL1.AND.(I.EQ.1.OR.I.EQ.2)) ELLOAD(K,I)=SUMU	00014850
	IF(K.GT.NEL1.AND.(I.EQ.3.OR.I.EQ.4)) ELLOAD(K,I-2)=SUMU	00014860
	NID=HELCON(K,I+1)	00014870
	WRITE(6,3239) NID,SUMU,SUMV	00014880
3239	FORMAT(2X,I8,5X,2(D9.3,2X))	00014890
510	CONTINUE	00014900
500	CONTINUE	00014910
C		00014920
C	COMPUTE ELEMENT FAILURE LOADS AND DETERMINE	00014930
C	CRITICAL ELEMENT TO CALCULATE JOINT FAILURE	00014940
C	LOAD	00014950

```

C      CALL FCRTT(APP,NEL1,NEL2,NDAM,IN,LTNCM,NAVD)
      FAILV=DABS(ELFAIL(IN,NDAM))
      IF(NSDLS.EQ.2) FAILV=2.*FAILV
      NID=NELCON(IN,1)
      WRITE(6,5555) NID,FAILV
5555  FORMAT(///,' FAILURE IS PREDICTED TO OCCUR IN ELEMENT',/,
      *' NUMBER',I5,' AT AN APPLIED JOINT LOAD VALUE ',/,
      *' OF ',D14.7,' LBS',/)
      IF(NDAM.EQ.1) WRITE(6,5556)
      IF(NDAM.EQ.2) WRITE(6,5557)
      IF(NDAM.EQ.3) WRITE(6,5558)
5556  FORMAT(' THE PREDICTED FAILURE MODE IS NET SECTION')
5557  FORMAT(' THE PREDICTED FAILURE MODE IS SHEAR-OUT ')
5558  FORMAT(' THE PREDICTED FAILURE MODE IS BEARING')
      STOP
      END

C
C
C      SUBROUTINE MGEO(HT,W,AST,J,IN,KJ,NEL,NCL)
C
      IMPLICIT REAL*8(A-H,O-Z)
      DIMENSION A1A(4),A2A(4)
      DIMENSION XB(200),YB(200),A1(200),A2(200)
      DIMENSION NELTYP(50),HTA(200)
      DIMENSION XC(5),YC(5)
      COMMON/CMT1/XB,YB,A1,A2,HTA
      COMMON/XC/YC,XC,YC
      COMMON/NTP/NELTYP

      DETERMINE EXTERIOR COLLOCATION POINTS AND
      STRESS BOUNDARY CONDITIONS CORRESPONDING
      TO THE NATURAL LOAD CASES

      NCS=5
      IF(NELTYP(IN).NE.2) NCS=3
      JK=0
      DO 15 I=1,4
      A1A(I)=0.
15  A2A(I)=0.
      A=(YC(2)-YC(1))*HT
      B=(XC(2)-XC(3))*HT
      IF(J.EQ.1) A1A(3)=1.0D0/A
      IF(J.EQ.1.AND.NELTYP(IN).NE.2) A1A(1)=1.0D0/A
      IF(J.EQ.2) A1A(2)=1.0D0/B
      IF(J.EQ.2.AND.NELTYP(IN).NE.2) A1A(4)=1.0D0/B
      IF(J.EQ.2.AND.NELTYP(IN).NE.2) GO TO 55
      IF(J.EQ.3) A1A(1)=1.0D0/A
      IF(J.EQ.4) A1A(4)=1.0D0/B
      IF(J.EQ.1.OR.J.EQ.3) AST=1.0D0/A
      IF(J.EQ.2.OR.J.EQ.4) AST=1.0D0/B
55  CONTINUE
      W=XC(3)-XC(2)
      IF(J.EQ.1.OR.J.EQ.3.OR.J.EQ.6) W=YC(2)-YC(1)
      DO 10 I=1,4
      X=XC(I)-XC(I+1)
      Y=YC(I+1)-YC(I)
      IF(X.EQ.0.) X=1.D-6
      IF(Y.EQ.0.) Y=1.D-6

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TH=DATAN2(X,Y)
TH=TH*180./DARCOS(-0.1D1)
DX=(XC(I+1)-XC(I))/(NCL+1)
DY=(YC(I+1)-YC(I))/(NCL+1)
DO 20 II=1,NCL
JK=JK+1
IF(I.EQ.1.OR.I.EQ.3) GO TO 23
YB(JK)=YC(I)
XB(JK)=XC(I)+DX*(II+.5)
IF(II.EQ.1) XB(JK)=XC(I)+(DX/2.)
GO TO 24
23 YB(JK)=YC(I)+DY*(II+.5)
IF(II.EQ.1) YB(JK)=YC(I)+(DY/2.)
XB(JK)=XC(I)
24 THTA(JK)=TH
A1(JK)=A1A(I)
A2(JK)=A2A(I)
IF(J.EQ.NCS.AND.(I.EQ.1.OR.I.EQ.3)) A1(JK)=(2*YB(JK)/W)*(3.0D0/A)
IF(J.EQ.(NCS+1).AND.(I.EQ.2.OR.I.EQ.4)) A1(JK)=(2*XB(JK)/W)*(3.0
*DO/3)
IF(J.EQ.(NCS+2).AND.(I.EQ.1.OR.I.EQ.3)) A2(JK)=2.0/DSQRT(A**2+B**2
*)
IF(J.EQ.(NCS+2).AND.(I.EQ.2.OR.I.EQ.4)) A2(JK)=-2.0/DSQRT(A**2+B**2
*)
20 CONTINUE
10 CONTINUE
RETURN
END

SUBROUTINE MCIR(W,AST,I,J,NCL)
IMPLICIT REAL*8(A-H,O-Z)
DIMENSION XB(200),YB(200),A1(200),A2(200)
DIMENSION THTA(200),NELTYP(50)
COMMON/CNT1/XB,YB,A1,A2,THTA
COMMON/NTP/NELTYP
COMMON/ELP/AX,DX
CCN=-1.0D0
RAD=DARCOS(CON)/180.
BSTR=DABS((2.*W*AST)/(DARCOS(CON)*AX))
IF(NELTYP(I).NE.2) BSTR=0.0D0
IF(NELTYP(I).EQ.2.AND.J.GT.5) BSTR=0.0D0
IC=4*NCL

DETERMINE INTERIOR COLLOCATION POINTS AND
STRESS BOUNDARY CONDITIONS

NBD=52
NBI=NBD/4
DO 1) K=1,4
CR=(K-1)*DARCOS(CON)/2.
DO 20 KI=1,NBI
IC=IC+1
THINC=(DARCOS(CON)/2.)/FLOAT(NBI)
A1(IC)=0.
A2(IC)=0.
THINC2=THINC/2.
THINC2=(KI-1)*THINC+CR
XB(IC)=AX*DCOS(TH)

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YB(IC)=AX*DSIN(TH)
THTA(IC)=TH/RAD
IF(J.GT.4) GO TO 20
IF(NELTYP(1).NE.2) GO TO 40
IF(J.EQ.1.AND.(K.EQ.1.OR.K.EQ.4)) A1(IC)=
  X-BSTR*DABS(DCOS(TH))
IF(J.EQ.2.AND.(K.EQ.3.OR.K.EQ.4)) A1(IC)=
  X-BSTR*DABS(DSIN(TH))
IF(J.EQ.3.AND.(K.EQ.2.OR.K.EQ.3)) A1(IC)=
  X-BSTR*DABS(DCOS(TH))
IF(J.EQ.4.AND.(K.EQ.1.OR.K.EQ.2)) A1(IC)=
  X-BSTR*DABS(DSIN(TH))
40 CONTINUE
20 CONTINUE
10 CONTINUE
  RETURN
  END

```

C
C
C

SUBROUTINE INFLN(WGHT,H,HRNK,J,KJ,I,NOPT)

```

IMPLICIT REAL*8(A-H,O-Z)
DIMENSION ELSTFF(50,10,10),WGHT(500),WK(150)
DIMENSION ELSTS(50,50,10),STSV(50),STSA(50,10)
DIMENSION AN(10,7),UVOUT(20)
DIMENSION PHI(3,7,400),STEMP(10,10),AO(10,3)
DIMENSION FINF(10,10),SINF(10,10),AINV(3,3)
DIMENSION APSX(500),APSY(500),APSXY(500)
DIMENSION H(2),XC(5),YC(5),NPLY(2)
DIMENSION IC(10)
DIMENSION A(10,10),ATEMP2(10,10)
COMMON/UV/UVOUT
COMMON/XCYC/XC,YC
COMMON/ELP/AX,BX,NOUT,NSTS
COMMON/ELS/ELSTFF,ELSTSS
COMMON/STS/STSV
COMMON/INF1/APSX,APSY,APSXY
COMMON/IYP/NPLY
COMMON/INV/AINV

```

C
C
C

COMPUTE ELEMENT STIFFNESS COEFFICIENTS

```

IF(J.GT.1) GO TO 200
DO 7 III=1,10
7 IC(III)=III
DO 100 IN1=1,3
DO 100 IN2=1,7
DO 100 IN3=1,400
100 PHI(IN1,IN2,IN3)=0.000
DO 444 N1=1,10
AO(N1,1)=0.000
AO(N1,2)=0.000
AO(N1,3)=0.000
DO 444 N2=1,7
444 AN(N1,N2)=0.000
DO 110 IN1=1,10
DO 110 IN2=1,10
SINF(IN1,IN2)=0.000
STEMP(IN1,IN2)=0.000

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110 FINF(IN1,IN2)=0.000
200 CONTINUE
C
C STRESSES AND DISPLACEMENTS ARE STORED
C FOR EACH LOAD CASE
DO 2107 KLK=1,8
2107 AN(KLK,J)=UVOUT(KLK)
IF(NRHK.EQ.5) GO TO 2221
IF(J.EQ.1) UVOUT(5)=UVOUT(9)
IF(J.EQ.1) UVOUT(7)=UVOUT(9)
IF(J.EQ.2) UVOUT(2)=UVOUT(16)
IF(J.EQ.2) UVOUT(8)=UVOUT(16)
IF(J.EQ.3) UVOUT(1)=UVOUT(13)
IF(J.EQ.3) UVOUT(3)=UVOUT(16)
IF(J.EQ.4) UVOUT(4)=UVOUT(12)
IF(J.EQ.4) UVOUT(6)=UVOUT(12)
2221 CONTINUE
IF(NRHK.EQ.7 AND J.LT.5) GO TO 371
AN(9,J)=(UVOUT(9)+UVOUT(13))/2.
AN(10,J)=(UVOUT(12)+UVOUT(16))/2.
GO TO 372
371 AN(9,J)=UVOUT(7+2*J)
AN(10,J)=UVOUT(8+2*J)
372 CONTINUE
DO 15 IS=1,NSTS
15 STSA(IS,J)=STGV(IS)
DO 10 IS=1,NGPT
PHI(1,J,IS)=APSX(IS)
10 CONTINUE
DO 20 IS=1,NGPT
PHI(2,J,IS)=APSY(IS)
20 CONTINUE
DO 30 IS=1,NGPT
PHI(3,J,IS)=APSYX(IS)
30 CONTINUE
IF(J.LT.NRHK) RETURN

INTEGRATION OF STRESSES
DO 1010 III=1,10
1010 CONTINUE
NTR=NRHK+3
DO 45 IK=1,NTR
DO 45 JK=1,NTR
45 FINF(IK,JK)=0.
H1=H/VJ)*NPLY(KJ)
DO 50 LI=1,NGPT
DO 50 LJ=1,3
DO 50 KI=1,NRHK
SUM=0.
DO 70 IL=1,3
SUM=SUM+H1*AINV(LJ,IL)*PHI(IL,KI,LI)
70 CONTINUE
STEMP(LJ,KI)=SUM
60 CONTINUE
DO 80 LK=1,NRHK
DO 80 LJ=1,NRHK
SUM=0.
DO 90 IL=1,3

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SUM=SUM+PHI(IL,LK,LI)*STEMP(IL,LJ)	00017360
90 CONTINUE	00017370
FINF(LK,LJ)=FINF(LK,LJ)+SUM*WQHT(LI)	00017380
80 CONTINUE	00017390
50 CONTINUE	00017400
DO 51 III=1, NRNK	00017410
DO 51 JJJ=1, NRNK	00017420
51 STMP(III, JJJ)=(FINF(III, JJJ)+FINF(JJJ, III))/2.0D0	00017430
DO 52 III=1, NRNK	00017440
DO 52 JJJ=1, NRNK	00017450
52 FINF(III, JJJ)=STEMP(III, JJJ)	00017460
CALL LINV2F(FINF, N, NK, 10, SINF, 4, WK, IER)	00017470
DO 410 IA=1, NTR	00017480
AO(IA, 1)=0.5+0.5*(-1.0)**(IA+1)	00017490
AO(IA, 2)=0.5+0.5*(-1.0)**(IA)	00017500
410 AO(IA, 3)=0.0D0	00017510
AO(1, 3)=DABS(YC(4))	00017520
AO(2, 3)=-DABS(XC(4))	00017530
AO(3, 3)=-DABS(YC(3))	00017540
AO(4, 3)=-DABS(XC(3))	00017550
AO(5, 3)=-DABS(YC(2))	00017560
AO(6, 3)=DABS(XC(2))	00017570
AO(7, 3)=DABS(YC(1))	00017580
AO(8, 3)=DABS(XC(1))	00017590
DO 420 KK=1, NTR	00017600
DO 420 LL=1, NRNK	00017610
SUM=0.0D0	00017620
DO 430 JJ=1, NRNK	00017630
430 SUM=SUM+AN(KK, JJ)*SINF(JJ, LL)	00017640
420 STMP(KK, LL)=SUM	00017650
DO 440 KK=1, NTR	00017660
N1=NTR-2	00017670
DO 440 JJ=N1, NTR	00017680
440 STMP(KK, JJ)=AO(KK, JJ-NRNK)	00017690
CALL LINV2F(STMP, NTR, 10, FINF, 4, WK, IER)	00017700
DO 450 II=1, NRNK	00017710
DO 450 JJ=1, NTR	00017720
SUM=0.0D0	00017730
DO 460 KK=1, NRNK	00017740
460 SUM=SUM+SINF(II, KK)*FINF(KK, JJ)	00017750
450 STMP(II, JJ)=SUM	00017760
DO 470 II=1, NTR	00017770
DO 470 JJ=1, NTR	00017780
SUM=0.0D0	00017790
DO 480 KK=1, NRNK	00017800
480 SUM=SUM+FINF(KK, II)*STEMP(KK, JJ)	00017810
ELSTFF(I, II, JJ)=SUM	00017820
A(II, JJ)=SUM	00017830
470 CONTINUE	00017840
DO 550 II=1, NRNK	00017850
DO 550 JJ=1, NTR	00017860
SUM=0.0D0	00017870
DO 560 KK=1, NRNK	00017880
560 SUM=SUM+SINF(II, KK)*FINF(KK, JJ)	00017890
550 STMP(II, JJ)=SUM	00017900
DO 570 II=1, NST3	00017910
DO 570 JJ=1, NTR	00017920
SUM=0.0D0	00017930
DO 580 KK=1, NRNK	00017940
580 SUM=SUM+A(II, KK)*STEMP(KK, JJ)	00017950

570	ELSTOP(I,II,1)=SUM	00017960
	RETURN	00017970
	END	00017980
C		00017990
C		00018000
C		00018010
	SUBROUTINE SMAX(HT,KJ,1)	00018020
	IMPLICIT REAL*8(A-H,O-Z)	00018030
	DIMENSION AINV(3,3),STULT(3,2),AVN(3)	00018040
	DIMENSION NV(3)	00018050
	DIMENSION PSMX(50,4),STM(3),CM(2)	00018060
	DIMENSION NPLY(2),NUMPLY(2),ANG(5,2),IPLY(100,2)	00018070
	DIMENSION E1(2),E2(2),Q12(2),V12(2),V21(2)	00018080
	COMMON/MD/E1,E2,Q12,V12,V21	00018090
	COMMON/LYP/NPLY,NUMPLY,ANG,IPLY	00018100
	COMMON/STMT/STM,CM	00018110
	COMMON/INV/AINV	00018120
	COMMON/SMX/PSMX	00018130
	COMMON/STN/STULT	00018140
	DATA CM/0.0	00018150
	IF(CM(KJ).EQ.0) GO TO 222	00018160
	PSMX(1,1)=STM(1)	00018170
	PSMX(1,2)=STM(2)	00018180
	PSMX(1,3)=STM(3)	00018190
	PSMX(1,4)=STM(2)	00018200
	RETURN	00018210
222	CONTINUE	00018220
C		00018230
C		00018240
C		00018250
	COMPUTE LAMINATE FAILURE LOADS BASED ON MAXIMUM	00018260
	FIBER STRAINS FOR EACH FAILURE MODE	00018270
	DO 100 K=1,3	00018280
	DO 10 II=1,3	00018290
	NV(II)=0	00018300
10	AVN(II)=0.000	00018310
	IF(K.EQ.1) NV(1)=1	00018320
	IF(K.EQ.2) NV(1)=-1	00018330
	IF(K.EQ.3) NV(3)=1	00018340
	DO 15 II=1,3	00018350
	DO 15 JJ=1,3	00018360
	AVN(II)=AVN(II)+AINV(II,JJ)*NV(JJ)	00018370
15	CONTINUE	00018380
	NP=NUMPLY(KJ)	00018390
	SMX=0.000	00018400
	PAO=DARCOS(-1.000)/180.00	00018410
	DO 25 II=1,NP	00018420
	TH=ANG(II,KJ)*RAD	00018430
	E1=DCOS(TH)*2*AVN(1)+AVN(2)*DSIN(TH)*2+	00018440
	*DCOS(TH)*DSIN(TH)*AVN(3)	00018450
	IF(K.EQ.1) GO TO 65	00018460
	EPRT=E1/STULT(2,KJ)	00018470
	GO TO 50	00018480
65	IF(K.EQ.2) GO TO 75	00018490
	EPRT=E1/STULT(1,KJ)	00018500
	GO TO 50	00018510
75	EPRT=E1/STULT(2,KJ)	00018520
50	CONTINUE	00018530
	IF(DABS(SMX).LT.DABS(EPRT)) SMX=EPRT	00018540
25	CONTINUE	00018550
	IF(DABS(SMX).GT.1.00-10) GO TO 555	

	PSMX(I,K)=STULT(3,KJ)*Q12(KJ)	00018560
	GO TO 100	00018570
555	CONTINUE	00018580
	PSMX(I,K)=DABS(1.0D0/SMX)	00018590
100	CONTINUE	00018600
	PSMX(I,4)=PSMX(1,2)	00018610
	RETURN	00018620
	END	00018630
		00018640
		00018650
		00018660
	SUBROUTINE POLY(W,AST,J,K,NCOL,LTNCM)	00018670
		00018680
		00018690
	IMPLICIT REAL*8(A-H,O-Z)	00018700
	DIMENSION XC(5),YC(5),A1(200),A2(200),XB(200)	00018710
	DIMENSION YB(200),T(200),A1A(4),A2A(4)	00018720
	COMMON/CHT1/XB,YB,A1,A2,T	00018730
	COMMON/XCYC/XC,YC	00018740
		00018750
	ARRAY COLLOCATION POINTS AROUND EXTERIOR	00018760
	BOUNDARY AND APPLY STRESS BOUNDARY	00018770
	CONDITION	00018780
		00018790
	DO 120 I=1,4	00018800
	A1A(I)=0.0	00018810
	A2A(I)=0.0	00018820
120	CONTINUE	00018830
	IF(LTNCM.EQ.1) A1A(I)=AST	00018840
	IF(LTNCM.EQ.2) A1A(I)=-AST	00018850
	J=0	00018860
	XC(5)=XC(1)	00018870
	YC(5)=YC(1)	00018880
	DO 10 I=1,4	00018890
	X=XC(I)-XC(I+1)	00018900
	Y=YC(I+1)-YC(I)	00018910
	IF(X.EQ.0.) X=1.D-6	00018920
	IF(Y.EQ.0.) Y=1.D-6	00018930
	TH=DATAH2(X,Y)	00018940
	TH=TH*180./DARCOS(-0.1D1)	00018950
	DX=(XC(I+1)-XC(I))/(NCOL+1)	00018960
	DY=(YC(I+1)-YC(I))/(NCOL+1)	00018970
	DO 20 II=1,NCOL	00018980
	J=J+1	00018990
	IF(I.EQ.1.OR.I.EQ.3) GO TO 23	00019000
	YB(J)=YC(I)	00019010
	XB(J)=XC(I)+DX*(II+.5)	00019020
	IF(II.EQ.1) XB(J)=XC(I)+(DX/2.)	00019030
	GO TO 24	00019040
23	CONTINUE	00019050
	YB(J)=YC(I)+DY*(II+.5)	00019060
	IF(II.EQ.1) YB(J)=YC(I)+(DY/2.)	00019070
	XB(J)=XC(I)	00019080
24	T(J)=TH	00019090
	A1(J)=A1A(I)	00019100
	A2(J)=A2A(I)	00019110
20	CONTINUE	00019120
10	CONTINUE	00019130
	RETURN	00019140
	END	00019150

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ARRAY COLLOCATION POINTS AROUND INNER BOUNDARY AND
APPLY BEARING LOAD IN A COSINUSOIDAL DISTRIBUTION

SUBROUTINE FIGECM(M,PHS,KJ,NOPT4,NCLL)

INTERNET FORUM - EVIDENCE GATHERING ANALYSIS BY A
TECHNICAL COLLECTION TECHNOLOGY

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C			00020360
C		CONSTANTS P1,P2,Q1,Q2 ARE NEEDED FOR STRESS CALCULATIONS	00020370
C		P1=AI(1,1)*R1**2+AI(1,2)-AI(1,3)*R1	00020380
		P2=AI(1,1)*R2**2+AI(1,2)-AI(1,3)*R2	00020390
		Q1=AI(2,2)/R1+AI(1,2)*R1-AI(2,3)	00020400
		Q2=AI(2,2)/R2+AI(1,2)*R2-AI(2,3)	00020410
C			00020420
C		INPUTS AIN1(I),AIN2(I) ETC. REFER TO BOUNDARY CONDITIONS	00020430
		NT4=4*NT	00020440
		NT8=8*NT	00020450
		NT8P4=8*NT+4	00020460
		NT8P2=8*NT+2	00020470
		NT8P1=8*NT+1	00020480
		NB2=2*NB	00020490
		NWK=NT8P1*(NT8P1+2)	00020500
		CALL CMAT(BC,CMCTCM,CMC,CM,RHS,GRHS,NT4,NT8,NT8P4,NT8P2,	00020510
		NT8P1,NB2,NWK,WA,NKK,NOPT4,KJ,NCLL)	00020520
		RETURN	00020530
		END	00020540
C			00020550
C			00020560
C			00020570
C		SUBROUTINE AMATRX(H,PHS,K)	00020580
C		ASSEMBLE THE A MATRIX	00020590
C			00020600
		IMPLICIT REAL*8(A-H,O-Z)	00020610
		DIMENSION A(3,3),ANO(5,2),H(2),NPLY(2),NUMPLY(2)	00020620
		DIMENSION E1(2),E2(2),Q12(2),V12(2),V21(2)	00020630
		DIMENSION IPLY(100,2)	00020640
		COMMON/MD/E1,E2,Q12,V12,V21	00020650
		COMMON/AMT/A	00020660
		COMMON/LYP/NPLY,NUMPLY,ANO,IPLY	00020670
		THKNES=NPLY(K)*H(K)	00020680
		DENO=1.-E2(K)*V12(K)*H2/E1(K)	00020690
		Q11=E1(K)/DENO	00020700
		Q22=E2(K)/DENO	00020710
		Q12=V12(K)*Q22	00020720
		Q21=Q12	00020730
		Q33=Q12(K)	00020740
		DO 10 I=1,3	00020750
		DO 10 J=1,3	00020760
10		A(I,J)=0.0D0	00020770
		NN=NPLY(K)	00020780
		T=H(K)	00020790
		DO 20 I=1,NN	00020800
		LP=IPLY(I,K)	00020810
		THTAI=(ANO(LP,K)+PHS)*DARCOS(-1.D0)/180.D0	00020820
		C=DCOS(THTAI)	00020830
		S=DSIN(THTAI)	00020840
		A(1,1)=(Q11*CMH4+2.*(Q12+2.*Q33)*CMCHS+S+Q22*SMH4)*T+A(1,1)	00020850
		A(2,2)=(Q11*SMH4+2.*(Q12+2.*Q33)*CMCHS+S+Q22*CMH4)*T+A(2,2)	00020860
		A(1,2)=((Q11+Q12-4.*Q33)*CMCHS+Q12*(CMH4+SMH4))*T+A(1,2)	00020870
		A(2,1)=A(1,2)	00020880
		A(3,3)=((Q11+Q22-2.*Q12-2.*Q33)*CMCHS+S+Q33*(CMH4+SMH4))*T+A(3,3)	00020890
		A(1,3)=((Q11-Q12-2.*Q33)*CMCHS+(Q12-Q22+2.*Q33)*SMH3*CM)*T+A(1,3)	00020900
		A(2,3)=((Q11-Q12-2.*Q33)*SMH3*CM+(Q12-Q22+2.*Q33)*CMH3*SM)*T+A(2,3)	00020910
		A(3,2)=A(2,3)	00020920
			00020930
			00020940
			00020950

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B=BX
CO=(0.0,1.0)
RB11=(Q1-P1*R1)/(A-CO*R1*B)
RB21=(Q2-P2*R2)/(A-CO*R2*B)
REALR1=R1
REALR2=R2
REALP1=P1
REALP2=P2
REALQ1=Q1
REALQ2=Q2
RRB11=RB11
RRB21=RB21
AIMGR1=CO*R1
AIMGR2=CO*R2
AIMGP1=CO*P1
AIMGP2=CO*P2
AIMGQ1=CO*Q1
AIMGQ2=CO*Q2
ARB11=CO*RB11
ARB21=CO*RB21
R1B=DCMPLX(REALR1,AIMGR1)
R2B=DCMPLX(REALR2,AIMGR2)
P1B=DCMPLX(REALP1,AIMGP1)
P2B=DCMPLX(REALP2,AIMGP2)
Q1B=DCMPLX(REALQ1,AIMGQ1)
Q2B=DCMPLX(REALQ2,AIMGQ2)
RB11B=DCMPLX(RRB11,ARB11)
RB21B=DCMPLX(RRB21,ARB21)
JJJ=0
DO 1000 I=1,NB
J=J+2
THTAI=THTA(I)*DARCO5(-1.000)/180.00
C=DCOS(THTAI)
S=DSIN(THTAI)
P11=C*P1+S*Q1
P12=C*P2+S*Q2
P21=-S*P1+C*Q1
P22=-S*P2+C*Q2
T11=(C*C*R1/R1+S*S-2.*C*S*R1)
T12=(C*C*R2/R2+S*S-2.*C*S*R2)
T21=(-C*C*R1/R1+C*S-(C*C-S*S)*R1)
T22=(-C*C*R2/R2+C*S-(C*C-S*S)*R2)
Z1=X(I)+R1*Y(I)
Z2=X(I)+R2*Y(I)
Z11=CDSORT(Z1*Z1-A*A-R1*R1*B*B)
Z22=CDSORT(Z2*Z2-A*A-R2*R2*B*B)
REAL1=Z11
AIMG1=-CO*Z11
IF(DABS(REAL1).LE.1.D-14)REAL1=0.000
IF(DABS(AIMG1).LE.1.D-16)AIMG1=0.000
Z11=DCMPLX(REAL1,AIMG1)
REAL2=Z22
AIMG2=-CO*Z22
IF(DABS(REAL2).LE.1.D-16)REAL2=0.000
IF(DABS(AIMG2).LE.1.D-16)AIMG2=0.000
Z22=DCMPLX(REAL2,AIMG2)
XETA1=(Z1+Z11)/(A-CO*R1*B)
IF(CDABS(XETA1).LT.0.999) GO TO 300
GO TO 310
300 Z11=-Z11
  
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XETA1=(Z1+Z11)/(A-COXR1*B)
310 XETA2=(Z2+Z22)/(A-COXR2*B)
    IF(CDABS(XETA2).LT.0.999) GO TO 320
    GO TO 330
320 Z22=-Z22
    XETA2=(Z2+Z22)/(A-COXR2*B)
330 CONTINUE
    JJJ=JJJ+1

C
C
C    NORMAL & TANGENTIAL STRESS BOUNDARY CONDITIONS ARE IMPOSED

    DO 5 N=1,NT
    NP=N
    CM(J-1,N)=NP*XETA1*NNPNT11/Z11
    CM(J-1,2*NT+N)=NP*XETA2*NNPNT12/Z22
    CM(J,N)=NP*XETA1*NNPNT21/Z11
    CM(J,2*NT+N)=NP*XETA2*NNPNT22/Z22
    NN=-N
    CM(J-1,NT+N)=NN*XETA1*NNNNT11/Z11
    CM(J-1,3*NT+N)=NN*XETA2*NNNNT12/Z22
    CM(J,NT+N)=NN*XETA1*NNNNT21/Z11
    CM(J,3*NT+N)=NN*XETA2*NNNNT22/Z22
5    CONTINUE
    CM(J-1,NT8+1)=T11/Z11
    CM(J-1,NT8+2)=T12/Z22
    CM(J,NT8+1)=T21/Z11
    CM(J,NT8+2)=T22/Z22
1000 CONTINUE
    DO 195 I=1,NB2
    DO 196 J=1,NT4
    REAL1=CM(I,J)
    AIMG1=-COXCM(I,J)
    IF(DABS(REAL1).LE.1.D-16)REAL1=0.0D0
    IF(DABS(AIMG1).LE.1.D-16)AIMG1=0.0D0
    CM(I,J)=DCMPLX(REAL1,AIMG1)
    AIMG2=-AIMG1
    CM(I,NT4+J)=DCMPLX(REAL1,AIMG2)
196 CONTINUE
195 CONTINUE
    DO 295 I=1,NB2
    DO 296 J=1,2
    REAL1=CM(I,NT8+J)
    AIMG1=-COXCM(I,NT8+J)
    IF(DABS(REAL1).LE.1.D-16)REAL1=0.0D0
    IF(DABS(AIMG1).LE.1.D-16)AIMG1=0.0D0
    CM(I,NT8+J)=DCMPLX(REAL1,AIMG1)
    AIMG2=-AIMG1
    CM(I,NT8+2+J)=DCMPLX(REAL1,AIMG2)
296 CONTINUE
295 CONTINUE
    SV11=(P2*Q1B-Q2*Q1B)/(Q1*P2-Q2*P1)
    SV12=(P2*Q2B-Q2*P2B)/(Q1*P2-Q2*P1)
    SV21=(Q1*P1B-Q1*P1B)/(Q1*P2-Q2*P1)
    SV22=(Q1*P2B-Q2*P1B)/(Q1*P2-Q2*P1)
    DO 139 I=1,NB2

C
C
C    IMPOSE RIGID BODY ROTATION CONDITION

    CM(I,2*NT+1)=-CM(I,1)*RB21/RB11+CM(I,2*NT+1)
    CM(I,4*NT+1)=-CM(I,1)*RB11B/RB11+CM(I,4*NT+1)

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CM(I,6*NT+1)=-CM(I,1)*RB21B/R311+CM(I,6*NT+1)
CM(I,1)=(0.0,0.0)
IMPOSE SINGLE-VALUEDNESS CONDITION
CM(I,NT8+3)=CM(I,NT8+1)*SV11+CM(I,NT8+3)
CM(I,NT8+4)=CM(I,NT8+1)*SV12+CM(I,NT8+4)
CM(I,NT8+3)=CM(I,NT8+2)*SV21+CM(I,NT8+3)
CM(I,NT8+4)=CM(I,NT8+2)*SV22+CM(I,NT8+4)
CM(I,NT8+1)=(0.0,0.0)
CM(I,NT8+2)=(0.0,0.0)
139 CONTINUE
DO 141 I=1,NB2
DO 142 J=2,NT8
142 CM(I,J-1)=CM(I,J)
CM(I,NT8)=CM(I,NT8+3)
CM(I,NT8+1)=CM(I,NT8+4)
141 CONTINUE
DO 95 I=1,NB2
DO 96 J=1,NT8P1
REAL1=CM(I,J)
AIM01=-CM(I,J)
IF(DABS(REAL1).LE.1.D-16)REAL1=0.0D0
IF(DABS(AIM01).LE.1.D-16)AIM01=0.0D0
CM(I,J)=DCMPLX(REAL1,AIM01)
AIM02=-AIM01
CM(I,J)=DCMPLX(REAL1,AIM02)
96 CONTINUE
95 CONTINUE
DO 100 I=1,NT8P1
DO 100 J=1,NT8P1
CSUM=(0.0,0.0)
DO 110 K=1,NB2
110 CSUM=CMC(K,I)*CM(K,J)+CSUM
CMCTCM(I,J)=CSUM
100 CONTINUE
3335 CONTINUE
DO 120 I=1,NB
J=I+2
BC(J-1)=AIN1(I)
120 BC(J)=AIN2(I)
DO 130 I=1,NT8P1
CSUM=(0.0,0.0)
DO 140 K=1,NB2
140 CSUM=CMC(K,I)*BC(K)+CSUM
130 RHS(I)=CSUM
IJOB=0
IF(NOPY4.EQ.5.AND.NCASE.GT.1) IJGB=2
M=1
CALL LEQ2C(CMCTCM,NT8P1,NT8P1,RHS,M,NT8P1,IJOB,WA,WKK,IER)
IF(IER.EQ.129) WRITE(6,11)
11 FORMAT(' TERMINAL ERROR(CMCTCM), IER = 129')
GRHS(1)=-(RHS(2*NT)*RB21+RHS(4*NT)*RB11B+RHS(6*NT)*RB21B)/RB11
GRHS(8*NT+1)=RHS(8*NT)*SV11+RHS(8*NT+1)*SV12
GRHS(8*NT+2)=RHS(8*NT)*SV21+RHS(8*NT+1)*SV22
DO 151 I=2,NT8
151 GRHS(I)=RHS(I-1)
C
C
C STRESS AND STRAIN CALCULATION

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RAD=DARCOS(-1.000)/180.00
IC=1
IC2=1
SUMU1=0.000
SUMV1=0.000
SUMU2=0.000
SUMV2=0.000
NADD=0
IF(NOPT4.EQ.1) GO TO 1195
IF(NTYPE.NE.2.OR.NCASE.GT.4) GO TO 1196
NADD=4*NCUL
NIC=1
DO 197 II=1,NADD
  XOUT(NOUT+NSTS+II)=X(II)
197 YOUT(NOUT+NSTS+II)=Y(II)
  MN=NOUT+NSTS+NADD
  XOUT(MN+1)=AX
  YOUT(MN+1)=0.000
  XOUT(MN+2)=0.000
  YOUT(MN+2)=-AX
  XOUT(MN+3)=AX*DCOS(177.00*RAD)
  YOUT(MN+3)=AX*DSIN(177.00*RAD)
  XOUT(MN+4)=0.000
  YOUT(MN+4)=AX
  NADD=NADD+4
  GO TO 1195
1196 CONTINUE
  NADD=8
  NIC=2
  MN=NOUT+NSTS
  DO 199 III=1,4
    ICM=5-III
    XOUT(MN+III)=XC(ICM)
    YOUT(MN+III)=YC(ICM)
199 CONTINUE
    MN=MN+4
    XOUT(MN+1)=AX
    YOUT(MN+1)=0.000
    XOUT(MN+2)=0.000
    YOUT(MN+2)=-AX
    XOUT(MN+3)=AX*DCOS(177.00*RAD)
    YOUT(MN+3)=AX*DSIN(177.00*RAD)
    XOUT(MN+4)=0.000
    YOUT(MN+4)=AX
1195 CONTINUE
    NRCF=NOUT
    IF(NOPT4.EQ.5) NRCF=NOUT+NSTS+NADD
    NINC=NSTS/4
    DO 190 K=1,NRCF
      Z1=XOUT(K)+R1*YOUT(K)
      Z2=XOUT(K)+R2*YOUT(K)
      Z11=CDSQRT(Z1*Z1-AXA-R1*NR1*NB)
      Z22=CDSQRT(Z2*Z2-AXA-R2*NR2*NB)
      XETA1=(Z1+Z11)/(A-CO*NR1*NB)
      IF(CDABS(XETA1).LT.0.999) GO TO 400
      GO TO 410
400 Z11=-Z11
      XETA1=(Z1+Z11)/(A-CO*NR1*NB)
410 XETA2=(Z2+Z22)/(A-CO*NR2*NB)
      IF(CDABS(XETA2).LT.0.999) GO TO 420

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GO T) 430
420 Z22=-Z22
XETA1:=(Z2+Z22)/(A-C0*R2*B)
430 CONTINUE
PHI1DP=(0.0,0.0)
PHI2DP=(0.0,0.0)
PHI1DN=(0.0,0.0)
PHI2DN=(0.0,0.0)
PHI1P=(0.0,0.0)
PHI2P=(0.0,0.0)
PHI1N=(0.0,0.0)
PHI2N=(0.0,0.0)
DO 170 N=1,NT
NP=N
NN=-N
PHI1DP=NP*XETA1*NN*GRHS(N)/Z11+PHI1DP
PHI1DN=NN*XETA1*NN*GRHS(N)/Z11+PHI1DN
PHI2DP=NP*XETA2*NN*GRHS(2*NT+N)/Z22+PHI2DP
PHI2DN=NN*XETA2*NN*GRHS(3*NT+N)/Z22+PHI2DN
PHI1P=XETA1*NN*GRHS(N)+PHI1P
PHI1N=XETA1*NN*GRHS(NT+N)+PHI1N
PHI2P=XETA2*NN*GRHS(2*NT+N)+PHI2P
PHI2N=XETA2*NN*GRHS(3*NT+N)+PHI2N
170 CONTINUE
PHI1D=PHI1DP+PHI1DN+GRHS(3*NT+1)/Z11
PHI2D=PHI2DP+PHI2DN+GRHS(8*NT+2)/Z22
PHI1=PHI1P+PHI1N+GRHS(8*NT+1)*CDLOG(XETA1)
PHI2=PHI2P+PHI2N+GRHS(8*NT+2)*CDLOG(XETA2)
SGMAX=2.*(R1*R1*PHI1D+R2*R2*PHI2D)
SGMAY=2.*(PHI1D+PHI2D)
SGMAXY=-2.*(R1*PHI1D+R2*PHI2D)
EPSX=AMAT(1,1)*SGMAX+AMAT(1,2)*SGMAY+AMAT(1,3)*SGMAXY
EPSY=AMAT(2,1)*SGMAX+AMAT(2,2)*SGMAY+AMAT(2,3)*SGMAXY
EPSXY=AMAT(3,1)*SGMAX+AMAT(3,2)*SGMAY+AMAT(3,3)*SGMAXY
U=2.*(P1*PHI1+P2*PHI2)
V=2.*(Q1*PHI1+Q2*PHI2)
PI=DARCO5(-1.00)
IF(XOUT(K).GT.0..AND.YOUT(K).GT.0.)
+TETAA=DATAN(YOUT(K)/XOUT(K))*180./PI
IF(XOUT(K).LT.0..AND.YOUT(K).GT.0.)
+TETAA=DATAN(YOUT(K)/XOUT(K))*180./PI+180.
IF(XOUT(K).LT.0..AND.YOUT(K).LT.0.)
+TETAA=DATAN(YOUT(K)/XOUT(K))*180./PI+180.
IF(YOUT(K).LT.0..AND.XOUT(K).GT.0.)
+TETAA=DATAN(YOUT(K)/XOUT(K))*180./PI+360.
C=DCOS(TETAA*PI/180.)
S=DSIN(TETAA*PI/180.)
SGMAR=C**2*SGMAX+S**2*SGMAY+2.*C*S*SGMAXY
SGMAT=S**2*SGMAX+C**2*SGMAY-2.*C*S*SGMAXY
SGMART=-C*S*SGMAX+C*S*SGMAY+(C**2-S**2)*SGMAXY
EPSR=C**2*EPSX+S**2*EPSY+C*S*EPSXY
EPST=S**2*EPSX+C**2*EPSY-C*S*EPSXY
EPSRT=2.*(-C*S*EPSX+C*S*EPSY+(C**2-S**2)*(EPSXY/2.))
UR=UXC+VXS
IF(NOPT4.EQ.5) GO TO 3338
KTHT(K)=TETAA
REPX(K)=EPSX
REPY(K)=EPSY
REPLY(K)=EPSXY
ASX(K)=SGMAX
00023960
00023970
00023980
00023990
00024000
00024010
00024020
00024030
00024040
00024050
00024060
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00024380
00024390
00024400
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00024450
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00024550

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ASXY(K)=SGMAXY	00024560
FUR(X)=UR	00024570
FTHT(K)=TETAA	00024580
FSMR(K)=SGMAR	00024590
3338 CONTINUE	00024600
IF(NOPT4.EQ.5.AND.K.GT.NOUT) GO TO 3339	00024610
APSX(K)=SGMAX	00024620
APSY(K)=SGMAY	00024630
APSY(K)=SGMAXY	00024640
3339 CONTINUE	00024650
IF(NOPT4.EQ.1) GO TO 190	00024660
IF(NOPT4.EQ.5.AND.K.LE.NOUT.OR.K.GT.(NOUT+NSTS)) GO TO 191	00024670
IF(IC2.LE.NINC) STSV(IC2)=SGMAX	00024680
IF(IC2.GT.NINC.AND.IC2.LE.(2*NINC)) STSV(IC2)=SGMAXY	00024690
IF(IC2.GT.(2*NINC).AND.IC2.LE.(3*NINC)) STSV(IC2)=SGMAX	00024700
IF(IC2.GT.(3*NINC).AND.IC2.LE.(4*NINC)) STSV(IC2)=SGMAX	00024710
IC2=IC2+1	00024720
GO TO 190	00024730
191 CONTINUE	00024740
IF(NIC.EQ.1) GO TO 192	00024750
IF(NOPT4.EQ.5.AND.X.LT.(NRCF-7)) GO TO 190	00024760
UVOUT(IC)=U	00024770
UVOUT(IC+1)=V	00024780
IC=IC+2	00024790
GO TO 190	00024800
192 CONTINUE	00024810
NC=NOUT+NSTS	00024820
IF(K.GT.NC.AND.K.LE.(NC+NCOL)) SUMU1=SUMU1+U	00024830
IF(K.GT.(NC+NCOL).AND.K.LE.(NC+2*NCOL)) SUMV1=SUMV1+V	00024840
IF(K.GT.(NC+2*NCOL).AND.K.LE.(NC+3*NCOL)) SUMU2=SUMU2+U	00024850
IF(K.GT.(NC+3*NCOL).AND.K.LE.(NC+4*NCOL)) SUMV2=SUMV2+V	00024860
NNC=NC+4*NCOL	00024870
IF(K.EQ.(NNC+1)) UVOUT(9)=U	00024880
IF(K.EQ.(NNC+1)) UVOUT(10)=V	00024890
IF(K.EQ.(NNC+2)) UVOUT(11)=U	00024900
IF(K.EQ.(NNC+2)) UVOUT(12)=V	00024910
IF(K.EQ.(NNC+3)) UVOUT(13)=U	00024920
IF(K.EQ.(NNC+3)) UVOUT(14)=V	00024930
IF(K.EQ.(NNC+4)) UVOUT(15)=U	00024940
IF(K.EQ.(NNC+4)) UVOUT(16)=V	00024950
190 CONTINUE	00024960
DISPLACEMENTS ARE AVERAGED OVER ELEMENT SIDES FOR	00024970
CERTAIN LOAD CASES	00024980
IF(NIC.NE.1) RETURN	00024990
SUMU1=SUMU1/FLOAT(NCOL)	00025000
SUMV1=SUMV1/FLOAT(NCOL)	00025010
SUMU2=SUMU2/FLOAT(NCOL)	00025020
SUMV2=SUMV2/FLOAT(NCOL)	00025030
UVOUT(1)=SUMU2	00025040
UVOUT(2)=SUMV2	00025050
UVOUT(3)=SUMU2	00025060
UVOUT(4)=SUMV1	00025070
UVOUT(5)=SUMU1	00025080
UVOUT(6)=SUMV1	00025090
UVOUT(7)=SUMU1	00025100
UVOUT(8)=SUMV2	00025110
RETURN	00025120
END	00025130
	00025140
	00025150

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SUBROUTINE FBOLT(ANGK,H,PSH,K)

FBOLT CALCULATES THE INDIVIDUAL PLY FOUNDATION
MODULI AND THE INDIVIDUAL PLY LOADS

IMPLICIT REAL*8(A-H,O-Z)
DIMENSION ATETAA(400),ANG(5,2),ASIGR(400),ASIGRT(400),H(2)
DIMENSION ASIG1(400),ASIG2(400),ASIG6(400),UR(400),ANGK(5,2)
DIMENSION FSMR(400),PLXPT(100)
DIMENSION IPLY(100,2),NPLY(2),NUMPLY(2)
DIMENSION FKI(100),PLX(100)
DIMENSION E11(2),E22(2),ESS(2),PMU12(2),PMU21(2)
COMMON/STRESS/ASIGR,ASIGRT,ASIG1,ASIG2,ASIG6
COMMON/ELP/AX,BX,HOUT
COMMON/FB1/BSTR,YSTR
COMMON/LYP/NPLY,NUMPLY,ANG,IPLY
COMMON/FB2/UR,ATETAA,FSMR
COMMON/MOD/E11,E22,ESS,PMU12,PMU21
COMMON/FCT/PLXPT
RAD=DARCOS(-0.1D1)/180.
THKTOT=NPLY(K)*H(K)
NN=NUMPLY(K)

CALCULATE DELEFF

WORK=0.
PLOADX=0.
IF(K.EQ.1) PLD=0.
DO 210 KK=1,NOUT
TH1=ATETAA(KK+1)-ATETAA(KK)
TH2=(ATETAA(KK)+ATETAA(KK+1))/2.
THETA=TH2*RAD
C=DCOS(THETA)
S=DSIN(THETA)
R=DSQRT(1./((C**2/AX**2+S**2/BX**2))
FORCE=((FSMR(KK)+FSMR(KK+1))/2.)*R*TH1*RAD*THKTOT
WORK=WORK+FORCE*.5*((UR(KK)+UR(KK+1))/2.)
PLOADX=PLOADX+FORCE*C

210 CONTINUE
PLD=PLD+PLOADX
DELEF=WORK/PLOADX

COMPUTE PLY STRESSES FROM LAMINATE STRAINS

(SIGMA)R,0,R0 = (Q)*(EPS)R,0,R0

NN=NPLY(K)
DO 100 J=1,NN
LP=IPLY(J,K)
THETA=(ANG(LP,K)+PSH)*RAD
I1=1

00025160
00025170
00025180
00025190
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00025680
00025690
00025700
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00025750

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LI2=NOUT
 NCAS=1
 CALL QMATX(RAD,THETA,K,LI1,LI2,NCAS)

INTEGRATE AROUND CIRCULAR BOUNDARY FOR
 INDIVIDUAL PLY LOADS AND COMPUTE FOUNDATION
 MODULI

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NNH=LI2-1
PLOADX=0.
WK=0.
DO 70 I=LI1,NNH
  TH1=ATETAA(I+1)-ATETAA(I)
  TH2=(ATE1AA(I)+ATETAA(I+1))/2.
  THETA=TH2*H*AD
  C=DCOS(THETA)
  S=DSIN(THETA)
  R=DSQRT(1./(C**2/AX**2+S**2/BX**2))
  FORCR=((ASIGR(I)+ASIGR(I+1))/2.)*R*TH1*RAD*H(K)
  FORCRT=((ASIORT(I)+ASIORT(I+1))/2.)*R*TH1*RAD*H(K)
  PLOADX=PLOADX+FORCR*C-FORCRT*S
70 CONTINUE
FKI(J)=DABS(PLOADX/(H(K)*DELEFF))
PLX(J+(K-1)*NPLY(1))=I*LOADX
100 CONTINUE
NT=NPLY(K)
NN=NPLY(K)
DO 310 I=1,NT
  DO 310 II=1,NN
    IF(IPLY(II,K).EQ.I) A'IGK(I,K)=FKI(II)
    IF(IPLY(II,K).EQ.I) PLXPT(II)=PLX(II+(K-1)*NPLY(1))
310 CONTINUE
NP=NPLY(K)
DO 311 I=1,NP
  AA=ANG(I,K)+PSH
311 CONTINUE
PLXTOT=0.000
IF(K.EQ.1) BLOAD=0.
TH=H(K)*NPLY(K)
BLOAD=(BSTR*DARCOS(-1.000)*BX*TH)/2.+BLOAD
IF(K.EQ.1) GO TO 611
NN=NPLY(1)+NPLY(2)
DO 212 I=1,NN
  PLXTOT=PLXTOT+PLX(I)
212 CONTINUE
611 CONTINUE
RETURN
END

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SUBROUTINE QMATX(RAD,THETA,K,LI1,LI2,NCAS)

QMATX PERFORMS BASIC STRESS AND STRAIN
 TRANSFORMATIONS

IMPLICIT REAL*8(A-H,O-Z)
 DIMENSION ASIGR(400),ASIORT(400),ASIG1(400),ASIG2(400),ASIG6(400)

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DIMENSION ATETAA(400),AEP5X(400),AEP5Y(400),AEP5XY(400) 00026360
DIMENSION E11(2),E22(2),ESS(2),PMU12(2),PMU21(2),SX(400),SXY(400) 00026370
DIMENSION AEP51(400) 00026380
DIMENSION ASX(400),ASXY(400) 00026390
COMMON/XXY1/ASX,ASXY 00026400
COMMON/MOD/E11,E22,ESS,PMU12,PMU21 00026410
COMMON/STRESS2/AEP51 00026420
COMMON/STRESS/ASIOR,ASIOR1,ASIOR2,ASIOR3,ASIOR4 00026430
COMMON/CMT/ATETAA,AEP5X,AEP5Y,AEP5XY 00026440
COMMON/PSC3/SX,SXY 00026450
J=0 00026460
Q11=E11(K)/(1.0-PMU12(K)*PMU21(K)) 00026470
Q12=(PMU21(K)*E11(K))/(1.0-PMU12(K)*PMU21(K)) 00026480
Q22=E22(K)/(1.0-PMU12(K)*PMU21(K)) 00026490
Q66=ESS(K) 00026500
C=DCOS(THETA) 00026510
S=DSIN(THETA) 00026520
BQ11=Q11*(C**4)+(2.*Q12*(2.*Q66))*C*(S**2)+(Q22*(S**4)) 00026530
BQ12=-(Q11+Q22-(4.*Q66))*C*(S**2)*(C**2)+(Q12*(S**4+C**4)) 00026540
BQ14=-(Q11-Q22-(2.*Q66))*C*(S**3)*(C**2)+(Q12-Q22+(2.*Q66))*C*(S**3)*C 00026550
BQ22=Q11*(S**4)+(2.*Q12*(2.*Q66))*C*(S**2)*(C**2)+(Q22*(C**4)) 00026560
BQ26=-(Q11-Q22-(2.*Q66))*C*(S**3)*(C**2)+(Q12-Q22+(2.*Q66))*C*(S**3)*C 00026570
BQ46=-(Q11+Q22-(2.*Q12+Q66))*C*(S**2)*(C**2)+(Q66*(C**4)+(S**4)) 00026580
DO 40 I=1,11,112 00026590
J=J+1 00026600
IF(NCAS.EQ.1) THETA=ATETAA(I)*RAD 00026610
C=DCOS(THETA) 00026620
S=DSIN(THETA) 00026630
SIGX=BQ11*AEP5X(I)+BQ12*AEP5Y(I)+BQ14*AEP5XY(I) 00026640
SIGY=BQ12*AEP5X(I)+BQ22*AEP5Y(I)+BQ26*AEP5XY(I) 00026650
SIGXY=BQ14*AEP5X(I)+BQ26*AEP5Y(I)+BQ46*AEP5XY(I) 00026660
SX(J)=SIGX 00026670
SXY(J)=SIGY 00026680
ASIGR(I)=SIGX*C**2+SIGY*S**2+2.*SIGXY*S*C 00026690
ASIGRT(I)=-SIGX*S*C+SIGY*C*S+SIGXY*(C**2-S**2) 00026700
ASIG1(J)=SIGX*C**2+SIGY*S**2+2.*SIGXY*S*C 00026710
ASIG2(J)=SIGX*S**2+SIGY*C**2-2.*SIGXY*S*C 00026720
ASIG6(J)=-C*S*SIGX+S*C*SIGY+(C**2-S**2)*SIGXY 00026730
AEP51(J)=AEP5X(I)*C**2+AEP5Y(I)*S**2+AEP5XY(I)*S*C 00026740
40 CONTINUE 00026750
RETURN 00026760
END 00026770

SUBROUTINE CENTD(H,FASSS,FASBS,P) 00026780
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00026890
00026900
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00026930
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00026950
IMPLICIT REAL*8(A-H,O-Z)
DIMENSION PLYK(100),BARK(100),BARU(100),F(100)
DIMENSION H(2),RF(2)
DIMENSION AII(100,100),A(2),B(2)
DIMENSION NPLY(2)
COMMON/PBB/PLYK,BARK,BARU
COMMON/RT/RF
COMMON/AFM/AII,F

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	COMMON/LYP/NPLY	00026960
	NNN=NPLY(1)+NPLY(2)	00026970
	SET UP THE CENTRAL DIFFERENCE EQUATIONS	00026980
	DO 3 I=1,100	00026990
	DO 3 J=1,100	00027000
	3 AII(I,J)=0.	00027010
	NECESSARY CONSTANTS ARE FORMED	00027020
	DO 7 I=1,...	00027030
	A(I)=H(I)**2/FASSS	00027040
	7 B(I)=H(I)**4/FASBS	00027050
	H12=H(1)/H(2)	00027060
	A1=H(1)**2/FASSS	00027070
	A2=H(2)**2/FASSS	00027080
	NP=NPLY(1)+NPLY(2)	00027090
	SHEAR AT TOP OF PANEL EQUALS ZERO	00027100
	AII(1,1)=1.	00027110
	AII(1,2)=-2.+A1*PLYK(2)	00027120
	AII(1,4)=2.+A1*PLYK(2)	00027130
	AII(1,5)=-1.	00027140
	F(1)=0.0	00027150
	MOMENT CONDITION AT TOP	00027160
	IF(RF(1).OE.1.D10) GO TO 50	00027170
	Z=1.	00027180
	R=RF(1)	00027190
	GO TO 60	00027200
50	Z=0.	00027210
	R=1.	00027220
60	AII(2,1)=R	00027230
	AII(2,2)=(Z**2.*H(1)*FASSS)*RM*(-2.-A1*PLYK(2)+(H(1)*H2	00027240
	*FASSS)/FASBS)	00027250
	AII(2,3)=-Z*(4.*H(1)*FASSS+(2*H(1)*H2*PLYK(1)*H(1)))	00027260
	AII(2,4)=Z**2.*H(1)*FASSS+RM*(2.+A1*PLYK(2)-(H(1)*H2	00027270
	*FASSS)/FASBS)	00027280
	AII(2,5)=-R	00027290
	F(2)=Z**2.*H(1)*H3*BARK(1)*BARU(1)	00027300
	GOVERNING EQUATIONS FOR THE TOP PLATE	00027310
	N2=NPLY(1)	00027320
	DO 55 J=1,N2	00027330
	I=J+2	00027340
	AII(I,J)=1.	00027350
	IF(J.EQ.1) GO TO 56	00027360
	AII(I,J+1)=-4.-A(1)*PLYK(J-1)	00027370
	GO TO 57	00027380
56	AII(I,J+1)=-4.-A(1)*PLYK(2)	00027390
57	AII(I,J+2)=6.+(2.*A(1)+B(1))*PLYK(J)	00027400
	IF(J.EQ.N2) GO TO 61	00027410
	AII(I,J+3)=-4.-A(1)*PLYK(J+1)	00027420
	GO TO 62	00027430
61	AII(I,J+3)=-4.-A(1)*PLYK(NPLY(1)-1)	00027440
		00027450
		00027460
		00027470
		00027480
		00027490
		00027500
		00027510
		00027520
		00027530
		00027540
		00027550

	62	ALL(I,J+4)=1.	00027560
		IF(J.EQ.1) GO TO 58	00027570
		IF(J.EQ.N2) GO TO 63	00027580
		F(I)=A(1)*BARK(J-1)*BARU(J-1)	00027590
		M=(2.*A(1)+B(1))*BARK(J)*BARU(J)	00027600
		M+A(1)*BARK(J+1)*BARU(J+1)	00027610
		GO TO 59	00027620
	58	F(I)=2.*A(1)*BARK(2)*BARU(2)	00027630
		M=(2.*A(1)+B(1))*BARK(1)*BARU(1)	00027640
		GO TO 59	00027650
	63	F(I)=2.*A(1)*BARK(NPLY(1)-1)*BARU(NPLY(1)-1)	00027660
		M=(2.*A(1)+B(1))*BARK(J)*BARU(J)	00027670
	59	CONTINUE	00027680
	55	CONTINUE	00027690
C		INTERFACE SHEAR ON TOP PLATE = P	00027700
C		I=NPLY(1)+3	00027710
		J=NPLY(1)	00027720
		ALL(I,J)=1.	00027730
		ALL(I,J+1)=-((2.+A1*PLYK(NPLY(1)-1))	00027740
		ALL(I,J+3)=2.+A1*PLYK(NPLY(1)-1)	00027750
		ALL(I,J+4)=-1.	00027760
		F(I)=(-2.*H(1)*M3*(P))/FASBS	00027770
C		SLOPE CONTINUITY	00027780
C		I=NPLY(1)+4	00027790
		J=NPLY(1)	00027800
		ALL(I,J)=1.	00027810
		ALL(I,J+1)=-((2.+A1*PLYK(NPLY(1)-1)-H(1)*M2*FASBS/FASBS)	00027820
		ALL(I,J+3)=2.+A1*PLYK(NPLY(1)-1)-H(1)*M2*FASBS/FASBS	00027830
		ALL(I,J+4)=-1.	00027840
		ALL(I,J+5)=-H12*M3	00027850
		ALL(I,J+6)=H12*M3*(2.+A2*PLYK(NPLY(1)+2)-H(2)*M2*FASBS/FASBS)	00027860
		ALL(I,J+8)=-H12*M3*(2.+A2*PLYK(NPLY(1)+2)-H(2)*M2*FASBS/FASBS)	00027870
		ALL(I,J+9)=H12*M3	00027880
		F(I)=0.	00027890
C		MOMENT CONTINUITY	00027900
C		I=NPLY(1)+5	00027910
		J=NPLY(1)+1	00027920
		ALL(I,J)=1.	00027930
		ALL(I,J+1)=-((2.+A1*PLYK(NPLY(1)))	00027940
		ALL(I,J+2)=1.	00027950
		ALL(I,J+5)=-H12*M2	00027960
		ALL(I,J+6)=H12*M2*(2.+A2*PLYK(NPLY(1)+1))	00027970
		ALL(I,J+7)=-H12*M2	00027980
		F(I)=A1*(BARK(NPLY(1))*BARU(NPLY(1))-BARK(NPLY(1)+1)*	00027990
		BARU(NPLY(1)+1))	00028000
C		INTERFACE SHEAR ON BOTTOM PLATE	00028010
C		I=NPLY(1)+6	00028020
		J=NPLY(1)+5	00028030
		ALL(I,J)=-1.	00028040
		ALL(I,J+1)=(2.+A2*PLYK(NPLY(1)+2))	00028050
		ALL(I,J+3)=-((2.+A2*PLYK(NPLY(1)+2))	00028060
		ALL(I,J+4)=1.	00028070
			00028080
			00028090
			00028100
			00028110
			00028120
			00028130
			00028140
			00028150

```

C
C
C
F(I)=2.*H(2)*NPLY(1)*NPLY(2)
GOVERNING EQUATIONS FOR THE BOTTOM PLATE
N1=NPLY(1)+7
N2=NPLY(1)+NPLY(2)+6
DO 70 I=N1,N2
  J=I-2
  AII(I,J)=1.
  IF(I.EQ.N1) GO TO 71
  AII(I,J-1)=-4.-A(2)*PLYK(J-5)
  AII(I,J)=2.
71 AII(I,J+1)=-4.-A(2)*PLYK(NPLY(1)+2)
72 AII(I,J+2)=6.+(2.*A(2)+B(2))*PLYK(J-4)
  IF(I.EQ.N2) GO TO 73
  AII(I,J-3)=-4.-A(2)*PLYK(J-3)
  AII(I,J-2)=-4.-A(2)*PLYK(J-5)
75 AII(I,J+3)=4.-A(2)*PLYK(J-5)
76 AII(I,J+4)=1.
  IF(I.EQ.N1) GO TO 73
  IF(I.EQ.N2) GO TO 77
  F(I)=A(2)*BARK(J-5)*BARU(J-5)
  X=(2.*A(2)+B(2))*BARK(J-4)*BARU(J-4)
  Y=A(2)*BARK(J-3)*BARU(J-3)
  GO TO 74
73 F(I)=2.*A(2)*BARK(NPLY(1)+2)*BARU(NPLY(1)+2)
  X=(2.*A(2)+B(2))*BARK(J-4)*BARU(J-4)
  GO TO 74
77 F(I)=2.*A(2)*BARK(J-5)*BARU(J-5)
  X=(2.*A(2)+B(2))*BARK(J-4)*BARU(J-4)
74 CONTINUE
70 CONTINUE
C
C
C
SHEAR ON BOTTOM PLATE EQUALS ZERO
NP=NPLY(1)+NPLY(2)
I=NP+7
J=NP+4
AII(I,J)=-1.
AII(I,J+1)=(2.+A2)*PLYK(NP-1)
AII(I,J+3)=-2.+A2*PLYK(NP-1)
AII(I,J+4)=1.
F(I)=0.
C
C
C
MOMENT BOUNDARY CONDITION ON BOTTOM PLATE
I=NP+8
IF(RF(2).GE.1.D10) GO TO 85
Z=1.
R=RF(2)
GO TO 95
85 Z=0.
R=1.
95 AII(I,J)=-R
  AII(I,J+1)=Z*(2.*H(2)*FASSS)+R*(2.+A2*PLYK(NP-1))
  X=H(2)*M2*FASSS/FASBS
  AII(I,J+2)=-Z*(4.*H(2)*FASSS+2.*H(2)*M3*PLYK(NP))
  AII(I,J+3)=Z*2.*H(2)*FASSS+R*(-2.-A2*PLYK(NP-1))
  X=H(2)*M2*FASSS/FASBS
  AII(I,J+4)=R
  
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	F(I)=Z*(2.*H(2)*X3+JARK(NP)*BARU(NP))	00028760
	RETURN	00028770
	END	00028780
		00028790
	SUBROUTINE SOLVE(H,P,U1,U2)	00028800
		00028810
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		00028840
		00028850
	IMPLICIT REAL*8(A-H,O-Z)	00028860
	DIMENSION A(100,100),B(100),NPLY(2),U(100),F(100)	00028870
	DIMENSION SX(100),PLYX(100),H(2)	00028880
	DIMENSION JARK(100),BARU(100)	00028890
	COMMON/PLP/NPLY	00028900
	COMMON/AFMA,A,F	00028910
	COMMON/P33/PLYX,JARK,BARU	00028920
		00028930
	SOLUTION OF THE SYSTEM: PA(U)=(3)	00028940
	NP=NPLY(1)+NPLY(2)+8	00028950
	DO 444 I=1,NP	00028960
444	B(I)=F(I)	00028970
		00028980
		00028990
	APPLYING GUASSIAN ELIMINATION TO THE	00029000
	MATRIX OF COEFFICIENTS	00029010
		00029020
	DO 2001 I=1,NP	00029030
	IR=I	00029040
2042	IF(A(IR,I).NE.0.) GO TO 2041	00029050
	IR=IR+1	00029060
	IF(IR.GT.NP) GO TO 2001	00029070
	GO TO 2042	00029080
2041	NN=IR+1	00029090
	DO 2002 L=NN,NP	00029100
	IF(DABS(A(L,I)).GT.1.D-30) GO TO 2009	00029110
	A(L,I)=0.	00029120
	GO TO 2002	00029130
2009	CF=-A(IR,I)/A(L,I)	00029140
	DO 2003 J=I,NP	00029150
	A(L,J)=A(L,J)+CF*A(IR,J)	00029160
	IF(DABS(A(L,J)).LT.1.D-30) A(L,J)=0.0	00029170
2003	CONTINUE	00029180
	B(L)=B(L)+CF*B(I)	00029190
2002	CONTINUE	00029200
2001	CONTINUE	00029210
		00029220
	BACK SUBSTITUTION	00029230
		00029240
	DO 2011 I=1,NP	00029250
	L=NP+1-I	00029260
	SUM=0.	00029270
	IF(A(L,L).EQ.0.) GO TO 2112	00029280
	N=L+1	00029290
	IF(N.GT.NP) GO TO 2013	00029300
	DO 2013 J=N,NP	00029310
	SUM=SUM-A(L,J)*SX(J)	00029320
2013	CONTINUE	00029330
	SX(L)=(B(L)+SUM)/A(L,L)	00029340
	GO TO 2011	00029350

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2112 CONTINUE
      SX(L)=0.
2011 CONTINUE
      PT=P
      N1=NPLY(1)+2
      N2=NPLY(1)+7
      NN=NPLY(1)+NPLY(2)+6
      DO 1444 I=3,N1
      J=I-2
      U(J)=SX(I)
1444 CONTINUE
      DO 1555 I=N2,NN
      J=I-6
      U(J)=SX(I)
1555 CONTINUE
      NP=NPLY(1)+NPLY(2)

      COMPUTE AVERAGE RELATIVE DISPLACEMENTS
      IN TOP AND BOTTOM PLATES

      U1=DABS(U(1)-U(NPLY(1)))/2.
      U2=DABS(U(NPLY(1)+1)-U(NPLY(1)+NPLY(2)))/2.
      RETURN
      END

      SUBROUTINE FCRT(APP,NEL1,NEL2,NDAM,IN,LTNCM,NAVD)
      IMPLICIT REAL*8(A-H,O-Z)
      DIMENSION ELSTFF(50,10,10),ELCTSS(50,50,10),U(200)
      DIMENSION OSSX(20),OSSW(20)
      DIMENSION NELDIS(50,5,2)
      DIMENSION PSMX(50,4),AVES(50,3),STRSS(50),DLT(10)
      DIMENSION ELFAIL(50,3),NELTYP(50)
      DIMENSION NELCON(50,6),NELCHA(50,6),NPLY(2)
      DIMENSION ELWDTH(50),ELTHK(50),ELLOAD(50,2)
      DIMENSION NELPLS(2,50),LYPN(50)
      COMMON/ELS/ELSTFF,ELSTSS
      COMMON/OSXW/OSSX,OSSW
      COMMON/NPLS/NELPLS,LYPN
      COMMON/NCH/NELCON,NELCHA,NELDIS
      COMMON/FCC/ELWDTH,ELTHK,ELLOAD
      COMMON/SMX/PSMX
      COMMON/LAMF/ELFAIL
      COMMON/DISP/U
      COMMON/NTP/NELTYP
      COMMON/LYP/NPLY

      DETERMINE ELEMENT FAILURE LOADS IN NET SECTION,
      SHEAROUT AND BEARING, AND LOCATE THE CRITICAL
      FASTENER LOCATION. JOINT STRENGTH IS DETERMINED
      FROM LOWEST ELEMENT FAILURE LOAD

      NELTOT=NEL1+NEL2
      NS=NAVD
      NSTS=4*NAVD
      DO 10 I=1,NELTOT
      NRNK=10
      IF(NELTYP(I).EQ.3) NRNK=8
      KJ=I
  
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IF(KJ.GT.NEL) KJ=KJ-NEL	C0029960
IF(NELTYP(I).EQ.1) GO TO 10	00029970
IC=0	00029980
DO 20 J=1,5	00029990
IC=IC+1	00030000
DLT(IC)=U(NELDIS(I,J,1))	00030010
IC=IC+1	00030020
DLT(IC)=U(NELDIS(I,J,2))	00030030
20 CONTINUE	00030040
DO 30 K=1,NSTS	00030050
SUM=0.000	00030060
DO 40 K2=1,NRNK	00030070
SUM=SUM+ELSTS3(I,K,K2)*DLT(K2)	00030080
40 CONTINUE	00030090
30 STRSS(K)=SUM	00030100
SUM1=0.000	00030110
SUM2=0.000	00030120
SUM3=0.000	00030130
SUM4=0.000	00030140
DO 50 J=1,NS	00030150
SUM1=SUM1+STRSS(J)	00030160
SUM2=SUM2+STRSS(J+NS)	00030170
SUM3=SUM3+STRSS(J+2*NS)	00030180
50 CONTINUE	00030190
NR=2*NS	00030200
DO 51 II=1,NS	00030210
51 SUM4=SUM4+STRSS(II+NS)*MOSSW(II)	00030220
AVES(I,1)=SUM1/NS	00030230
AVES(I,2)=SUM2/NS	00030240
AVES(I,3)=SUM3/NS	00030250
IF(I.LE.NEL1) THK=ELTHK(I)*NELPLS(1,LYPN(I))	00030260
IF(I.GT.NEL1) THK=ELTHK(I)*NELPLS(2,LYPN(I))	00030270
ELD=(SUM4/2.00)*ELHDTH(I)*THK	00030280
PRATIO=DABS((ELLOAD(I,1)+ELLOAD(I,2))/ELD)	00030290
C C C SCALE AVERAGE STRESSES	00030300
AVES(I,1)=AVES(I,1)*PRATIO	00030310
AVES(I,2)=AVES(I,2)*PRATIO	00030320
AVES(I,3)=AVES(I,3)*PRATIO	00030330
10 CONTINUE	00030340
C C C COMPUTE JOINT FAILURE LOADS BASED ON ELEMENT LOADS	00030350
DO 100 I=1,NELTOT	00030360
IF(NELTYP(I).EQ.1) GO TO 100	00030370
DO 110 J=1,3	00030380
N=J+1	00030390
IF(J.EQ.1.AND.LTNCM.EQ.1) N=1	00030400
IF(J.EQ.1.AND.LTNCM.EQ.2) N=2	00030410
ELFAIL(I,J)=DABS(APP*PSMX(I,N)/AVES(I,J))	00030420
110 CONTINUE	00030430
100 CONTINUE	00030440
C C C SEARCH FOR LOWEST JOINT FAILURE LOAD	00030450
INNS=0	00030460
FNS=1.0D10	00030470
INCO=0	00030480
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C	SUBROUTINE LEQT2F (A,M,N,IA,B,IDOT,WKAREA,IER)	00031160
C	DIMENSION A(IA,1),B(IA,1),WKAREA(1)	00031170
C	DOUBLE PRECISION A,B,WKAREA,D1,D2,WA	00031180
C	FIRST EXECUTABLE STATEMENT	00031190
C	INITIALIZE IER	00031200
	IER=0	00031210
	JER=0	00031220
	J = NXN+1	00031230
	K = J+N	00031240
	MM = K+N	00031250
	KK = 0	00031260
	MM1 = MM-1	00031270
	JJ=1	00031280
	DO 5 L=1,N	00031290
	DO 5 I=1,N	00031300
	WKAREA(JJ)=A(I,L)	00031310
	JJ=JJ+1	00031320
5	CONTINUE	00031330
C	DECOMPOSE A	00031340
	CALL LUDATN (WKAREA,N,N,A,IA,IDOT,D1,D2,WKAREA(J),WKAREA(K),	00031350
	WA,IER)	00031360
	IF (IER.GT.128) GO TO 25	00031370
	IF (IDOT.EQ. 0 .OR. IER .NE. 0) KK = 1	00031380
	DO 15 I = 1,M	00031390
C	PERFORMS THE ELIMINATION PART OF	00031400
C	AX = B	00031410
C	CALL LUELMM (A,IA,N,B(1,I),WKAREA(J),WKAREA(MM))	00031420
C	REFINEMENT OF SOLUTION TO AX = B	00031430
	IF (KK .NE. 0)	00031440
	CALL LUREFN (WKAREA,N,N,A,IA,B(1,I),IDOT,WKAREA(J),WKAREA(MM),	00031450
	WKAREA(K),WKAREA(K),JER)	00031460
	DO 10 II=1,N	00031470
	B(II,I) = WKAREA(MM1+II)	00031480
10	CONTINUE	00031490
	IF (JER.NE.0) GO TO 20	00031500
15	CONTINUE	00031510
	GO TO 25	00031520
20	IER = 131	00031530
25	JJ=1	00031540
	DO 30 J = 1,N	00031550
	DO 30 I = 1,M	00031560
	A(I,J)=WKARFA(JJ)	00031570
	JJ=JJ+1	00031580
30	CONTINUE	00031590
	IF (IER.EQ. 0) GO TO 9005	00031600
9000	CONTINUE	00031610
	CALL UERTST (IER,6HLEQT2F)	00031620
9005	RETURN	00031630
	END	00031640
C		00031650
C		00031660
C		00031670
C		00031680
C		00031690
	SUBROUTINE LUDATF (A,LU,N,IA,IDOT,D1,D2,IPVT,EQUIL,WA,IER)	00031700
	DIMENSION A(IA,1),IU(IA,1),IPVT(1),EQUIL(1)	00031710
	DOUBLE PRECISION A,LU,D1,D2,EQUIL,WA,ZERO,ONE,FOUR,SIXTN,SIXTH,	00031720
	RN,WREL,BIGA,BIG,P,SUM,AI,WI,T,TEST,Q	00031730
	DATA ZERO,ONE,FOUR,SIXTN,SIXTH/0.DO,1.DO,4.DO,	00031740
		00031750

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      16.DO,.0625DO/
      FIRST EXECUTABLE STATEMENT
      INITIALIZATION
C
C
      IER = 0
      RN = N
      WREL = ZERO
      D1 = ONE
      D2 = ZERO
      BIGA = ZERO
      DO 10 I=1,N
        BIG = ZERO
        DO 5 J=1,N
          P = A(I,J)
          L(I,J) = P
          P = DABS(P)
          IF (P .GT. BIG) BIG = P
5      CONTINUE
      IF (BIG .GT. BIGA) BIGA = BIG
      IF (BIG .EQ. ZERO) GO TO 110
      EQUI(I) = ONE/BIG
10 CONTINUE
      DO 105 J=1,N
        JMI = J-1
        IF (JMI .LT. 1) GO TO 40
C
C      DO 35 I=1,JMI
        SUM = LU(I,J)
        IM1 = I-1
        IF (IDOT .EQ. 0) GO TO 25
C
C      AI = DABS(SUM)
        WI = ZERO
        IF (IM1 .LT. 1) GO TO 20
        DO 15 K=1,IM1
          T = LU(I,K)*LU(K,J)
          SUM = SUM-T
          WI = WI+DABS(T)
15      CONTINUE
        LU(I,J) = SUM
20      WI = WI+DABS(SUM)
        IF (AI .EQ. ZERO) AI = BIGA
        TEST = WI/AI
        IF (TEST .GT. WREL) WREL = TEST
        GO TO 35
C
C      IF (IM1 .LT. 1) GO TO 35
        DO 30 K=1,IM1
          SUM = SUM-LU(I,K)*LU(K,J)
30      CONTINUE
        LU(I,J) = SUM
35      CONTINUE
        P = ZERO
40
C
C      COMPUTE U(J,J) AND L(I,J), I=J+1,...
      DO 70 I=J,N
        SUM = LU(I,J)
        IF (IDOT .EQ. 0) GO TO 55
C
C      AI = DABS(SUM)
        WI = ZERO
        IF (JMI .LT. 1) GO TO 50

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    DC 45 K=1,JM1
      T = LU(I,K)*LU(K,J)
      SUM = SUM-T
      WI = WI+DABS(T)
45    CONTINUE
      LU(I,J) = SUM
50    WI = WI+DABS(SUM)
      IF (AI .EQ. ZERO) AI = BIOA
      TEST = WI/AY
      IF (TEST .GT. WREL) WREL = TEST
      GO TO 65
C    WITHOUT ACCURACY TEST
55    IF (JM .LT. 1) GO TO 65
      DO 60 K=1,JM1
        SUM = SUM-LU(I,K)*LU(K,J)
60    CONTINUE
      LU(I,J) = SUM
65    Q = EQUIL(I)*DABS(SUM)
      IF (P .GE. Q) GO TO 70
      P = Q
      IMAX = I
70    CONTINUE
C    TEST FOR ALGORITHMIC SINGULARITY
      IF (RH+P .EQ. RH) GO TO 110
      IF (J .EQ. IMAX) GO TO 80
C    INTERCHANGE ROWS J AND IMAX
      D1 = -D1
      DO 75 K=1,N
        P = LU(IMAX,K)
        LU(IMAX,K) = LU(J,K)
        LU(J,K) = P
75    CONTINUE
      EQUIL(IMAX) = EQUIL(J)
80    IPVT(J) = IMAX
      D1 = D1*LU(J,J)
85    IF (DABS(D1) .LE. ONE) GO TO 90
      D1 = D1*SIXTH
      D2 = D2+FOUR
      GO TO 85
90    IF (DABS(D1) .GE. SIXTH) GO TO 95
      D1 = D1*SIXTH
      D2 = D2-FOUR
      GO TO 90
95    CONTINUE
      JPI = J+1
      IF (JPI .GT. N) GO TO 105
C    DIVIDE BY PIVOT ELEMENT U(J,J)
      P = LU(J,J)
      DO 100 I=JPI,N
        LU(I,J) = LU(I,J)/P
100    CONTINUE
105    CONTINUE
C    PERFORM ACCURACY TEST
      IF (IDGT .EQ. 0) GO TO 9005
      P = 3*N+3
      WA = P*WREL
      IF (WA+10.D0**(-IDGT) .NE. WA) GO TO 9005
      IER = 34
      GO TO 9000
C    ALGORITHMIC SINGULARITY

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110	IER = 129	00032960
	D1 = ZERO	00032970
	D2 = ZERO	00032980
9000	CONTINUE	00032990
	PRINT ERRJR	00033000
	CALL UERTST(IER,6HLUDATF)	00033010
9005	RETURN	00033020
	END	00033030
C		00033040
C		00033050
	SUBROUTINE LUELMN (A,IA,N,B,APVT,X)	00033060
C		00033070
	DIMENSION A(IA,1),B(1),APVT(1),X(1)	00033080
	DOUBLE PRECISION A,B,X,SUM,APVT	00033090
C		00033100
	FIRST EXECUTABLE STATEMENT	00033110
	SOLVE LY = B FOR Y	00033120
	DO 5 I=1,N	00033130
5	X(I) = B(I)	00033140
	IW = 0	00033150
	DO 20 I=1,N	00033160
	IP = APVT(I)	00033170
	SUM = X(IP)	00033180
	X(IP) = X(I)	00033190
	IF (IW .EQ. 0) GO TO 15	00033200
	IM1 = I-1	00033210
	DO 10 J=IW,IM1	00033220
	SUM = SUM-A(I,J)*X(J)	00033230
10	CONTINUE	00033240
	GO TO 20	00033250
15	IF (SUM .NE. 0.D0) IW = I	00033260
20	X(I) = SUM	00033270
C		00033280
	SOLVE UX = Y FOR X	00033290
	DO 30 IB=1,N	00033300
	I = N+1-IB	00033310
	IP1 = I+1	00033320
	SUM = X(I)	00033330
	IF (IP1 .GT. N) GO TO 30	00033340
	DO 25 J=IP1,N	00033350
	SUM = SUM-A(I,J)*X(J)	00033360
25	CONTINUE	00033370
30	X(I) = SUM/A(I,I)	00033380
	RETURN	00033390
	END	00033400
C		00033410
C		00033420
	SUBROUTINE LUREFN (A,IA,N,UL,IUL,B,IDGT,APVT,X,RES,DX,IER)	00033430
C		00033440
	DIMENSION A(IA,1),UL(IUL,1),B(1),X(1),RES(1),DX(1)	00033450
	DIMENSION APVT(1)	00033460
	DIMENSION ACCXT(2)	00033470
	DOUBLE PRECISION A,ACCXT,B,UL,X,RES,DX,ZERO,XNORM,DXNORM,APVT	00033480
	DATA ITMAX/75/,ZERO/0.D0/	00033490
C		00033500
	FIRST EXECUTABLE STATEMENT	00033510
	IER=0	00033520
	XNORM = ZERO	00033530
	DO 10 I=1,N	00033540
	XNORM = DMAX1(XNORM,DABS(X(I)))	00033550
10	CONTINUE	
	IF (XNORM .NE. ZERO) GO TO 20	
	IDGT = 50	

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    IF (IEQDF.EQ.1) WRITE(IUNIT,40) IER,NAMEQ,IEQ,NAMUPK      00034160
    IF (IEQDF.EQ.1) WRITE(IUNIT,40) IER,NAMUPK                00034170
    GO TO 30                                                    00034180
10 IF (IER.LE.32) GO TO 15                                     00034190
C   PRINT WARNING MESSAGE                                     00034200
    IF (LEVEL.LT.3) GO TO 30                                    00034210
    IF (IEQDF.EQ.1) WRITE(IUNIT,45) IER,NAMEQ,IEQ,NAMUPK      00034220
    IF (IEQDF.EQ.0) WRITE(IUNIT,45) IER,NAMUPK                 00034230
    GO TO 30                                                    00034240
15 CONTINUE                                                    00034250
C   CHECK FOR UERSET CALL                                     00034260
    DO 20 I=1,6                                                 00034270
      IF (NAMUPK(I).NE.NAMSET(I)) GO TO 25                     00034280
20 CONTINUE                                                    00034290
    LEVOLD = LEVEL                                              00034300
    LEVEL = IER                                                 00034310
    IER = LEVOLD                                                00034320
    IF (LEVEL.LT.0) LEVEL = 4                                   00034330
    IF (LEVEL.GT.4) LEVEL = 4                                   00034340
    GO TO 30                                                    00034350
25 CONTINUE                                                    00034360
    IF (LEVEL.LT.4) GO TO 30                                    00034370
C   PRINT NON-DEFINED MESSAGE                                  00034380
    IF (IEQDF.EQ.1) WRITE(IUNIT,50) IER,NAMEQ,IEQ,NAMUPK      00034390
    IF (IEQDF.EQ.0) WRITE(IUNIT,50) IER,NAMUPK                 00034400
30 IEQDF = 0                                                    00034410
    RETURN                                                      00034420
35 FORMAT(19H *** TERMINAL ERROR,10X,7H( IER = ,I3,          00034430
1   20H) FROM IMSL ROUTINE ,6A1,A1,6A1)                      00034440
40 FORMAT(27H *** WARNING WITH FIX ERROR,2X,7H( IER = ,I3,    00034450
1   20H) FROM IMSL ROUTINE ,6A1,A1,6A1)                      00034460
45 FORMAT(18H *** WARNING ERROR,11X,7H( IER = ,I3,           00034470
1   20H) FROM IMSL ROUTINE ,6A1,A1,6A1)                      00034480
50 FORMAT(20H *** UNDEFINED ERROR,9X,7H( IER = ,I5,          00034490
1   20H) FROM IMSL ROUTINE ,6A1,A1,6A1)                      00034500
C   SAVE P FOR P = R CASE                                     00034510
C   P IS THE PAGE NAMUPK                                     00034520
C   R IS THE ROUTINE NAMUPK                                  00034530
55 IEQDF = 1                                                    00034540
    DO 60 I=1,6                                                 00034550
60 NAMEQ(I) = NAMUPK(I)                                         00034560
65 RETURN                                                       00034570
    END                                                         00034580
C   SUBROUTINE UGETIO(IOP1,NIN,NOUT)                          00034590
C   SPECIFICATIONS FOR ARGUMENTS                             00034600
C   INTEGER IOP1,NIN,NOUT                                     00034610
C   SPECIFICATIONS FOR LOCAL VARIABLES                       00034620
C   INTEGER NIND,NOUTD                                        00034630
C   DATA NIND/S/,NOUTD/S/                                  00034640
C   FIRST EXECUTABLE STATEMENT                              00034650
    IF (IOP1.EQ.3) GO TO 10                                     00034660
    IF (IOP1.EQ.2) GO TO 5                                     00034670
    IF (IOP1.EQ.1) GO TO 9005                                  00034680
    NIN = NIND                                                  00034690
    NOUT = NOUTD                                                00034700
    GO TO 9005                                                  00034710
    9005                                                        00034720
    9005                                                        00034730
    9005                                                        00034740
    9005                                                        00034750

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5 NIND = NIN
GO TO 9005
10 NOUTD = NOUT
9005 RETURN
END

```

C
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C

SUBROUTINE VXADD(A,ACC)

C
C
C

DOUBLE PRECISION A,ACC(2)

SPECIFICATIONS FOR ARGUMENTS

C
C
C

DOUBLE PRECISION X,Y,Z,ZZ

SPECIFICATIONS FOR LOCAL VARIABLES

FIRST EXECUTABLE STATEMENT

X = ACC(1)

Y = A

IF (DABS(ACC(1)).GE.DABS(A)) GO TO 1

X = A

Y = ACC(1)

C
C
C

1 Z = X+Y

ZZ = (X-Z)+Y

COMPUTE Z+ZZ = ACC(1)+A EXACTLY

COMPUTE ZZ+ACC(2) USING DOUBLE
PRECISION ARITHMETIC

C
C
C

ZZ = ZZ+ACC(2)

COMPUTE ACC(1)+ACC(2) = Z+ZZ EXACTLY

C
C
C

ACC(1) = Z+ZZ

ACC(2) = (Z-ACC(1))+ZZ

RETURN

END

C
C
C

SUBROUTINE VXMUL (A,B,ACC)

C
C
C

DOUBLE PRECISION A,B,ACC(2)

SPECIFICATIONS FOR ARGUMENTS

SPECIFICATIONS FOR LOCAL VARIABLES

DOUBLE PRECISION X,HA,TA,HB,TB

INTEGER IX(2),I

LOGICAL L1

EQUIVALENCE

DATA

LX(8),LI(4)

(X,LX(1),IX(1)),(I,LI(1))

I/O

SPLIT A = HA+TA

B = HB+TB

FIRST EXECUTABLE STATEMENT

C
C
C

X = A

LI(4) = LX(5)

IX(2) = 0

I = (I/16)*16

LX(5) = LI(4)

HA=X

TA=A-HA

X = B

LI(4) = LX(5)

IX(2) = 0

I = (I/16)*16

LX(5) = LI(4)

HB = X

TB = B-HB

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C		RADIX THE BASE OF THE FLOATING-POINT	00035960
C		NUMBER SYSTEM USED	00035970
	DATA	RINFP/27FFFFFFF/	00035980
	DATA	REPSP/200100000/	00035990
	DATA	RADIX/16.0/	00036000
	DATA	RE: 1/234100000000000000/	00036010
	DATA	ZERO/0.000/ONE/1.000/	00036020
C		ZRPOLY USES SINGLE PRECISION	00036030
C		CALCULATIONS FOR SCALING, BOUNDS	00036040
C		AND ERROR CALCULATIONS.	00036050
C		FIRST EXECUTABLE STATEMENT	00036060
	IER = 0		00036070
	IF (NDEG .GT. 100 .OR. NDEG .LT. 1) GO TO 165		00036080
	ETA = REPSR1		00036090
	AME = ETA		00036100
	RMRE = ETA		00036110
	RLO = REPSP/ETA		00036120
C		INITIALIZATION OF CONSTANTS FOR	00036130
C		SHIFT ROTATION	00036140
	XX = .7071068		00036150
	YY = -XX		00036160
	SINR = .9975641		00036170
	COSR = -.06975647		00036180
	N = NDEG		00036190
	NN = N+1		00036200
C		ALGORITHM FAILS IF THE LEADING	00036210
C		COEFFICIENT IS ZERO.	00036220
	IF (A(1).NE.ZERO) GO TO 5		00036230
	IER = 130		00036240
	GO TO 9000		00036250
C		REMOVE THE ZEROS AT THE ORIGIN IF	00036260
C		ANY	00036270
	5 IF (A(NN).NE.ZERO) GO TO 10		00036280
	J = NDEG-N+1		00036290
	JJ = J+NDEG		00036300
	Z(J) = ZERO		00036310
	Z(JJ) = ZERO		00036320
	NN = NN-1		00036330
	N = N-1		00036340
	IF (NN.EQ.1) GO TO 9005		00036350
	GO TO 5		00036360
C		MAKE A COPY OF THE COEFFICIENTS	00036370
	10 DO 15 I=1,NN		00036380
	P(I) = A(I)		00036390
	15 CONTINUE		00036400
C		START THE ALGORITHM FOR ONE ZERO	00036410
	20 IF (N.GT.2) GO TO 30		00036420
	IF (N.LT.1) GO TO 9005		00036430
C		CALCULATE THE FINAL ZERO OR PAIR OF	00036440
C		ZEROS	00036450
	IF (N.EQ.2) GO TO 25		00036460
	Z(NDEG) = -P(2)/P(1)		00036470
	Z(NDEG+NDEG) = ZERO		00036480
	GO TO 145		00036490
	25 CALL ZRPOLI (P(1),P(2),P(3),Z(NDEG-1),Z(NDEG+NDEG-1),Z(NDEG),		00036500
	1 Z(NDEG+NDEG))		00036510
	GO TO 145		00036520
		FIND LARGEST AND SMALLEST MODULI OF	00036530
		COEFFICIENTS.	00036540
	30 RMAX = 0.		00036550

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**ZERO OR ZEROS AND RETURN TO THE
MAIN ALGORITHM.**

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C      DOUBLE PRECISION P(101),QP(101),RK(101),QK(101),SVK(101) 00038360
C      DOUBLE PRECISION SR,S1,U,V,RA,RD,C,D,A1,A2,A3, 00038370
1      A6,A7,E,F,O,H,SZR,SZI,RLZR,RLZI, 00038380
2      SVU,SVV,UI,VI,S,ZERO 00038390
C      LOGICAL VPASS,SPASS,VTRY,STRY 00038400
C      COMMON /ZRPQLJ/ P,QP,RK,QK,SVK,SR,S1,U,V,RA,RD,C,D,A1,A2,A3,A6, 00038410
1      A7,E,F,O,H,SZR,SZI,RLZR,RLZI,ETA,ARE,RMRE,N,NH 00038420
C      DATA ZERO/0.0D0/ 00038430
C      NZ = 0 00038440
C      FIRST EXECUTABLE STATEMENT 00038450
C      COMPUTES UP TO L2 FIXED SHIFT 00038460
C      K-POLYNOMIALS, TESTING FOR 00038470
C      CONVERGENCE IN THE LINEAR OR 00038480
C      QUADRATIC CASE. INITIATES ONE OF 00038490
C      THE VARIABLE SHIFT ITERATIONS AND 00038500
C      RETURNS WITH THE NUMBER OF ZEROS 00038510
C      FOUND. 00038520
C      L2 - LIMIT OF FIXED SHIFT STEPS 00038530
C      NZ -NUMBER OF ZEROS FOUND 00038540
C      BETAV = .25 00038550
C      BETAS = .25 00038560
C      OSS = SR 00038570
C      OVV = V 00038580
C      EVALUATE POLYNOMIAL BY SYNTHETIC 00038590
C      DIVISION 00038600
C      CALL ZRPQLH (NH,U,V,P,QP,RA,RD) 00038610
C      CALL ZRPQLE (ITYPE) 00038620
C      DO 40 J=1,L2 00038630
C      CALCULATE NEXT K POLYNOMIAL AND 00038640
C      ESTIMATE V 00038650
C      CALL ZRPQLF (ITYPE) 00038660
C      CALL ZRPQLE (ITYPE) 00038670
C      CALL ZRPQLO (ITYPE,UI,VI) 00038680
C      VV = VI 00038690
C      ESTIMATE S 00038700
C      SS = 0. 00038710
C      IF (RK(N).NE.ZERO) SS = -P(NN)/RK(N) 00038720
C      TV = 1. 00038730
C      TS = 1. 00038740
C      IF (J.EQ.1.OR.ITYPE.EQ.3) GO TO 35 00038750
C      COMPUTE RELATIVE MEASURES OF 00038760
C      CONVERGENCE OF S AND V SEQUENCES 00038770
C      IF (VV.NE.0.) TV = ABS((VV-OVV)/VV) 00038780
C      IF (SS.NE.0.) TS = ABS((SS-USS)/SS) 00038790
C      IF DECREASING, MULTIPLY TWO MOST 00038800
C      RECENT CONVERGENCE MEASURES 00038810
C      TVV = 1. 00038820
C      IF (TV.LT.OTV) TVV = TV*OTV 00038830
C      TSS = 1. 00038840
C      IF (TS.LT.OTS) TSS = TS*OTS 00038850
C      COMPARE WITH CONVERGENCE CRITERIA 00038860
C      VPASS = TVV.LT.BETAV 00038870
C      SPASS = TSS.LT.BETAS 00038880
C      IF (.NOT.(SPASS.OR.VPASS)) GO TO 35 00038890
C      AT LEAST ONE SEQUENCE HAS PASSED THE 00038900
C      CONVERGENCE TEST. STORE VARIABLES 00038910
C      BEFORE ITERATING 00038920
C      SVU = U 00038930
C      SVV = V 00038940
C      DO 5 I=1,N 00038950

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5	SVK(I) = RK(I)		00038960
	S = SS		00038970
C		CHOOSE ITERATION ACCORDING TO THE	00038980
C		FASTEST CONVERGING SEQUENCE	00038990
	VTRY = .FALSE.		00039000
	STRY = .FALSE.		00039010
10	IF (SPASS.AND.(.NOT.VPASS).OR.TSS.LT.TVV)) GO TO 20		00039020
	CALL ZRPQLC (UI,VI,NZ)		00039030
	IF (NZ.GT.0) RETURN		00039040
C		QUADRATIC ITERATION HAS FAILED. FLAG	00039050
C		THAT IT HAS BEEN TRIED AND	00039060
C		DECREASE THE CONVERGENCE	00039070
		CRITERION.	00039080
	VTRY = .TRUE.		00039090
	BETAV = BETAV*.25		00039100
C		TRY LINEAR ITERATION IF IT HAS NOT	00039110
C		BEEN TRIED AND THE S SEQUENCE IS	00039120
C		CONVERGING	00039130
	IF (STRY.OR.(.NOT.SPASS)) GO TO 25		00039140
	DO 15 I=1,N		00039150
15	RK(I) = SVK(I)		00039160
20	CALL ZRPQLD (S,NZ,IFLAG)		00039170
	IF (NZ.GT.0) RETURN		00039180
C		LINEAR ITERATION HAS FAILED. FLAG	00039190
C		THAT IT HAS BEEN TRIED AND	00039200
		DECREASE THE CONVERGENCE CRITERION	00039210
	STRY = .TRUE.		00039220
	BETAS = BETAS*.25		00039230
	IF (IFLAG.EQ.0) GO TO 25		00039240
C		IF LINEAR ITERATION SIGNALS AN	00039250
C		ALMOST DOUBLE REAL ZERO ATTEMPT	00039260
C		QUADRATIC ITERATION	00039270
	UI = -(S+S)		00039280
	VI = S*S		00039290
	GO TO 10		00039300
C		RESTORE VARIABLES	00039310
25	U = SVU		00039320
	V = SVV		00039330
	DO 30 I=1,N		00039340
30	RK(I) = SVK(I)		00039350
C		TRY QUADRATIC ITERATION IF IT HAS	00039360
C		NOT BEEN TRIED AND THE V SEQUENCE	00039370
C		IS CONVERGING	00039380
	IF (VPASS.AND.(.NOT.VTRY)) GO TO 10		00039390
C		RECOMPUTE QP AND SCALAR VALUES TO	00039400
C		CONTINUE THE SECOND STAGE	00039410
	CALL ZRPQLH (NN,U,V,P,QP,RA,RB)		00039420
	CALL ZRPQLE (ITYPE)		00039430
35	QVV = VV		00039440
	QSS = SS		00039450
	QTV = TV		00039460
	QTS = TS		00039470
40	CONTINUE		00039480
	RETURN		00039490
	END		00039500
C			00039510
C			00039520
C			00039530
	SUBROUTINE ZRPQLC (UU,VV,NZ)		00039540
C		SPECIFICATIONS FOR ARGUMENTS	00039550

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C      INTEGER          NZ          00039560
      DOUBLE PRECISION  UU,VV      00039570
C      SPECIFICATIONS FOR LOCAL VARIABLES
      INTEGER          N,NN,J,I,ITYPE 00039580
      REAL             ARE,EE,ETA,OMP,RELSTP,RMP,RMRE,T,ZM 00039590
      DOUBLE PRECISION P(101),QP(101),RK(101),QK(101),SVK(101) 00039600
      DOUBLE PRECISION SR,SI,U,V,RA,RB,C,D,A1,A2,A3, 00039610
      1                A6,A7,E,F,O,H,SZR,SZI,RLZR,RLZI, 00039620
      2                UI,VI,ZERO,PT01,ONE 00039630
      LOGICAL          TRIED 00039640
      COMMON /ZRPQLJ/ P,QP,RK,QK,SVK,SR,SI,U,V,RA,RB,C,D,A1,A2,A3,A6, 00039650
      1                A7,E,F,O,H,SZR,SZI,RLZR,RLZI,ETA,ARE,RMRE,N,NN 00039660
      DATA            ZERO,PT01,ONE/0.000,0.0100,1.000/ 00039670
C      FIRST EXECUTABLE STATEMENT 00039680
      NZ = 0 00039690
C      VARIABLE-SHIFT K-POLYNOMIAL 00039700
C      ITERATION FOR A QUADRATIC FACTOR 00039710
C      CONVERGES ONLY IF THE ZEROS ARE 00039720
C      EQUIMODULAR OR NEARLY SO 00039730
C      UU,VV - COEFFICIENTS OF STARTING 00039740
C      QUADRATIC 00039750
C      NZ - NUMBER OF ZERO FOUND 00039760
      TRIED = .FALSE. 00039770
      U = UU 00039780
      V = VV 00039790
      J = 0 00039800
C      MAIN LOOP 00039810
C      3 CALL ZRPQLI (ONE,U,V,SZR,SZI,RLZR,RLZI) 00039820
C      RETURN IF ROOTS OF THE QUADRATIC ARE 00039830
C      REAL AND NOT CLOSE TO MULTIPLE OR 00039840
C      NEARLY EQUAL AND OF OPPOSITE SIGN 00039850
C      IF ( DABS(DABS(SZR)-DABS(RLZR)).GT.PT01*DABS(RLZR)) RETURN 00039860
C      EVALUATE POLYNOMIAL BY QUADRATIC 00039870
C      SYNTHETIC DIVISION 00039880
C      CALL ZRPQLH (NN,U,V,P,QP,RA,RB) 00039890
C      RMP = DABS(RA-SZR*RB)+DABS(SZI*RB) 00039900
C      COMPUTE A RIGOROUS BOUND ON THE 00039910
C      ROUNDING ERROR IN EVALUATING P 00039920
      ZM = SQRT(ABS(SNOL(V))) 00039930
      EE = 2.*ABS(SNOL(QP(1))) 00039940
      T = -SZR*RB 00039950
      DO 10 I=2,N 00039960
      10 EE = EE*ZM+ABS(SNOL(QP(I))) 00039970
      EE = EE*ZM+ABS(SNOL(RA)+T) 00039980
      EE = (5.*RMRE+4.*ARE)*EE-(5.*RMRE+2.*ARE)*M(ABS(SNOL(RA)+T)+ 00039990
      1    ABS(SNOL(RB))*ZM)+2.*ARE*ABS(T) 00040000
C      ITERATION HAS CONVERGED SUFFICIENTLY 00040010
C      IF THE POLYNOMIAL VALUE IS LESS 00040020
C      THAN 20 TIMES THIS BOUND 00040030
      IF (RMP.GT.20.*EE) GO TO 15 00040040
      NZ = 2 00040050
      RETURN 00040060
C      15 J = J+1 00040070
C      STOP ITERATION AFTER 20 STEPS 00040080
      IF (J.GT.20) RETURN 00040090
      IF (J.LT.2) GO TO 25 00040100
      IF (RELSTP.GT..01.OR.RMP.LT.OMP.OR.TRIED) GO TO 25 00040110
C      A CLUSTER APPEARS TO BE STALLING THE 00040120
C      CONVERGENCE. FIVE FIXED SHIFT 00040130
C      STEPS ARE TAKEN WITH A U,V CLOSE 00040140
      00040150
  
```

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C      IF (RELSTP.LT.EYA) RELSTP = ETA      TO THE CLUSTER      00040160
RELSTP = SQRT(RELSTP)                      00040170
U = U-UXRELSTP                             00040180
V = V+VXRELSTP                             00040190
CALL ZRPQLH (NN,U,V,P,QP,RA,RB)           00040200
DO 20 I=1,5                                00040210
    CALL ZRPQLE (ITYPE)                     00040220
    CALL ZRPQLF (ITYPE)                     00040230
20 CONTINUE                                00040240
    TRIED = .TRUE.                          00040250
    J = 0                                    00040260
25 OMP = RMP                                00040270
                                           00040280
C                                           CALCULATE NEXT K POLYNOMIAL AND NEW
C                                           U AND V      00040290
                                           00040300
    CALL ZRPQLE (ITYPE)                     00040310
    CALL ZRPQLF (ITYPE)                     00040320
    CALL ZRPQLE (ITYPE)                     00040330
    CALL ZRPQLO (ITYPE,UI,VI)               00040340
                                           00040350
C                                           IF VI IS ZERO THE ITERATION IS NOT
C                                           CONVERGING      00040360
                                           00040370
    IF (VI.EQ.ZERO) RETURN                  00040380
    RELSTP = DABS((VI-V)/VI)                00040390
    U = UI                                  00040400
    V = VI                                  00040410
    GO TO 5                                  00040420
END                                          00040430
                                           00040440
C                                           00040450
C                                           00040460
SUBROUTINE ZRPQLD (SSS,NZ,IFLAG)            00040470
C                                           SPECIFICATIONS FOR ARGUMENTS      00040480
INTEGER NZ,IFLAG                           00040490
DOUBLE PRECISION SSS                        00040500
C                                           SPECIFICATIONS FOR LOCAL VARIABLES
INTEGER N,NN,J,X                           00040510
REAL ARE,EE,ETA,OMP,RMP,RMS,RMRE           00040520
DOUBLE PRECISION P(101),QP(101),RK(101),QK(101),SVK(101) 00040530
DOUBLE PRECISION SR,SI,U,V,RA,RB,C,D,A1,A2,A3, 00040540
A6,A7,E,F,G,H,SZR,SZI,RLZR,RLZI,          00040550
1 PV,RKV,T,S,ZERO,PT001                    00040560
2 COMMON /ZRPQLJ/ P,QP,RK,QK,SVK,SR,SI,U,V,RA,RB,C,D,A1,A2,A3,A6, 00040570
A7,E,F,G,H,SZR,SZI,RLZR,RLZI,ETA,ARE,RMRE,N,NN 00040580
1 DATA ZERO/0.000/,PT001/0.00100/        00040590
                                           VARIABLE-SHIFT M POLYNOMIAL
                                           ITERATION FOR A REAL ZERO SSS -
                                           STARTING ITERATE      00040610
NZ - NUMBER OF ZERO FOUND                  00040620
IFLAG - FLAG TO INDICATE A PAIR OF        00040630
ZEROS NEAR REAL AXIS                      00040640
FIRST EXECUTABLE STATEMENT                00040650
                                           00040660
NZ = 0                                     00040670
S = SSS                                   00040680
IFLAG = 0                                 00040690
J = 0                                     00040700
C                                           MAIN LOOP      00040710
C 5 PV = P(1)                             00040720
                                           EVALUATE P AT S      00040730
QP(1) = PV                                00040740
DO 10 I=7,NN                              00040750
  
```

```

    PV = PV*S+P(I)
    QP(I) = PV
10  CONTINUE
    RMP = DABS(PV)
C
C      COMPUTE A RIGOROUS BOUND ON THE
      ERROR IN EVALUATING P
    RMS = DABS(S)
    EE = (RMRE/(ARE+RMRE))*ABS(SNGL(QP(1)))
    DO 15 I=2,N
15  EE = EE*RMS+ABS(SNGL(QP(I)))
C
C      ITERATION HAS CONVERGED SUFFICIENTLY
      IF THE POLYNOMIAL VALUE IS LESS
      THAN 20 TIMES THIS BOUND
    IF (RMP.GT.20.*((ARE+RMRE)*EE-RMRE*RMP)) GO TO 20
    NZ = 1
    SZR = S
    SZI = ZERO
    RETURN
20  J = J+1
C
C      STOP ITERATION AFTER 10 STEPS
    IF (J.GT.10) RETURN
    IF (J.LT.2) GO TO 25
    IF (DABS(T).GT.PT001*DABS(S-T).OR.RMP.LE.OMP) GO TO 25
C
C      A CLUSTER OF ZEROS NEAR THE REAL
      AXIS HAS BEEN ENCOUNTERED RETURN
      WITH IFLAG SET TO INITIATE A
      QUADRATIC ITERATION
    IFLAG = 1
    SSS = S
    RETURN
C
C      RETURN IF THE POLYNOMIAL VALUE HAS
      INCREASED SIGNIFICANTLY
25  OMP = RMP
C
C      COMPUTE T, THE NEXT POLYNOMIAL, AND
      THE NEW ITERATE
    RKV = RK(1)
    QK(1) = RKV
    DO 30 I=2,N
      RKV = RKV*S+RK(I)
      QK(I) = RKV
30  CONTINUE
    IF (DABS(RKV).LE.DABS(RK(N))*10.*ETA) GO TO 40
C
C      USE THE SCALED FORM OF THE
      RECURRENCE IF THE VALUE OF K AT S
      IS NONZERO
    T = -PV/RKV
    RK(1) = QP(1)
    DO 35 I=2,N
35  RK(I) = T*QK(I-1)+QP(I)
    GO TO 20
C
C      USE UNSCALED FORM
40  RK(1) = ZERO
    DO 45 I=2,N
45  RK(I) = QK(I-1)
50  RKV = RK(1)
    DO 55 I=2,N
55  RKV = RKV*S+RK(I)
    T = ZERO
    IF (DABS(RKV).GT.DABS(RK(N))*10.*ETA) T = -PV/RKV
    S = S+T

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 00041000
 00041010
 00041020
 00041030
 00041040
 00041050
 00041060
 00041070
 00041080
 00041090
 00041100
 00041110
 00041120
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 00041190
 00041200
 00041210
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 00041290
 00041300
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      GO TO 5
      END
      IMSL ROUTINE NAME - ZRPQLE
      -----
      COMPUTER - IBM/DOUBLE
      LATEST REVISION - JANUARY 1, 1978
      SUBROUTINE ZRPQLE (ITYPE)
      INTEGER ITYPE
      INTEGER N, NN
      REAL ARE, ETA, FMRE
      DOUBLE PRECISION P(101), QF(101), RK(101), QK(101), SVK(101)
      DOUBLE PRECISION SR, SI, U, V, RA, RB, C, D, A1, A2, A3,
      COMMON /ZRPQLJ/ A6, A7, E, F, G, H, SZR, SZI, RLZR, RLZI,
      P, QP, RK, QK, SVK, SP, SI, U, V, RA, RB, C, D, A1, A2, A3, A6,
      A7, E, F, G, H, SZR, SZI, RLZR, RLZI, ETA, ARE, RMRE, N, NN
      THIS ROUTINE CALCULATES SCALAR
      QUANTITIES USED TO COMPUTE THE
      NEXT K POLYNOMIAL AND NEW
      ESTIMATES OF THE QUADRATIC
      COEFFICIENTS
      ITYPE - INTEGER VARIABLE SET HERE
      INDICATING HOW THE CALCULATIONS
      ARE NORMALIZED TO AVOID OVERFLOW
      SYNTHETIC DIVISION OF K BY THE
      QUADRATIC 1, U, V
      FIRST EXECUTABLE STATEMENT
      CALL ZRPQLH (N, U, V, RK, QK, C, D)
      IF (DABS(C).GT.DABS(RK(N))*100.*ETA) GO TO 5
      IF (DABS(D).GT.DABS(RK(N-1))*100.*ETA) GO TO 5
      ITYPE = 3
      TYPE=3 INDICATES THE QUADRATIC IS
      ALMOST A FACTOR OF K
      RETURN
      5 IF (DABS(D).LT.DABS(C)) GO TO 10
      ITYPE = 2
      TYPE=2 INDICATES THAT ALL FORMULAS
      ARE DIVIDED BY D
      F = RA/D
      F = C/D
      G = U*RB
      H = V*RB
      A3 = (RA+G)*E+H*(RB/D)
      A1 = RB*F-RA
      A7 = (F+U)*RA+H
      RETURN
      10 ITYPE = 1
      TYPE=1 INDICATES THAT ALL FORMULAS
      ARE DIVIDED BY C
      E = RA/C
      F = D/C
      G = U*E
  
```

[illegible]

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C   LATEST REVISION      - JANUARY 1, 1978                                00042560
C                                                                    00042570
C   SUBROUTINE ZRPQLG (ITYPE,UU,VV)                                00042580
C   SPECIFICATIONS FOR ARGUMENTS                                00042590
C   INTEGER              ITYPE                                00042600
C   DOUBLE PRECISION     UU,VV                                00042610
C   SPECIFICATIONS FOR LOCAL VARIABLES                            00042620
C   INTEGER              N,NN                                00042630
C   REAL                 ARE,ETA,RMRE                        00042640
C   DOUBLE PRECISION     P(101),QP(101),RK(101),QK(101),SVK(101) 00042650
C   DOUBLE PRECISION     SR,SI,U,V,RA,RB,C,D,A1,A2,A3,        00042660
C   1                     A6,A7,E,F,G,H,SZR,SZI,RLZR,RLZI,      00042670
C   2                     A4,A5,B1,B2,C1,C2,C3,C4,TEMP,ZERO      00042680
C   COMMON /ZRPQLJ/      P,GP,RK,QK,SVK,SR,SI,U,V,RA,RB,C,D,A1,A2,A3,A6, 00042690
C   1                     A7,E,F,G,H,SZR,SZI,RLZR,RLZI,ETA,ARE,RMRE,N,NN 00042700
C   DATA                ZERO/0.0D0/                            00042710
C                                                                    00042720
C   COMPUTE NEW ESTIMATES OF THE                                00042730
C   QUADRATIC COEFFICIENTS USING THE                            00042740
C   SCALARS COMPUTED IN ZRPQLG                                00042750
C   USE FORMULAS APPROPRIATE TO SETTING                        00042760
C   OF TYPE.                                                    00042770
C   FIRST EXECUTABLE STATEMENT                                00042780
C                                                                    00042790
C   IF (ITYPE.EQ.3) GO TO 15                                00042800
C   IF (ITYPE.EQ.2) GO TO 5                                    00042810
C   A4 = RA+UMRB+H*F                                          00042820
C   A5 = C+(U+V*F)*D                                          00042830
C   GO TO 10                                                  00042840
C   5 A4 = (RA+G)*F+H                                          00042850
C   A5 = (F+U)*C+V*D                                          00042860
C                                                                    00042870
C   EVALUATE NEW QUADRATIC COEFFICIENTS.                      00042880
C                                                                    00042890
C   10 B1 = -RK(N)/P(NN)                                       00042900
C   B2 = -(RK(N-1)+B1*P(N))/P(NN)                             00042910
C   C1 = V*B2*A1                                              00042920
C   C2 = B1*A7                                                00042930
C   C3 = B1*B1*A3                                             00042940
C   C4 = C1-C2-C3                                             00042950
C   TEMP = A5+B1*A4-C4                                        00042960
C   IF (TEMP.EQ.ZERO) GO TO 15                                00042970
C   UU = U-(U*(C3+C2)+V*(B1*A1+B2*A7))/TEMP                  00042980
C   VV = V*(1+C4/TEMP)                                        00042990
C   RETURN                                                    00043000
C   IF TYPE=3 THE QUADRATIC IS ZEROED                          00043010
C   15 UU = ZERO                                              00043020
C   VV = ZERO                                                 00043030
C   RETURN                                                    00043040
C   END                                                        00043050
C                                                                    00043060
C   SUBROUTINE ZRPQLH (NN,U,V,P,Q,RA,RB)                      00043070
C   SPECIFICATIONS FOR ARGUMENTS                                00043080
C   INTEGER              NN                                00043090
C   DOUBLE PRECISION     P(NN),Q(NN),U,V,RA,RB              00043100
C   SPECIFICATIONS FOR LOCAL VARIABLES                            00043110
C   INTEGER              I                                00043120
C   DOUBLE PRECISION     C                                00043130
C   DIVIDES P BY THE QUADRATIC 1,U,V                          00043140
C   PLACING THE QUOTIENT IN Q AND THE                          00043150
C   REMAINDER IN A,B

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C	RB = P(1)	FIRST EXECUTABLE STATEMENT	00043160
	Q(1) = RB		00043170
	RA = P(2)-UMRB		00043180
	Q(2) = RA		00043190
	DO 5 I=3,NH		00043200
	C = P(I)-UMRA-VMRB		00043210
	Q(I) = C		00043220
	RB = RA		00043230
	RA = C		00043240
	5 CONTINUE		00043250
	RETURN		00043260
	END		00043270
C			00043280
C			00043290
C	IMSL ROUTINE NAME - ZRPQLI		00043300
C			00043310
C			00043320
C			00043330
C			00043340
C	COMPUTER - IBM/DOUBLE		00043350
C			00043360
C	LATEST REVISION - JANUARY 1, 1978		00043370
C			00043380
C	SUBROUTINE ZRPQLI (RA,B1,C,SR,SI,RLR,RLI)		00043390
C		SPECIFICATIONS FOR ARGUMENTS	00043400
C	DOUBLE PRECISION RA,B1,C,SR,SI,RLR,RLI		00043410
C		SPECIFICATIONS FOR LOCAL VARIABLES	00043420
C	DOUBLE PRECISION RB,D,E,ZERO,ONE,TWO		00043430
C	DATA ZERO,ONE,TWO/0.000,1.000,2.000/		00043440
C		CALCULATE THE ZEROS OF THE QUADRATIC	00043450
C		FORMULA, MODIFIED TO AVOID	00043460
C		OVERFLOW. IS USED TO FIND THE	00043470
C		LARGER ZERO IF THE ZEROS ARE REAL	00043480
C		AND BOTH ZEROS ARE COMPLEX.	00043490
C		THE SMALLER REAL ZERO IS FOUND	00043500
C		DIRECTLY FROM THE PRODUCT OF THE	00043510
C		ZEROS C/A	00043520
C		FIRST EXECUTABLE STATEMENT	00043530
C	IF (RA.NE.ZERO) GO TO 10		00043540
C	SR = ZERO		00043550
C	IF (B1.NE.ZERO) SR = -C/B1		00043560
C	RLR = ZERO		00043570
C	5 SI = ZERO		00043580
C	RLI = ZERO		00043590
C	RETURN		00043600
C	10 IF (C.NE.ZERO) GO TO 15		00043610
C	SR = ZERO		00043620
C	RLR = -B1/RA		00043630
C	GO TO 5		00043640
C		COMPUTE DISCRIMINANT AVOIDING	00043650
C		OVERFLOW	00043660
C	15 RB = B1/TWO		00043670
C	IF (DABS(RB).LT.DABS(C)) GO TO 20		00043680
C	E = ONE-(RA/RB)*(C/RB)		00043690
C	D = DSQRT(DABS(E))*DABS(RB)		00043700
C	GO TO 25		00043710
C	20 E = RA		00043720
C	IF (C.LT.?ERO) E = -RA		00043730
C			00043740
C			00043750

```

E = RB*(RB/DABS(C))-E
D = DSQRT(DABS(E))*DSQRT(DABS(C))
25 IF (E.LT.ZERO) GO TO 30
C REAL ZEROS
IF (RB.GE.ZERO) D = -D
RLR = (-RB+D)/RA
SR = ZERO
IF (RLR.NE.ZERO) SR = (C/RLR)/RA
GO TO 5
C COMPLEX CONJUGATE ZEROS
30 SR = -RB/RA
RLR = SR
SI = DABS(D/RA)
RLI = -SI
RETURN
END
C
C
C
C SUBROUTINE LEQ2C (A,N,IA,B,M,IB,IJOB,WA,WK,IER)
C COMPLEX*16 A(IA,1),B(IB,1),WA(N,1),TEMPA,TEMPC,TEMPC
C DOUBLE PRECISION WK(N),TA(2),TB(2),TC(2)
C DOUBLE PRECISION AR,AI,BR,BI,CR,CI,DXNORM,XNORM,ZERO
C DOUBLE PRECISION ACC(2)
C EQUIVALENCE (TA(1),TEMPA),(TB(1),TEMPB),(TC(1),TEMPC),
C (TA(1),AR),(TA(2),AI),(TB(1),BR),(TB(2),BI),
C (TC(1),CR),(TC(2),CI)
C DATA ZERO/0.0D0/
C DATA ITMAX/50/
C FIRST EXECUTABLE STATEMENT
IER = 0
N1 = N+1
N2 = N+2
IF (IJOB .EQ. 2) GO TO 15
C SAVE MATRIX A
DO 10 I = 1,N
DO 5 J = 1,N
WA(I,J) = A(I,J)
5 CONTINUE
10 CONTINUE
C FACTOR MATRIX A
CALL LEQ1C(WA,N,N,B,M,IB,1,WK,IER)
IF (IER .NE. 0) GO TO 9000
IF (IJOB .EQ. 1) GO TO 9005
C SAVE THE RIGHT HAND SIDES
15 DO 65 J = 1,M
DO 20 I = 1,N
WA(I,N1) = B(I,J)
20 CONTINUE
C OBTAIN A SOLUTION
CALL LEQ1C(WA,N,N,WA(1,N1),1,N,2,WK,IER)
C COMPUTE THE NORM OF THE SOLUTION
XNORM = ZERO
DO 25 I = 1,N
TEMPA = WA(I,N1)
XNORM = DMAX1(XNORM,DABS(AR),DABS(AI))
25 CONTINUE
IF (XNORM .EQ. ZERO) GO TO 65
C COMPUTE RESIDUALS

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DO 50 ITER = 1,ITMAX
DO 40 I = 1,N
  TEMPB = B(I,J)
  ACC(1) = 0.000
  ACC(2) = 0.000
  CALL VXADD(BR,ACC)
DO 30 JJ = 1,N
  TEMPA = A(I,JJ)
  TEMPB = WA(JJ,N1)
  CALL VXMUL(-AR,BR,ACC)
  CALL VXMUL(AI,BI,ACC)
30  CONTINUE
  CALL VXSTO(ACC,CR)
  TEMPB = B(I,J)
  ACC(1) = 0.000
  ACC(2) = 0.000
  CALL VXADD(BI,ACC)
DO 35 JJ = 1,N
  TEMPA = A(I,JJ)
  TEMPB = WA(JJ,N1)
  CALL VXMUL(-AR,BI,ACC)
  CALL VXMUL(-BR,AI,ACC)
35  CONTINUE
  CALL VXSTO(ACC,CI)
  WA(I,N2) = TEMPC
40  CONTINUE
  CALL LEQ1C(WA,N,N,WA(1,N2),1,N,2,WK,IER)
  DXNORM = ZERO
C      UPDATE THE SOLUTION
DO 45 I = 1,N
  WA(I,N1) = WA(I,N1)+WA(I,N2)
  TEMPA = WA(I,N2)
  DXNORM = DMAX1(DXNORM,DABS(AR),DABS(AI))
45  CONTINUE
  IF (XNORM+DXNORM .EQ. XNORM) GO TO 55
50  CONTINUE
  IER = 130
C      STORE THE SOLUTION
55  DO 60 JK = 1,N
  B(JK,J) = WA(JK,N1)
60  CONTINUE
  IF (IER .NE. 0) GO TO 9000
65  CONTINUE
  GO TO 9005
9000 CONTINUE
  CALL UERTST(IER,6HLEQ2C )
9005 RETURN
END
C
C
C
C      SUBROUTINE LEQ1C (A,N,IA,B,M,IB,IJOB,WA,IER)
C      SPECIFICATIONS FOR ARGUMENTS
C      INTEGER N,IA,M,IB,IJOB,IER
C      COMPLEX*16 A(IA,N),B(IB,M)
C      DOUBLE PRECISION WA(N)
C      SPECIFICATIONS FOR LOCAL VARIABLES
C      DOUBLE PRECISION P,Q,ZERO,ONE,T(2),RN,BIG
C      COMPLEX*16 SUM,TEMP
C      INTEGER I,J,JM1,IM1,K,IMAX,JF1,IW,N1

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	EQUIVALENCE	(SUM,T(1))	00044960
	DATA	ZERO/0.000/,ONE/1.00/	00044970
C		INITIALIZATION	00044980
C		FIRST EXECUTABLE STATEMENT	00044990
	IER = 0		00045000
	IF (IJOB .EQ. 2) GO TO 75		00045010
	RN = N		00045020
C		FIND EQUILIBRATION FACTORS	00045030
	DO 10 I=1,N		00045040
	BIG = ZERO		00045050
	DO 5 J=1,N		00045060
	TEMP = A(I,J)		00045070
	P = CDABS(TEMP)		00045080
	IF (P .GT. BIG) BIG = P		00045090
5	CONTINUE		00045100
	IF (BIG .EQ. ZERO) GO TO 105		00045110
	WA(I) = ONE/BIG		00045120
10	CONTINUE		00045130
C		L-U DECOMPOSITION	00045140
	DO 70 J = 1,N		00045150
	JM1 = J-1		00045160
	IF (JM1 .LT. 1) GO TO 25		00045170
C		COMPUTE U(I,J), I=1,...,J-1	00045180
	DO 20 I=1,JM1		00045190
	SUM = A(I,J)		00045200
	IM1 = I-1		00045210
	IF (IM1 .LT. 1) GO TO 20		00045220
	DO 15 K=1,IM1		00045230
	SUM = SUM-A(I,K)*A(K,J)		00045240
15	CONTINUE		00045250
	A(I,J) = SUM		00045260
20	CONTINUE		00045270
25	P = ZERO		00045280
C		COMPUTE U(J,J) AND L(I,J), I=J+1,...,	00045290
	DO 45 I=J,N		00045300
	SUM = A(I,J)		00045310
	IF (JM1 .LT. 1) GO TO 40		00045320
	DO 35 K=1,JM1		00045330
	SUM = SUM-A(I,K)*A(K,J)		00045340
35	CONTINUE		00045350
	A(I,J) = SUM		00045360
40	Q = WA(I)*CDABS(SUM)		00045370
	IF (P .GE. Q) GO TO 45		00045380
	P = Q		00045390
	IMAX = I		00045400
45	CONTINUE		00045410
C		TEST FOR ALGORITHMIC SINGULARITY	00045420
	Q = RN+P		00045430
	IF (Q .EQ. RN) GO TO 105		00045440
	IF (J .EQ. IMAX) GO TO 60		00045450
C		INTERCHANGE ROWS J AND IMAX	00045460
	DO 50 K=1,N		00045470
	TEMP = A(IMAX,K)		00045480
	A(IMAX,K) = A(J,K)		00045490
	A(J,K) = TEMP		00045500
50	CONTINUE		00045510
	WA(IMAX) = WA(J)		00045520
60	WA(J) = IMAX		00045530
	JP1 = J+1		00045540
	IF (JP1 .GT. N) GO TO 70		00045550

C	TEMP = A(J,J)	DIVIDE BY PIVOT ELEMENT U(J,J)	00045560
	DO 65 I = JP1,N		00045570
	A(I,J) = A(I,J)/TEMP		00045580
65	CONTINUE		00045590
70	CONTINUE		00045600
75	IF (IJOB .EQ. 1) GO TO 9005		00045610
	DO 103 K = 1,M		00045620
C		SOLVE UX = Y FOR X	00045630
	IW = 0		00045640
	DO 90 I = 1,N		00045650
	IMAX = WA(I)		00045660
	SUM = B(IMAX,K)		00045670
	B(IMAX,K) = B(I,K)		00045680
	IF (IW .EQ. 0) GO TO 85		00045690
	IM1 = I-1		00045700
	DO 80 J = IW,IM1		00045710
	SUM = SUM - A(I,J)*B(J,K)		00045720
80	CONTINUE		00045730
	GO TO 88		00045740
85	IF (T(1) .NE. ZERO .OR. T(2) .NE. ZERO) IW = I		00045750
88	B(I,K) = SUM		00045760
90	CONTINUE		00045770
C		SOLVE LY = B FOR Y	00045780
	N1 = N+1		00045790
	DO 100 IW = 1,N		00045800
	I = N1-IW		00045810
	JP1 = I+1		00045820
	SUM = B(I,K)		00045830
	IF (JP1 .GT. N) GO TO 98		00045840
	DO 95 J = JP1,N		00045850
	SUM = SUM - A(I,J)*B(J,K)		00045860
95	CONTINUE		00045870
98	B(I,K) = SUM/A(I,I)		00045880
100	CONTINUE		00045890
103	CONTINUE		00045900
	GO TO 9005		00045910
C		ALGORITHMIC SINGULARITY	00045920
105	IER = 129		00045930
9000	CONTINUE		00045940
C		PRINT ERROR	00045950
	CALL UERTST(IER,6HLE9T1C)		00045960
9005	RETURN		00045970
	END		00045980
			00045990

SUPPLEMENTARY

INFORMATION



DEPARTMENT OF THE AIR FORCE
AIR FORCE WRIGHT AERONAUTICAL LABORATORIES (AFWL)
WRIGHT-PATTERSON AIR FORCE BASE, OHIO 45433-6543

REPLY TO
ATTN OF: IMST (513/255-7466)

1 May 1987

SUBJECT: Correction to AFWAL Technical Reports, AFWAL-TR-86-3034
and 86-3035

ALL ADDRESSES

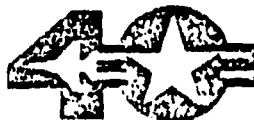
1. Please delete the second paragraph in the NOTICE page affixed to the inside cover of AFWAL-TR-86-3034, "Strength Analysis of Laminated and Metallic Plates Bolted Together by Many Fasteners" and AFWAL-TR-86-3035, "Design Guide for Bolted Joints in Composite Structures."
2. Please contact the undersigned if you have any questions regarding this letter.

G. Doben
G. DOBEN
Chief, Scientific & Tech Info Gp
Information Services Branch

cc: AFWAL/FIBRA
(V. Venkayya)

AD-B108123

UNITED STATES AIR FORCE



SEPTEMBER 18, 1947